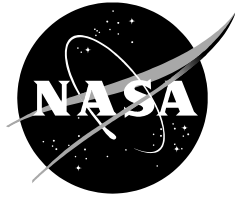


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HISTORY OF ON-ORBIT SATELLITE FRAGMENTATIONS 15th Edition

Orbital Debris Program Office

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Preface to the Fifteenth Edition

The first edition of the History of On-Orbit Satellite Fragmentations was published by Teledyne Brown Engineering (TBE) in August 1984, under the sponsorship of the NASA Johnson Space Center and with the cooperation of the United States Air Force Space Command and the U.S. Army Ballistic Missile Command. The objective was to bring together information about the 75 satellites, which had at that time experienced noticeable breakups. This update encompasses all known satellite fragmentations. This update is published by the NASA Johnson Space Center Orbital Debris Program Office (ODPO).

The information cut-off date for this edition is 4 July 2018.

Since the 14th edition (information cut-off date of 1 August 2007, published in June 2008) there have been 41 identified on-orbit breakups and 18 anomalous events (new or discerned), for a historical total of 242 fragmentations and 78 anomalous events. This activity, in addition to launch activity, has resulted in an approximately 36% increase in the number of cataloged resident space objects since 1 August 2007, which includes on-orbit and decayed objects. More significantly, breakup and anomalous debris accounted for 65% of the catalog growth observed since the last edition. The reason for these large increases was the first accidental collision of two intact spacecraft, Iridium 33 and Cosmos 2251, on 10 February 2009 and the continued cataloging of debris created by the intentional destruction of the *Fengyun 1C* spacecraft on 11 January 2007. There is no reason to believe that any of these three debris clouds have been cataloged completely, to a limiting RCS, as of this writing.

The current authors would like to recognize the substantial contributions of the authors of previous editions of this document. In addition, the assistance of personnel of U.S. Strategic Command, Air Force Space Command, and Naval Network and Space Operations Command (formerly Naval Space Command) has been vital to the present work. Finally, special thanks to Mr. Chris Ostrom of the NASA Orbital Debris Program Office for his thorough review of this 15th edition.

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ACRONYMS

| | |
|-------------------|---|
| ADCOM | (USAF) Aerospace Defense Command |
| AFB | Air Force Base |
| AFSPC | Air Force Space Command |
| AFSSS | Air Force Space Surveillance System (formerly NAVSPASUR) |
| AN/FPS-85 | See FPS-85 |
| Asc | Ascending |
| BMEWS | Ballistic Missile Early Warning System |
| CIS | Commonwealth of Independent States (see also USSR) |
| Dsc | Descending |
| ESA | European Space Agency |
| ESRO | European Space Research Organization |
| FPS-85 | Phased-array UHF radar at Eglin AFB, Florida |
| GEO | Geosynchronous Orbit (orbit category) |
| GEODSS | Ground-based Electro-Optical Deep-space Surveillance System |
| JSC | Johnson Space Center (NASA) |
| LEO | Low Earth Orbit, up to 2000 km altitude (orbit category) |
| NASA | National Aeronautics and Space Administration |
| NAVSPASUR | Naval Space Surveillance System |
| NAVSPOC | Naval Space Operations Center |
| NORAD | North American Aerospace Defense Command |
| PARCS | Phased-array UHF radar at Cavalier AFB, North Dakota: the Perimeter Acquisition Radar Attack Characterization System |
| RAE | The Royal Aerospace Establishment |
| R/B | Rocket Body or Rocket Booster |
| RCS | Radar Cross Section |
| RORSAT | Radar Ocean Reconnaissance Satellite |
| RSO | Resident Space Object |
| SATRAK | PC compatible astrodynamics toolkit |
| SCC | formerly Space Computational Center (obsolete); now Space Control Center |
| SOZ | Sistema Obespechanya Zapuska (Proton-K Block DM attitude/ullage motor unit) |
| SSN | Space Surveillance Network |
| TBE | Teledyne Brown Engineering |
| TLE | Two-Line Element Set |
| USSPACECOM | United States Space Command |
| USSR/CIS | Union of Soviet Socialist Republics/Commonwealth of Independent States (after 1991) |

SYMBOLS

| | |
|------------|--|
| ΔP | The maximum observed change in the orbital period [min]. |
| ΔI | The maximum observed change in the inclination [°]. |

1.0 INTRODUCTION

Since the first serious satellite fragmentation occurred in June 1961, which instantaneously increased the total Earth satellite population by more than 400%, the issue of space operations within the finite region of space around the Earth has been the subject of increasing interest and concern. The prolific satellite fragmentations of the 1970s and the marked increase in the number of fragmentations in the 1980s served to widen international research into the characteristics and consequences of such events. Continued events in all orbits in later years make definition and historical accounting of those events crucial to future research. Large, manned space stations and the growing number of operational robotic satellites demand a better understanding of the hazards of the dynamic Earth satellite population.

The contribution of satellite fragmentations to the growth of the Earth satellite population is complex and varied. A slight majority of detectable fragmentation debris has already fallen out of orbit, and the effects of 38% of all breakups have completely disappeared. On the other hand, just 10 of the 5385 space missions flown since 1957 are responsible for 33% of all cataloged artificial Earth satellites presently in orbit (Figure 1.0-1). Moreover, the sources of four of these 10 fragmentations were discarded rocket bodies that had operated as designed, but later broke up. Modern debris mitigation best practices would have prevented these six events. The remaining six fragmentations are diverse in character. The oldest, the fragmentation of Cosmos 1275, is assessed by Russian authorities to have been caused by a battery fragmentation. Two, USA 109 and NOAA 16, share a similar spacecraft bus. More recently, the intentional fragmentation of the *Fengyun 1C* meteorological payload (1999-025) by an Anti-Satellite (ASAT) weapon and the first accidental collision of large intact spacecraft, Cosmos 2251 (1993-036) and Iridium 33 (1997-051), together account for over 30% of all cataloged resident space objects (RSOs). The breakup fragments associated with these three spacecraft account for almost 13% of all objects cataloged since the launch of Sputnik 1 on 4 October 1957.

The primary factors affecting the growth of the true Earth satellite population are the international space launch rate, satellite fragmentations, and solar activity. As of 4 July 2018, breakup debris have surpassed half of the cataloged Earth satellite population, as illustrated in Figure 1.0-2. Also, approximately three out of every four payloads are no longer operational and constitute a separate, but statistically important class of orbital debris.

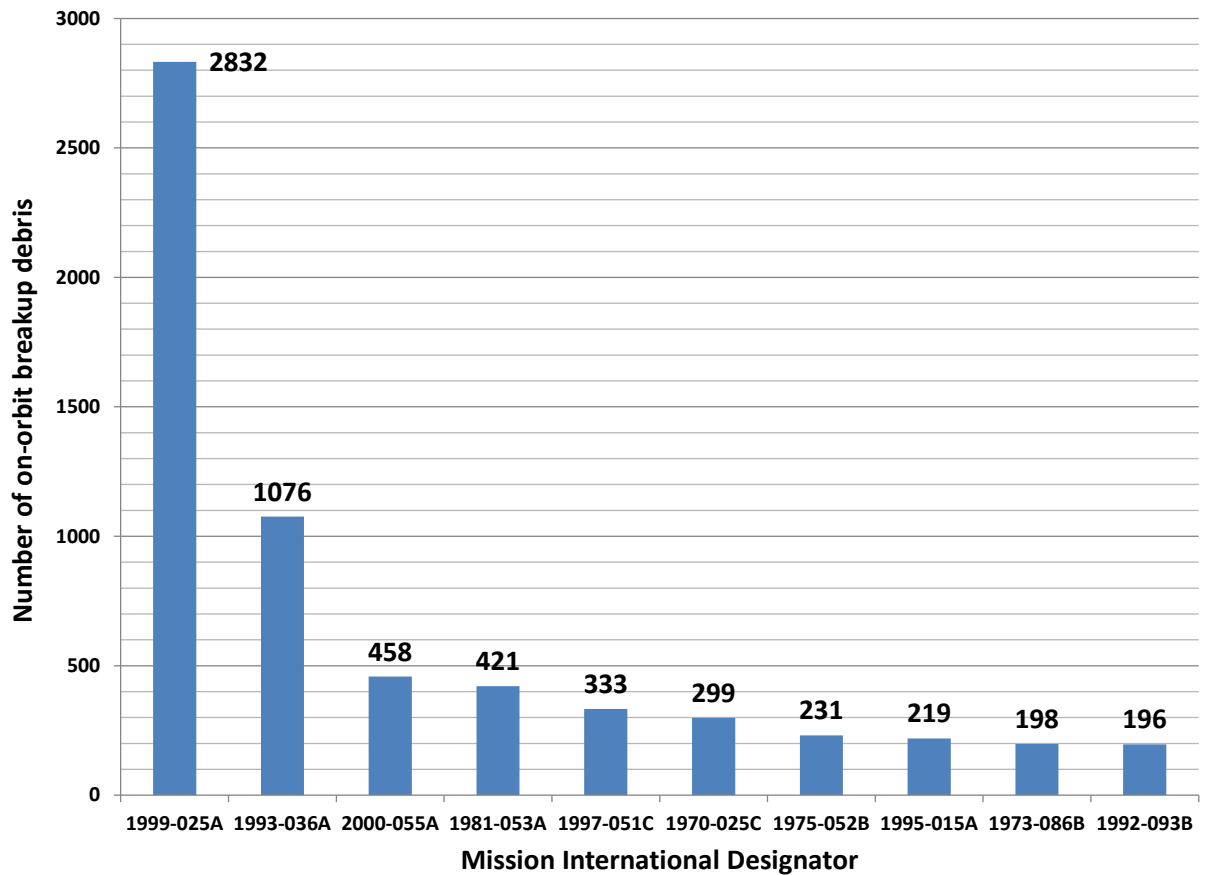


Figure 1.0-1. Magnitude of the 10 largest debris clouds *in orbit* as of 4 July 2018.
Note that cataloging is on-going, accounting for increased totals for some missions relative to the 14th edition of this book.

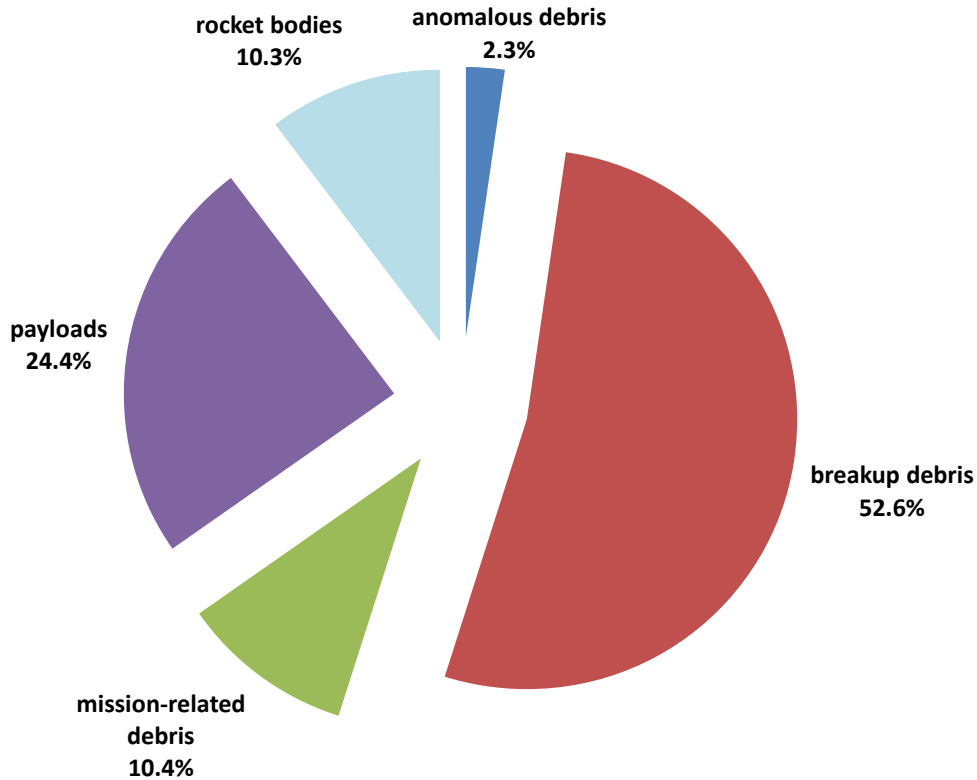


Figure 1.0-2. Relative segments of the cataloged *in-orbit* Earth satellite population.

1.1 Definition of Terms

In this volume, satellite fragmentations are categorized by their assessed nature and, to a lesser degree, by their effect on the near-Earth space environment. A **satellite breakup** is the usually destructive disassociation of an orbital payload, rocket body, or structure, often with a wide range of ejecta velocities. A satellite breakup may be accidental or the result of intentional actions, *e.g.*, due to a propulsion system malfunction or a space weapons test, respectively. An **anomalous event** is the unplanned separation, usually at low velocity, of one or more detectable objects from a satellite, which remains essentially intact. Anomalous events can be caused by material deterioration of items such as thermal blankets, protective shields, or solar panels, or by the impact of small particles. As a general rule, a satellite breakup will produce considerably more debris, both trackable and non-trackable, than an anomalous event. From one perspective, satellite breakups may be viewed as a measure of the effects of man's activity on the environment, while anomalous events may be a measure of the effects of the environment on man-made objects.

Mission-related debris result from the intentional release of objects, usually in small numbers, during normal on-orbit operations. Objects ejected during the deployment, activation, and de-orbit of payloads and during manned operations are examples of mission-related debris. Usually, mission-related debris from a single launch are few in number, but extreme examples occasionally arise, such as the 323 objects from the *Mir* space station or the more than 140 objects from the Westford Needles experiment. Although mission-related debris represent a non-trivial portion (approximately 10%) of all satellites in orbit today and, therefore, are a legitimate subject in the

study of methods to retard the growth of the Earth satellite population, identification of the thousands of mission-related debris events is beyond the scope of this report.

1.2 Information Sources and Accuracy

A number of data sources were employed in the compilation of this volume. However, nearly all are derived from observations collected by the U.S. Space Surveillance Network (US SSN). Due to the variety of sources and geodetic models used to create satellite orbital element sets, all altitudes cited within this volume are presented to the nearest 5 km, referenced to a mean Earth of radius 6378.135 km. The accuracy of the data presented is not of adequate fidelity for precision analysis, although it is appropriate for the anticipated uses of this text. Complete base element sets are provided, but manipulation of these data, in particular satellite orbit propagation, should be performed only with validated, Air Force Space Command (AFSPC)-derived software, such as the PC-compatible SATRAK astrodynamics toolkit. Long-term propagation of these elements is not appropriate regardless of the propagation technique applied and is, therefore, discouraged.

Although all fragmentations are described by the number of debris cataloged and the number of cataloged debris remaining in orbit, these parameters are poor measures of merit and should be used with extreme caution when undertaking comparative analyses. The sensitivity of the SSN and hence the degree to which debris will be detected and cataloged are highly dependent upon satellite altitude and to a lesser degree on satellite inclination. Additionally, historical cataloging practices have changed over the years. Past practices have included cataloging all debris objects associated with a breakup, even if they had already decayed; cataloging almost no pieces from a low altitude breakup when decay of most of the cloud was imminent; and cataloging objects as they were created, regardless of status. These different practices have resulted in an inconsistent historical record.

As a rule of thumb, low altitude, cataloged debris are assessed to be larger than 10 cm in diameter. At higher altitudes objects less than 1 m in diameter may be undetectable. Individual object sensitivities may vary dramatically from this simple generalization. Debris counts for fragmentations occurring in highly elliptical orbits near 63° inclination (*Molniya*-type) are traditionally low, in part due to stable perigees situated deep in the Southern Hemisphere and often beyond SSN coverage. In February of 2007 a *Briz-M* upper stage broke up into over 1000 observed fragments. However, due to the elliptical nature of the stage's orbit which impeded the SSN's ability to detect, to identify, and to catalog the debris, only 102 fragments had been cataloged, of which 92 remain on orbit. Similar outcomes may occur in practice with other *Briz-M* upper stage fragmentations. During a special surveillance session in 1987, as many as 250 uncataloged objects were observed in low inclination, highly elliptical orbits, but reliable tracking and parent identification were not achieved. The disclosure by the Russian Government of the Ekran 2 battery explosion on 25 June 1978 is the first known fragmentation in geostationary orbit. This event was not detected by the SSN, but since the event, four pieces have entered the catalog. Cataloging errors, *e.g.*, identification of an object with the wrong parent satellite, normally are not explicitly noted in this volume since many errors have been or may be corrected.

For fragmentations at very low altitudes, *i.e.*, below 400 km, much of the debris may reenter before detection, identification, and cataloging can be completed. For example, when the debris cloud from Cosmos 1813 passed over a single SSN radar, 846 individual fragments could be discerned.

However, the total number of debris officially cataloged only reached 195. Likewise, more than 380 fragments are known to have been injected into Earth orbits (an equal number probably were sent on reentry trajectories) following the USA 19 test, but only 18 debris were entered into the official satellite catalog.

1.3 Environment Overview

To place the debris population component of the orbital environment in context for the reader, it is useful to review the general orbital environment in the near Earth and near geosynchronous regions. Differentiation of the population by source, object type, and orbit type are also included below.

1.3.1 ON-ORBIT SPATIAL DENSITY

The spatial density of resident space objects is a common means of describing the space object environment and is adopted here. Spatial density (objects per unit volume) represents the effective number of spacecraft and other objects as a function of altitude. Effective number, rather than the simple counting of objects, is used because many objects traverse the altitude regions of interest yet contribute little to the local collision hazard, *e.g.*, geosynchronous transfer orbits. Such orbits exhibit an effective contribution to the environment at any given altitude of up to two orders of magnitude less than an object in a circular orbit within this same altitude interval. Thus, circular orbits at or near an orbit of interest normally dominate the hazard environment. The following figure portrays the near Earth (up to 2000 km altitude) environment categorized by intact or debris object types. The densities are subdivided into 5-km altitude intervals and graphed linearly. The epoch of the source data, a US SSN Two-Line Element (TLE) set, is 2 October 2018.

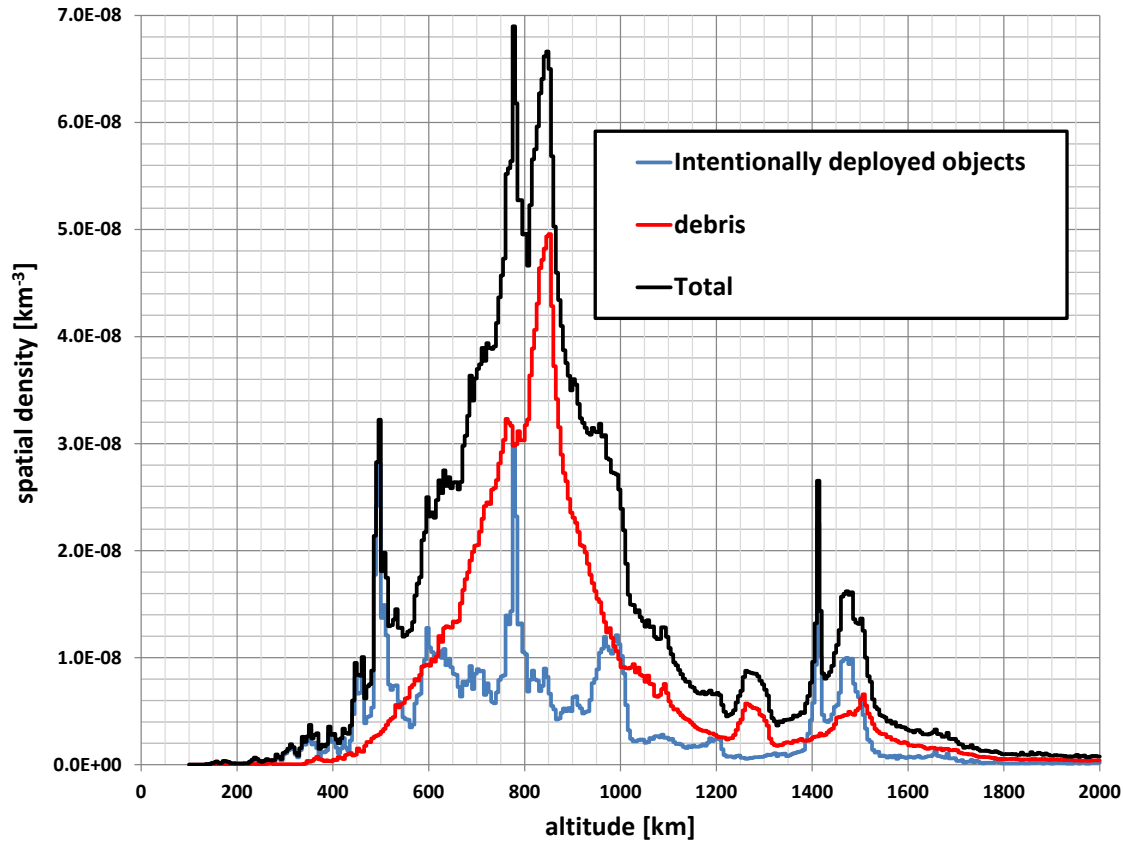


Figure 1.3.1-1. The near Earth (up to 2000 km) altitude population.

The peak near 890 km is due principally to the January 2007 intentional destruction of the *Fengyun IC* weather spacecraft while the peak centered at approximately 770 km altitude was created by the February 2009 accidental collision of Iridium 33 (active) and Cosmos 2251 (derelict) communication spacecraft. Also clearly visible in this figure are other high-density regions of space. The satellite constellations deployed in LEO in the late 1990s are clearly evident: the IRIDIUM constellation inhabits the altitude region at and about 780 km altitude, while the GLOBALSTAR constellation inhabits the region from 1410-1420 km. Other spacecraft constellations, such as the USSR/CIS communications and navigation constellations, are also visible near 1480 km and 950 km, respectively.

The geosynchronous altitude environment increased in both importance and number of RSOs over the course of the early 21st century. Fig. 1.3.1-2 shows the geosynchronous altitude using a logarithmic, vertical spatial density axis and altitude intervals of 5 km. Only objects with an inclination less than 15 degrees were included. Consequently, the spatial density values assume all spacecraft are contained within 15 degrees latitude from the equator. Because high inclination orbits normally do not penetrate this true geosynchronous region, the assumption is appropriate to best categorize the spatial density of this region.

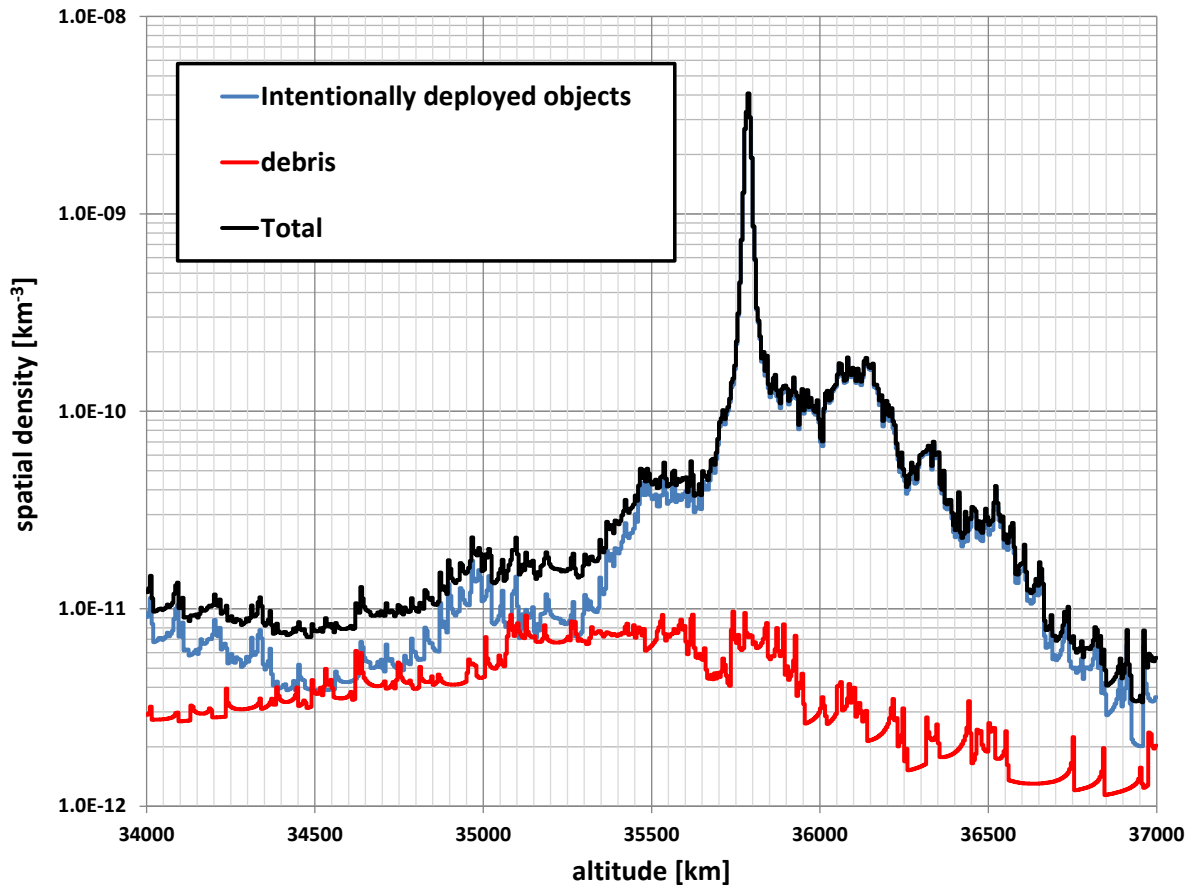


Figure 1.3.1-2. The geosynchronous altitude population.

Because the LEO spatial density chart averages over all inclinations and the GEO spatial density chart averages over inclinations between ± 15 degrees, collision rates are not linearly related to the spatial density at any given altitude. Rather, collision rates will vary not only with the spatial density, but also with the inclination-dependent relative velocity. Altitudes dominated by high inclination (70 - 110°) orbits yield a significantly higher collision rate as compared to those populated by lower inclination orbits. Objects in these inclinations can collide at near head-on engagement geometries with objects in complementary inclinations. The exception to this general rule is provided by the commercial constellations in LEO and spacecraft in GEO. The commercial constellations are maintained in precise orbital planes; hence, their expected collision rate would be versus the “background” population only. Therefore, the spikes representing the IRIDIUM and GLOBALSTAR constellations do not present the inordinate collision risk implied by a casual examination. Similarly, the GEO environment is characterized by low collision velocities (< 1.5 km/s) due to the relative motion between controlled and uncontrolled objects.

1.3.2 POPULATION DISTRIBUTION

The distribution of objects by type (*e.g.*, spacecraft, rocket bodies) and source (U.S., the People’s Republic of China, *etc.*) is germane to this discussion since objects are not randomly distributed among these categories. To display the distribution statistics, the 4 July 2018 U.S. Satellite Catalog was categorized by these nominal variables. The reader should note that absolute numbers in these

statistics are fluid, in the sense that on-going analyses can sometimes move a given debris object from one parent breakup to another. This is particularly noted for the highly populated sun-synchronous orbital region, and accounts for seven number discrepancies when one compares the numerical count of objects in a given debris cloud with the number implied by the highest piece tag associated with the cloud's international designator. Tentative identifications are also present in the public satellite catalog, and are indicated by an asterisk (*) appearing in the "common name" field. Finally, some identifications require in-depth analysis and interpretation, and the understanding of a particular launch or debris cloud may evolve over time.

TABLE 1.3.2 SOURCE VS. TYPE ACCOUNTING

| on-orbit | | | | | | | | | |
|-------------------------------|------|-------|--------|------|-------|-------|-------------|-------|-------|
| | US | CIS | France | PRC | India | Japan | ESRO/ESA | Other | TOTAL |
| payloads | 1646 | 1436 | 63 | 309 | 87 | 160 | 71 | 849 | 4621 |
| rocket bodies | 596 | 949 | 145 | 125 | 35 | 51 | 7 | 40 | 1948 |
| mission-related debris | 734 | 725 | 148 | 270 | 10 | 41 | 13 | 20 | 1961 |
| breakup debris | 3024 | 3315 | 192 | 3258 | 72 | 13 | 27 | 52 | 9953 |
| anomalous debris | 310 | 113 | 1 | 0 | 0 | 0 | 11 | 2 | 437 |
| TOTAL | 6310 | 6538 | 549 | 3962 | 204 | 265 | 129 | 963 | 18920 |
| | | | | | | | | | |
| decayed or beyond Earth orbit | | | | | | | | | |
| | US | CIS | France | PRC | India | Japan | ESRO/ESA | Other | TOTAL |
| payloads | 1098 | 2000 | 9 | 88 | 12 | 68 | 29 | 83 | 3387 |
| rocket bodies | 744 | 2534 | 76 | 141 | 17 | 69 | 6 | 15 | 3602 |
| mission-related debris | 839 | 5772 | 155 | 254 | 10 | 130 | 9 | 102 | 7271 |
| breakup debris | 3531 | 4842 | 487 | 845 | 300 | 39 | 12 | 35 | 10091 |
| anomalous debris | 171 | 68 | 5 | 3 | 2 | 2 | 1 | 1 | 253 |
| TOTAL | 6383 | 15216 | 732 | 1331 | 341 | 308 | 57 | 236 | 24604 |
| | | | | | | | | | |
| | | | | | | | GRAND TOTAL | | 43524 |

Several salient features are apparent in this table. Debris is dominant among all source variables and the majority of debris (and all other categories of resident space objects) are due to space activities of the U.S., CIS, and PRC. However, individual events from other space-faring nations have also contributed greatly to the local environment in several sun-synchronous orbital regimes. An example is provided by the 1986 fragmentation of the Ariane SPOT-1/Viking rocket body.

A net increase of almost 6800 objects on-orbit (over 11,500 total) has been observed since the 14th edition of this book was published in 2008. The majority of the on-orbit objects were from three debris clouds: the intentional destruction of the PRC's *Fengyun 1C* spacecraft and the accidental collision of the US Iridium 33 and Russian Cosmos 2251 spacecraft. Table 1.3.3 shows the net increase or decrease in objects since the 14th edition. A discouraging feature of Table 1.3.3 is that almost every object type showed a net increase of on-orbit objects.

TABLE 1.3.3 SOURCE VS. TYPE ACCOUNTING – NET CHANGE SINCE 14TH EDITION

| on-orbit | | | | | | | | | |
|-------------------------------|------|------|--------|------|-------|-------|-------------|-------|-------|
| | US | CIS | France | PRC | India | Japan | ESRO/ESA | Other | TOTAL |
| payloads | 583 | 112 | 19 | 248 | 54 | 57 | 35 | 462 | 1570 |
| rocket bodies | 54 | 112 | 48 | 88 | 27 | 16 | 1 | 13 | 359 |
| mission-related debris | -45 | 218 | 56 | 208 | 9 | 5 | 1 | 15 | 467 |
| breakup debris | 1358 | 1791 | 66 | 943 | -25 | 11 | 9 | 17 | 4170 |
| anomalous debris | 166 | 31 | -2 | 0 | 0 | 0 | 11 | 2 | 208 |
| TOTAL | 2116 | 2264 | 187 | 1487 | 65 | 89 | 57 | 509 | 6774 |
| | | | | | | | | | |
| decayed or beyond Earth orbit | | | | | | | | | |
| | US | CIS | France | PRC | India | Japan | ESRO/ESA | Other | TOTAL |
| payloads | 298 | 140 | 1 | 38 | 3 | 46 | 11 | 34 | 571 |
| rocket bodies | 113 | 167 | 21 | 73 | 9 | 16 | 1 | 8 | 408 |
| mission-related debris | 132 | 222 | 32 | 141 | 2 | 49 | 1 | 48 | 627 |
| breakup debris | 696 | 1570 | 13 | 666 | 51 | 17 | 8 | 31 | 3052 |
| anomalous debris | 21 | 63 | 3 | 3 | 2 | 0 | 1 | 1 | 94 |
| TOTAL | 1260 | 2162 | 70 | 921 | 67 | 128 | 22 | 122 | 4752 |
| | | | | | | | | | |
| | | | | | | | GRAND TOTAL | | 11526 |

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2.0 SATELLITE BREAKUPS

This section summarizes the current breakup environment and describes each individual breakup. Each breakup is presented in a two-page format. New classes of breakup types have tended to fuel the background breakup rate, replacing classes of breakups from older on-orbit practices such as the well-known Delta second stage rocket body failures.

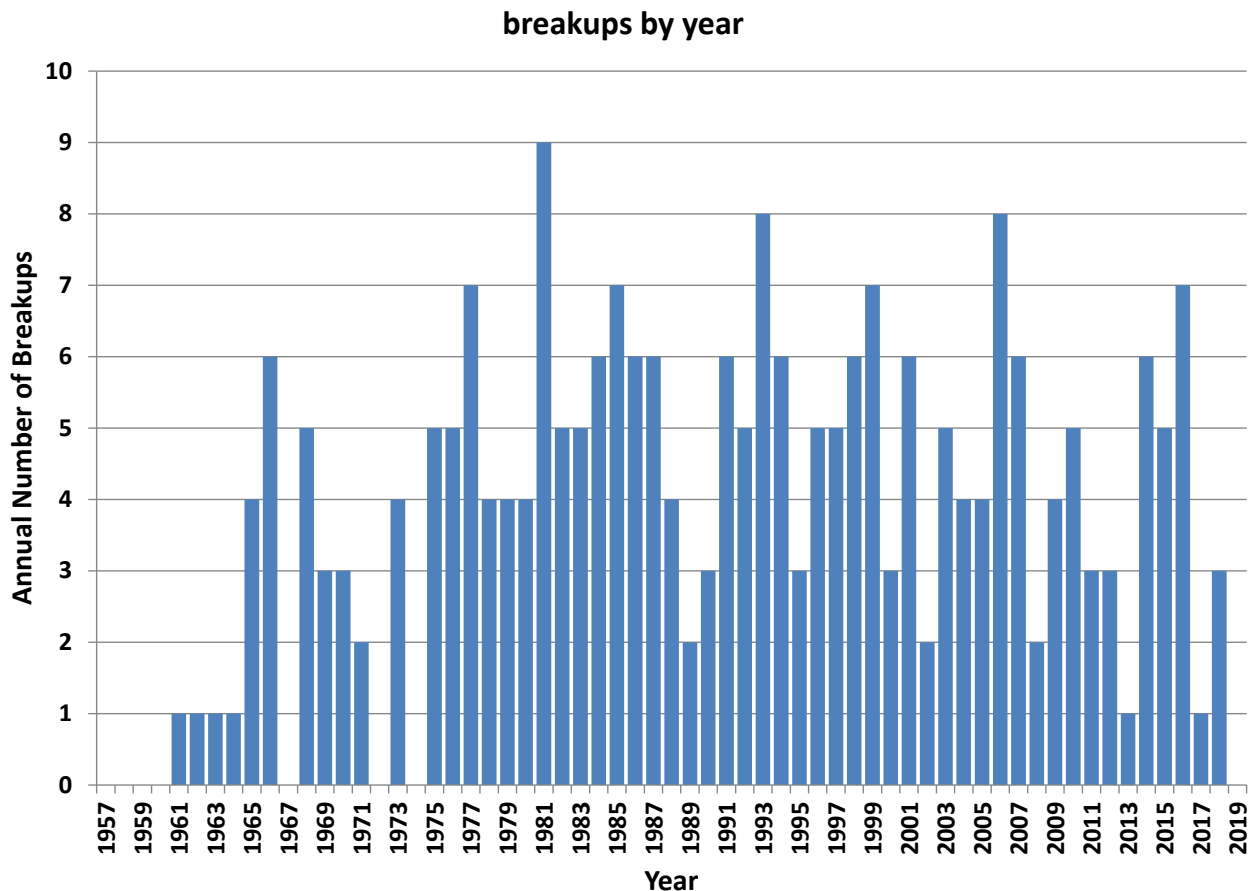


Figure 2.0-1. Number of breakups by year since 1961. Note that some older debris-producing events have been recognized as breakups since the publication of the 14th edition.

2.1 Background and Status

By far the most important category of man-made, on-orbit objects is satellite breakups, which now account for 53% of the total cataloged on-orbit Earth satellite population of 18,920 Earth-orbiting objects. Since 1957, a total of 242 satellites are believed to have broken up (Tables 2.1 and 2.2). Breakups due to aerodynamic forces at or near reentry are treated separately from breakups caused by other factors, because aerodynamic breakups occur at the end of the satellite lifetime and, therefore, contribute nothing toward the orbital environment past the very near term. Only a fraction of these breakups are even detected because of the short remaining lifetime of the object

and its debris. Sixteen additional breakups of this aerodynamic nature that have been detected, and these events are discussed in Chapter 4 and omitted from data included in this chapter.

The primary causes of satellite breakups (Figure 2.1-1) are propulsion-related events and deliberate actions, although the cause for almost one in four breakups remains uncertain. This document will continue to carry breakup causes as unknown until a strong case can be made for one of the other cause classifications. Deliberate actions, often associated with activities related to national security, were formerly the most frequently occurring class, although only one such event occurred during the decade from 1997 until the *Fengyun 1C* event in January 2007. On average, the resulting debris from deliberate actions are short-lived (Figures 2.1-2 and 2.1-3), the exception being *Fengyun 1C*. Propulsion-related breakups, currently the most frequent class, include catastrophic malfunctions during orbital injection or maneuvers, subsequent explosions based on residual propellants, and failures of active attitude control systems. Breakups of rocket bodies due to propulsion failures are usually more prolific and produce longer-lived debris than the intentional destruction of payloads, often due to the higher altitudes of the malfunctioning rocket bodies rather than the mechanics of the explosive event. Breakups of the CIS' Blok-DM *Sistema Obespecheniya Zapuska* (SOZ) ullage motors are segregated from other members of the propulsion ensemble due to their high probability of fragmentation, which tends to overinflate the propulsion category to some degree; omitting the SOZ breakups *equates* the non-SOZ propulsion and deliberate categories. Although it may appear obvious that a rocket body breakup should be classified under the "Propulsion" category, rocket body events are carried as "Unknown" until a failure mechanism can be confidently identified for that rocket body design and is associated with a given rocket body event.

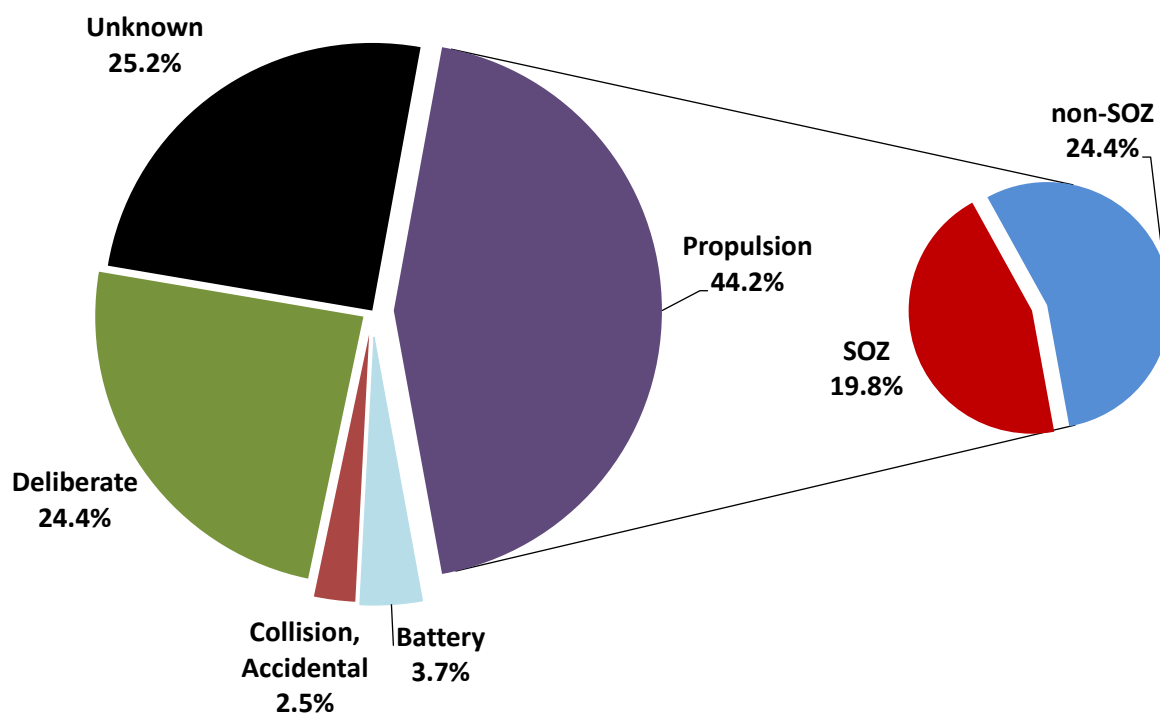


Figure 2.1-1. Causes of known satellite breakups.

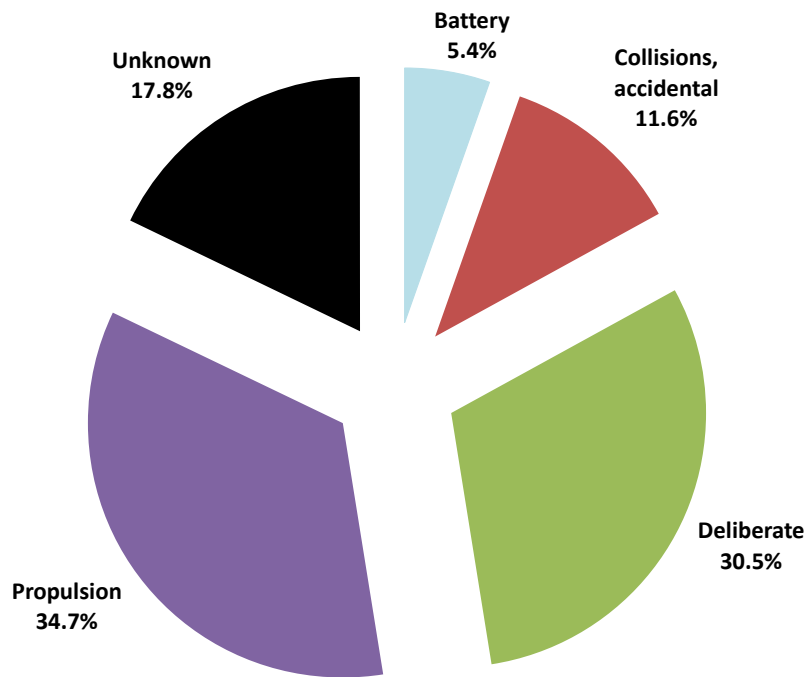


Figure 2.1-2. Proportion of all cataloged satellite breakup debris.

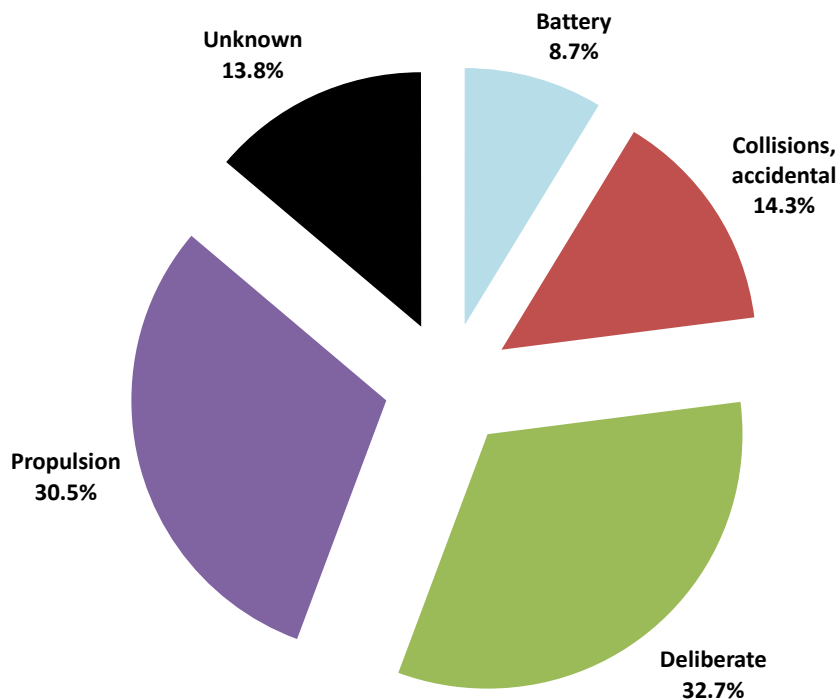


Figure 2.1-3. Proportion of cataloged satellite breakup debris *remaining in orbit*.

The rate of satellite breakups increased noticeably in the 1970s and has continued through the 1990s and into the new millennium at an average pace of approximately four fragmentations per year since 2001. The recent low-activity years of 2013 and 2017 appear as statistical anomalies in

this regard. Increased awareness of potential hazards has resulted in positive actions to mitigate or eliminate many known breakup causes, *e.g.*, Delta second stages, weapons testing, and Cosmos 862-type events. Together, these three programs were responsible for more than a quarter of all satellite breakups in the decade of the 1980s. The quick response of *Arianespace* and the European Space Agency to the breakup of an Ariane third stage in 1986 is indicative of a desire by most space-faring organizations to operate in near-Earth space responsibly. Today, new series of boosters and satellites have resulted in new breakup sources, such as the multiple fragmentations of *Briz-M* upper stages. Also, the intentional destruction of the *Fengyun 1C* spacecraft has continued to increase the percentage of debris from deliberate events as compared to the previous edition of this book, as over 900 additional debris have been officially cataloged.

Figures 2.1-4 and -5 illustrate that an absolute majority of the satellite breakup debris total and debris remaining in orbit today have originated from payloads. In previous editions of this book, up to the 14th, rocket bodies always had more total and on-orbit debris than payloads (in the case of on-orbit, by a ratio of almost 3:1), but the three major fragmentation clouds created between 2007 and 2009 account for the new dominance of payload debris.

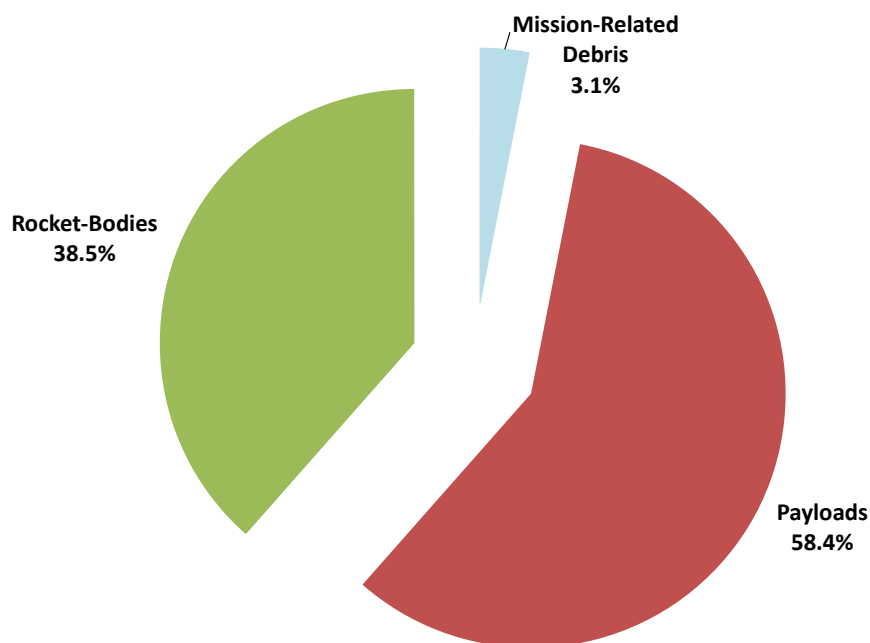


Figure 2.1-4. Sources of all cataloged satellite breakup debris by satellite type.

This dominance is even more pronounced when considering only on-orbit debris. In this case, payload debris now accounts for almost two-thirds of all cataloged debris.

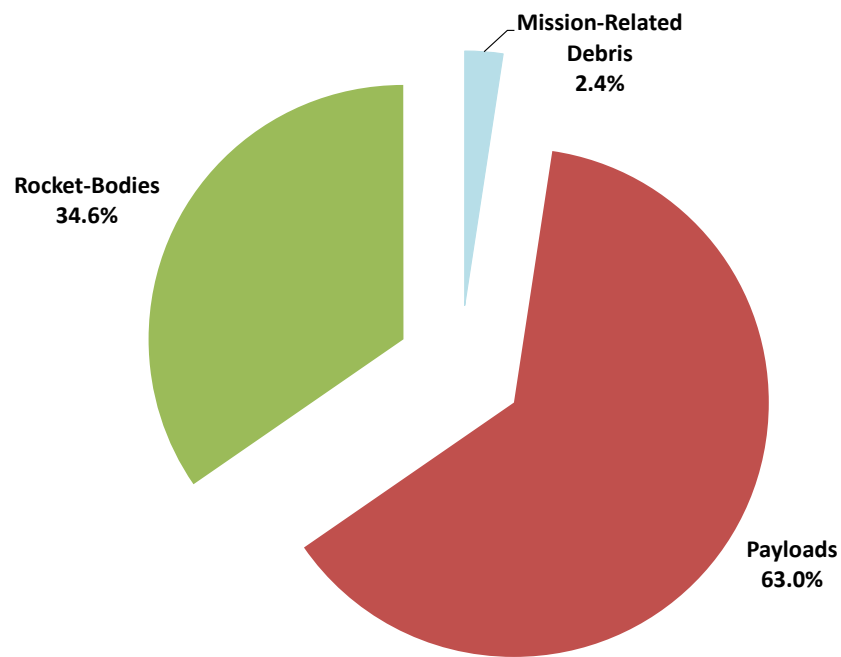


Figure 2.1-5. Sources of satellite breakup debris *in orbit* by satellite type.

TABLE 2.1 HISTORY OF SATELLITE BREAKUPS BY LAUNCH DATE

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | BREAKUP DATE | DEBRIS CATALOGED | DEBRIS ON-ORBIT | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) | ASSESSED CAUSE | ADDITIONAL INFORMATION |
|-----------------------|--------------------------|---------------------|-------------|--------------|------------------|-----------------|-------------|--------------|-------------------|----------------|------------------------|
| TRANSIT 4A R/B | 1961-015C | 118 | 29-Jun-61 | 29-Jun-61 | 296 | 172 | 995 | 880 | 66.8 | PROPULSION | ABLESTAR STAGE |
| SPUTNIK 29 | 1962-057A | 443 | 24-Oct-62 | 29-Oct-62 | 24 | 0 | 260 | 200 | 65.1 | PROPULSION | MOLNIYA FINAL STAGE |
| ATLAS CENTAUR 2 | 1963-047A | 694 | 27-Nov-63 | 27-Nov-63 | 19 | 8 | 1785 | 475 | 30.3 | PROPULSION | CENTAUR STAGE |
| COSMOS 50 | 1964-070A | 919 | 28-Oct-64 | 5-Nov-64 | 96 | 0 | 220 | 175 | 51.2 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 57 | 1965-012A | 1093 | 22-Feb-65 | 22-Feb-65 | 167 | 0 | 425 | 165 | 64.8 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 61-63 R/B | 1965-020D | 1270 | 15-Mar-65 | 15-Mar-65 | 147 | 18 | 1825 | 260 | 56.1 | UNKNOWN | COSMOS SECOND STAGE |
| OV2-1/LCS 2 R/B | 1965-082DM | 1822 | 15-Oct-65 | 15-Oct-65 | 473 | 32 | 790 | 710 | 32.2 | PROPULSION | TITAN TRANSTAGE |
| COSMOS 95 | 1965-088A | 1706 | 4-Nov-65 | 15-Jan-66 | 1 | 0 | 300 | 180 | 48.4 | UNKNOWN | |
| OV2-3/ et al. R/B | 1965-108A | 1863 | 21-Dec-65 | 21-Dec-65 | 108 | 101 | 33660 | 165 | 26.4 | PROPULSION | TITAN TRANSTAGE |
| OPS 3031 | 1966-012C | 2015 | 15-Feb-66 | 15-Feb-66 | 38 | 0 | 270 | 150 | 96.5 | UNKNOWN | INFLATABLE SPHERE |
| GEMINI 9 ATDA R/B | 1966-046B | 2188 | 1-Jun-66 | Jun-66 | 51 | 0 | 275 | 240 | 28.8 | UNKNOWN | ATLAS CORE STAGE |
| PAGEOS | 1966-056A | 2253 | 24-Jun-66 | 12-Jul-75 | 79 | 1 | 5170 | 3200 | 85.3 | UNKNOWN | INFLATABLE SPHERE |
| AS-203 | 1966-059A | 2289 | 5-Jul-66 | 5-Jul-66 | 34 | 0 | 215 | 185 | 32.0 | DELIBERATE | SATURN S-IVB STAGE |
| COSMOS U-1 | 1966-088A | 2437 | 17-Sep-66 | 17-Sep-66 | 52 | 0 | 855 | 140 | 49.6 | DELIBERATE | SELF-DESTRUCT |
| COSMOS U-2 | 1966-101A | 2536 | 2-Nov-66 | 2-Nov-66 | 41 | 0 | 885 | 145 | 49.6 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 192 | 1967-116A | 3047 | 23-Nov-67 | 30-Aug-09 | 4 | 1 | 715 | 710 | 74.0 | UNKNOWN | |
| COSMOS 199 | 1968-003A | 3099 | 16-Jan-68 | 24-Jan-68 | 3 | 0 | 355 | 200 | 65.6 | DELIBERATE | SELF-DESTRUCT |
| APOLLO 6 R/B | 1968-025B | 3171 | 4-Apr-68 | 13-Apr-68 | 16 | 0 | 360 | 200 | 32.6 | PROPULSION | SATURN S-IVB STAGE |
| OV2-5 R/B | 1968-081E | 3432 | 26-Sep-68 | 21-Feb-92 | 29 | 29 | 35810 | 35100 | 11.9 | PROPULSION | TITAN TRANSTAGE |
| COSMOS 248 | 1968-090A | 3503 | 19-Oct-68 | 1-Nov-68 | 5 | 0 | 545 | 475 | 62.2 | DELIBERATE | DEBRIS IMPACT |
| COSMOS 249 | 1968-091A | 3504 | 20-Oct-68 | 20-Oct-68 | 108 | 39 | 2165 | 490 | 62.3 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 252 | 1968-097A | 3530 | 1-Nov-68 | 1-Nov-68 | 139 | 40 | 2140 | 535 | 62.3 | DELIBERATE | SELF-DESTRUCT |
| ESSA 8 (TOS F) R/B | 1968-114B | 3616 | 15-Dec-68 | 15-Nov-73 | 18 | 18 | 1462 | 1413 | 101.6 | PROPULSION | DELTA SECOND STAGE |
| OPS 0757 (TACSAT) R/B | 1969-013B | 3692 | 9-Feb-69 | 28-Feb-18 | 19 | 19 | 37257 | 35886 | 6.2 | UNKNOWN | TITAN TRANSTAGE |
| METEOR 1-1 R/B | 1969-029B | 3836 | 26-Mar-69 | 28-Mar-69 | 37 | 0 | 850 | 460 | 81.2 | UNKNOWN | VOSTOK FINAL STAGE |
| INTELSAT 3 F-5 R/B | 1969-064B | 4052 | 26-Jul-69 | 26-Jul-69 | 22 | 0 | 5445 | 270 | 30.4 | PROPULSION | TE 364-4 STAGE |
| OPS 7613 R/B | 1969-082AB | 4159 | 30-Sep-69 | 4-Oct-69 | 261 | 63 | 940 | 905 | 70.0 | UNKNOWN | AGENA D STAGE |
| NIMBUS 4 R/B | 1970-025C | 4367 | 8-Apr-70 | 17-Oct-70 | 441 | 299 | 1085 | 1065 | 99.9 | UNKNOWN | AGENA D STAGE |
| COSMOS 374 | 1970-089A | 4594 | 23-Oct-70 | 23-Oct-70 | 99 | 20 | 2130 | 530 | 62.9 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 375 | 1970-091A | 4598 | 30-Oct-70 | 30-Oct-70 | 47 | 15 | 2100 | 525 | 62.8 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 397 | 1971-015A | 4964 | 25-Feb-71 | 25-Feb-71 | 116 | 44 | 2200 | 575 | 65.8 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 462 | 1971-106A | 5646 | 3-Dec-71 | 3-Dec-71 | 25 | 0 | 1800 | 230 | 65.7 | DELIBERATE | SELF-DESTRUCT |
| LANDSAT 1 R/B | 1972-058B | 6127 | 23-Jul-72 | 22-May-75 | 244 | 47 | 910 | 635 | 98.3 | PROPULSION | DELTA SECOND STAGE |

TABLE 2.1 HISTORY OF SATELLITE BREAKUPS BY LAUNCH DATE (CONT'D)

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | BREAKUP DATE | DEBRIS CATALOGED | DEBRIS ON-ORBIT | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) | ASSESSED CAUSE | ADDITIONAL INFORMATION |
|-----------------|--------------------------|---------------------|-------------|--------------|------------------|-----------------|-------------|--------------|-------------------|-----------------------|------------------------|
| SALYUT 2 R/B | 1973-017B | 6399 | 3-Apr-73 | 3-Apr-73 | 25 | 0 | 245 | 195 | 51.5 | PROPULSION | PROTON-K THIRD STAGE |
| COSMOS 554 | 1973-021A | 6432 | 19-Apr-73 | 6-May-73 | 193 | 0 | 350 | 170 | 72.9 | DELIBERATE | SELF-DESTRUCT |
| NOAA 3 R/B | 1973-086B | 6921 | 6-Nov-73 | 28-Dec-73 | 220 | 198 | 1510 | 1500 | 102.1 | PROPULSION | DELTA SECOND STAGE |
| DMSP 5B F5 R/B | 1974-015B | 7219 | 16-Mar-74 | 17-Jan-05 | 7 | 7 | 885 | 775 | 99.1 | COLLISION, ACCIDENTAL | HIT BY DEBRIS (26207) |
| NOAA 4 R/B | 1974-089D | 7532 | 15-Nov-74 | 20-Aug-75 | 185 | 160 | 1460 | 1445 | 101.7 | PROPULSION | DELTA SECOND STAGE |
| COSMOS 699 | 1974-103A | 7587 | 24-Dec-74 | 17-Apr-75 | 50 | 0 | 445 | 425 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| LANDSAT 2 R/B | 1975-004B | 7616 | 22-Jan-75 | 9-Feb-76 | 207 | 32 | 915 | 740 | 97.8 | PROPULSION | DELTA SECOND STAGE |
| NIMBUS 6 R/B | 1975-052B | 7946 | 12-Jun-75 | 1-May-91 | 307 | 231 | 1105 | 1095 | 99.6 | PROPULSION | DELTA SECOND STAGE |
| COSMOS 758 | 1975-080A | 8191 | 5-Sep-75 | 6-Sep-75 | 76 | 0 | 325 | 175 | 67.1 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 777 | 1975-102A | 8416 | 29-Oct-75 | 25-Jan-76 | 62 | 0 | 440 | 430 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 838 | 1976-063A | 8932 | 2-Jul-76 | 17-May-77 | 40 | 0 | 445 | 415 | 65.1 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 839 | 1976-067A | 9011 | 8-Jul-76 | 29-Sep-77 | 70 | 68 | 2100 | 980 | 65.9 | BATTERY | |
| COSMOS 844 | 1976-072A | 9046 | 22-Jul-76 | 25-Jul-76 | 248 | 0 | 355 | 170 | 67.1 | DELIBERATE | SELF-DESTRUCT |
| NOAA 5 R/B | 1976-077B | 9063 | 29-Jul-76 | 24-Dec-77 | 184 | 174 | 1520 | 1505 | 102.0 | PROPULSION | DELTA SECOND STAGE |
| COSMOS 862 | 1976-105A | 9495 | 22-Oct-76 | 15-Mar-77 | 13 | 5 | 39645 | 765 | 63.2 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 880 | 1976-120A | 9601 | 9-Dec-76 | 27-Nov-78 | 49 | 0 | 620 | 550 | 65.8 | BATTERY | |
| COSMOS 884 | 1976-123A | 9614 | 17-Dec-76 | 29-Dec-76 | 2 | 0 | 320 | 170 | 65.0 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 886 | 1976-126A | 9634 | 27-Dec-76 | 27-Dec-76 | 76 | 58 | 2295 | 595 | 65.8 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 903 | 1977-027A | 9911 | 11-Apr-77 | 8-Jun-78 | 6 | 3 | 39035 | 1325 | 63.2 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 917 | 1977-047A | 10059 | 16-Jun-77 | 30-Mar-79 | 14 | 12 | 38725 | 1645 | 62.9 | DELIBERATE | SELF-DESTRUCT |
| HIMAWARI 1 R/B | 1977-065B | 10144 | 14-Jul-77 | 14-Jul-77 | 177 | 59 | 2025 | 535 | 29.0 | PROPULSION | DELTA SECOND STAGE |
| COSMOS 931 | 1977-068A | 10150 | 20-Jul-77 | 24-Oct-77 | 6 | 5 | 39665 | 680 | 62.9 | DELIBERATE | SELF-DESTRUCT |
| EKRAN 2 | 1977-092A | 10365 | 20-Sep-77 | 23-Jun-78 | 5 | 5 | 35800 | 35785 | 0.1 | BATTERY | |
| COSMOS 970 | 1977-121A | 10531 | 21-Dec-77 | 21-Dec-77 | 70 | 64 | 1140 | 945 | 65.8 | DELIBERATE | SELF-DESTRUCT |
| LANDSAT 3 R/B | 1978-026C | 10704 | 5-Mar-78 | 27-Jan-81 | 248 | 153 | 910 | 900 | 98.8 | PROPULSION | DELTA SECOND STAGE |
| COSMOS 1030 | 1978-083A | 11015 | 6-Sep-78 | 10-Oct-78 | 13 | 9 | 39760 | 665 | 62.8 | DELIBERATE | SELF-DESTRUCT |
| NIMBUS 7 R/B | 1978-098B | 11081 | 24-Oct-78 | 26-Dec-81 | 2 | 1 | 955 | 935 | 99.3 | PROPULSION | DELTA SECOND STAGE |
| COSMOS 1045 R/B | 1978-100D | 11087 | 26-Oct-78 | 9-May-88 | 42 | 32 | 1705 | 1685 | 82.6 | PROPULSION | TSYKLON THIRD STAGE |
| P-78/SOLWIND | 1979-017A | 11278 | 24-Feb-79 | 13-Sep-85 | 285 | 0 | 545 | 515 | 97.6 | COLLISION, DELIBERATE | HYPERVELOCITY IMPACT |
| COSMOS 1094 | 1979-033A | 11333 | 18-Apr-79 | 17-Sep-79 | 1 | 0 | 405 | 380 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 1109 | 1979-058A | 11417 | 27-Jun-79 | Feb-80 | 19 | 8 | 39425 | 960 | 63.3 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1124 | 1979-077A | 11509 | 28-Aug-79 | 9-Sep-79 | 5 | 5 | 39795 | 570 | 63.0 | DELIBERATE | SELF-DESTRUCT |
| SATCOM 3 | 1979-101A | 11635 | 7-Dec-79 | 11-Dec-79 | 21 | 21 | 35776 | 204 | 23.7 | PROPULSION | AKM FAILURE |
| CAT R/B | 1979-104B | 11659 | 24-Dec-79 | Apr-80 | 31 | 24 | 33140 | 180 | 17.9 | PROPULSION | ARIANE 1 FINAL STAGE |

TABLE 2.1 HISTORY OF SATELLITE BREAKUPS BY LAUNCH DATE (CONT'D)

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | BREAKUP DATE | DEBRIS CATALOGED | DEBRIS ON-ORBIT | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) | ASSESSED CAUSE | ADDITIONAL INFORMATION |
|-----------------------------|--------------------------|---------------------|-------------|--------------|------------------|-----------------|-------------|--------------|-------------------|----------------|------------------------|
| COSMOS 1167 | 1980-021A | 11729 | 14-Mar-80 | 15-Jul-81 | 12 | 0 | 450 | 355 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 1174 | 1980-030A | 11765 | 18-Apr-80 | 18-Apr-80 | 46 | 4 | 1660 | 380 | 66.1 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1188 | 1980-050A | 11844 | 14-Jun-80 | 26-Aug-80 | 8 | 6 | 39630 | 735 | 62.9 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1191 | 1980-057A | 11871 | 2-Jul-80 | 14-May-81 | 11 | 11 | 39255 | 1110 | 62.6 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1217 | 1980-085A | 12032 | 24-Oct-80 | 12-Feb-83 | 10 | 8 | 38830 | 1530 | 65.2 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1220 | 1980-089A | 12054 | 4-Nov-80 | 20-Jun-82 | 83 | 1 | 885 | 570 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 1247 | 1981-016A | 12303 | 19-Feb-81 | 20-Oct-81 | 7 | 6 | 39390 | 970 | 63.0 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1260 | 1981-028A | 12364 | 20-Mar-81 | 8-May-82 | 68 | 0 | 750 | 450 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 1261 | 1981-031A | 12376 | 31-Mar-81 | Apr-81 | 10 | 10 | 39765 | 610 | 63.0 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1275 | 1981-053A | 12504 | 4-Jun-81 | 24-Jul-81 | 479 | 421 | 1015 | 960 | 83.0 | BATTERY | |
| COSMOS 1278 | 1981-058A | 12547 | 19-Jun-81 | Dec-86 | 3 | 0 | 37690 | 2665 | 67.1 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1285 | 1981-071A | 12627 | 4-Aug-81 | 21-Nov-81 | 25 | 25 | 40100 | 720 | 63.1 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1286 | 1981-072A | 12631 | 4-Aug-81 | 29-Sep-82 | 2 | 0 | 325 | 300 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 1305 R/B | 1981-088F | 12827 | 11-Sep-81 | 11-Sep-81 | 8 | 8 | 13795 | 605 | 62.8 | PROPULSION | MOLNIYA FINAL STAGE |
| COSMOS 1306 | 1981-089A | 12828 | 14-Sep-81 | 12-Jul-82 | 8 | 0 | 405 | 380 | 64.9 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 1317 | 1981-108A | 12933 | 31-Oct-81 | 25-28 Jan-84 | 11 | 11 | 39055 | 1315 | 62.8 | DELIBERATE | SELF-DESTRUCT |
| METEOR 2-8 | 1982-025A | 13113 | 25-Mar-82 | 29-May-99 | 53 | 53 | 960 | 935 | 82.5 | UNKNOWN | |
| COSMOS 1348 | 1982-029A | 13124 | 7-Apr-82 | 2-Sep-84 | 10 | 10 | 39200 | 1185 | 62.8 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1355 | 1982-038A | 13150 | 29-Apr-82 | 8-Aug-83 | 29 | 0 | 395 | 360 | 65.1 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 1375 | 1982-055A | 13259 | 6-Jun-82 | 21-Oct-85 | 61 | 58 | 1000 | 990 | 65.8 | BATTERY | |
| COSMOS 1405 | 1982-088A | 13508 | 4-Sep-82 | 20-Dec-83 | 32 | 0 | 340 | 310 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 1423 R/B | 1982-115E | 13696 | 8-Dec-82 | 8-Dec-82 | 29 | 0 | 425 | 235 | 62.9 | PROPULSION | MOLNIYA FINAL STAGE |
| ASTRON ULLAGE MOTOR | 1983-020B | 13902 | 23-Mar-83 | 3-Sep-84 | 1 | 0 | 1230 | 220 | 51.5 | PROPULSION | PROTON-K BLOCK DM SOZ |
| NOAA 8 | 1983-022A | 13923 | 28-Mar-83 | 30-Dec-85 | 5 | 1 | 830 | 805 | 98.6 | BATTERY | |
| COSMOS 1456 | 1983-038A | 14034 | 25-Apr-83 | 13-Aug-83 | 4 | 0 | 39630 | 730 | 63.3 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1461 | 1983-044A | 14064 | 7-May-83 | 11-Mar-85 | 187 | 14 | 890 | 570 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 1481 | 1983-070A | 14182 | 8-Jul-83 | 9-Jul-83 | 7 | 7 | 39225 | 625 | 62.9 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1484 | 1983-075A | 14207 | 24-Jul-83 | 18-Oct-93 | 49 | 0 | 595 | 550 | 97.5 | UNKNOWN | |
| COSMOS 1519-21 ULLAGE MOTOR | 1983-127H | 14608 | 29-Dec-83 | 4-Feb-91 | 8 | 3 | 18805 | 340 | 51.9 | PROPULSION | PROTON-K BLOCK DM SOZ |
| PALAPA B2 R/B | 1984-011E | 14693 | 3-Feb-84 | 6-Feb-84 | 3 | 0 | 285 | 275 | 28.5 | PROPULSION | PAM-D UPPER STAGE |
| WESTAR 6 R/B | 1984-011F | 14694 | 3-Feb-84 | 3-Feb-84 | 14 | 0 | 310 | 305 | 28.5 | PROPULSION | PAM-D UPPER STAGE |
| COSMOS 1588 | 1984-083A | 15167 | 7-Aug-84 | 23-Feb-86 | 45 | 0 | 440 | 410 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 1603 ULLAGE MOTOR | 1984-106F | 15338 | 28-Sep-84 | 5-Sep-92 | 23 | 2 | 845 | 835 | 66.6 | PROPULSION | PROTON-K BLOCK DM SOZ |

TABLE 2.1 HISTORY OF SATELLITE BREAKUPS BY LAUNCH DATE (CONT'D)

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | BREAKUP DATE | DEBRIS CATALOGED | DEBRIS ON-ORBIT | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) | ASSESSED CAUSE | ADDITIONAL INFORMATION |
|-----------------------------|--------------------------|---------------------|-------------|--------------|------------------|-----------------|-------------|--------------|-------------------|-----------------------|------------------------|
| SPACENET 2/MARECS B2 R/B | 1984-114C | 15388 | 10-Nov-84 | 20-Nov-84 | 3 | 2 | 35960 | 325 | 7.0 | PROPULSION | ARIANE 3 FINAL STAGE |
| COSMOS 1646 | 1985-030A | 15653 | 18-Apr-85 | 20-Nov-87 | 24 | 0 | 410 | 385 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 1650-52 ULLAGE MOTOR | 1985-037G | 15714 | 17-May-85 | 29-Nov-98 | 4 | 2 | 18620 | 320 | 52.0 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 1654 | 1985-039A | 15734 | 23-May-85 | 21-Jun-85 | 18 | 0 | 300 | 185 | 64.9 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1656 ULLAGE MOTOR | 1985-042E | 15773 | 30-May-85 | 5-Jan-88 | 6 | 6 | 860 | 810 | 66.6 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 1682 | 1985-082A | 16054 | 19-Sep-85 | 18-Dec-86 | 23 | 0 | 475 | 385 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 1691 (1695) | 1985-094B | 16139 | 9-Oct-85 | 22-Nov-85 | 21 | 18 | 1415 | 1410 | 82.6 | BATTERY | |
| COSMOS 1703 R/B | 1985-108B | 16263 | 22-Nov-85 | 4-May-06 | 50 | 2 | 640 | 610 | 82.5 | PROPULSION | TSYKLON THIRD STAGE |
| COSMOS 1710-12 ULLAGE MOTOR | 1985-118L | 16446 | 24-Dec-85 | 29-Dec-91 | 17 | 10 | 18885 | 655 | 65.3 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 1714 R/B | 1985-121F | 16439 | 28-Dec-85 | 28-Dec-85 | 2 | 0 | 830 | 165 | 71.0 | PROPULSION | ZENIT SECOND STAGE |
| SPOT 1 R/B | 1986-019C | 16615 | 22-Feb-86 | 13-Nov-86 | 498 | 31 | 835 | 805 | 98.7 | PROPULSION | ARIANE 1 FINAL STAGE |
| COSMOS 1769 | 1986-059A | 16895 | 4-Aug-86 | 21-Sep-87 | 4 | 0 | 445 | 310 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| USA 19 | 1986-069A | 16937 | 5-Sep-86 | 5-Sep-86 | 13 | 0 | 745 | 210 | 39.1 | COLLISION, DELIBERATE | HYPERVELOCITY IMPACT |
| USA 19 R/B | 1986-069B | 16938 | 5-Sep-86 | 5-Sep-86 | 5 | 0 | 610 | 220 | 22.8 | COLLISION, DELIBERATE | HYPERVELOCITY IMPACT |
| COSMOS 1813 | 1987-004A | 17297 | 15-Jan-87 | 29-Jan-87 | 195 | 0 | 415 | 360 | 72.8 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1823 | 1987-020A | 17535 | 20-Feb-87 | 17-Dec-87 | 150 | 76 | 1525 | 1480 | 73.6 | BATTERY | |
| COSMOS 1866 | 1987-059A | 18184 | 9-Jul-87 | 26-Jul-87 | 9 | 0 | 255 | 155 | 67.1 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1869 | 1987-062A | 18214 | 16-Jul-87 | 27-Nov-97 | 2 | 1 | 635 | 605 | 83.0 | UNKNOWN | |
| METEOR 2-16 R/B | 1987-068B | 18313 | 18-Aug-87 | 15-Feb-98 | 108 | 42 | 960 | 940 | 82.6 | PROPULSION | TSYKLON THIRD STAGE |
| AUSSAT K3/ECS 4 R/B | 1987-078C | 18352 | 16-Sep-87 | 16-19 Sep-87 | 4 | 1 | 36515 | 245 | 6.9 | PROPULSION | ARIANE 3 FINAL STAGE |
| COSMOS 1883-85 ULLAGE MOTOR | 1987-079G | 18374 | 16-Sep-87 | 1-Dec-96 | 14 | 11 | 19120 | 335 | 64.9 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 1883-85 ULLAGE MOTOR | 1987-079H | 18375 | 16-Sep-87 | 23-Apr-03 | 42 | 11 | 18540 | 755 | 65.2 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 1906 | 1987-108A | 18713 | 26-Dec-87 | 31-Jan-88 | 37 | 0 | 265 | 245 | 82.6 | DELIBERATE | SELF-DESTRUCT |
| EKRAN 17 ULLAGE MOTOR | 1987-109E | 18719 | 27-Dec-87 | 22-May-97 | 1 | 0 | 22975 | 310 | 46.6 | PROPULSION | PROTON-K BLOCK DM SOZ |
| METEOR 2-17 | 1988-005A | 18820 | 30-Jan-88 | 21-Jun-05 | 45 | 45 | 960 | 930 | 82.5 | UNKNOWN | |
| DMSP 5D-2 F9 (USA 29) | 1988-006A | 18822 | 3-Feb-88 | 14-17 Dec-12 | 10 | 3 | 810 | 800 | 98.8 | UNKNOWN | |
| COSMOS 1916 | 1988-007A | 18823 | 3-Feb-88 | 27-Feb-88 | 1 | 0 | 230 | 150 | 64.8 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1934 | 1988-023A | 18985 | 22-Mar-88 | 23-Dec-91 | 3 | 3 | 1010 | 950 | 83.0 | COLLISION, ACCIDENTAL | HIT BY DEBRIS (13475) |
| INTELSAT 513 R/B | 1988-040B | 19122 | 17-May-88 | 9-Jul-02 | 8 | 8 | 35445 | 535 | 7.0 | PROPULSION | ARIANE 2 R/B |
| COSMOS 1970-72 ULLAGE MOTOR | 1988-085F | 19535 | 16-Sep-88 | 4-Aug-03 | 79 | 10 | 18515 | 720 | 65.3 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 1970-72 ULLAGE MOTOR | 1988-085G | 19537 | 16-Sep-88 | 9-Mar-99 | 1 | 1 | 18950 | 300 | 64.6 | PROPULSION | PROTON-K BLOCK DM SOZ |

TABLE 2.1 HISTORY OF SATELLITE BREAKUPS BY LAUNCH DATE (CONT'D)

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | BREAKUP DATE | DEBRIS CATALOGED | DEBRIS ON-ORBIT | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) | ASSESSED CAUSE | ADDITIONAL INFORMATION |
|-----------------------------|--------------------------|---------------------|-------------|--------------|------------------|-----------------|-------------|--------------|-------------------|----------------|--------------------------|
| SKYNET 4B/ASTRA 1A R/B | 1988-109C | 19689 | 11-Dec-88 | 17-Feb-98 | 18 | 17 | 35875 | 435 | 7.3 | PROPULSION | ARIANE 4 H10 FINAL STAGE |
| COSMOS 1987-89 ULLAGE MOTOR | 1989-001G | 19755 | 10-Jan-89 | 3-Aug-98 | 16 | 6 | 19055 | 340 | 64.9 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 1987-89 ULLAGE MOTOR | 1989-001H | 19856 | 10-Jan-89 | 13-Nov-03 | 1 | 1 | 18740 | 710 | 65.4 | PROPULSION | PROTON-K BLOCK DM SOZ |
| GORIZONT 17 ULLAGE MOTOR | 1989-004E | 19771 | 26-Jan-89 | 17-Dec-92 | 1 | 0 | 17575 | 195 | 46.7 | PROPULSION | PROTON-K BLOCK DM SOZ |
| INTELSAT 515 R/B | 1989-006B | 19773 | 27-Jan-89 | 1-Jan-01 | 87 | 86 | 35720 | 510 | 8.4 | PROPULSION | ARIANE 2 R/B |
| COSMOS 2022-24 ULLAGE MOTOR | 1989-039G | 20081 | 31-May-89 | 10-Jun-06 | 120 | 43 | 18410 | 655 | 65.1 | PROPULSION | PROTON-K BLOCK DM SOZ |
| GORIZONT 18 ULLAGE MOTOR | 1989-052F | 20116 | 5-Jul-89 | 12-Jan-93 | 2 | 0 | 36745 | 260 | 46.8 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2030 | 1989-054A | 20124 | 12-Jul-89 | 28-Jul-89 | 1 | 0 | 215 | 150 | 67.1 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 2031 | 1989-056A | 20136 | 18-Jul-89 | 31-Aug-89 | 9 | 0 | 365 | 240 | 50.5 | DELIBERATE | SELF-DESTRUCT |
| COBE R/B | 1989-089B | 20323 | 18-Nov-89 | 3-Dec-06 | 26 | 1 | 790 | 685 | 97.1 | UNKNOWN | DELTA SECOND STAGE |
| COSMOS 2053 R/B | 1989-100B | 20390 | 27-Dec-89 | 18-Apr-99 | 26 | 0 | 485 | 475 | 73.5 | PROPULSION | TSYKLON THIRD STAGE |
| COSMOS 2054 ULLAGE MOTOR | 1989-101E | 20399 | 27-Dec-89 | Jul-92 | 14 | 2 | 27650 | 345 | 47.1 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2079-81 ULLAGE MOTOR | 1990-045F | 20630 | 19-May-90 | 17-Nov-11 | 1 | 1 | 18620 | 420 | 65.0 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2079-81 ULLAGE MOTOR | 1990-045G | 20631 | 19-May-90 | 28-Mar-99 | 1 | 1 | 19065 | 405 | 64.8 | PROPULSION | PROTON-K BLOCK DM SOZ |
| FENGYUN 1-2 R/B | 1990-081D | 20791 | 3-Sep-90 | 4-Oct-90 | 103 | 75 | 895 | 880 | 98.9 | PROPULSION | CZ-4 FINAL STAGE |
| COSMOS 2101 | 1990-087A | 20828 | 1-Oct-90 | 30-Nov-90 | 4 | 0 | 280 | 195 | 64.8 | DELIBERATE | SELF-DESTRUCT |
| GORIZONT 22 ULLAGE MOTOR | 1990-102E | 20957 | 23-Nov-90 | 14-Dec-95 | 2 | 1 | 13105 | 170 | 46.5 | PROPULSION | PROTON-K BLOCK DM SOZ |
| USA 68 | 1990-105A | 20978 | 1-Dec-90 | 1-Dec-90 | 29 | 1 | 850 | 610 | 98.9 | PROPULSION | TE-M-364-15 UPPER STAGE |
| COSMOS 2109-11 ULLAGE MOTOR | 1990-110G | 21012 | 8-Dec-90 | 21-Feb-03 | 1 | 1 | 18805 | 645 | 65.4 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2109-11 ULLAGE MOTOR | 1990-110H | 21013 | 8-Dec-90 | 14-Mar-98 | 2 | 2 | 18995 | 520 | 65.1 | PROPULSION | PROTON-K BLOCK DM SOZ |
| ITALSAT 1/EUTELSAT 2 F2 R/B | 1991-003C | 21057 | 15-Jan-91 | Apr/May 96 | 15 | 12 | 30930 | 235 | 6.7 | PROPULSION | ARIANE 4 H10 FINAL STAGE |
| COSMOS 2125-32 R/B | 1991-009J | 21108 | 12-Feb-91 | 5-Mar-91 | 112 | 112 | 1725 | 1460 | 74.0 | PROPULSION | COSMOS SECOND STAGE |
| COSMOS 2133 ULLAGE MOTOR | 1991-010D | 21114 | 12-Feb-91 | 7-May-94 | 4 | 0 | 21805 | 225 | 46.6 | PROPULSION | PROTON-K BLOCK DM SOZ |
| ASTRA 1B/MOP 2 R/B | 1991-015C | 21141 | 2-Mar-91 | 27-Apr-94 | 11 | 7 | 17630 | 205 | 6.8 | PROPULSION | ARIANE 4 H10 FINAL STAGE |
| COSMOS 2139-41 ULLAGE MOTOR | 1991-025F | 21220 | 4-Apr-91 | 8-Mar-09 | 33 | 11 | 18535 | 465 | 64.9 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2139-41 ULLAGE MOTOR | 1991-025G | 21226 | 4-Apr-91 | 16-Jun-01 | 1 | 1 | 18960 | 300 | 64.5 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2157-62 R/B | 1991-068G | 21734 | 28-Sep-91 | 9-Oct-99 | 40 | 40 | 1485 | 1410 | 82.6 | PROPULSION | TSYKLON THIRD STAGE |
| COSMOS 2163 | 1991-071A | 21741 | 9-Oct-91 | 6-Dec-91 | 1 | 0 | 260 | 185 | 64.8 | DELIBERATE | SELF-DESTRUCT |
| INTELSAT 601 R/B | 1991-075B | 21766 | 29-Oct-91 | 24-Dec-01 | 13 | 8 | 28505 | 230 | 7.2 | PROPULSION | ARIANE 4 H10 FINAL STAGE |
| USA 73 (DMSP 5D2 F11) | 1991-082A | 21798 | 28-Nov-91 | 15-Apr-04 | 85 | 65 | 850 | 830 | 98.7 | UNKNOWN | |

TABLE 2.1 HISTORY OF SATELLITE BREAKUPS BY LAUNCH DATE (CONT'D)

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | BREAKUP DATE | DEBRIS CATALOGED | DEBRIS ON-ORBIT | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) | ASSESSED CAUSE | ADDITIONAL INFORMATION |
|-----------------------------|--------------------------|---------------------|-------------|--------------|------------------|-----------------|-------------|--------------|-------------------|-----------------------|---------------------------|
| TELECOM 2B/INMARSAT 2 R/B | 1992-021C | 21941 | 15-Apr-92 | 21-Apr-93 | 18 | 17 | 34080 | 235 | 4.0 | PROPULSION | ARIANE 4 H10+ FINAL STAGE |
| INSAT 2A/EUTELSAT 2F4 R/B | 1992-041C | 22032 | 9-Jul-92 | Feb-02 | 2 | 2 | 26550 | 250 | 7.0 | PROPULSION | ARIANE 4 H10 FINAL STAGE |
| COSMOS 2204-06 ULLAGE MOTOR | 1992-047G | 22066 | 30-Jul-92 | 10-Jul-04 | 34 | 12 | 18820 | 415 | 64.9 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2204-06 ULLAGE MOTOR | 1992-047H | 22067 | 30-Jul-92 | 8-Nov-94 | 4 | 2 | 19035 | 480 | 64.8 | PROPULSION | PROTON-K BLOCK DM SOZ |
| GORIZONT 27 ULLAGE MOTOR | 1992-082F | 22250 | 27-Nov-92 | 14-Jul-01 | 1 | 0 | 5340 | 145 | 46.5 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2224 ULLAGE MOTOR | 1992-088F | 22274 | 17-Dec-92 | ~22-Apr-05 | 1 | 0 | 21140 | 200 | 46.7 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2225 | 1992-091A | 22280 | 22-Dec-92 | 18-Feb-93 | 6 | 0 | 280 | 225 | 64.9 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 2227 R/B | 1992-093B | 22285 | 25-Dec-92 | 26-Dec-92 | 279 | 196 | 855 | 845 | 71.0 | PROPULSION | ZENIT-2 SECOND STAGE |
| COSMOS 2237 R/B | 1993-016B | 22566 | 26-Mar-93 | 28-Mar-93 | 104 | 100 | 850 | 840 | 71.0 | PROPULSION | ZENIT-2 SECOND STAGE |
| COSMOS 2238 | 1993-018A | 22585 | 30-Mar-93 | 1-Dec-94 | 1 | 0 | 305 | 210 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 2243 | 1993-028A | 22641 | 27-Apr-93 | 27-Apr-93 | 1 | 0 | 225 | 180 | 70.4 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 2251 | 1993-036A | 22675 | 16-Jun-93 | 10-Feb-09 | 1668 | 1076 | 800 | 775 | 74.0 | COLLISION, ACCIDENTAL | COLLIDED WITH IRIDIUM 33 |
| COSMOS 2259 | 1993-045A | 22716 | 14-Jul-93 | 25-Jul-93 | 1 | 0 | 320 | 175 | 67.1 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 2262 | 1993-057A | 22789 | 7-Sep-93 | 18-Dec-93 | 1 | 0 | 295 | 170 | 64.9 | DELIBERATE | SELF-DESTRUCT |
| GORIZONT 29 ULLAGE MOTOR | 1993-072E | 22925 | 18-Nov-93 | 6-Sep-00 | 1 | 0 | 11215 | 140 | 46.7 | PROPULSION | PROTON-K BLOCK DM SOZ |
| CLEMENTINE R/B | 1994-004B | 22974 | 25-Jan-94 | 7-Feb-94 | 1 | 0 | 295 | 240 | 67.0 | PROPULSION | |
| STEP II R/B | 1994-029B | 23106 | 19-May-94 | 3-Jun-96 | 754 | 82 | 820 | 585 | 82.0 | PROPULSION | PEGASUS HAPS |
| COSMOS 2282 ULLAGE MOTOR | 1994-038F | 23174 | 6-Jul-94 | 21-Oct-95 | 2 | 0 | 34930 | 280 | 47.0 | PROPULSION | PROTON-K BLOCK DM SOZ |
| ELEKTRO ULLAGE MOTOR | 1994-069E | 23338 | 31-Oct-94 | 11-May-95 | 1 | 0 | 35465 | 155 | 46.9 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2294-96 ULLAGE MOTOR | 1994-076G | 23402 | 20-Nov-94 | 7-Jun-14 | 2 | 2 | 18990 | 420 | 65.0 | PROPULSION | PROTON-K BLOCK DM SOZ |
| RS-15 R/B | 1994-085B | 23440 | 26-Dec-94 | 26-Dec-94 | 26 | 23 | 2200 | 1880 | 64.8 | UNKNOWN | ROKOT THIRD STAGE |
| USA 109 (DMSP 5D-2 F13) | 1995-015A | 23533 | 24-Mar-95 | 3-Feb-15 | 236 | 219 | 840 | 840 | 98.8 | BATTERY | Operational at event time |
| COSMOS 2313 | 1995-028A | 23596 | 8-Jun-95 | 26-Jun-97 | 13 | 0 | 325 | 210 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| CERISE | 1995-033B | 23606 | 7-Jul-95 | 24-Jul-96 | 2 | 1 | 675 | 665 | 98.1 | COLLISION, ACCIDENTAL | HIT BY DEBRIS (18208) |
| COSMOS 2316-18 ULLAGE MOTOR | 1995-037K | 23631 | 24-Jul-95 | 21-Nov-00 | 1 | 0 | 18085 | 150 | 64.4 | PROPULSION | PROTON-K BLOCK DM SOZ |
| RADUGA 33 R/B | 1996-010D | 23797 | 19-Feb-96 | 19-Feb-96 | 1 | 0 | 36505 | 240 | 48.7 | PROPULSION | PROTON-K BLOCK DM |
| GORIZONT 32 ULLAGE MOTOR | 1996-034F | 23887 | 25-May-96 | 13-Dec-99 | 1 | 0 | 5605 | 145 | 46.5 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2343 | 1997-024A | 24805 | 15-May-97 | 16-Sep-97 | 1 | 0 | 285 | 225 | 65.0 | DELIBERATE | SELF-DESTRUCT |
| IRIDIUM 33 | 1997-051C | 24946 | 14-Sep-97 | 10-Feb-09 | 628 | 333 | 780 | 775 | 86.4 | COLLISION, ACCIDENTAL | COLLIDED WITH COSMOS 2251 |
| KUPON ULLAGE MOTOR | 1997-070F | 25054 | 12-Nov-97 | 14-Feb-07 | 7 | 4 | 14160 | 260 | 46.6 | PROPULSION | PROTON-K BLOCK DM SOZ |

TABLE 2.1 HISTORY OF SATELLITE BREAKUPS BY LAUNCH DATE (CONT'D)

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | BREAKUP DATE | DEBRIS CATALOGED | DEBRIS ON-ORBIT | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) | ASSESSED CAUSE | ADDITIONAL INFORMATION |
|-------------------------------|--------------------------|---------------------|-------------|--------------|------------------|-----------------|-------------|--------------|-------------------|-----------------------|------------------------|
| COSMOS 2347 | 1997-079A | 25088 | 9-Dec-97 | 22-Nov-99 | 9 | 0 | 410 | 230 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| ASIASAT 3 R/B | 1997-086D | 25129 | 24-Dec-97 | 25-Dec-97 | 1 | 0 | 35995 | 270 | 51.0 | PROPULSION | PROTON-K BLOCK DM |
| COMETS R/B | 1998-011B | 25176 | 21-Feb-98 | 21-Feb-98 | 1 | 0 | 1880 | 245 | 30.0 | PROPULSION | H-II SECOND STAGE |
| ARGOS/ORSTED/SUN SAT R/B | 1999-008D | 25637 | 23-Feb-99 | 28-Apr-14 | 8 | 1 | 840 | 635 | 96.5 | UNKNOWN | DELTA 2 SECOND STAGE |
| FENGYUN 1C | 1999-025A | 25730 | 10-May-99 | 11-Jan-07 | 3442 | 2832 | 865 | 845 | 98.6 | COLLISION, DELIBERATE | HYPERVELOCITY IMPACT |
| CBERS 1 | 1999-057A | 25940 | 14-Oct-99 | 18-Feb-07 | 88 | 53 | 780 | 770 | 98.2 | UNKNOWN | |
| CBERS 1/SACI 1 R/B | 1999-057C | 25942 | 14-Oct-99 | 11-Mar-00 | 344 | 150 | 745 | 725 | 98.5 | PROPULSION | CZ-4 FINAL STAGE |
| COSMOS 2367 | 1999-072A | 26040 | 26-Dec-99 | 21-Nov-01 | 17 | 0 | 415 | 405 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 2371 ULLAGE MOTOR | 2000-036E | 26398 | 4-Jul-00 | ~1-Sep-06 | 1 | 0 | 21320 | 220 | 46.9 | PROPULSION | PROTON-K BLOCK DM SOZ |
| NOAA 16 | 2000-055A | 26536 | 21-Sep-00 | 25-Nov-15 | 458 | 458 | 858 | 842 | 98.9 | UNKNOWN | |
| TES R/B | 2001-049D | 26960 | 22-Oct-01 | 19-Dec-01 | 372 | 72 | 675 | 550 | 97.9 | PROPULSION | PSLV FINAL STAGE |
| COSMOS 2383 | 2001-057A | 27053 | 21-Dec-01 | 28-Feb-04 | 14 | 0 | 400 | 220 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 2392 ULLAGE MOTOR | 2002-037E | 27474 | 25-Jul-02 | 1-Jun-05 | 61 | 0 | 835 | 255 | 63.7 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2392 ULLAGE MOTOR | 2002-037F | 27475 | 25-Jul-02 | 29-Oct-04 | 1 | 0 | 840 | 235 | 63.6 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2399 | 2003-035A | 27856 | 12-Aug-03 | 9-Dec-03 | 22 | 0 | 250 | 175 | 64.9 | DELIBERATE | SELF-DESTRUCT |
| ALOS-1 R/B | 2006-002B | 28932 | 24-Jan-06 | 8-Aug-06 | 24 | 3 | 700 | 550 | 98.2 | UNKNOWN | H-IIA SECOND STAGE |
| ARABSAT 4 BRIZ-M R/B | 2006-006B | 28944 | 28-Feb-06 | 19-Feb-07 | 102 | 92 | 14705 | 495 | 51.5 | PROPULSION | PROTON-K BRIZ-M STAGE |
| YAOGAN 1 | 2006-015A | 29092 | 26-Apr-06 | 4-Feb-10 | 8 | 2 | 630 | 625 | 97.9 | UNKNOWN | |
| COSMOS 2421 | 2006-026A | 29247 | 25-Jun-06 | 14-Mar-08 | 509 | 0 | 420 | 400 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| IGS 3A R/B | 2006-037B | 29394 | 11-Sep-06 | 28-Dec-06 | 10 | 0 | 490 | 430 | 97.2 | UNKNOWN | H-IIA SECOND STAGE |
| COSMOS 2423 | 2006-039A | 29402 | 14-Sep-06 | 17-Nov-06 | 31 | 0 | 285 | 200 | 64.9 | DELIBERATE | SELF-DESTRUCT |
| DMSP 5D-3 F17 R/B | 2006-050B | 29523 | 4-Nov-06 | 4-Nov-06 | 65 | 0 | 865 | 830 | 98.8 | UNKNOWN | DELTA IV SECOND STAGE |
| USA 193 | 2006-057A | 29651 | 14-Dec-06 | 21-Feb-08 | 175 | 0 | 255 | 245 | 58.5 | COLLISION, DELIBERATE | HYPERVELOCITY IMPACT |
| COSMOS 2424-2426 ULLAGE MOTOR | 2006-062G | 29680 | 25-Dec-06 | 27-Jul-16 | 9 | 8 | 19088 | 426 | 64.8 | PROPULSION | PROTON-K BLOCK DM SOZ |
| BEIDOU 1D R/B | 2007-003B | 30324 | 2-Feb-07 | 2-Feb-07 | 39 | 36 | 41900 | 235 | 25.0 | UNKNOWN | CZ-3A FINAL STAGE |
| IGS 4A/4B R/B DEBRIS | 2007-005E | 30590 | 24-Feb-07 | 23-Dec-10 | 4 | 0 | 440 | 430 | 97.3 | UNKNOWN | H-IIA DEBRIS |
| COSMOS 2428 | 2007-029A | 31792 | 29-Jun-07 | 10-May-14 | 10 | 2 | 860 | 845 | 71.0 | UNKNOWN | |
| COSMOS 2431-33 ULLAGE MOTOR | 2007-052F | 32280 | 26-Oct-07 | 13-Aug-14 | 25 | 25 | 18790 | 730 | 65.0 | PROPULSION | PROTON-K BLOCK DM SOZ |
| USA 197 R/B | 2007-054B | 32288 | 11-Nov-07 | 11-Nov-07 | 32 | 32 | 1575 | 220 | 29.0 | UNKNOWN | DELTA IV SECOND STAGE |
| COSMOS 2434-36 ULLAGE MOTOR | 2007-065G | 32399 | 25-Dec-07 | 18-Aug-11 | 1 | 1 | 18965 | 540 | 65.0 | PROPULSION | PROTON-K BLOCK DM SOZ |
| AMC 14 BRIZ-M R/B | 2008-011B | 32709 | 14-Mar-08 | 13-Oct-10 | 116 | 115 | 26565 | 645 | 48.9 | PROPULSION | PROTON-K BRIZ-M STAGE |

TABLE 2.1 HISTORY OF SATELLITE BREAKUPS BY LAUNCH DATE (CONT'D)

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | BREAKUP DATE | DEBRIS CATALOGED | DEBRIS ON-ORBIT | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) | ASSESSED CAUSE | ADDITIONAL INFORMATION |
|-----------------------------------|--------------------------|---------------------|-------------|--------------|------------------|-----------------|-------------|--------------|-------------------|-----------------------|--|
| COSMOS 2442-44 ULLAGE MOTOR | 2008-046H | 33385 | 25-Sep-08 | 20-May-14 | 11 | 10 | 18720 | 865 | 65.0 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2447-49 ULLAGE MOTOR | 2008-067G | 33472 | 25-Dec-08 | 26-Mar-16 | 4 | 4 | 18840 | 682 | 65.4 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2447-49 ULLAGE MOTOR | 2008-067H | 33473 | 25-Dec-08 | 1-Jun-16 | 12 | 12 | 18786 | 709 | 65.3 | | |
| BEIDOU G2 | 2009-018A | 34779 | 14-Apr-09 | 29-Jun-16 | 1 | 1 | 36137 | 35384 | 4.7 | UNKNOWN | |
| COSMOS 2459-61 ULLAGE MOTOR | 2010-007G | 36406 | 1-Mar-10 | 9-Jul-14 | 11 | 11 | 18750 | 770 | 65.0 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2459-61 ULLAGE MOTOR | 2010-007H | 36407 | 1-Mar-10 | 22-May-18 | 11 | 11 | 18929 | 602 | 65.1 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2464-2466 ULLAGE MOTOR | 2010-041G | 37143 | 2-Sep-10 | 3-Sep-17 | 10 | 10 | 18684 | 756 | 65.2 | PROPULSION | PROTON-K BLOCK DM SOZ |
| CHINASAT 6A R/B | 2010-042B | 37151 | 4-Sep-10 | 4-Sep-10 | 30 | 29 | 41894 | 194 | 25.2 | UNKNOWN | |
| BEIDOU G4 R/B | 2010-057B | 37211 | 1-Nov-10 | 1-Nov-10 | 57 | 55 | 35780 | 160 | 20.5 | UNKNOWN | CZ-3C FINAL STAGE |
| SL-23 DEBRIS | 2011-037B | 37756 | 18-Jul-11 | 3/4-Aug-15 | 1 | 1 | 3649 | 428 | 51.4 | UNKNOWN | Fregat-SB SBB propellant tank |
| NIGCOMSAT 1R R/B | 2011-077B | 38015 | 19-Dec-11 | ~21-Dec-11 | 39 | 39 | 41715 | 230 | 24.3 | UNKNOWN | CZ-3B FINAL STAGE |
| BEIDOU G5 R/B | 2012-008B | 38092 | 24-Feb-12 | 26-Feb-12 | 38 | 37 | 35950 | 150 | 20.7 | UNKNOWN | CZ-3C FINAL STAGE |
| NIMIQ 6 R/B | 2012-026B | 38343 | 17-May-12 | 23-Dec-15 | 11 | 11 | 34592 | 10408 | 12.0 | UNKNOWN | Briz-M CORE |
| TELKOM 3 / EXPRESS MD2 BRIZ-M R/B | 2012-044C | 38746 | 6-Aug-12 | 16-Oct-12 | 113 | 20 | 5010 | 265 | 49.9 | PROPULSION | PROTON-K BRIZ-M STAGE |
| CASSIOPE R/B | 2013-055B | 39266 | 29-Sep-13 | 29-Sep-13 | 16 | 1 | 1490 | 320 | 81.0 | PROPULSION | FALCON 9 SECOND STAGE |
| SENTINEL 1A | 2014-016A | 39634 | 3-Apr-14 | 23-Aug-16 | 8 | 3 | 698 | 696 | 98.2 | COLLISION, ACCIDENTAL | Solar array impact by small MMOD |
| PROGRESS-M 27M R/B | 2015-024B | 40620 | 28-Apr-15 | 28-Apr-15 | 21 | 0 | 181 | 169 | 51.7 | UNKNOWN | Soyuz 2.1a (3); separation with PROGRESS M-27M |
| COSMOS 2513 R/B | 2015-075B | 41122 | 13-Dec-15 | 16-Jan-16 | 7 | 7 | 35777 | 33426 | 0.2 | UNKNOWN | Briz-M CORE |
| ASTRO H (HITOMI) | 2016-012A | 41337 | 17-Feb-16 | 26-Mar-16 | 13 | 10 | 578 | 563 | 31.0 | PROPULSION | likely structural failure mechanism |
| FREGAT DEB (TANK) | 2017-086C | 43089 | 26-Dec-17 | 12-Feb-18 | 5 | 2 | 4070 | 277 | 50.4 | UNKNOWN | SL-23 Fregat SB SBB propellant tank |

1. NA = NOT AVAILABLE

2. BREAKUP DATE AND ORBIT ARE FOR FIRST EVENT ONLY IF MULTIPLE EVENTS OCCURRED

3. DOES NOT INCLUDE SATELLITE BREAKUPS IF VEHICLE WAS ON REENTRY TRAJECTORY AT THE TIME OF THE EVENT

TABLE 2.2 HISTORY OF SATELLITE BREAKUPS BY EVENT DATE

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | BREAKUP DATE | DEBRIS CATALOGED | DEBRIS ON-ORBIT | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) | ASSESSED CAUSE | ADDITIONAL INFORMATION |
|--------------------|--------------------------|---------------------|-------------|--------------|------------------|-----------------|-------------|--------------|-------------------|----------------|------------------------|
| TRANSIT 4A R/B | 1961-015C | 118 | 29-Jun-61 | 29-Jun-61 | 296 | 172 | 995 | 880 | 66.8 | PROPULSION | ABLESTAR STAGE |
| SPUTNIK 29 | 1962-057A | 443 | 24-Oct-62 | 29-Oct-62 | 24 | 0 | 260 | 200 | 65.1 | PROPULSION | MOLNIYA FINAL STAGE |
| ATLAS CENTAUR 2 | 1963-047A | 694 | 27-Nov-63 | 27-Nov-63 | 19 | 8 | 1785 | 475 | 30.3 | PROPULSION | CENTAUR STAGE |
| COSMOS 50 | 1964-070A | 919 | 28-Oct-64 | 5-Nov-64 | 96 | 0 | 220 | 175 | 51.2 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 57 | 1965-012A | 1093 | 22-Feb-65 | 22-Feb-65 | 167 | 0 | 425 | 165 | 64.8 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 61-63 R/B | 1965-020D | 1270 | 15-Mar-65 | 15-Mar-65 | 147 | 18 | 1825 | 260 | 56.1 | UNKNOWN | COSMOS SECOND STAGE |
| OV2-1/LCS 2 R/B | 1965-082DM | 1822 | 15-Oct-65 | 15-Oct-65 | 473 | 32 | 790 | 710 | 32.2 | PROPULSION | TITAN TRANSTAGE |
| OV2-3/ et al. R/B | 1965-108A | 1863 | 21-Dec-65 | 21-Dec-65 | 108 | 101 | 33660 | 165 | 26.4 | PROPULSION | TITAN TRANSTAGE |
| COSMOS 95 | 1965-088A | 1706 | 4-Nov-65 | 15-Jan-66 | 1 | 0 | 300 | 180 | 48.4 | UNKNOWN | |
| OPS 3031 | 1966-012C | 2015 | 15-Feb-66 | 15-Feb-66 | 38 | 0 | 270 | 150 | 96.5 | UNKNOWN | INFLATABLE SPHERE |
| GEMINI 9 ATDA R/B | 1966-046B | 2188 | 1-Jun-66 | Jun-66 | 51 | 0 | 275 | 240 | 28.8 | UNKNOWN | ATLAS CORE STAGE |
| AS-203 | 1966-059A | 2289 | 5-Jul-66 | 5-Jul-66 | 34 | 0 | 215 | 185 | 32.0 | DELIBERATE | SATURN S-IVB STAGE |
| COSMOS U-1 | 1966-088A | 2437 | 17-Sep-66 | 17-Sep-66 | 52 | 0 | 855 | 140 | 49.6 | DELIBERATE | SELF-DESTRUCT |
| COSMOS U-2 | 1966-101A | 2536 | 2-Nov-66 | 2-Nov-66 | 41 | 0 | 885 | 145 | 49.6 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 199 | 1968-003A | 3099 | 16-Jan-68 | 24-Jan-68 | 3 | 0 | 355 | 200 | 65.6 | DELIBERATE | SELF-DESTRUCT |
| APOLLO 6 R/B | 1968-025B | 3171 | 4-Apr-68 | 13-Apr-68 | 16 | 0 | 360 | 200 | 32.6 | PROPULSION | SATURN S-IVB STAGE |
| COSMOS 249 | 1968-091A | 3504 | 20-Oct-68 | 20-Oct-68 | 108 | 39 | 2165 | 490 | 62.3 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 248 | 1968-090A | 3503 | 19-Oct-68 | 1-Nov-68 | 5 | 0 | 545 | 475 | 62.2 | DELIBERATE | DEBRIS IMPACT |
| COSMOS 252 | 1968-097A | 3530 | 1-Nov-68 | 1-Nov-68 | 139 | 40 | 2140 | 535 | 62.3 | DELIBERATE | SELF-DESTRUCT |
| METEOR 1-1 R/B | 1969-029B | 3836 | 26-Mar-69 | 28-Mar-69 | 37 | 0 | 850 | 460 | 81.2 | UNKNOWN | VOSTOK FINAL STAGE |
| INTELSAT 3 F-5 R/B | 1969-064B | 4052 | 26-Jul-69 | 26-Jul-69 | 22 | 0 | 5445 | 270 | 30.4 | PROPULSION | TE 364-4 STAGE |
| OPS 7613 R/B | 1969-082AB | 4159 | 30-Sep-69 | 4-Oct-69 | 261 | 63 | 940 | 905 | 70.0 | UNKNOWN | AGENA D STAGE |
| NIMBUS 4 R/B | 1970-025C | 4367 | 8-Apr-70 | 17-Oct-70 | 441 | 299 | 1085 | 1065 | 99.9 | UNKNOWN | AGENA D STAGE |
| COSMOS 374 | 1970-089A | 4594 | 23-Oct-70 | 23-Oct-70 | 99 | 20 | 2130 | 530 | 62.9 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 375 | 1970-091A | 4598 | 30-Oct-70 | 30-Oct-70 | 47 | 15 | 2100 | 525 | 62.8 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 397 | 1971-015A | 4964 | 25-Feb-71 | 25-Feb-71 | 116 | 44 | 2200 | 575 | 65.8 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 462 | 1971-106A | 5646 | 3-Dec-71 | 3-Dec-71 | 25 | 0 | 1800 | 230 | 65.7 | DELIBERATE | SELF-DESTRUCT |
| SALYUT 2 R/B | 1973-017B | 6399 | 3-Apr-73 | 3-Apr-73 | 25 | 0 | 245 | 195 | 51.5 | PROPULSION | PROTON-K THIRD STAGE |
| COSMOS 554 | 1973-021A | 6432 | 19-Apr-73 | 6-May-73 | 193 | 0 | 350 | 170 | 72.9 | DELIBERATE | SELF-DESTRUCT |
| ESSA 8 (TOS F) R/B | 1968-114B | 3616 | 15-Dec-68 | 15-Nov-73 | 18 | 18 | 1462 | 1413 | 101.6 | PROPULSION | DELTA SECOND STAGE |
| NOAA 3 R/B | 1973-086B | 6921 | 6-Nov-73 | 28-Dec-73 | 220 | 198 | 1510 | 1500 | 102.1 | PROPULSION | DELTA SECOND STAGE |
| COSMOS 699 | 1974-103A | 7587 | 24-Dec-74 | 17-Apr-75 | 50 | 0 | 445 | 425 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| LANDSAT 1 R/B | 1972-058B | 6127 | 23-Jul-72 | 22-May-75 | 244 | 47 | 910 | 635 | 98.3 | PROPULSION | DELTA SECOND STAGE |
| PAGEOS | 1966-056A | 2253 | 24-Jun-66 | 12-Jul-75 | 79 | 1 | 5170 | 3200 | 85.3 | UNKNOWN | INFLATABLE SPHERE |

TABLE 2.2 HISTORY OF SATELLITE BREAKUPS BY EVENT DATE (CONT'D)

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | BREAKUP DATE | DEBRIS CATALOGED | DEBRIS ON-ORBIT | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) | ASSESSED CAUSE | ADDITIONAL INFORMATION |
|-----------------|--------------------------|---------------------|-------------|--------------|------------------|-----------------|-------------|--------------|-------------------|----------------|------------------------|
| NOAA 4 R/B | 1974-089D | 7532 | 15-Nov-74 | 20-Aug-75 | 185 | 160 | 1460 | 1445 | 101.7 | PROPULSION | DELTA SECOND STAGE |
| COSMOS 758 | 1975-080A | 8191 | 5-Sep-75 | 6-Sep-75 | 76 | 0 | 325 | 175 | 67.1 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 777 | 1975-102A | 8416 | 29-Oct-75 | 25-Jan-76 | 62 | 0 | 440 | 430 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| LANDSAT 2 R/B | 1975-004B | 7616 | 22-Jan-75 | 9-Feb-76 | 207 | 32 | 915 | 740 | 97.8 | PROPULSION | DELTA SECOND STAGE |
| COSMOS 844 | 1976-072A | 9046 | 22-Jul-76 | 25-Jul-76 | 248 | 0 | 355 | 170 | 67.1 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 886 | 1976-126A | 9634 | 27-Dec-76 | 27-Dec-76 | 76 | 58 | 2295 | 595 | 65.8 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 884 | 1976-123A | 9614 | 17-Dec-76 | 29-Dec-76 | 2 | 0 | 320 | 170 | 65.0 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 862 | 1976-105A | 9495 | 22-Oct-76 | 15-Mar-77 | 13 | 5 | 39645 | 765 | 63.2 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 838 | 1976-063A | 8932 | 2-Jul-76 | 17-May-77 | 40 | 0 | 445 | 415 | 65.1 | UNKNOWN | COSMOS 699 CLASS |
| HIMAWARI 1 R/B | 1977-065B | 10144 | 14-Jul-77 | 14-Jul-77 | 177 | 59 | 2025 | 535 | 29.0 | PROPULSION | DELTA SECOND STAGE |
| COSMOS 839 | 1976-067A | 9011 | 8-Jul-76 | 29-Sep-77 | 70 | 68 | 2100 | 980 | 65.9 | BATTERY | |
| COSMOS 931 | 1977-068A | 10150 | 20-Jul-77 | 24-Oct-77 | 6 | 5 | 39665 | 680 | 62.9 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 970 | 1977-121A | 10531 | 21-Dec-77 | 21-Dec-77 | 70 | 64 | 1140 | 945 | 65.8 | DELIBERATE | SELF-DESTRUCT |
| NOAA 5 R/B | 1976-077B | 9063 | 29-Jul-76 | 24-Dec-77 | 184 | 174 | 1520 | 1505 | 102.0 | PROPULSION | DELTA SECOND STAGE |
| COSMOS 903 | 1977-027A | 9911 | 11-Apr-77 | 8-Jun-78 | 6 | 3 | 39035 | 1325 | 63.2 | DELIBERATE | SELF-DESTRUCT |
| EKRAN 2 | 1977-092A | 10365 | 20-Sep-77 | 23-Jun-78 | 5 | 5 | 35800 | 35785 | 0.1 | BATTERY | |
| COSMOS 1030 | 1978-083A | 11015 | 6-Sep-78 | 10-Oct-78 | 13 | 9 | 39760 | 665 | 62.8 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 880 | 1976-120A | 9601 | 9-Dec-76 | 27-Nov-78 | 49 | 0 | 620 | 550 | 65.8 | BATTERY | |
| COSMOS 917 | 1977-047A | 10059 | 16-Jun-77 | 30-Mar-79 | 14 | 12 | 38725 | 1645 | 62.9 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1124 | 1979-077A | 11509 | 28-Aug-79 | 9-Sep-79 | 5 | 5 | 39795 | 570 | 63.0 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1094 | 1979-033A | 11333 | 18-Apr-79 | 17-Sep-79 | 1 | 0 | 405 | 380 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| SATCOM 3 | 1979-101A | 11635 | 7-Dec-79 | 11-Dec-79 | 21 | 21 | 35776 | 204 | 23.7 | PROPULSION | AKM FAILURE |
| COSMOS 1109 | 1979-058A | 11417 | 27-Jun-79 | Feb-80 | 19 | 8 | 39425 | 960 | 63.3 | DELIBERATE | SELF-DESTRUCT |
| CAT R/B | 1979-104B | 11659 | 24-Dec-79 | Apr-80 | 31 | 24 | 33140 | 180 | 17.9 | PROPULSION | ARIANE 1 FINAL STAGE |
| COSMOS 1174 | 1980-030A | 11765 | 18-Apr-80 | 18-Apr-80 | 46 | 4 | 1660 | 380 | 66.1 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1188 | 1980-050A | 11844 | 14-Jun-80 | 26-Aug-80 | 8 | 6 | 39630 | 735 | 62.9 | DELIBERATE | SELF-DESTRUCT |
| LANDSAT 3 R/B | 1978-026C | 10704 | 5-Mar-78 | 27-Jan-81 | 248 | 153 | 910 | 900 | 98.8 | PROPULSION | DELTA SECOND STAGE |
| COSMOS 1261 | 1981-031A | 12376 | 31-Mar-81 | Apr-81 | 10 | 10 | 39765 | 610 | 63.0 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1191 | 1980-057A | 11871 | 2-Jul-80 | 14-May-81 | 11 | 11 | 39255 | 1110 | 62.6 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1167 | 1980-021A | 11729 | 14-Mar-80 | 15-Jul-81 | 12 | 0 | 450 | 355 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 1275 | 1981-053A | 12504 | 4-Jun-81 | 24-Jul-81 | 479 | 421 | 1015 | 960 | 83.0 | BATTERY | |
| COSMOS 1305 R/B | 1981-088F | 12827 | 11-Sep-81 | 11-Sep-81 | 8 | 8 | 13795 | 605 | 62.8 | PROPULSION | MOLNIYA FINAL STAGE |
| COSMOS 1247 | 1981-016A | 12303 | 19-Feb-81 | 20-Oct-81 | 7 | 6 | 39390 | 970 | 63.0 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1285 | 1981-071A | 12627 | 4-Aug-81 | 21-Nov-81 | 25 | 25 | 40100 | 720 | 63.1 | DELIBERATE | SELF-DESTRUCT |

TABLE 2.2 HISTORY OF SATELLITE BREAKUPS BY EVENT DATE (CONT'D)

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | BREAKUP DATE | DEBRIS CATALOGED | DEBRIS ON-ORBIT | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) | ASSESSED CAUSE | ADDITIONAL INFORMATION |
|--------------------------|--------------------------|---------------------|-------------|--------------|------------------|-----------------|-------------|--------------|-------------------|-----------------------|------------------------|
| NIMBUS 7 R/B | 1978-098B | 11081 | 24-Oct-78 | 26-Dec-81 | 2 | 1 | 955 | 935 | 99.3 | PROPULSION | DELTA SECOND STAGE |
| COSMOS 1260 | 1981-028A | 12364 | 20-Mar-81 | 8-May-82 | 68 | 0 | 750 | 450 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 1220 | 1980-089A | 12054 | 4-Nov-80 | 20-Jun-82 | 83 | 1 | 885 | 570 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 1306 | 1981-089A | 12828 | 14-Sep-81 | 12-Jul-82 | 8 | 0 | 405 | 380 | 64.9 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 1286 | 1981-072A | 12631 | 4-Aug-81 | 29-Sep-82 | 2 | 0 | 325 | 300 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 1423 R/B | 1982-115E | 13696 | 8-Dec-82 | 8-Dec-82 | 29 | 0 | 425 | 235 | 62.9 | PROPULSION | MOLNIYA FINAL STAGE |
| COSMOS 1217 | 1980-085A | 12032 | 24-Oct-80 | 12-Feb-83 | 10 | 8 | 38830 | 1530 | 65.2 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1481 | 1983-070A | 14182 | 8-Jul-83 | 9-Jul-83 | 7 | 7 | 39225 | 625 | 62.9 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1355 | 1982-038A | 13150 | 29-Apr-82 | 8-Aug-83 | 29 | 0 | 395 | 360 | 65.1 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 1456 | 1983-038A | 14034 | 25-Apr-83 | 13-Aug-83 | 4 | 0 | 39630 | 730 | 63.3 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1405 | 1982-088A | 13508 | 4-Sep-82 | 20-Dec-83 | 32 | 0 | 340 | 310 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 1317 | 1981-108A | 12933 | 31-Oct-81 | 25-28 Jan-84 | 11 | 11 | 39055 | 1315 | 62.8 | DELIBERATE | SELF-DESTRUCT |
| WESTAR 6 R/B | 1984-011F | 14694 | 3-Feb-84 | 3-Feb-84 | 14 | 0 | 310 | 305 | 28.5 | PROPULSION | PAM-D UPPER STAGE |
| PALAPA B2 R/B | 1984-011E | 14693 | 3-Feb-84 | 6-Feb-84 | 3 | 0 | 285 | 275 | 28.5 | PROPULSION | PAM-D UPPER STAGE |
| COSMOS 1348 | 1982-029A | 13124 | 7-Apr-82 | 2-Sep-84 | 10 | 10 | 39200 | 1185 | 62.8 | DELIBERATE | SELF-DESTRUCT |
| ASTRON ULLAGE MOTOR | 1983-020B | 13902 | 23-Mar-83 | 3-Sep-84 | 1 | 0 | 1230 | 220 | 51.5 | PROPULSION | PROTON-K BLOCK DM SOZ |
| SPACENET 2/MARECS B2 R/B | 1984-114C | 15388 | 10-Nov-84 | 20-Nov-84 | 3 | 2 | 35960 | 325 | 7.0 | PROPULSION | ARIANE 3 FINAL STAGE |
| COSMOS 1461 | 1983-044A | 14064 | 7-May-83 | 11-Mar-85 | 187 | 14 | 890 | 570 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 1654 | 1985-039A | 15734 | 23-May-85 | 21-Jun-85 | 18 | 0 | 300 | 185 | 64.9 | DELIBERATE | SELF-DESTRUCT |
| P-78/SOLWIND | 1979-017A | 11278 | 24-Feb-79 | 13-Sep-85 | 285 | 0 | 545 | 515 | 97.6 | COLLISION, DELIBERATE | HYPERVELOCITY IMPACT |
| COSMOS 1375 | 1982-055A | 13259 | 6-Jun-82 | 21-Oct-85 | 61 | 58 | 1000 | 990 | 65.8 | BATTERY | |
| COSMOS 1691 (1695) | 1985-094B | 16139 | 9-Oct-85 | 22-Nov-85 | 21 | 18 | 1415 | 1410 | 82.6 | BATTERY | |
| COSMOS 1714 R/B | 1985-121F | 16439 | 28-Dec-85 | 28-Dec-85 | 2 | 0 | 830 | 165 | 71.0 | PROPULSION | ZENIT SECOND STAGE |
| NOAA 8 | 1983-022A | 13923 | 28-Mar-83 | 30-Dec-85 | 5 | 1 | 830 | 805 | 98.6 | BATTERY | |
| COSMOS 1588 | 1984-083A | 15167 | 7-Aug-84 | 23-Feb-86 | 45 | 0 | 440 | 410 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| USA 19 | 1986-069A | 16937 | 5-Sep-86 | 5-Sep-86 | 13 | 0 | 745 | 210 | 39.1 | COLLISION, DELIBERATE | HYPERVELOCITY IMPACT |
| USA 19 R/B | 1986-069B | 16938 | 5-Sep-86 | 5-Sep-86 | 5 | 0 | 610 | 220 | 22.8 | COLLISION, DELIBERATE | HYPERVELOCITY IMPACT |
| SPOT 1 R/B | 1986-019C | 16615 | 22-Feb-86 | 13-Nov-86 | 498 | 31 | 835 | 805 | 98.7 | PROPULSION | ARIANE 1 FINAL STAGE |
| COSMOS 1278 | 1981-058A | 12547 | 19-Jun-81 | Dec-86 | 3 | 0 | 37690 | 2665 | 67.1 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1682 | 1985-082A | 16054 | 19-Sep-85 | 18-Dec-86 | 23 | 0 | 475 | 385 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 1813 | 1987-004A | 17297 | 15-Jan-87 | 29-Jan-87 | 195 | 0 | 415 | 360 | 72.8 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1866 | 1987-059A | 18184 | 9-Jul-87 | 26-Jul-87 | 9 | 0 | 255 | 155 | 67.1 | DELIBERATE | SELF-DESTRUCT |
| AUSSAT K3/ECS 4 R/B | 1987-078C | 18352 | 16-Sep-87 | 16-19 Sep-87 | 4 | 1 | 36515 | 245 | 6.9 | PROPULSION | ARIANE 3 FINAL STAGE |

TABLE 2.2 HISTORY OF SATELLITE BREAKUPS BY EVENT DATE (CONT'D)

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | BREAKUP DATE | DEBRIS CATALOGED | DEBRIS ON-ORBIT | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) | ASSESSED CAUSE | ADDITIONAL INFORMATION |
|-----------------------------|--------------------------|---------------------|-------------|--------------|------------------|-----------------|-------------|--------------|-------------------|-----------------------|---------------------------|
| COSMOS 1769 | 1986-059A | 16895 | 4-Aug-86 | 21-Sep-87 | 4 | 0 | 445 | 310 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 1646 | 1985-030A | 15653 | 18-Apr-85 | 20-Nov-87 | 24 | 0 | 410 | 385 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 1823 | 1987-020A | 17535 | 20-Feb-87 | 17-Dec-87 | 150 | 76 | 1525 | 1480 | 73.6 | BATTERY | |
| COSMOS 1656 ULLAGE MOTOR | 1985-042E | 15773 | 30-May-85 | 5-Jan-88 | 6 | 6 | 860 | 810 | 66.6 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 1906 | 1987-108A | 18713 | 26-Dec-87 | 31-Jan-88 | 37 | 0 | 265 | 245 | 82.6 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1916 | 1988-007A | 18823 | 3-Feb-88 | 27-Feb-88 | 1 | 0 | 230 | 150 | 64.8 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1045 R/B | 1978-100D | 11087 | 26-Oct-78 | 9-May-88 | 42 | 32 | 1705 | 1685 | 82.6 | PROPULSION | TSYKLON THIRD STAGE |
| COSMOS 2030 | 1989-054A | 20124 | 12-Jul-89 | 28-Jul-89 | 1 | 0 | 215 | 150 | 67.1 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 2031 | 1989-056A | 20136 | 18-Jul-89 | 31-Aug-89 | 9 | 0 | 365 | 240 | 50.5 | DELIBERATE | SELF-DESTRUCT |
| FENGYUN 1-2 R/B | 1990-081D | 20791 | 3-Sep-90 | 4-Oct-90 | 103 | 75 | 895 | 880 | 98.9 | PROPULSION | CZ-4 FINAL STAGE |
| COSMOS 2101 | 1990-087A | 20828 | 1-Oct-90 | 30-Nov-90 | 4 | 0 | 280 | 195 | 64.8 | DELIBERATE | SELF-DESTRUCT |
| USA 68 | 1990-105A | 20978 | 1-Dec-90 | 1-Dec-90 | 29 | 1 | 850 | 610 | 98.9 | PROPULSION | TE-M-364-15 UPPER STAGE |
| COSMOS 1519-21 ULLAGE MOTOR | 1983-127H | 14608 | 29-Dec-83 | 4-Feb-91 | 8 | 3 | 18805 | 340 | 51.9 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2125-32 R/B | 1991-009J | 21108 | 12-Feb-91 | 5-Mar-91 | 112 | 112 | 1725 | 1460 | 74.0 | PROPULSION | COSMOS SECOND STAGE |
| NIMBUS 6 R/B | 1975-052B | 7946 | 12-Jun-75 | 1-May-91 | 307 | 231 | 1105 | 1095 | 99.6 | PROPULSION | DELTA SECOND STAGE |
| COSMOS 2163 | 1991-071A | 21741 | 9-Oct-91 | 6-Dec-91 | 1 | 0 | 260 | 185 | 64.8 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1934 | 1988-023A | 18985 | 22-Mar-88 | 23-Dec-91 | 3 | 3 | 1010 | 950 | 83.0 | COLLISION, ACCIDENTAL | HIT BY DEBRIS (13475) |
| COSMOS 1710-12 ULLAGE MOTOR | 1985-118L | 16446 | 24-Dec-85 | 29-Dec-91 | 17 | 10 | 18885 | 655 | 65.3 | PROPULSION | PROTON-K BLOCK DM SOZ |
| OV2-5 R/B | 1968-081E | 3432 | 26-Sep-68 | 21-Feb-92 | 29 | 29 | 35810 | 35100 | 11.9 | PROPULSION | TITAN TRANSTAGE |
| COSMOS 2054 ULLAGE MOTOR | 1989-101E | 20399 | 27-Dec-89 | Jul-92 | 14 | 2 | 27650 | 345 | 47.1 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 1603 ULLAGE MOTOR | 1984-106F | 15338 | 28-Sep-84 | 5-Sep-92 | 23 | 2 | 845 | 835 | 66.6 | PROPULSION | PROTON-K BLOCK DM SOZ |
| GORIZONT 17 ULLAGE MOTOR | 1989-004E | 19771 | 26-Jan-89 | 17-Dec-92 | 1 | 0 | 17575 | 195 | 46.7 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2227 R/B | 1992-093B | 22285 | 25-Dec-92 | 26-Dec-92 | 279 | 196 | 855 | 845 | 71.0 | PROPULSION | ZENIT-2 SECOND STAGE |
| GORIZONT 18 ULLAGE MOTOR | 1989-052F | 20116 | 5-Jul-89 | 12-Jan-93 | 2 | 0 | 36745 | 260 | 46.8 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2225 | 1992-091A | 22280 | 22-Dec-92 | 18-Feb-93 | 6 | 0 | 280 | 225 | 64.9 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 2237 R/B | 1993-016B | 22566 | 26-Mar-93 | 28-Mar-93 | 104 | 100 | 850 | 840 | 71.0 | PROPULSION | ZENIT-2 SECOND STAGE |
| TELECOM 2B/INMARSAT 2 R/B | 1992-021C | 21941 | 15-Apr-92 | 21-Apr-93 | 18 | 17 | 34080 | 235 | 4.0 | PROPULSION | ARIANE 4 H10+ FINAL STAGE |
| COSMOS 2243 | 1993-028A | 22641 | 27-Apr-93 | 27-Apr-93 | 1 | 0 | 225 | 180 | 70.4 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 2259 | 1993-045A | 22716 | 14-Jul-93 | 25-Jul-93 | 1 | 0 | 320 | 175 | 67.1 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1484 | 1983-075A | 14207 | 24-Jul-83 | 18-Oct-93 | 49 | 0 | 595 | 550 | 97.5 | UNKNOWN | |
| COSMOS 2262 | 1993-057A | 22789 | 7-Sep-93 | 18-Dec-93 | 1 | 0 | 295 | 170 | 64.9 | DELIBERATE | SELF-DESTRUCT |
| CLEMENTINE R/B | 1994-004B | 22974 | 25-Jan-94 | 7-Feb-94 | 1 | 0 | 295 | 240 | 67.0 | PROPULSION | |

TABLE 2.2 HISTORY OF SATELLITE BREAKUPS BY EVENT DATE (CONT'D)

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | BREAKUP DATE | DEBRIS CATALOGED | DEBRIS ON-ORBIT | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) | ASSESSED CAUSE | ADDITIONAL INFORMATION |
|-----------------------------|--------------------------|---------------------|-------------|--------------|------------------|-----------------|-------------|--------------|-------------------|-----------------------|--------------------------|
| ASTRA 1B/MOP 2 R/B | 1991-015C | 21141 | 2-Mar-91 | 27-Apr-94 | 11 | 7 | 17630 | 205 | 6.8 | PROPULSION | ARIANE 4 H10 FINAL STAGE |
| COSMOS 2133 ULLAGE MOTOR | 1991-010D | 21114 | 12-Feb-91 | 7-May-94 | 4 | 0 | 21805 | 225 | 46.6 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2204-06 ULLAGE MOTOR | 1992-047H | 22067 | 30-Jul-92 | 8-Nov-94 | 4 | 2 | 19035 | 480 | 64.8 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2238 | 1993-018A | 22585 | 30-Mar-93 | 1-Dec-94 | 1 | 0 | 305 | 210 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| RS-15 R/B | 1994-085B | 23440 | 26-Dec-94 | 26-Dec-94 | 26 | 23 | 2200 | 1880 | 64.8 | UNKNOWN | ROKOT THIRD STAGE |
| ELEKTRO ULLAGE MOTOR | 1994-069E | 23338 | 31-Oct-94 | 11-May-95 | 1 | 0 | 35465 | 155 | 46.9 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2282 ULLAGE MOTOR | 1994-038F | 23174 | 6-Jul-94 | 21-Oct-95 | 2 | 0 | 34930 | 280 | 47.0 | PROPULSION | PROTON-K BLOCK DM SOZ |
| GORIZONT 22 ULLAGE MOTOR | 1990-102E | 20957 | 23-Nov-90 | 14-Dec-95 | 2 | 1 | 13105 | 170 | 46.5 | PROPULSION | PROTON-K BLOCK DM SOZ |
| RADUGA 33 R/B | 1996-010D | 23797 | 19-Feb-96 | 19-Feb-96 | 1 | 0 | 36505 | 240 | 48.7 | PROPULSION | PROTON-K BLOCK DM |
| ITALSAT 1/EUTELSAT 2 F2 R/B | 1991-003C | 21057 | 15-Jan-91 | Apr/May 96 | 15 | 12 | 30930 | 235 | 6.7 | PROPULSION | ARIANE 4 H10 FINAL STAGE |
| STEP II R/B | 1994-029B | 23106 | 19-May-94 | 3-Jun-96 | 754 | 82 | 820 | 585 | 82.0 | PROPULSION | PEGASUS HAPS |
| CERISE | 1995-033B | 23606 | 7-Jul-95 | 24-Jul-96 | 2 | 1 | 675 | 665 | 98.1 | COLLISION, ACCIDENTAL | HIT BY DEBRIS (18208) |
| COSMOS 1883-85 ULLAGE MOTOR | 1987-079G | 18374 | 16-Sep-87 | 1-Dec-96 | 14 | 11 | 19120 | 335 | 64.9 | PROPULSION | PROTON-K BLOCK DM SOZ |
| EKRAN 17 ULLAGE MOTOR | 1987-109E | 18719 | 27-Dec-87 | 22-May-97 | 1 | 0 | 22975 | 310 | 46.6 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2313 | 1995-028A | 23596 | 8-Jun-95 | 26-Jun-97 | 13 | 0 | 325 | 210 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 2343 | 1997-024A | 24805 | 15-May-97 | 16-Sep-97 | 1 | 0 | 285 | 225 | 65.0 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 1869 | 1987-062A | 18214 | 16-Jul-87 | 27-Nov-97 | 2 | 1 | 635 | 605 | 83.0 | UNKNOWN | |
| ASIASAT 3 R/B | 1997-086D | 25129 | 24-Dec-97 | 25-Dec-97 | 1 | 0 | 35995 | 270 | 51.0 | PROPULSION | PROTON-K BLOCK DM |
| METEOR 2-16 R/B | 1987-068B | 18313 | 18-Aug-87 | 15-Feb-98 | 108 | 42 | 960 | 940 | 82.6 | PROPULSION | TSYKLON THIRD STAGE |
| SKYNET 4B/ASTRA 1A R/B | 1988-109C | 19689 | 11-Dec-88 | 17-Feb-98 | 18 | 17 | 35875 | 435 | 7.3 | PROPULSION | ARIANE 4 H10 FINAL STAGE |
| COMETS R/B | 1998-011B | 25176 | 21-Feb-98 | 21-Feb-98 | 1 | 0 | 1880 | 245 | 30.0 | PROPULSION | H-II SECOND STAGE |
| COSMOS 2109-11 ULLAGE MOTOR | 1990-110H | 21013 | 8-Dec-90 | 14-Mar-98 | 2 | 2 | 18995 | 520 | 65.1 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 1987-89 ULLAGE MOTOR | 1989-001G | 19755 | 10-Jan-89 | 3-Aug-98 | 16 | 6 | 19055 | 340 | 64.9 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 1650-52 ULLAGE MOTOR | 1985-037G | 15714 | 17-May-85 | 29-Nov-98 | 4 | 2 | 18620 | 320 | 52.0 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 1970-72 ULLAGE MOTOR | 1988-085G | 19537 | 16-Sep-88 | 9-Mar-99 | 1 | 1 | 18950 | 300 | 64.6 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2079-81 ULLAGE MOTOR | 1990-045G | 20631 | 19-May-90 | 28-Mar-99 | 1 | 1 | 19065 | 405 | 64.8 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2053 R/B | 1989-100B | 20390 | 27-Dec-89 | 18-Apr-99 | 26 | 0 | 485 | 475 | 73.5 | PROPULSION | TSYKLON THIRD STAGE |
| METEOR 2-8 | 1982-025A | 13113 | 25-Mar-82 | 29-May-99 | 53 | 53 | 960 | 935 | 82.5 | UNKNOWN | |
| COSMOS 2157-62 R/B | 1991-068G | 21734 | 28-Sep-91 | 9-Oct-99 | 40 | 40 | 1485 | 1410 | 82.6 | PROPULSION | TSYKLON THIRD STAGE |
| COSMOS 2347 | 1997-079A | 25088 | 9-Dec-97 | 22-Nov-99 | 9 | 0 | 410 | 230 | 65.0 | UNKNOWN | COSMOS 699 CLASS |

TABLE 2.2 HISTORY OF SATELLITE BREAKUPS BY EVENT DATE (CONT'D)

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | BREAKUP DATE | DEBRIS CATALOGED | DEBRIS ON-ORBIT | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) | ASSESSED CAUSE | ADDITIONAL INFORMATION |
|-----------------------------|--------------------------|---------------------|-------------|--------------|------------------|-----------------|-------------|--------------|-------------------|-----------------------|--------------------------|
| GORIZONT 32 ULLAGE MOTOR | 1996-034F | 23887 | 25-May-96 | 13-Dec-99 | 1 | 0 | 5605 | 145 | 46.5 | PROPULSION | PROTON-K BLOCK DM SOZ |
| CBERS 1/SACI 1 R/B | 1999-057C | 25942 | 14-Oct-99 | 11-Mar-00 | 344 | 150 | 745 | 725 | 98.5 | PROPULSION | CZ-4 FINAL STAGE |
| GORIZONT 29 ULLAGE MOTOR | 1993-072E | 22925 | 18-Nov-93 | 6-Sep-00 | 1 | 0 | 11215 | 140 | 46.7 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2316-18 ULLAGE MOTOR | 1995-037K | 23631 | 24-Jul-95 | 21-Nov-00 | 1 | 0 | 18085 | 150 | 64.4 | PROPULSION | PROTON-K BLOCK DM SOZ |
| INTELSAT 515 R/B | 1989-006B | 19773 | 27-Jan-89 | 1-Jan-01 | 87 | 86 | 35720 | 510 | 8.4 | PROPULSION | ARIANE 2 R/B |
| COSMOS 2139-41 ULLAGE MOTOR | 1991-025G | 21226 | 4-Apr-91 | 16-Jun-01 | 1 | 1 | 18960 | 300 | 64.5 | PROPULSION | PROTON-K BLOCK DM SOZ |
| GORIZONT 27 ULLAGE MOTOR | 1992-082F | 22250 | 27-Nov-92 | 14-Jul-01 | 1 | 0 | 5340 | 145 | 46.5 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2367 | 1999-072A | 26040 | 26-Dec-99 | 21-Nov-01 | 17 | 0 | 415 | 405 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| TES R/B | 2001-049D | 26960 | 22-Oct-01 | 19-Dec-01 | 372 | 72 | 675 | 550 | 97.9 | PROPULSION | PSLV FINAL STAGE |
| INTELSAT 601 R/B | 1991-075B | 21766 | 29-Oct-91 | 24-Dec-01 | 13 | 8 | 28505 | 230 | 7.2 | PROPULSION | ARIANE 4 H10 FINAL STAGE |
| INSAT 2A/EUTELSAT 2F4 R/B | 1992-041C | 22032 | 9-Jul-92 | Feb-02 | 2 | 2 | 26550 | 250 | 7.0 | PROPULSION | ARIANE 4 H10 FINAL STAGE |
| INTELSAT 513 R/B | 1988-040B | 19122 | 17-May-88 | 9-Jul-02 | 8 | 8 | 35445 | 535 | 7.0 | PROPULSION | ARIANE 2 R/B |
| COSMOS 2109-11 ULLAGE MOTOR | 1990-110G | 21012 | 8-Dec-90 | 21-Feb-03 | 1 | 1 | 18805 | 645 | 65.4 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 1883-85 ULLAGE MOTOR | 1987-079H | 18375 | 16-Sep-87 | 23-Apr-03 | 42 | 11 | 18540 | 755 | 65.2 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 1970-72 ULLAGE MOTOR | 1988-085F | 19535 | 16-Sep-88 | 4-Aug-03 | 79 | 10 | 18515 | 720 | 65.3 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 1987-89 ULLAGE MOTOR | 1989-001H | 19856 | 10-Jan-89 | 13-Nov-03 | 1 | 1 | 18740 | 710 | 65.4 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2399 | 2003-035A | 27856 | 12-Aug-03 | 9-Dec-03 | 22 | 0 | 250 | 175 | 64.9 | DELIBERATE | SELF-DESTRUCT |
| COSMOS 2383 | 2001-057A | 27053 | 21-Dec-01 | 28-Feb-04 | 14 | 0 | 400 | 220 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| USA 73 (DMSP 5D2 F11) | 1991-082A | 21798 | 28-Nov-91 | 15-Apr-04 | 85 | 65 | 850 | 830 | 98.7 | UNKNOWN | |
| COSMOS 2204-06 ULLAGE MOTOR | 1992-047G | 22066 | 30-Jul-92 | 10-Jul-04 | 34 | 12 | 18820 | 415 | 64.9 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2392 ULLAGE MOTOR | 2002-037F | 27475 | 25-Jul-02 | 29-Oct-04 | 1 | 0 | 840 | 235 | 63.6 | PROPULSION | PROTON-K BLOCK DM SOZ |
| DMSP 5B F5 R/B | 1974-015B | 7219 | 16-Mar-74 | 17-Jan-05 | 7 | 7 | 885 | 775 | 99.1 | COLLISION, ACCIDENTAL | HIT BY DEBRIS (26207) |
| COSMOS 2224 ULLAGE MOTOR | 1992-088F | 22274 | 17-Dec-92 | ~22-Apr-05 | 1 | 0 | 21140 | 200 | 46.7 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2392 ULLAGE MOTOR | 2002-037E | 27474 | 25-Jul-02 | 1-Jun-05 | 61 | 0 | 835 | 255 | 63.7 | PROPULSION | PROTON-K BLOCK DM SOZ |
| METEOR 2-17 | 1988-005A | 18820 | 30-Jan-88 | 21-Jun-05 | 45 | 45 | 960 | 930 | 82.5 | UNKNOWN | |
| COSMOS 1703 R/B | 1985-108B | 16263 | 22-Nov-85 | 4-May-06 | 50 | 2 | 640 | 610 | 82.5 | PROPULSION | TSYKLON THIRD STAGE |
| COSMOS 2022-24 ULLAGE MOTOR | 1989-039G | 20081 | 31-May-89 | 10-Jun-06 | 120 | 43 | 18410 | 655 | 65.1 | PROPULSION | PROTON-K BLOCK DM SOZ |
| ALOS-1 R/B | 2006-002B | 28932 | 24-Jan-06 | 8-Aug-06 | 24 | 3 | 700 | 550 | 98.2 | UNKNOWN | H-IIA SECOND STAGE |
| COSMOS 2371 ULLAGE MOTOR | 2000-036E | 26398 | 4-Jul-00 | ~1-Sep-06 | 1 | 0 | 21320 | 220 | 46.9 | PROPULSION | PROTON-K BLOCK DM SOZ |
| DMSP 5D-3 F17 R/B | 2006-050B | 29523 | 4-Nov-06 | 4-Nov-06 | 65 | 0 | 865 | 830 | 98.8 | UNKNOWN | DELTA IV SECOND STAGE |

TABLE 2.2 HISTORY OF SATELLITE BREAKUPS BY EVENT DATE (CONT'D)

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | BREAKUP DATE | DEBRIS CATALOGED | DEBRIS ON-ORBIT | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) | ASSESSED CAUSE | ADDITIONAL INFORMATION |
|-----------------------------------|--------------------------|---------------------|-------------|--------------|------------------|-----------------|-------------|--------------|-------------------|-----------------------|---------------------------|
| COSMOS 2423 | 2006-039A | 29402 | 14-Sep-06 | 17-Nov-06 | 31 | 0 | 285 | 200 | 64.9 | DELIBERATE | SELF-DESTRUCT |
| COBE R/B | 1989-089B | 20323 | 18-Nov-89 | 3-Dec-06 | 26 | 1 | 790 | 685 | 97.1 | UNKNOWN | DELTA SECOND STAGE |
| IGS 3A R/B | 2006-037B | 29394 | 11-Sep-06 | 28-Dec-06 | 10 | 0 | 490 | 430 | 97.2 | UNKNOWN | H-IIA SECOND STAGE |
| FENGYUN 1C | 1999-025A | 25730 | 10-May-99 | 11-Jan-07 | 3442 | 2832 | 865 | 845 | 98.6 | COLLISION, DELIBERATE | HYPERVELOCITY IMPACT |
| BEIDOU 1D R/B | 2007-003B | 30324 | 2-Feb-07 | 2-Feb-07 | 39 | 36 | 41900 | 235 | 25.0 | UNKNOWN | CZ-3A FINAL STAGE |
| KUPON ULLAGE MOTOR | 1997-070F | 25054 | 12-Nov-97 | 14-Feb-07 | 7 | 4 | 14160 | 260 | 46.6 | PROPULSION | PROTON-K BLOCK DM SOZ |
| CBERS 1 | 1999-057A | 25940 | 14-Oct-99 | 18-Feb-07 | 88 | 53 | 780 | 770 | 98.2 | UNKNOWN | |
| ARABSAT 4 BRIZ-M R/B | 2006-006B | 28944 | 28-Feb-06 | 19-Feb-07 | 102 | 92 | 14705 | 495 | 51.5 | PROPULSION | PROTON-K BRIZ-M STAGE |
| USA 197 R/B | 2007-054B | 32288 | 11-Nov-07 | 11-Nov-07 | 32 | 32 | 1575 | 220 | 29.0 | UNKNOWN | DELTA IV SECOND STAGE |
| USA 193 | 2006-057A | 29651 | 14-Dec-06 | 21-Feb-08 | 175 | 0 | 255 | 245 | 58.5 | COLLISION, DELIBERATE | HYPERVELOCITY IMPACT |
| COSMOS 2421 | 2006-026A | 29247 | 25-Jun-06 | 14-Mar-08 | 509 | 0 | 420 | 400 | 65.0 | UNKNOWN | COSMOS 699 CLASS |
| COSMOS 2251 | 1993-036A | 22675 | 16-Jun-93 | 10-Feb-09 | 1668 | 1076 | 800 | 775 | 74.0 | COLLISION, ACCIDENTAL | COLLIDED WITH IRIDIUM 33 |
| IRIDIUM 33 | 1997-051C | 24946 | 14-Sep-97 | 10-Feb-09 | 628 | 333 | 780 | 775 | 86.4 | COLLISION, ACCIDENTAL | COLLIDED WITH COSMOS 2251 |
| COSMOS 2139-41 ULLAGE MOTOR | 1991-025F | 21220 | 4-Apr-91 | 8-Mar-09 | 33 | 11 | 18535 | 465 | 64.9 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 192 | 1967-116A | 3047 | 23-Nov-67 | 30-Aug-09 | 4 | 1 | 715 | 710 | 74.0 | UNKNOWN | |
| YAOGAN 1 | 2006-015A | 29092 | 26-Apr-06 | 4-Feb-10 | 8 | 2 | 630 | 625 | 97.9 | UNKNOWN | |
| CHINASAT 6A R/B | 2010-042B | 37151 | 4-Sep-10 | 4-Sep-10 | 30 | 29 | 41894 | 194 | 25.2 | UNKNOWN | |
| AMC 14 BRIZ-M R/B | 2008-011B | 32709 | 14-Mar-08 | 13-Oct-10 | 116 | 115 | 26565 | 645 | 48.9 | PROPULSION | PROTON-K BRIZ-M STAGE |
| BEIDOU G4 R/B | 2010-057B | 37211 | 1-Nov-10 | 1-Nov-10 | 57 | 55 | 35780 | 160 | 20.5 | UNKNOWN | CZ-3C FINAL STAGE |
| IGS 4A/4B R/B DEBRIS | 2007-005E | 30590 | 24-Feb-07 | 23-Dec-10 | 4 | 0 | 440 | 430 | 97.3 | UNKNOWN | H-IIA DEBRIS |
| COSMOS 2434-36 ULLAGE MOTOR | 2007-065G | 32399 | 25-Dec-07 | 18-Aug-11 | 1 | 1 | 18965 | 540 | 65.0 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2079-81 ULLAGE MOTOR | 1990-045F | 20630 | 19-May-90 | 17-Nov-11 | 1 | 1 | 18620 | 420 | 65.0 | PROPULSION | PROTON-K BLOCK DM SOZ |
| NIGCOMSAT 1R R/B | 2011-077B | 38015 | 19-Dec-11 | ~21-Dec-11 | 39 | 39 | 41715 | 230 | 24.3 | UNKNOWN | CZ-3B FINAL STAGE |
| BEIDOU G5 R/B | 2012-008B | 38092 | 24-Feb-12 | 26-Feb-12 | 38 | 37 | 35950 | 150 | 20.7 | UNKNOWN | CZ-3C FINAL STAGE |
| TELKOM 3 / EXPRESS MD2 BRIZ-M R/B | 2012-044C | 38746 | 6-Aug-12 | 16-Oct-12 | 113 | 20 | 5010 | 265 | 49.9 | PROPULSION | PROTON-K BRIZ-M STAGE |
| DMSP 5D-2 F9 (USA 29) | 1988-006A | 18822 | 3-Feb-88 | 14-17 Dec-12 | 10 | 3 | 810 | 800 | 98.8 | UNKNOWN | |
| CASSIOPE R/B | 2013-055B | 39266 | 29-Sep-13 | 29-Sep-13 | 16 | 1 | 1490 | 320 | 81.0 | PROPULSION | FALCON 9 SECOND STAGE |
| ARGOS/ORSTED/SUNSAT R/B | 1999-008D | 25637 | 23-Feb-99 | 28-Apr-14 | 8 | 1 | 840 | 635 | 96.5 | UNKNOWN | DELTA 2 SECOND STAGE |
| COSMOS 2428 | 2007-029A | 31792 | 29-Jun-07 | 10-May-14 | 10 | 2 | 860 | 845 | 71.0 | UNKNOWN | |
| COSMOS 2442-44 ULLAGE MOTOR | 2008-046H | 33385 | 25-Sep-08 | 20-May-14 | 11 | 10 | 18720 | 865 | 65.0 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2294-96 ULLAGE MOTOR | 1994-076G | 23402 | 20-Nov-94 | 7-Jun-14 | 2 | 2 | 18990 | 420 | 65.0 | PROPULSION | PROTON-K BLOCK DM SOZ |

TABLE 2.2 HISTORY OF SATELLITE BREAKUPS BY EVENT DATE (CONT'D)

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | BREAKUP DATE | DEBRIS CATALOGED | DEBRIS ON-ORBIT | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) | ASSESSED CAUSE | ADDITIONAL INFORMATION |
|-------------------------------|--------------------------|---------------------|-------------|--------------|------------------|-----------------|-------------|--------------|-------------------|-----------------------|--|
| COSMOS 2459-61 ULLAGE MOTOR | 2010-007G | 36406 | 1-Mar-10 | 9-Jul-14 | 11 | 11 | 18750 | 770 | 65.0 | PROPULSION | PROTON-K BLOCK DM SOZ |
| COSMOS 2431-33 ULLAGE MOTOR | 2007-052F | 32280 | 26-Oct-07 | 13-Aug-14 | 25 | 25 | 18790 | 730 | 65.0 | PROPULSION | PROTON-K BLOCK DM SOZ |
| USA 109 (DMSP 5D-2 F13) | 1995-015A | 23533 | 24-Mar-95 | 3-Feb-15 | 236 | 219 | 840 | 840 | 98.8 | BATTERY | Operational at event time |
| PROGRESS-M 27M R/B | 2015-024B | 40620 | 28-Apr-15 | 28-Apr-15 | 21 | 0 | 181 | 169 | 51.7 | UNKNOWN | Soyuz 2.1a (3); separation with PROGRESS M-27M |
| SL-23 DEBRIS | 2011-037B | 37756 | 18-Jul-11 | 3/4-Aug-15 | 1 | 1 | 3649 | 428 | 51.4 | UNKNOWN | Fregat-SB SBB propellant tank |
| NOAA 16 | 2000-055A | 26536 | 21-Sep-00 | 25-Nov-15 | 458 | 458 | 858 | 842 | 98.9 | UNKNOWN | |
| NIMIQ 6 R/B | 2012-026B | 38343 | 17-May-12 | 23-Dec-15 | 11 | 11 | 34592 | 10408 | 12.0 | UNKNOWN | Briz-M CORE |
| COSMOS 2513 R/B | 2015-075B | 41122 | 13-Dec-15 | 16-Jan-16 | 7 | 7 | 35777 | 33426 | 0.2 | UNKNOWN | Briz-M CORE |
| COSMOS 2447-49 ULLAGE MOTOR | 2008-067G | 33472 | 25-Dec-08 | 26-Mar-16 | 4 | 4 | 18840 | 682 | 65.4 | PROPULSION | PROTON-K BLOCK DM SOZ |
| ASTRO H (HITOMI) | 2016-012A | 41337 | 17-Feb-16 | 26-Mar-16 | 13 | 10 | 578 | 563 | 31.0 | PROPULSION | likely structural failure mechanism |
| COSMOS 2447-49 ULLAGE MOTOR | 2008-067H | 33473 | 25-Dec-08 | 1-Jun-16 | 12 | 12 | 18786 | 709 | 65.3 | PROPULSION | PROTON-K BLOCK DM SOZ |
| BEIDOU G2 | 2009-018A | 34779 | 14-Apr-09 | 29-Jun-16 | 1 | 1 | 36137 | 35384 | 4.7 | UNKNOWN | |
| COSMOS 2424-2426 ULLAGE MOTOR | 2006-062G | 29680 | 25-Dec-06 | 27-Jul-16 | 9 | 8 | 19088 | 426 | 64.8 | PROPULSION | PROTON-K BLOCK DM SOZ |
| SENTINEL 1A | 2014-016A | 39634 | 3-Apr-14 | 23-Aug-16 | 8 | 3 | 698 | 696 | 98.2 | COLLISION, ACCIDENTAL | Solar array impact by small MMOD |
| COSMOS 2464-2466 ULLAGE MOTOR | 2010-041G | 37143 | 2-Sep-10 | 3-Sep-17 | 10 | 10 | 18684 | 756 | 65.2 | PROPULSION | PROTON-K BLOCK DM SOZ |
| FREGAT DEB (TANK) | 2017-086C | 43089 | 26-Dec-17 | 12-Feb-18 | 5 | 2 | 4070 | 277 | 50.4 | UNKNOWN | SL-23 Fregat SB SBB propellant tank |
| OPS 0757 (TACSAT) R/B | 1969-013B | 3692 | 9-Feb-69 | 28-Feb-18 | 19 | 19 | 37257 | 35886 | 6.2 | UNKNOWN | TITAN TRANSTAGE |
| COSMOS 2459-61 ULLAGE MOTOR | 2010-007H | 36407 | 1-Mar-10 | 22-May-18 | 11 | 11 | 18929 | 602 | 65.1 | PROPULSION | PROTON-K BLOCK DM SOZ |

1. NA = NOT AVAILABLE

2. BREAKUP DATE AND ORBIT ARE FOR FIRST EVENT ONLY IF MULTIPLE EVENTS OCCURRED

3. DOES NOT INCLUDE SATELLITE BREAKUPS IF VEHICLE WAS ON REENTRY TRAJECTORY AT THE TIME OF THE EVENT

2.2 Identified Satellite Breakups

The remainder of this section devotes two pages to each identified satellite breakup. Each satellite is listed by common name, international designator, and satellite number. The satellite is then described in terms of type, ownership, launch date, and physical characteristics. The third grouping defines the breakup event by time, location, altitude, and assessed cause. In almost all cases, the calculated time of the event has been determined by the US SSN. Next, the last available element set for the satellite prior to the breakup is provided.

Contents of the pre- or post-event elements are described in Table 2.2-1. The epoch time's format consists of the last two digits of a year (YY) followed by a fractional day of year (DDD.DDDDDDDD). Three propagation scheme drag coefficients are available in a TLE, which form the basis of the orbital element data presented in this subsection. Though not all TLEs possess data for all three, they are described here for completeness; these are denoted as drag coefficients peculiar to the US SSN SGP, SGP4, and SGP8 orbit propagators. The data items $\dot{n}/2$ (pronounced "n dot over two") and $\ddot{n}/6$ (pronounced "n double dot over six") refer to the first and second order time derivatives of the mean motion n and represent phenomenological series expansion coefficient fits to the observed change in mean motion. The SGP4 propagator is the accepted standard for orbit propagation.

TABLE 2.2-1. TLE NUMERICAL DATA, AS INCORPORATED INTO THIS SECTION'S "PRE-/POST-EVENT ELEMENTS" FOR ALL FRAGMENTATION EVENTS.

| <i>DATA ITEM</i> | <i>FORMAT/UNITS</i> |
|--|--|
| Epoch time | YYDDD.DDDDDDDD |
| $\dot{n}/2$ (SGP) <u>or</u> B (SGP8) | [rev/day ²] <u>or</u> [m ² /kg] |
| $\ddot{n}/6$ (SGP) | [rev/day ³] |
| B* (SGP4) | [1/Earth radii] |
| Eccentricity e | [-] |
| Inclination i | [°] |
| Right ascension of ascending node Ω | [°] |
| Argument of perigee ω | [°] |
| Mean anomaly M | [°] |
| Mean motion n | [rev/day] |

If the breakup occurred soon after launch or after a maneuver and before an element set could be generated, the most appropriate post-event element set is given. The maximum observed changes in the orbital period (ΔP) and inclination (ΔI), referenced to the parent's pre-event element set, are then summarized. The reader is reminded that for a given event, the magnitudes of the resultant ΔP and ΔI are a function of the satellite's latitude and altitude. Comparisons of these values from one event to another cannot be made directly. Additionally, inclination changes measure only one portion of the fragmentation orbital plane change. Changes in Right Ascension also occur in most events and can account for some plane change fragmentation energy.

A general summary of the event, actions leading to the event, debris cataloging progress, and evaluations of the event are collected under the Comments heading. Documents that relate directly to the subject breakup or to breakups of satellites of this type are then listed. Gabbard diagrams of the early debris cloud prior to the effects of perturbations, if the data were available, are reconstructed. These diagrams often include uncataloged as well as cataloged debris data. When used correctly, Gabbard diagrams can provide important insights into the features of the fragmentation.

SATELLITE DATA

TYPE: Ablestar Stage
OWNER: US
LAUNCH DATE: 29.18 Jun 1961
DRY MASS (KG): 625
MAIN BODY: Flared cylinder; 1.6 m diameter by 4.8 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event
ENERGY SOURCES: On-board propellants, range safety device

EVENT DATA

| | | | |
|-----------|-------------|-----------------|-----------------|
| DATE: | 29 Jun 1961 | LOCATION: | 28N, 254E (dsc) |
| TIME: | 0608 GMT | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | 990 km | | |

POST-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 61187.36647288 | MEAN ANOMALY: | 72.1786 |
| RIGHT ASCENSION: | 79.1120 | MEAN MOTION: | 13.86864257 |
| INCLINATION: | 66.8199 | MEAN MOTION DOT/2: | .0 |
| ECCENTRICITY: | .0078181 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 288.2398 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 15.5 min
MAXIMUM ΔI : 1.3 deg

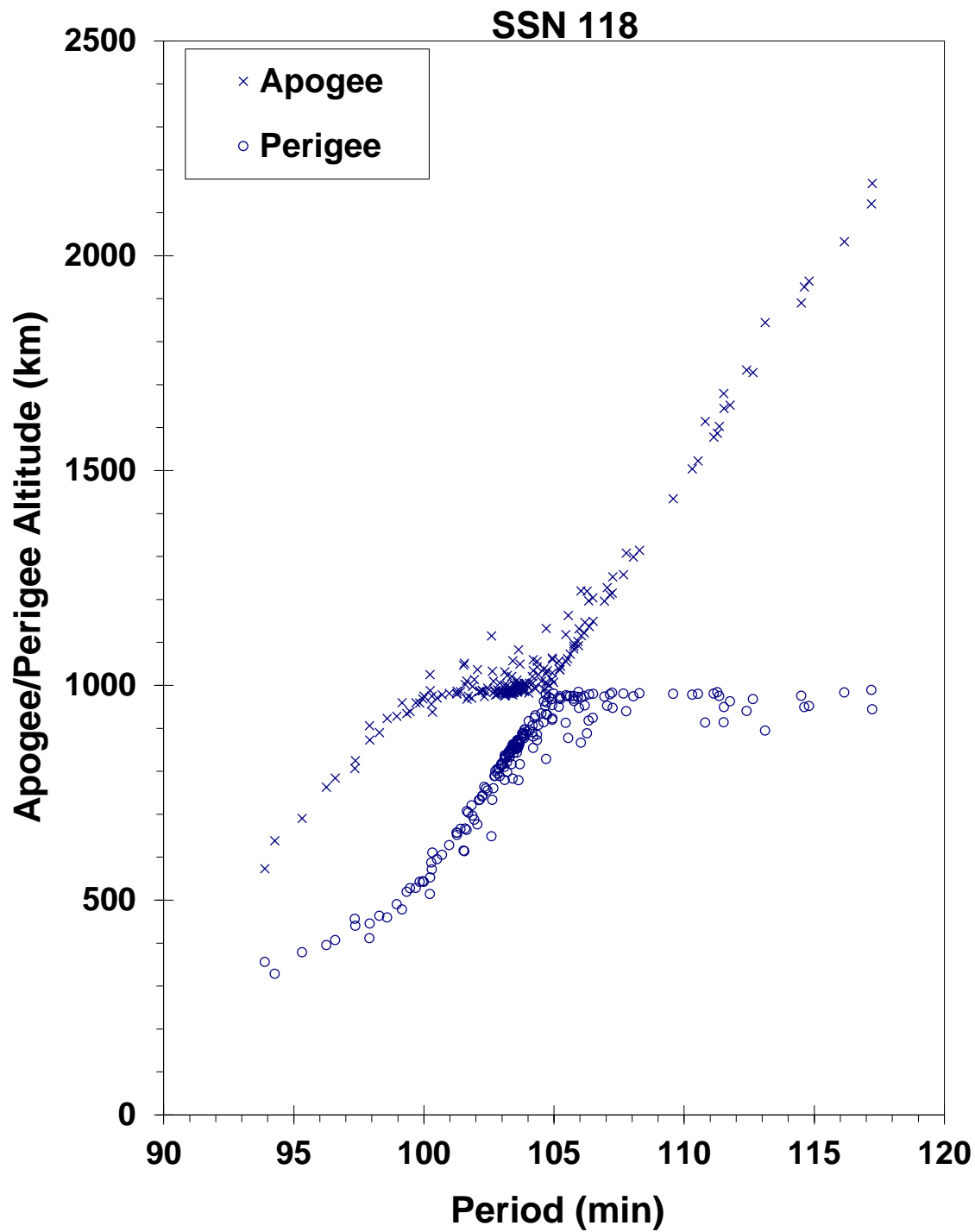
COMMENTS

This is the first known satellite fragmentation. The Ablestar stage performed two main burns and a small payload separation retro burn to successfully deploy three payloads (Transit 4A, Injun, and Solrad 3), although the Injun and Solrad 3 satellites did not separate from one another as planned. The event occurred approximately 77 minutes after orbital insertion and was photographically imaged by the Organ Pass, NM, Baker-Nunn camera system. Fragmentation coincided with cessation of the 378 MHz beacon on the Ablestar stage at 06:08:10 GMT. At the time of the event, 100 kg of hypergolic propellants remained on board. This was the first time an Ablestar stage did not vent the fuel tank during payload separation. After a thorough investigation, fuel venting was recommended for future missions. No reliable elements are available prior to the event. Elements above are for one of the payloads with parameters believed to be very similar to those for the Ablestar at the time of the event.

REFERENCE DOCUMENTS

Transit 4-A Ablestar Vehicle Fragmentation Study (Preliminary), Report TOR-930 (2102)-6, Flight Test Planning and Evaluation Department, Transit Program Office, USAF Systems Command, Inglewood, 28 August 1961.

Description, Operation and Performance of Ablestar Stage AJ10-104S, S/N 008 (Transit 4-A), T.W. Fehr and J.K. Stark, Report No. 2102, Spacecraft Division, Aerojet-General Corporation, Azusa, October 1961.



Transit 4A R/B debris cloud of 201 cataloged fragments in May 1964 as reconstructed from the US SSN database.

SATELLITE DATA

| | |
|-------------------|---|
| TYPE: | Payload and R/B(s) (?) |
| OWNER: | CIS |
| LAUNCH DATE: | 24.75 Oct 1962 |
| DRY MASS (KG): | 1500 |
| MAIN BODY: | Cylinder; 2.6 m diameter by 7.15 m length |
| MAJOR APPENDAGES: | None |
| ATTITUDE CONTROL: | Unknown at time of event |
| ENERGY SOURCES: | On-board propellants |

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 29 Oct 1962 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | ~200 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 62297.80327270 | MEAN ANOMALY: | 229.0409 |
| RIGHT ASCENSION: | 336.4972 | MEAN MOTION: | 16.15589719 |
| INCLINATION: | 65.1128 | MEAN MOTION DOT/2: | .01124103 |
| ECCENTRICITY: | .0044520 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 92.2650 | BSTAR: | .0 |

DEBRIS CLOUD DATA

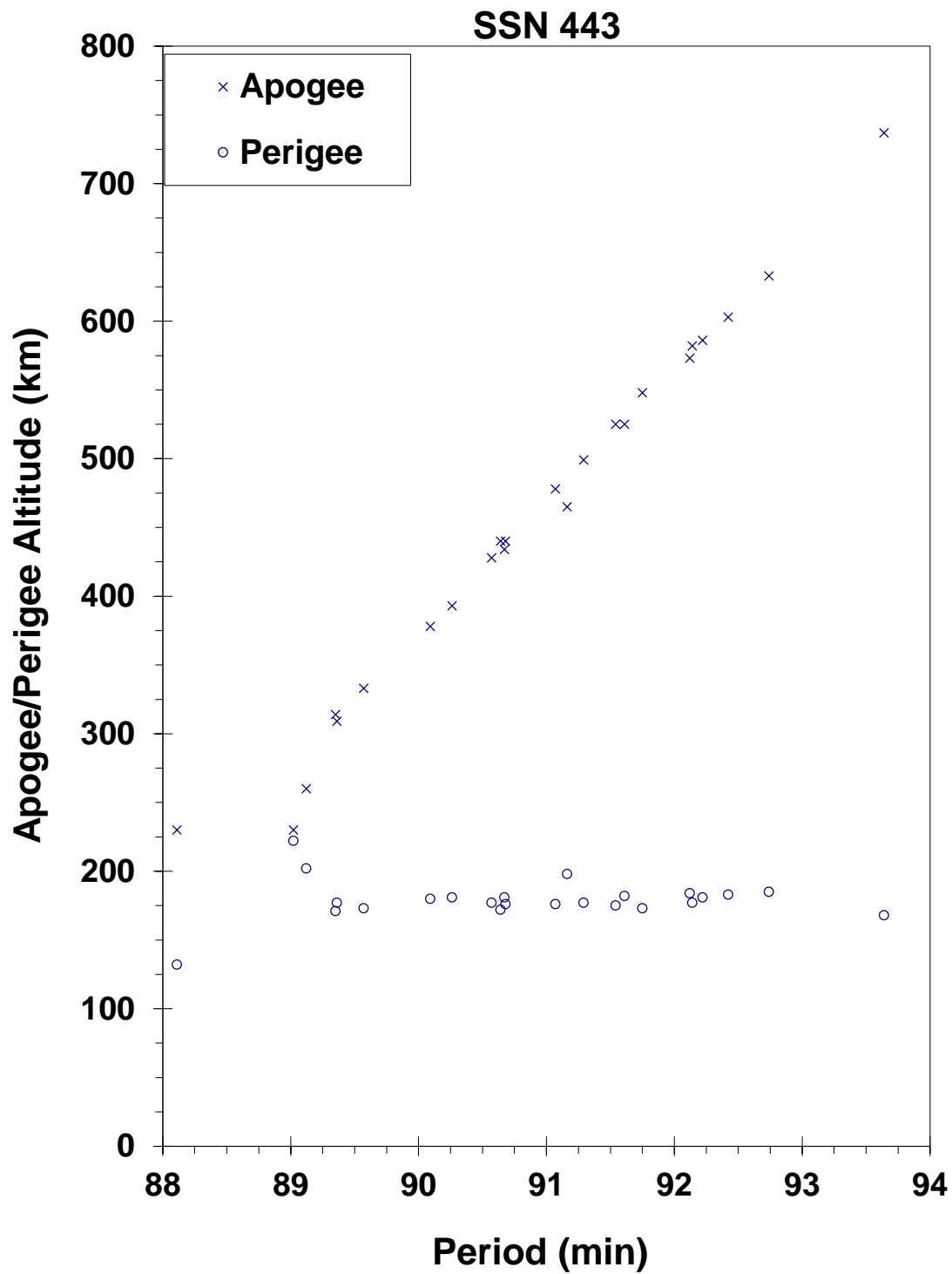
| | |
|----------------------|---------|
| MAXIMUM ΔP : | Unknown |
| MAXIMUM ΔI : | 0.6 deg |

COMMENTS

Sputnik 29 (also known as Sputnik 22) was not acknowledged at launch by the USSR and was probably a Mars probe that failed to leave Earth orbit. This was apparently the fourth orbital failure of the Molniya third stage since 25 August 1962. No Molniya orbital (3rd) stage nor final (4th) stage was cataloged after launch. Possible that orbital and final stages never separated. Sputnik 29 was officially decayed 29 October 1962 but no debris were cataloged before 11 November. Consequently, ΔP cannot be calculated. Source of the fragmentation was probably the fully-fueled Molniya final stage.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Sputnik 29 debris cloud of 24 fragments cataloged by mid-December 1962 as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Centaur Stage
OWNER: US
LAUNCH DATE: 27.79 Nov 1963
DRY MASS (KG): 4600
MAIN BODY: Cylinder; 3 m diameter by 9 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Unknown at time of the event
ENERGY SOURCES: Unknown

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 27 Nov 1963 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

POST-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 63336.85832214 | MEAN ANOMALY: | 213.1623 |
| RIGHT ASCENSION: | 135.1828 | MEAN MOTION: | 13.34437775 |
| INCLINATION: | 30.3440 | MEAN MOTION DOT/2: | .00003262 |
| ECCENTRICITY: | .0869282 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 151.8246 | BSTAR: | .0 |

DEBRIS CLOUD DATA

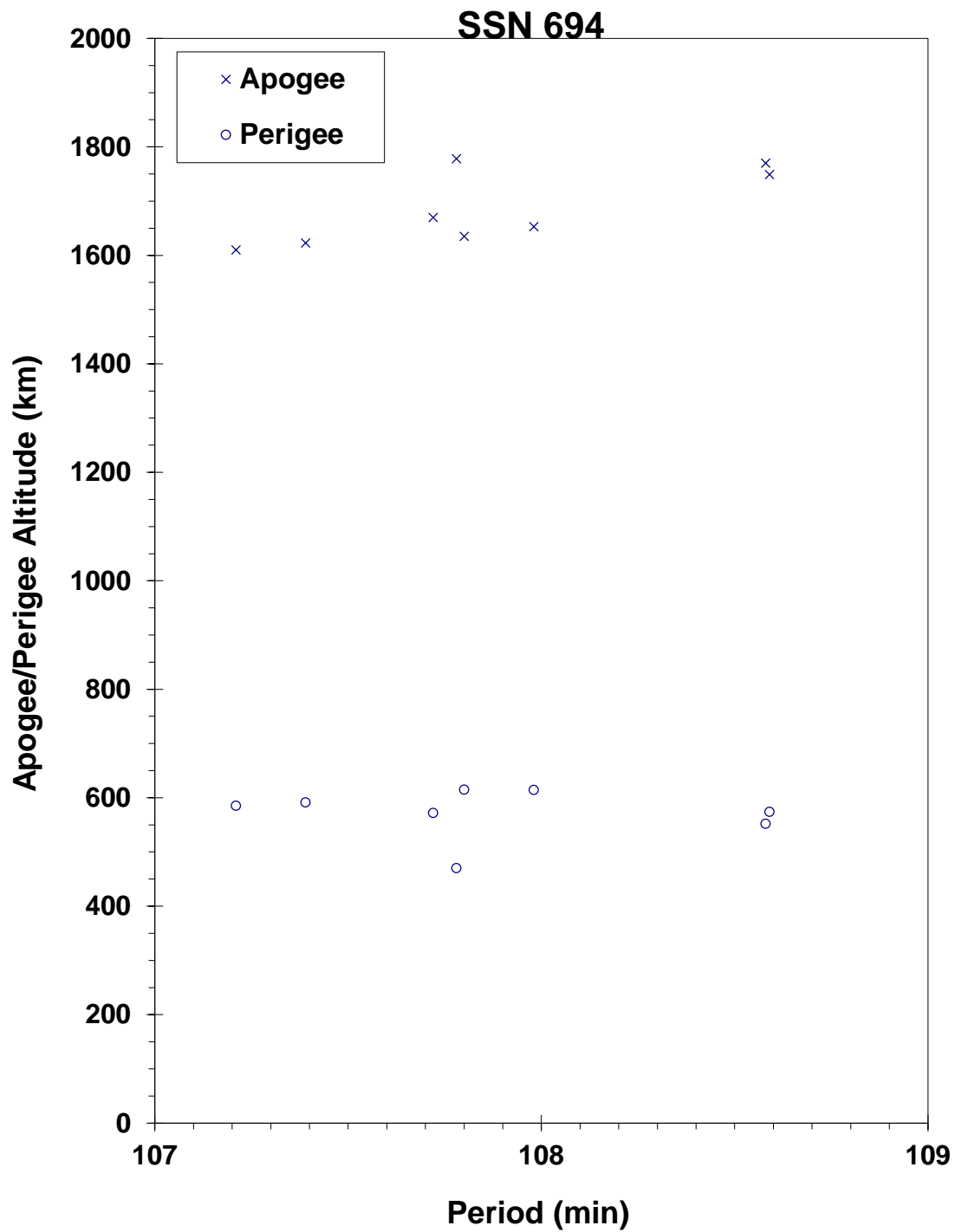
MAXIMUM ΔP : 0.9 min
MAXIMUM ΔI : 0.4 deg

COMMENTS

First Centaur stage to reach Earth orbit. No payload was carried. After orbital insertion, residual liquid hydrogen vaporized, resulting in an increase in tank pressurization. Venting via an aft tube then induced a pin-wheel tumble that reached 48 rpm a little more than 1 hour after launch. At the beginning of the third orbit insulation blankets around the Centaur stage were thrown off. Subsequent Centaur missions were not subject to this phenomenon that was caused by the unique configuration of Atlas Centaur 2. First six fragments were cataloged within 1 week of launch. Centaur stage retains large radar cross-section, while all debris are substantially smaller.

REFERENCE DOCUMENT

Supplementary Information on AC-2 Post-Injection Flight Events, W.S. Hicks, Memorandum BXN63-521, 27 December 1963.



Atlas Centaur 2 debris cloud of 8 fragments 5 months after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 28.45 Oct 1964
DRY MASS (KG): 4750
MAIN BODY: Sphere-cylinder; 2.4 m diameter by 4.3 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, 10 kg TNT explosive charge

EVENT DATA

| | | | |
|-----------|------------|-----------------|------------|
| DATE: | 5 Nov 1964 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | ~200 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 64303.72916435 | MEAN ANOMALY: | 46.7488 |
| RIGHT ASCENSION: | 198.5952 | MEAN MOTION: | 16.23335350 |
| INCLINATION: | 51.2318 | MEAN MOTION DOT/2: | .00269057 |
| ECCENTRICITY: | .0034483 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 312.9624 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

Spacecraft was destroyed after a malfunction prevented reentry and landing in the Soviet Union. Event occurred on the anticipated day of recovery. All debris were cataloged without elements. A probable fragment from this event reentered on 12 November 1964, landing in Malawi. See cited reference below.

REFERENCE DOCUMENTS

The Examination of a Sample of Space Debris, P.H.H. Bishop and K.F. Rogers, Technical Report 65165, Royal Aircraft Establishment, Farnborough Hants, August 1965.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 22.32 Feb 1965
DRY MASS (KG): 5500
MAIN BODY: Sphere-cylinder; 2.4 m diameter by 6 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, 10 kg TNT explosive charge

EVENT DATA

DATE: 22 Feb 1965 LOCATION: 64N, 80E (asc)
TIME: 0957 GMT ASSESSED CAUSE: Deliberate
ALTITUDE: 380 km

POST-EVENT ELEMENTS

EPOCH: 65056.64509999 MEAN ANOMALY: 293.2095
RIGHT ASCENSION: 288.1532 MEAN MOTION: 15.92461677
INCLINATION: 64.7411 MEAN MOTION DOT/2: .01501524
ECCENTRICITY: .0182240 MEAN MOTION DOT DOT/6: .0048063
ARG. OF PERIGEE: 68.7266 BSTAR: .0

DEBRIS CLOUD DATA

MAXIMUM ΔP : 4.4 min
MAXIMUM ΔI : 0.9 deg

COMMENTS

Cosmos 57 was an unmanned precursor for the manned Voskhod 2 mission that took place in March 1965. Spacecraft fragmented a little more than 2 hours after launch when operational ground instructions were misinterpreted by the on-board command system and the self-destruct system was activated. No elements available for Cosmos 57, but the rocket body elements are provided above. The Royal Aircraft Establishment published the following parameters for Cosmos 57 for 22.4 February: 165 km by 427 km, 64.74 degree inclination, 64 degree argument of perigee. A total of 35 debris were cataloged without elements. Event may have occurred a little later than the time calculated above.

REFERENCE DOCUMENTS

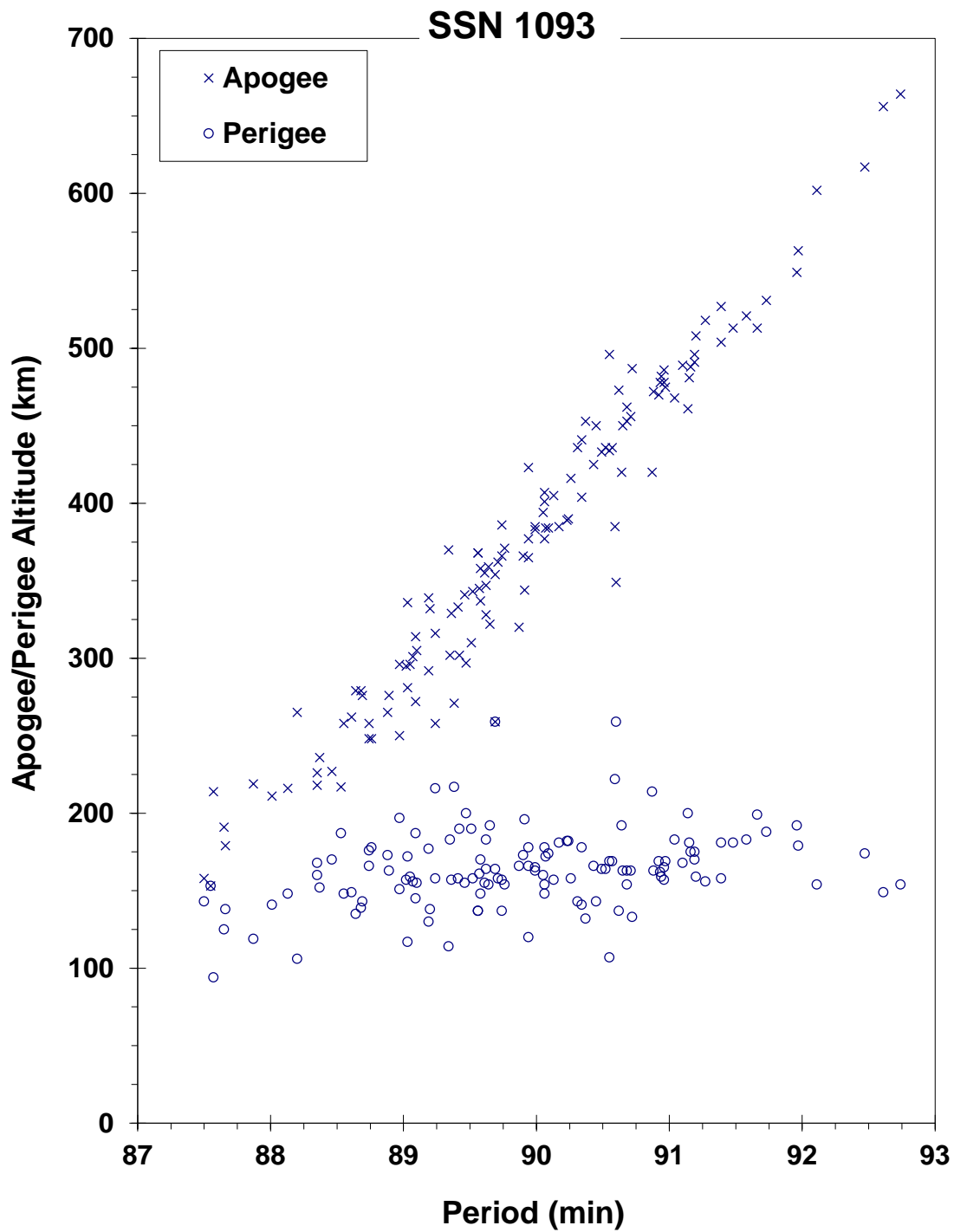
The 1093 Breakup, D.J. Watson, BMEWS-ADC Systems Engineering Memorandum BSM-1000-16, 16 June 1965.

"To Save Man: A Conversation with the General Designer of Life-Support and Rescue Systems, Hero of Socialist Labor G.I. Severin", Pravda, Moscow, 26 June 1989, p. 4.

"Pages From a Diary: He Soared Freely Above the Earth", Sovetskaya Rossiya, Moscow, 17 March 1990, p. 6.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

"The Kamanin Diaries 1964-1966", B. Hendrickx, Journal of the Interplanetary Society, Vol. 51, 1998, pp. 421-422.



Cosmos 57 debris cloud of 133 fragments cataloged within 1 month of the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Cosmos Second Stage
OWNER: CIS
LAUNCH DATE: 15.46 Mar 1965
DRY MASS (KG): 1600
MAIN BODY: Cylinder; 2.4 m diameter by 5 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: Unknown

EVENT DATA

DATE: 15 Mar 1965 LOCATION: 51S, 162E (dsc)
TIME: 1714 GMT ASSESSED CAUSE: Unknown
ALTITUDE: 1640 km

POST-EVENT ELEMENTS

EPOCH: 65074.89183830 MEAN ANOMALY: 265.7165
RIGHT ASCENSION: 357.3218 MEAN MOTION: 13.57884745
INCLINATION: 56.0538 MEAN MOTION DOT/2: .00231832
ECCENTRICITY: .1056119 MEAN MOTION DOT DOT/6: .0
ARG. OF PERIGEE: 106.1560 BSTAR: .0

DEBRIS CLOUD DATA

MAXIMUM ΔP : 10.3 min
MAXIMUM ΔI : 0.4 deg

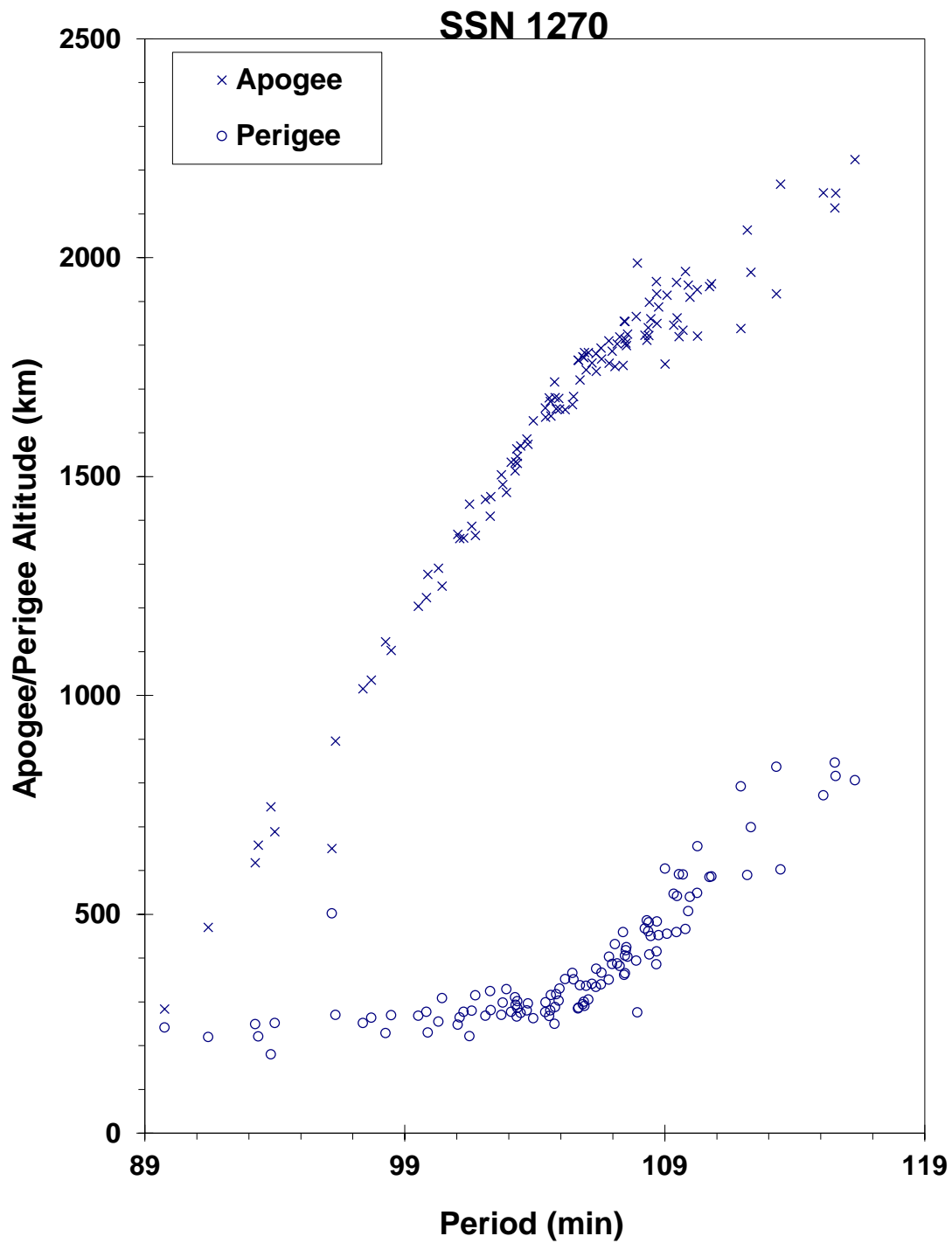
COMMENTS

This is the first confirmed case of the fragmentation of the Cosmos 3 (SL-8 or C-1) second stage. This was the third mission to deploy three payloads and was a repeat of the Cosmos 54-56 mission 3 weeks earlier. The event occurred a little more than 6 hours after the successful deployment of the three payloads. Elements above are the first developed for the rocket body and are about 4 hours after the event. Official debris cataloging did not begin for 6 weeks.

REFERENCE DOCUMENTS

"Fragmentations of Asteroids and Artificial Satellites in Orbit", W. Wiesel, Icarus, Vol. 34, 1978, pp. 99-116.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 61-63 R/B debris cloud of 113 fragments 8 months after the event as reconstructed from the US SSN database.

SATELLITE DATA

| | |
|-------------------|---|
| TYPE: | Payload |
| OWNER: | CIS |
| LAUNCH DATE: | 4.23 Nov 1965 |
| DRY MASS (KG): | 400 |
| MAIN BODY: | Ellipsoid; 1.2 m diameter by 1.8 m length |
| MAJOR APPENDAGES: | Unknown |
| ATTITUDE CONTROL: | Unknown |
| ENERGY SOURCES: | Unknown |

EVENT DATA

| | | | |
|-----------|-------------|-----------------|---------|
| DATE: | 15 Jan 1966 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------|------------------------|-------------|
| EPOCH: | 66009.5 | MEAN ANOMALY: | Unknown |
| RIGHT ASCENSION: | Unknown | MEAN MOTION: | 16.09757275 |
| INCLINATION: | 48.39 | MEAN MOTION DOT/2: | Unknown |
| ECCENTRICITY: | 0.009282 | MEAN MOTION DOT DOT/6: | Unknown |
| ARG. OF PERIGEE: | 77 | BSTAR: | Unknown |

DEBRIS CLOUD DATA

| | |
|----------------------|---------|
| MAXIMUM ΔP : | Unknown |
| MAXIMUM ΔI : | Unknown |

COMMENTS

Pre-event elements were taken from RAE Table of Earth Satellites. Cosmos 95 was placed into a low Earth orbit on 4 November 1965. Within 2 weeks nearly two dozen debris had been detected and were later cataloged. However, the nature of the debris, i.e. breakup versus operational, was not determined. The last of these debris decayed naturally by 6 January 1966. Russian records indicate that a breakup may have occurred on 15 January 1966, 3 days before the 400 kg spacecraft itself reentered. No other information on this event has been discovered, and no debris remains in orbit.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Titan 3C-4 Transtage
OWNER: US
LAUNCH DATE: 15.72 Oct 1965
DRY MASS (KG): 2500
MAIN BODY: Cylinder; 3 m diameter by 6 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|-------------|-----------------|-----------------|
| DATE: | 15 Oct 1965 | LOCATION: | 22S, 108E (asc) |
| TIME: | 1820 GMT | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | 740 km | | |

POST-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 65361.23126396 | MEAN ANOMALY: | 237.1066 |
| RIGHT ASCENSION: | 21.5316 | MEAN MOTION: | 14.54928550 |
| INCLINATION: | 32.1697 | MEAN MOTION DOT/2: | .00000268 |
| ECCENTRICITY: | .0072678 | MEAN MOTION DOT DOT/6: | .071801 |
| ARG. OF PERIGEE: | 123.6068 | BSTAR: | .0 |

DEBRIS CLOUD DATA

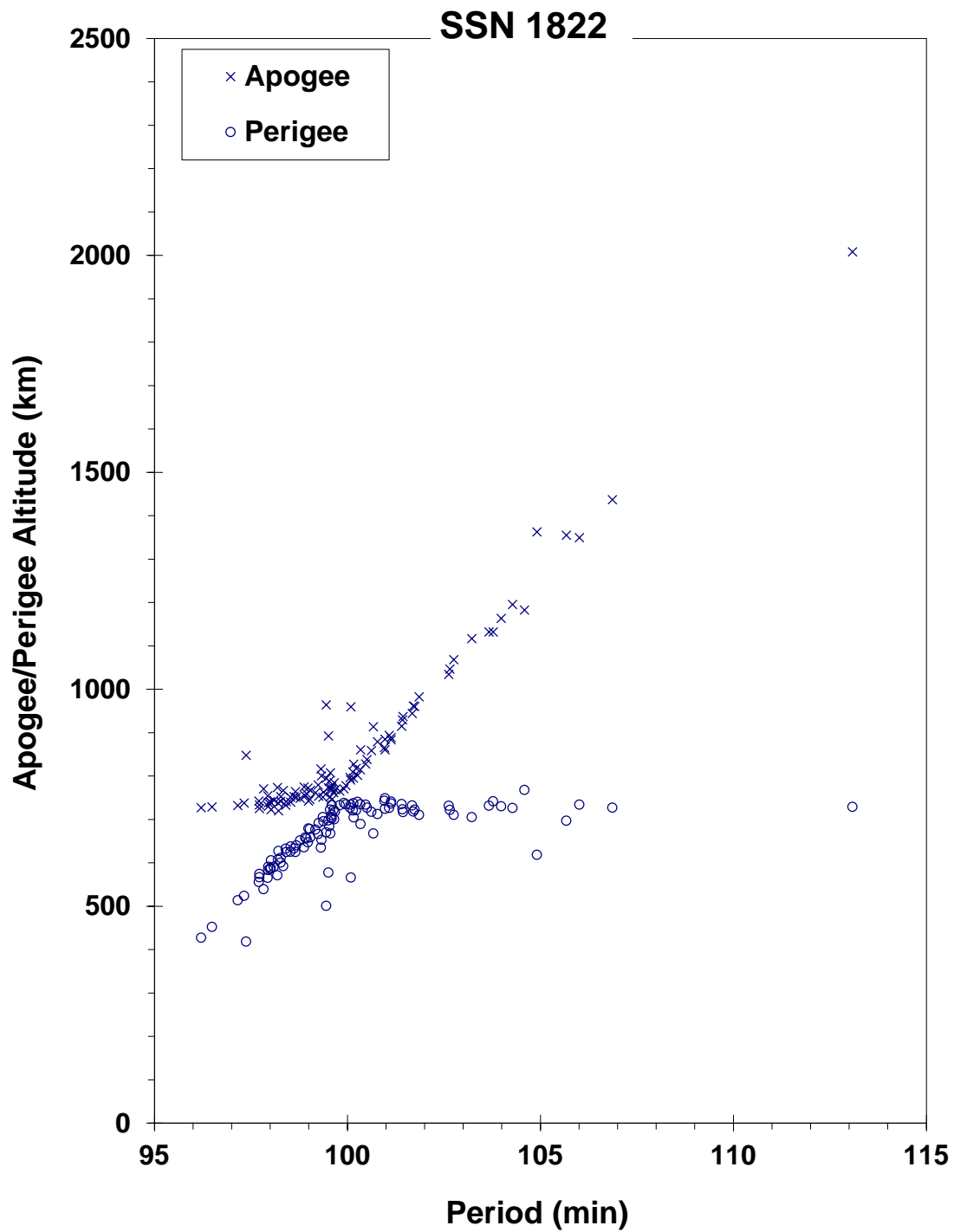
MAXIMUM ΔP : 4.1 min
MAXIMUM ΔI : 1.4 deg

COMMENTS

This was the second test of the Titan 3C-4 Transtage with AJ10-138 engine using hypergolic propellants. Event occurred one-half revolution after launch following second ignition that may have been accompanied with vehicle tumbling. LCS 2 payload was to have been deployed at 735 km circular while OV2-1 was to have been released later in an orbit of 735 km by about 7400 km. Transtage also malfunctioned on next mission in December 1965. The main remnant of the rocket body was recently identified as Satellite No. 1822 (1965-082DM). Previous editions of this book had identified the main rocket body remnant as Satellite No. 1640 (1965-082B).

REFERENCE DOCUMENT

TRW Space Log, Winter 1965-66, Vol. 5, No. 4, T.L. Branigan, ed., TRW Systems, Redondo Beach, 1966, pp. 15-17.



OV2-1/LCS 2 R/B debris cloud of 103 cataloged fragments 6 weeks after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Titan IIIC Transtage Rocket Body
OWNER: US
LAUNCH DATE: 21 Dec 1965
DRY MASS (KG): 2,555
MAIN BODY: 3.048 m diameter x ~5 m long right circular cylinder
MAJOR APPENDAGES: none
ATTITUDE CONTROL: 3-axis
ENERGY SOURCES: on-board propellants; pressure vessels; chemical batteries

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 21 Dec 1965 | LOCATION: | unknown |
| TIME: | unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | unknown | | |

POST-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 69013.84962483 | MEAN ANOMALY: | 2.6587 |
| RIGHT ASCENSION: | 206.0825 | MEAN MOTION: | 4.07936348 |
| INCLINATION: | 26.4899 | MEAN MOTION DOT/2: | 0.00014521 |
| ECCENTRICITY: | 0.6007336 | MEAN MOTION DOT DOT/6: | 0.00000 |
| ARG. OF PERIGEE: | 346.5872 | BSTAR: | 0.0097405 |

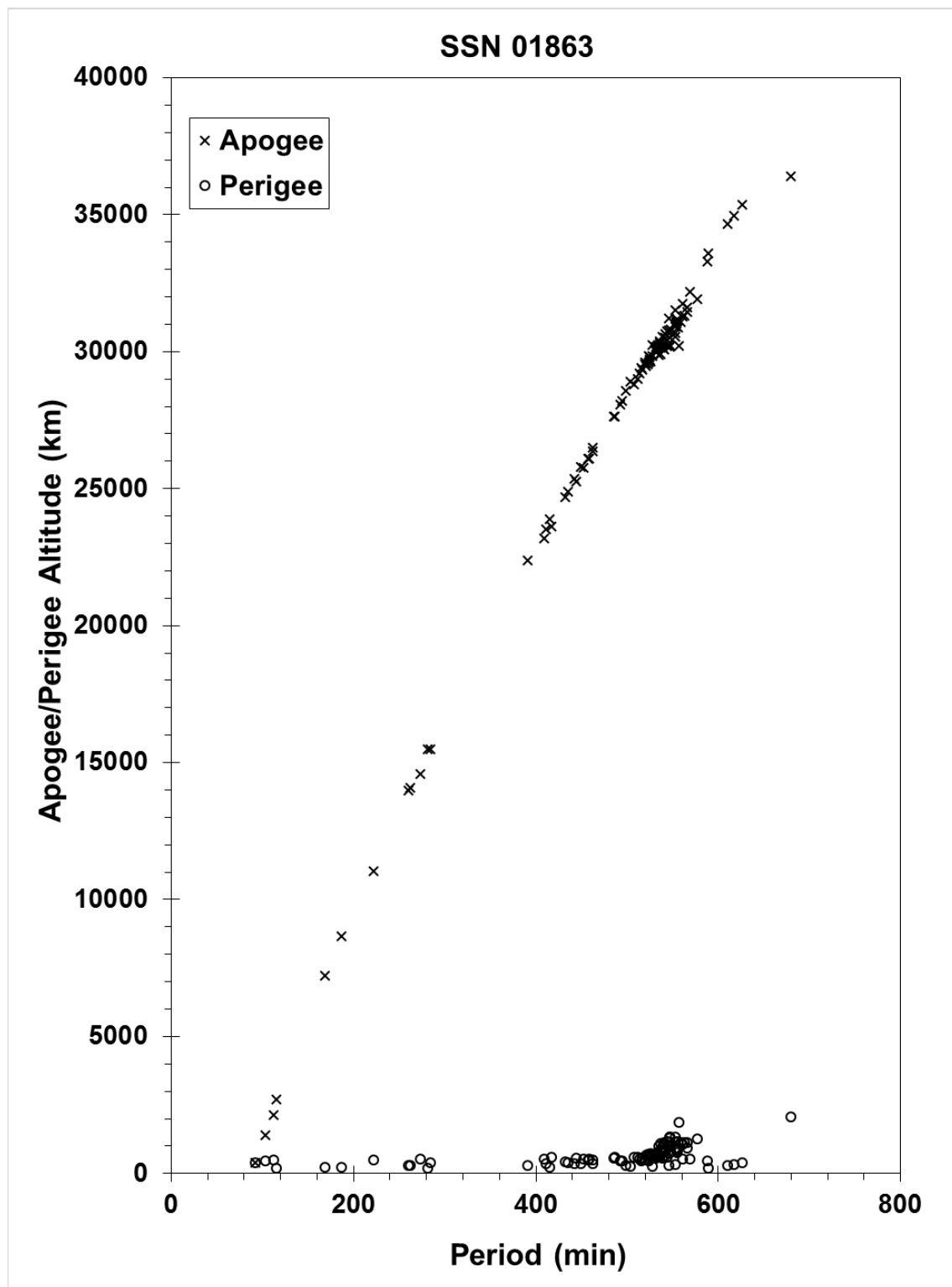
DEBRIS CLOUD DATA

MAXIMUM ΔP : unknown
MAXIMUM ΔI : unknown

COMMENTS

The cause of this event is assessed to be propulsion-related. The 13th ed. of the History of On-Orbit Satellite Fragmentations described this event as a provisional fragmentation.

REFERENCE DOCUMENT



Titan Transtage R/B cloud; some data cataloged up to 49 years after event.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 15.85 Feb 1966
DRY MASS (KG): 4
MAIN BODY: Sphere; 0.3 m diameter
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: Unknown

EVENT DATA

| | | | |
|-----------|-------------|-----------------|---------|
| DATE: | 15 Feb 1966 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | ~200 km | | |

POST-EVENT ELEMENTS

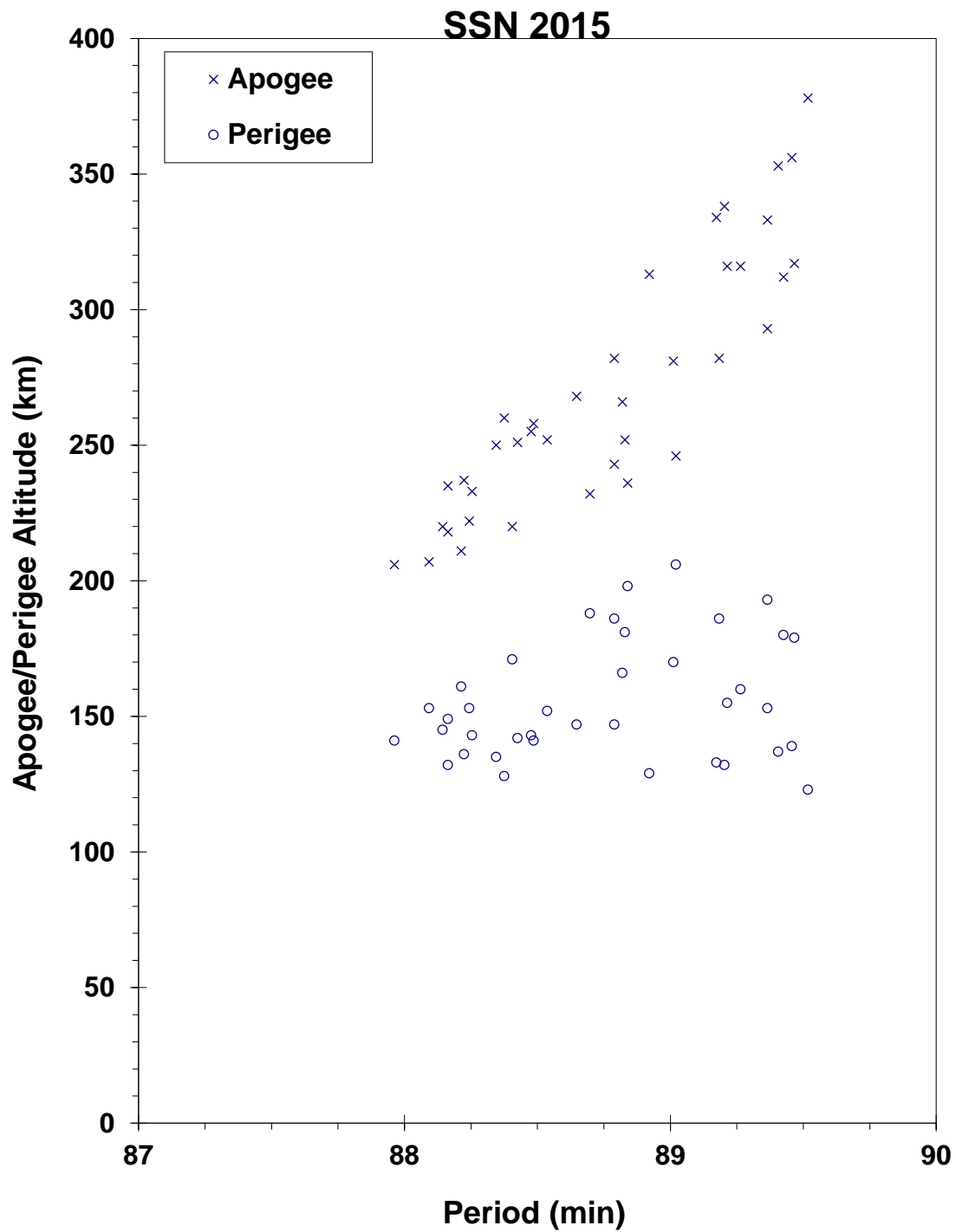
| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 66047.01671304 | MEAN ANOMALY: | 234.6777 |
| RIGHT ASCENSION: | 148.6481 | MEAN MOTION: | 16.20030654 |
| INCLINATION: | 96.5380 | MEAN MOTION DOT/2: | .01298049 |
| ECCENTRICITY: | .0108362 | MEAN MOTION DOT DOT/6: | .0053719 |
| ARG. OF PERIGEE: | 126.3670 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : 0.6 deg

COMMENTS

OPS 3031 was an inflated sphere also known as Bluebell 2. It was deployed from satellite 2012, which was an Agena D stage carrying a separate payload. Elements above are for satellite 2012. Debris cataloging began 19 February after many debris had already decayed. Consequently, ΔP cannot be calculated. OPS 3031 and all debris decayed within 1 week of launch.



OPS 3031 debris cloud of 38 fragments as initially cataloged
by the US SSN during February 1966.

SATELLITE DATA

TYPE: Atlas Core Stage
OWNER: US
LAUNCH DATE: 1.63 Jun 1966
DRY MASS (KG): 3400
MAIN BODY: Cylinder; 3 m diameter by 20 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: Unknown

EVENT DATA

| | | | |
|----------|--------------|-----------------|---------|
| DATE: | Mid-Jun 1966 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Unknown |
| ALTITUDE | ~250 km | | |

PRE-EVENT ELEMENTS

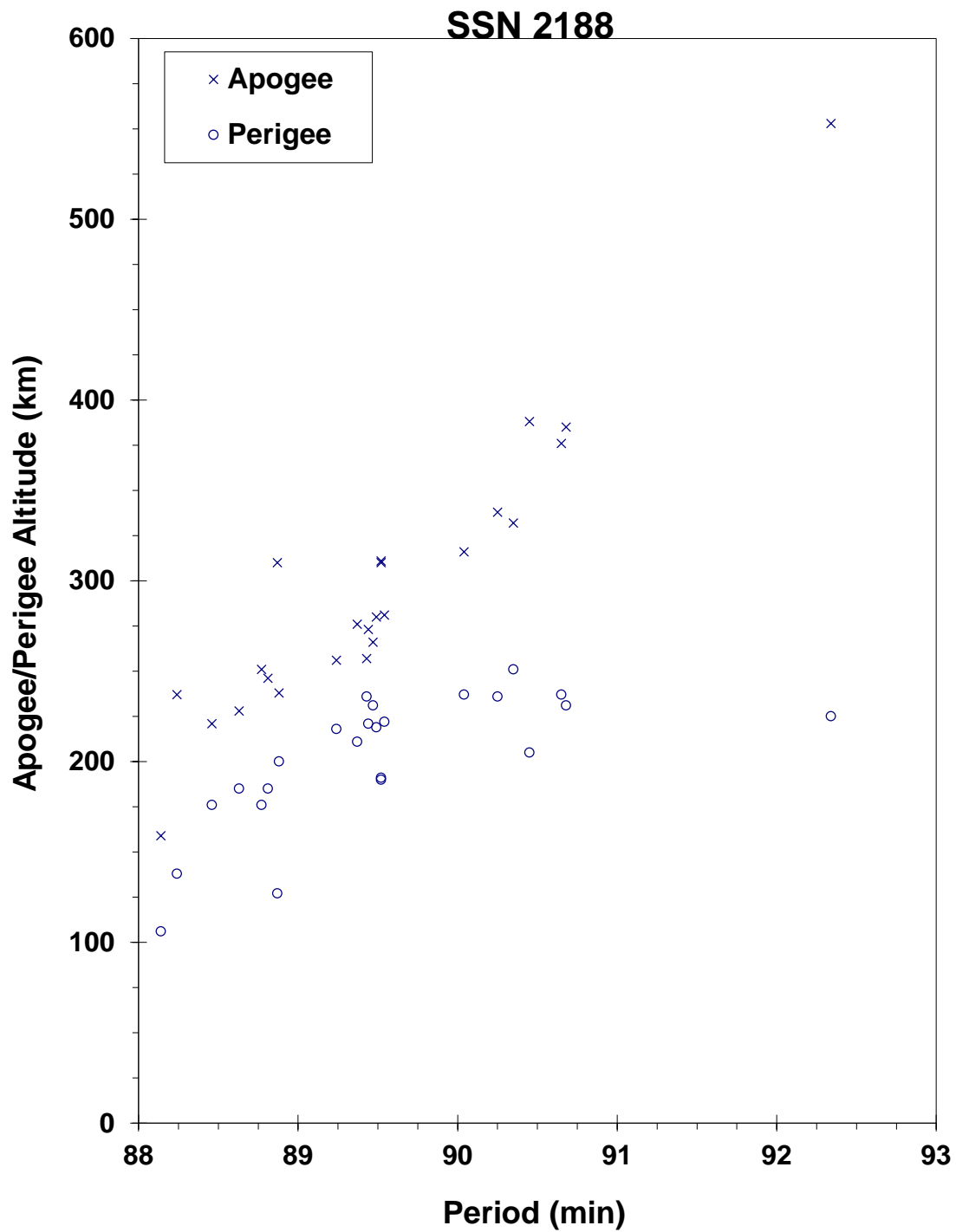
| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 66164.96883397 | MEAN ANOMALY: | 224.9775 |
| RIGHT ASCENSION: | 223.9064 | MEAN MOTION: | 16.05545399 |
| INCLINATION: | 28.7968 | MEAN MOTION DOT/2: | .00654808 |
| ECCENTRICITY: | .0025152 | MEAN MOTION DOT DOT/6: | .0010778 |
| ARG. OF PERIGEE: | 135.2510 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 5.5 min
MAXIMUM ΔI : 1.5 deg

COMMENTS

This stage successfully deployed the Augmented Target Docking Adapter (ATDA) for the Gemini 9 mission. The elements above are the last available for the rocket body. Debris cataloging began on 21 June. Debris decay dates ranged from 21 June to 4 July with the rocket body officially decaying on 22 June. A review of NASA archives for this mission revealed no documented anomaly with the Atlas booster. Discussions in 1989 with General Dynamics personnel involved in the mission (Mr. Phil Genser of General Dynamics, San Diego) also failed to uncover any knowledge of the event. Pressure relief valves should have relieved pressurization increases, particularly in the oxygen tank. Possible failure of the oxygen relief valve could not be ruled out.



Gemini 9 ATDA R/B debris cloud of 24 fragments cataloged between 21 and 24 June as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 24.01 Jun 1966
DRY MASS (KG): 55
MAIN BODY: Sphere; 30 m diameter
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: None

EVENT DATA (1)

| | | | |
|-----------|-------------|-----------------|-----------------|
| DATE: | 12 Jul 1975 | LOCATION: | 67N, 135E (dsc) |
| TIME: | 2248 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 5145 km | | |

PRE-EVENT ELEMENTS (1)

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 75192.78059719 | MEAN ANOMALY: | 67.9594 |
| RIGHT ASCENSION: | 238.7429 | MEAN MOTION: | 7.99684492 |
| INCLINATION: | 85.2811 | MEAN MOTION DOT/2: | .00001217 |
| ECCENTRICITY: | .0931904 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 281.8264 | BSTAR: | .77087 |

EVENT DATA (2)

| | | | |
|-----------|-------------|-----------------|---------|
| DATE: | 20 Jan 1976 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS (2)

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 76019.86486339 | MEAN ANOMALY: | 305.5539 |
| RIGHT ASCENSION: | 209.8639 | MEAN MOTION: | 8.00368182 |
| INCLINATION: | 85.0720 | MEAN MOTION DOT/2: | .0 |
| ECCENTRICITY: | .1179567 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 66.4633 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 0.1 min*
MAXIMUM ΔI : 0.7 deg*

*Based on 1st event data

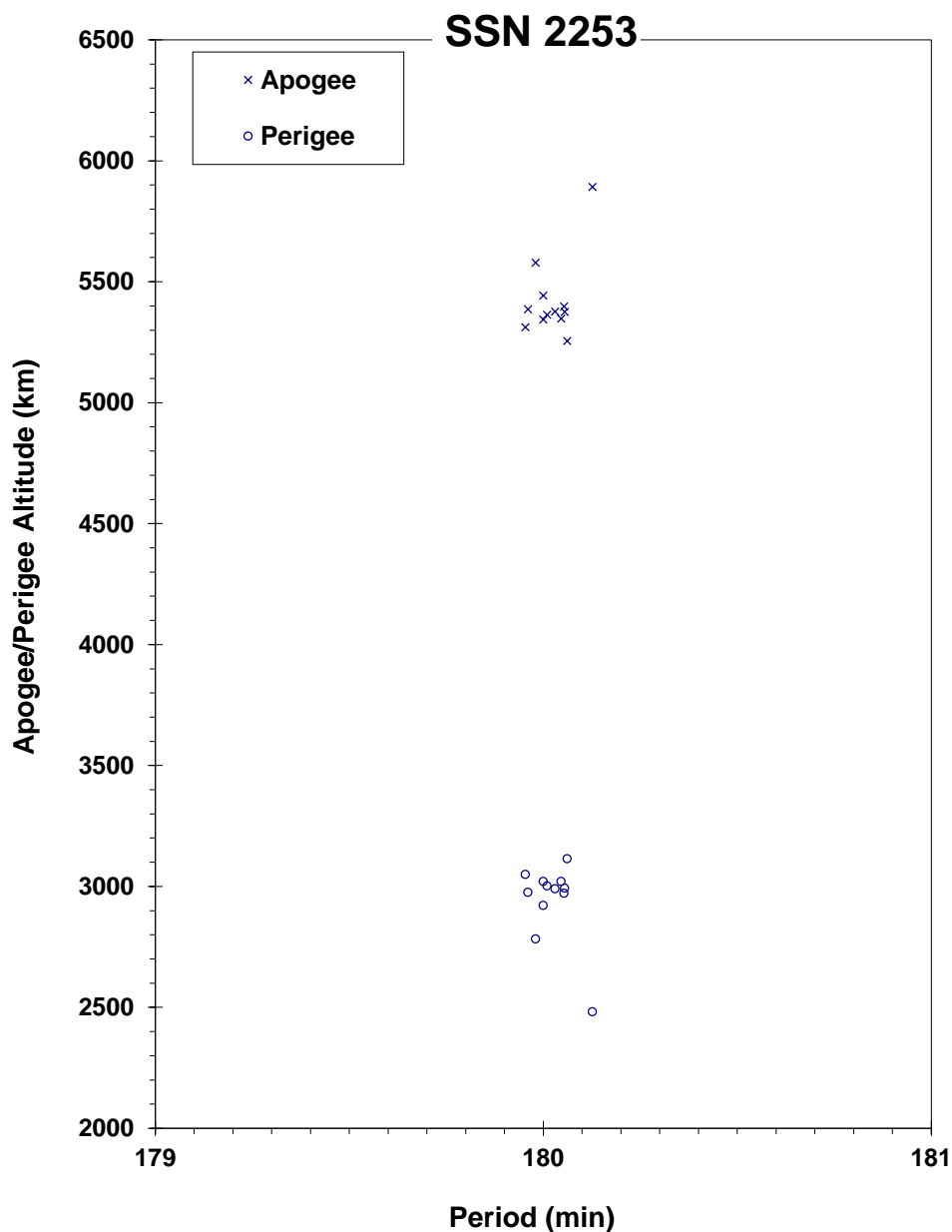
COMMENTS

PAGEOS (Passive Geodetic Earth-Orbiting Satellite) was an inflated balloon made of thin Mylar with an aluminum coating. The first fragmentation event occurred 9 years after launch and resulted in 11 new cataloged objects. The second event was detected by D.G. King-Hele of the RAE, and NAVSPASUR confirmed 44 additional fragments. By August 1976 no additional debris had been cataloged but 19 objects were being tracked in orbits with mean motions near 8 and eccentricities between 0.16 and 0.34. Due to the character of PAGEOS and its subsequent debris, natural perturbations had little effect on orbital period but strongly increased eccentricity by simultaneously lowering perigee and raising apogee. About 10 September 1976 one of the 19 unofficial objects is believed to have broken up into perhaps more than 250 new pieces, none of which were cataloged prior to reentry. Eighteen objects were later cataloged during 7-8 October 1976. On the first anniversary of the second fragmentation (20 Jan 1977), 45 fragments were cataloged without elements and immediately decayed administratively. Additional fragmentations are suspected

to have taken place in June 1978, September 1984, and December 1985. Historically, radar tracking of PAGEOS debris has been extremely difficult and cross-tagging frequent. Cause for the second and subsequent events may be material deterioration under environmental stress. A suspected PAGEOS fragment, SSN 5994, which was cataloged as a Westford Needles object, fragmented on 8 September 1995 and again on 14 September 1995 with 12 associated objects.

REFERENCE DOCUMENT

Spacetrack System Data Related to Some Non-Routine Events Through May 1981, J.R. Gabbard, Technical Memorandum 81-6, DCS/Plans, Hdqtrs NORAD/ADCOM, Colorado Springs, 30 June 1981.



PAGEOS debris cloud of 12 fragments 5 weeks after the first event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Saturn SIVB Stage
OWNER: US
LAUNCH DATE: 5.62 Jul 1966
DRY MASS (KG): 26,600
MAIN BODY: Cylinder; 6.6 m diameter by 28.3 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: Attitude control and pressurization systems

EVENT DATA

DATE: 5 Jul 1966
TIME: 2111 GMT
ALTITUDE: 205 km
LOCATION: 20N, 277E (dsc)
ASSESSED CAUSE: Deliberate

PRE-EVENT ELEMENTS

EPOCH: 66186.73481847
RIGHT ASCENSION: 5.5870
INCLINATION: 31.9810
ECCENTRICITY: .0022272
ARG. OF PERIGEE: 6.1632
MEAN ANOMALY: 353.9219
MEAN MOTION: 16.27379993
MEAN MOTION DOT/2: .03796193
MEAN MOTION DOT DOT/6: .17429
BSTAR: .0

DEBRIS CLOUD DATA

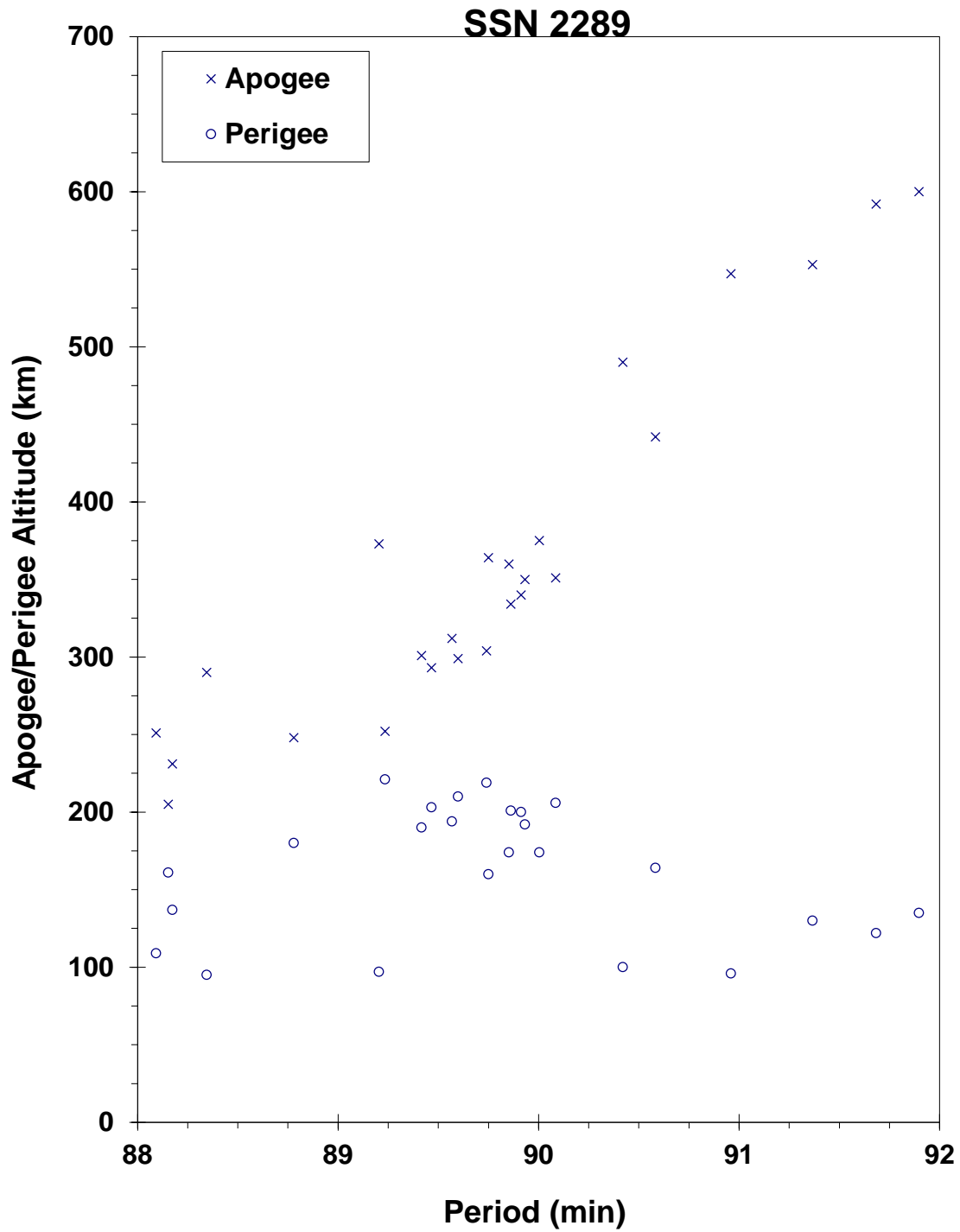
MAXIMUM ΔP : 3.5 min
MAXIMUM ΔI : 1.4 deg

COMMENTS

This was the second flight of the SIVB stage. After orbital insertion, the vehicle was intentionally subjected to dynamic integrity tests, including high gravity loadings during attitude control maneuvers and high pressure tests. The vehicle finally broke up after exceeding structural design limits with a propellant tank bulkhead differential pressure in excess of 23.7 N/cm^2 . The fragmentation occurred early on the fifth revolution. Elements for the first fragments were not cataloged until 8 July.

REFERENCE DOCUMENT

Saturn AS-203 Evaluation Bulletin, No. 2, R-AERO-F-142-66, J.P. Lindberg, NASA Marshall Space Flight Center, Alabama, 21 July 1966.



AS-203 debris cloud of 25 fragments using orbits developed within 1 week of the event as reconstructed from the US SSN database.

SATELLITE DATA

| | |
|-------------------|---|
| TYPE: | Unknown |
| OWNER: | CIS |
| LAUNCH DATE: | 17.94 Sep 1966 |
| DRY MASS (KG): | Unknown |
| MAIN BODY: | Cone-cylinder; 1.5 m diameter by 6 m length |
| MAJOR APPENDAGES: | None |
| ATTITUDE CONTROL: | Unknown |
| ENERGY SOURCES: | Explosive device |

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 17 Sep 1966 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | ~300 km | | |

POST-EVENT ELEMENTS

| | | | |
|------------------|---------|------------------------|--------|
| EPOCH: | 66261.0 | MEAN ANOMALY: | 283 |
| RIGHT ASCENSION: | 338 | MEAN MOTION: | 14.879 |
| INCLINATION: | 49.63 | MEAN MOTION DOT/2: | .0 |
| ECCENTRICITY: | .063 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 83 | BSTAR: | .0 |

DEBRIS CLOUD DATA

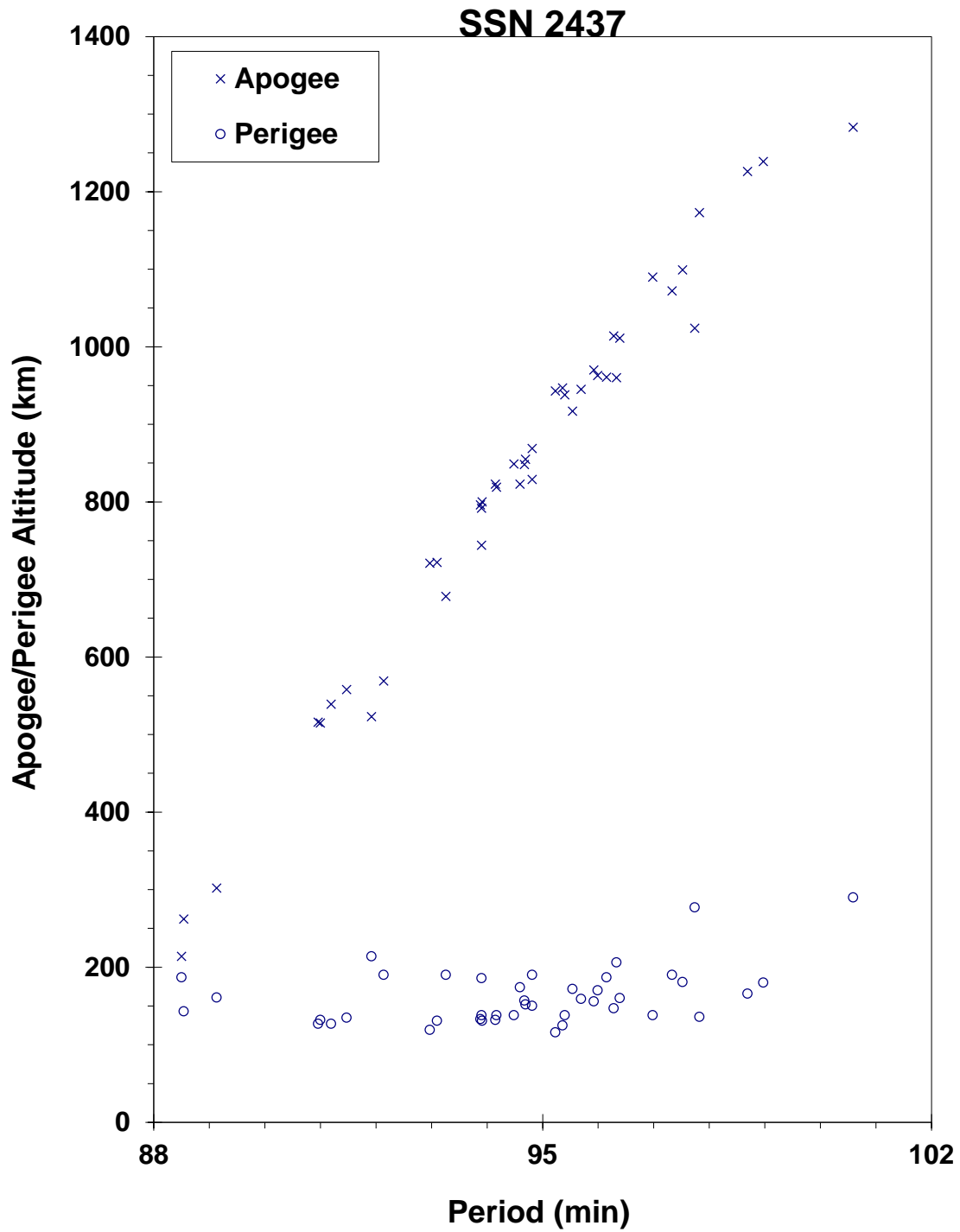
| | |
|----------------------|---------|
| MAXIMUM ΔP : | Unknown |
| MAXIMUM ΔI : | Unknown |

COMMENTS

This was the first of two missions of this type flown in 1966 and not acknowledged by the USSR. The identity of the parent orbit is uncertain. Satellite 2437 was the first cataloged fragment. The above elements are taken or derived from the RAE Table of Earth Satellites. The debris distribution is consistent with a fragmentation near 300 km. Failure of the payload led to immediate activation of the self-destruct system.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos U-1 debris cloud of 43 fragments cataloged by 5 October 1966 as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Unknown
OWNER: CIS
LAUNCH DATE: 2.03 Nov 1966
DRY MASS (KG): Unknown
MAIN BODY: Cone-cylinder; 1.5 m diameter by 6 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Unknown
ENERGY SOURCES: Explosive device

EVENT DATA

DATE: 2 Nov 1966
TIME: Unknown
ALTITUDE: ~225 km
LOCATION: Unknown
ASSESSED CAUSE: Deliberate

POST-EVENT ELEMENTS

EPOCH: 66309.99121234
RIGHT ASCENSION: 35.2944
INCLINATION: 49.5617
ECCENTRICITY: .05339049
ARG. OF PERIGEE: 100.3324
MEAN ANOMALY: 265.7893
MEAN MOTION: 15.17033022
MEAN MOTION DOT/2: .01866914
MEAN MOTION DOT DOT/6: .0043309
BSTAR: .0

DEBRIS CLOUD DATA

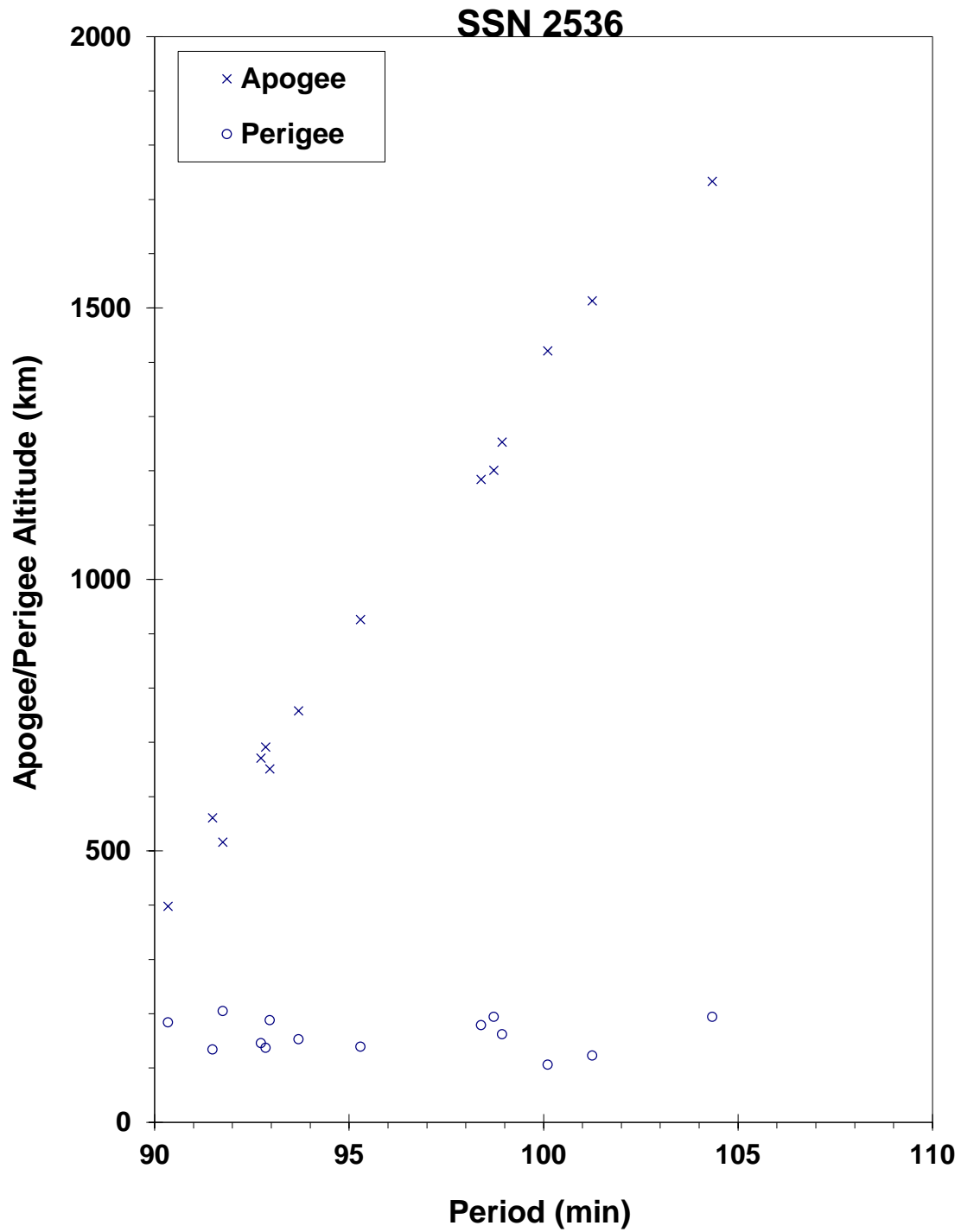
MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

This was the second mission of this type flown in 1966 and not acknowledged by the USSR. No elements were cataloged until 3 days after the launch. The identity of the parent orbit is uncertain. Satellite 2536 was the first object cataloged and was near the center of the debris cloud. The debris distribution is consistent with a fragmentation near 225 km. Failure of the payload led to immediate activation of the self-destruct system.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos U-2 debris cloud composed of 14 different orbits as developed by the US SSN within 1 week of the event.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 23.96 Nov 1967
DRY MASS (KG): 750
MAIN BODY: 2.035 m diameter x 2 m long right circular cylinder
MAJOR APPENDAGES: gravity gradient stabilization boom
ATTITUDE CONTROL: gravity gradient (passive)
ENERGY SOURCES: pressurized payload compartment; chemical batteries

EVENT DATA

DATE: 30 Aug 2009 LOCATION: 33.4S, 18.0E (asc)
TIME: 2150 GMT ASSESSED CAUSE: Unknown
ALTITUDE: 728.2. km

PRE-EVENT ELEMENTS

EPOCH: 09241.21736132 MEAN ANOMALY: 353.9164
RIGHT ASCENSION: 137.1436 MEAN MOTION: 14.53942052
INCLINATION: 74.0172 MEAN MOTION DOT/2: 0.00000028
ECCENTRICITY: 0.0005259 MEAN MOTION DOT DOT/6: 0.0
ARG. OF PERIGEE: 6.2065 BSTAR: 0.000013364

DEBRIS CLOUD DATA

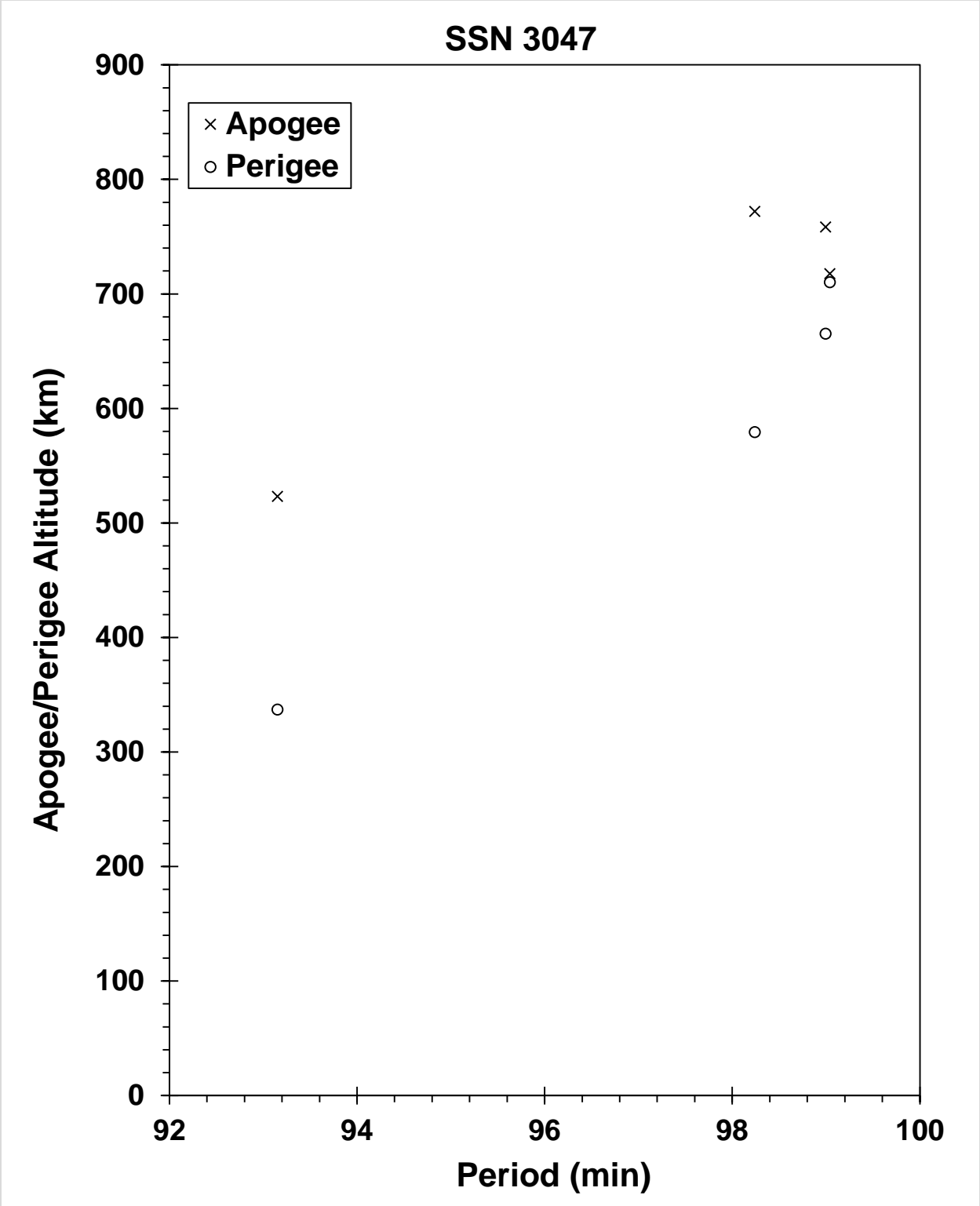
MAXIMUM ΔP : 5.9 min
MAXIMUM ΔI : 0.4 deg

COMMENTS

Cosmos 192 was a prototype or early spacecraft in the Tsyklon low altitude navigation satellite system. The spacecraft utilized the versatile KAUR-1 bus (*Kosmicheskii Apparat Universalnogo Ryada-1*, (Космический Аппарат Универсального Ряда), which can be translated as Spacecraft Bus from the Standardized Line-1). This bus consists of a central pressurized cylinder for the payload, an outer cylinder mounting solar cells (the Solar Battery), and a gravity gradient boom for passive stabilization. The cause of this event, occurring almost 42 years after launch, is unknown.

REFERENCE DOCUMENT

“Old Spacecraft Suffers Minor Fragmentation”, The Orbital Debris Quarterly News, NASA JSC, October 2009.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv13i4.pdf>.



Four cataloged fragments three weeks after the event.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 16.50 Jan 1968
DRY MASS (KG): 5500
MAIN BODY: Sphere-cylinder; 2.4 m diameter by 6.0 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, 10 kg TNT explosive charge

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 24 Jan 1968 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 68024.25242706 | MEAN ANOMALY: | 305.4920 |
| RIGHT ASCENSION: | 247.4278 | MEAN MOTION: | 15.98596524 |
| INCLINATION: | 65.6289 | MEAN MOTION DOT/2: | 0.00196964 |
| ECCENTRICITY: | 0.0118074 | MEAN MOTION DOT DOT/6: | 0 |
| ARG. OF PERIGEE: | 55.7254 | BSTAR: | 0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

Spacecraft was destroyed after a malfunction prevented reentry and landing in the Soviet Union.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Saturn SIVB Stage
OWNER: US
LAUNCH DATE: 4.50 Apr 1968
DRY MASS (KG): 30,000
MAIN BODY: Cylinder; 6.6 m diameter by 30 m length (?)
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: On-board propellants

EVENT DATA

DATE: 13 Apr 1968
TIME: 1054 GMT
ALTITUDE: 330 km
LOCATION: 32N, 245E (asc)
ASSESSED CAUSE: Propulsion

PRE-EVENT ELEMENTS

EPOCH: 68103.56521409
RIGHT ASCENSION: 177.3270
INCLINATION: 32.5869
ECCENTRICITY: .0120930
ARG. OF PERIGEE: 208.3921
MEAN ANOMALY: 151.0074
MEAN MOTION: 15.97292993
MEAN MOTION DOT/2: .00302835
MEAN MOTION DOT DOT/6: .0
BSTAR: .0

DEBRIS CLOUD DATA

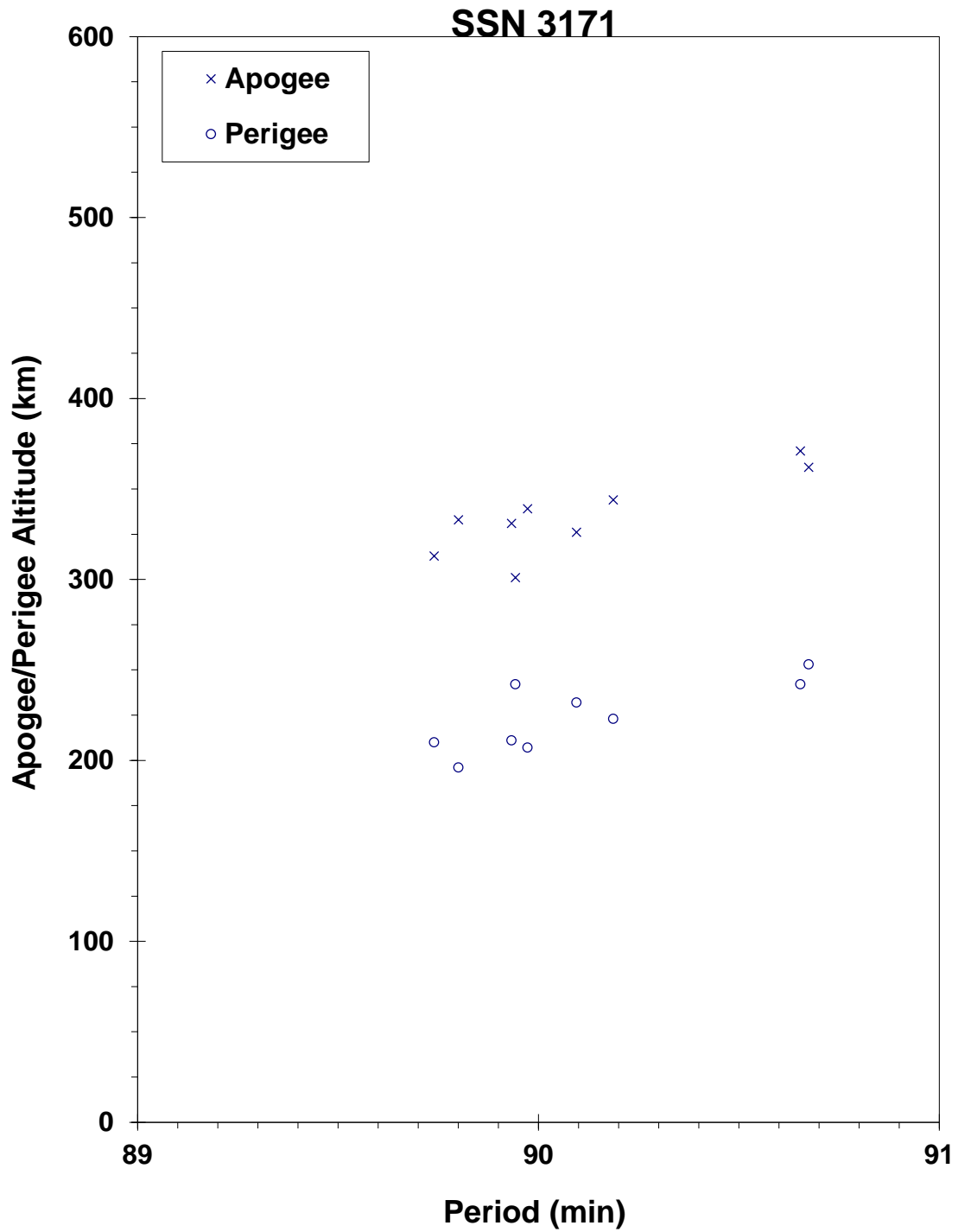
MAXIMUM ΔP : 0.7 min
MAXIMUM ΔI : 0.1 deg

COMMENTS

This Saturn SIVB Stage was fitted with an 11,800 kg mock Lunar Module (LM). The SIVB stage was programmed for a second firing to place the Apollo 6 vehicle into a more eccentric orbit, but the restart did not occur. The Apollo 6 payload was separated, leaving the SIVB stage and the LM in a low Earth orbit. Vaporization and venting of residual liquid oxygen induced a tumble to the SIVB stage that reached 30 rpm by 13 April. On this date the axial loads on the LM attach strap fittings and support struts were exceeded, resulting in separation of the LM from the SIVB along with numerous debris. Five fragments were cataloged without elements.

REFERENCE DOCUMENT

Apollo 6 Mission Anomaly Report No. 6, Unexpected Structural Indications During Launch Phase (Review Copy), MSC-PT-R-68-22, prepared by Apollo 6 Mission Evaluation Team, Marshall Space Flight Center, Alabama, and Manned Spacecraft Center, Texas, 1968.



Apollo 6 R/B debris cloud of 9 fragments 4 days after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Titan 3C Transtage
OWNER: US
LAUNCH DATE: 26.32 Sep 1968
DRY MASS (KG): 2500
MAIN BODY: Cylinder; 3 m diameter by 6 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------------|
| DATE: | 21 Feb 1992 | LOCATION: | Unknown (~ 197E) |
| TIME: | 0931 GMT | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | ~ 35600 | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 92043.23217642 | MEAN ANOMALY: | 284.5600 |
| RIGHT ASCENSION: | 21.8025 | MEAN MOTION: | 1.01459126 |
| INCLINATION: | 11.9035 | MEAN MOTION DOT/2: | .00000174 |
| ECCENTRICITY: | .0084771 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 76.2786 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

This was the second major fragmentation of a Titan 3C Transtage (the first was 1965-082DM). This transtage released ERS-28 (also known as OV5-2) in highly eccentric transfer orbit, then released LES-6 and ERS-21 (also known as OV5-4) in synchronous orbit, before slightly decelerating and releasing OV2-5 into a slightly lower orbit. This rocket body successfully completed its mission and remained on-orbit for 281 months before fragmenting. Mr. Bob Brock, operating the Maui GEODSS sensor, observed this transtage as it fragmented, liberating a reported 20 objects.

REFERENCE DOCUMENTS

TRW Space Log, Winter 1968-69 edition, Vol. 8, No. 4, H. T. Seaborn, ed., TRW Systems Group, Redondo Beach, pp. 32-35.

“Debris in Geosynchronous Orbits”, A.F. Pensa et al, Space Forum, special issue, 1st International Workshop on Space Debris, Moscow, October 1995.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 19.18 Oct 1968
DRY MASS (KG): 1400
MAIN BODY: Irregular; 1.8 m by 4.2 m
MAJOR APPENDAGES: Unknown
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants

EVENT DATA

DATE: 1 Nov 1968
TIME: 0412 GMT
ALTITUDE: 540 km
LOCATION: 55N, 104E (dsc)
ASSESSED CAUSE: Deliberate

PRE-EVENT ELEMENTS

EPOCH: 68304.83833772
RIGHT ASCENSION: 82.2502
INCLINATION: 62.2495
ECCENTRICITY: .0050333
ARG. OF PERIGEE: 298.4670
MEAN ANOMALY: 61.1261
MEAN MOTION: 15.19330723
MEAN MOTION DOT/2: .00016932
MEAN MOTION DOT DOT/6: .0
BSTAR: .0

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : 0.1 deg

COMMENTS

Cosmos 248 was the target of rendezvous for the Cosmos 249 and Cosmos 252 tests. Calculations suggest the few fragments detected from Cosmos 248 were released within 10 minutes of the Cosmos 252 event that took place in the vicinity of Cosmos 248. The four observed fragments were not cataloged until 4-6 weeks after the event, preventing an accurate assessment of the event due to drag effects. It is possible that the Cosmos 248 event occurred immediately after the rendezvous and was a direct result of interaction with Cosmos 252 debris.

REFERENCE DOCUMENTS

"Artificial Satellite Break-Ups (Part 2): Soviet Anti-Satellite Program", N.L. Johnson, Journal of the British Interplanetary Society, August 1983, pp. 357-362.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 20.17 Oct 1968
DRY MASS (KG): 1400
MAIN BODY: Irregular; 1.8 m by 4.2 m
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

DATE: 20 Oct 1968
TIME: 1427 GMT
ALTITUDE: 1995 km
LOCATION: 57S, 181E (asc)
ASSESSED CAUSE: Deliberate

POST-EVENT ELEMENTS

EPOCH: 68294.85197372
RIGHT ASCENSION: 118.4255
INCLINATION: 62.3313
ECCENTRICITY: .1088260
ARG. OF PERIGEE: 76.6147
MEAN ANOMALY: 295.3555
MEAN MOTION: 12.83515528
MEAN MOTION DOT/2: .0
MEAN MOTION DOT DOT/6: .0
BSTAR: .0

DEBRIS CLOUD DATA

MAXIMUM ΔP : 3.9 min
MAXIMUM ΔI : 0.4 deg

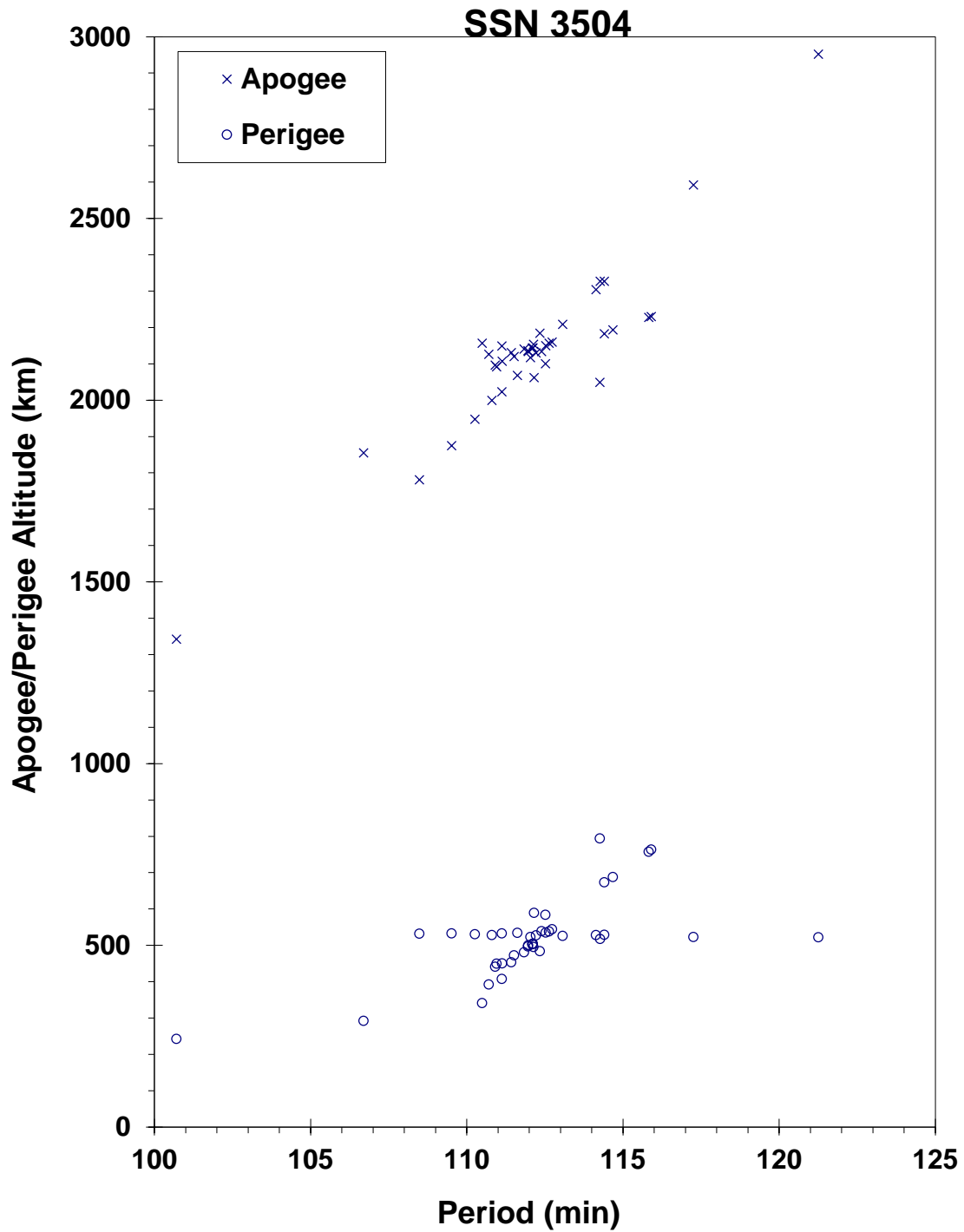
COMMENTS

Cosmos 249 was the first of a class of maneuverable spacecraft flown to rendezvous within 4 hours with another Cosmos satellite. In 9 of 20 such missions, orbital debris clouds were created by the active spacecraft, and in one case a passive (target) spacecraft also spawned a few fragments. Fragmentations occurred either in the vicinity of the passive satellite or a few hours after the rendezvous. In the case of Cosmos 249, the spacecraft was launched on a two-revolution rendezvous with Cosmos 248. After a close approach, Cosmos 249 continued on before its warhead was intentionally fired. The elements above are the first available for the final orbit. Some debris from Cosmos 249 and Cosmos 252 have been cross-tagged.

REFERENCE DOCUMENTS

"Artificial Satellite Break-Ups (Part 2): Soviet Anti-Satellite Program", N.L. Johnson, Journal of the British Interplanetary Society, August 1983, pp. 357-362.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 249 cataloged debris cloud of 43 fragments 4 months after the event as reconstructed from the US SSN database. Cross-tagging with Cosmos 252 debris is evident.

SATELLITE DATA

| | |
|-------------------|--|
| TYPE: | Payload |
| OWNER: | CIS |
| LAUNCH DATE: | 1.02 Nov 1968 |
| DRY MASS (KG): | 1400 |
| MAIN BODY: | Irregular; 1.8 m by 4.2 m |
| MAJOR APPENDAGES: | None |
| ATTITUDE CONTROL: | Active, 3-axis |
| ENERGY SOURCES: | On-board propellants, explosive charge |

EVENT DATA

| | | | |
|-----------|------------|-----------------|----------------|
| DATE: | 1 Nov 1968 | LOCATION: | 58N, 34E (asc) |
| TIME: | 0402 GMT | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | 535 km | | |

POST-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 68306.70122094 | MEAN ANOMALY: | 297.5777 |
| RIGHT ASCENSION: | 76.5565 | MEAN MOTION: | 12.81276799 |
| INCLINATION: | 62.3351 | MEAN MOTION DOT/2: | .00811969 |
| ECCENTRICITY: | .1040368 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 73.6953 | BSTAR: | .0 |

DEBRIS CLOUD DATA

| | |
|----------------------|---------|
| MAXIMUM ΔP : | 8.7 min |
| MAXIMUM ΔI : | 0.5 deg |

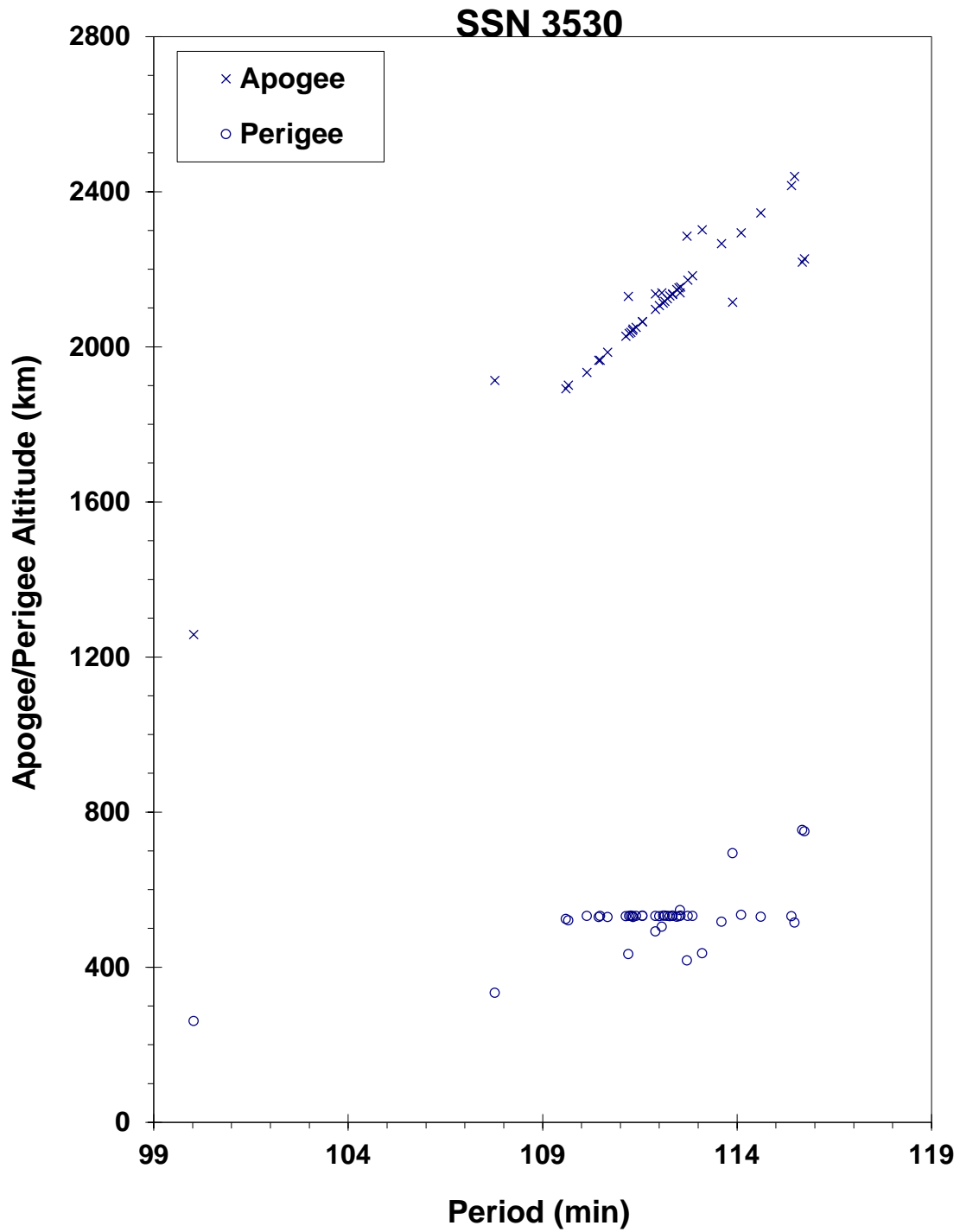
COMMENTS

Cosmos 252 was launched on a two-revolution rendezvous with Cosmos 248. The fragmentation occurred in the vicinity of Cosmos 248. Cosmos 252 was part of the test series begun with Cosmos 249. Elements above are for the orbit of the spacecraft after final maneuver, which took place immediately before fragmentation. Some debris from Cosmos 249 and Cosmos 252 have been cross-tagged.

REFERENCE DOCUMENTS

"Artificial Satellite Break-Ups (Part 2): Soviet Anti-Satellite Program", N.L. Johnson, Journal of the British Interplanetary Society, August 1983, pp. 357-362.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 252 cataloged debris cloud of 43 fragments 4 months after the event as reconstructed from the US SSN database. Cross-tagging with the Cosmos 249 cloud is evident.

SATELLITE DATA

TYPE: Delta 1 (model N) second stage Rocket Body
OWNER: US
LAUNCH DATE: 15.72 Dec 1968
DRY MASS (KG): 765
MAIN BODY: cylinder-nozzle; 1.4 m diameter x 4.8 m long right circular cylinder
MAJOR APPENDAGES: none
ATTITUDE CONTROL: none at time of event
ENERGY SOURCES: on-board propellants; pressure vessels; chemical batteries

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 15 Nov 1973 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 73318.68417963 | MEAN ANOMALY: | 337.0891 |
| RIGHT ASCENSION: | 26.4043 | MEAN MOTION: | 12.56868935 |
| INCLINATION: | 101.6374 | MEAN MOTION DOT/2: | 0.00000260 |
| ECCENTRICITY: | 0.0031347 | MEAN MOTION DOT DOT/6: | 7.694E-8 |
| ARG. OF PERIGEE: | 23.1544 | BSTAR: | 0.00000 |

DEBRIS CLOUD DATA

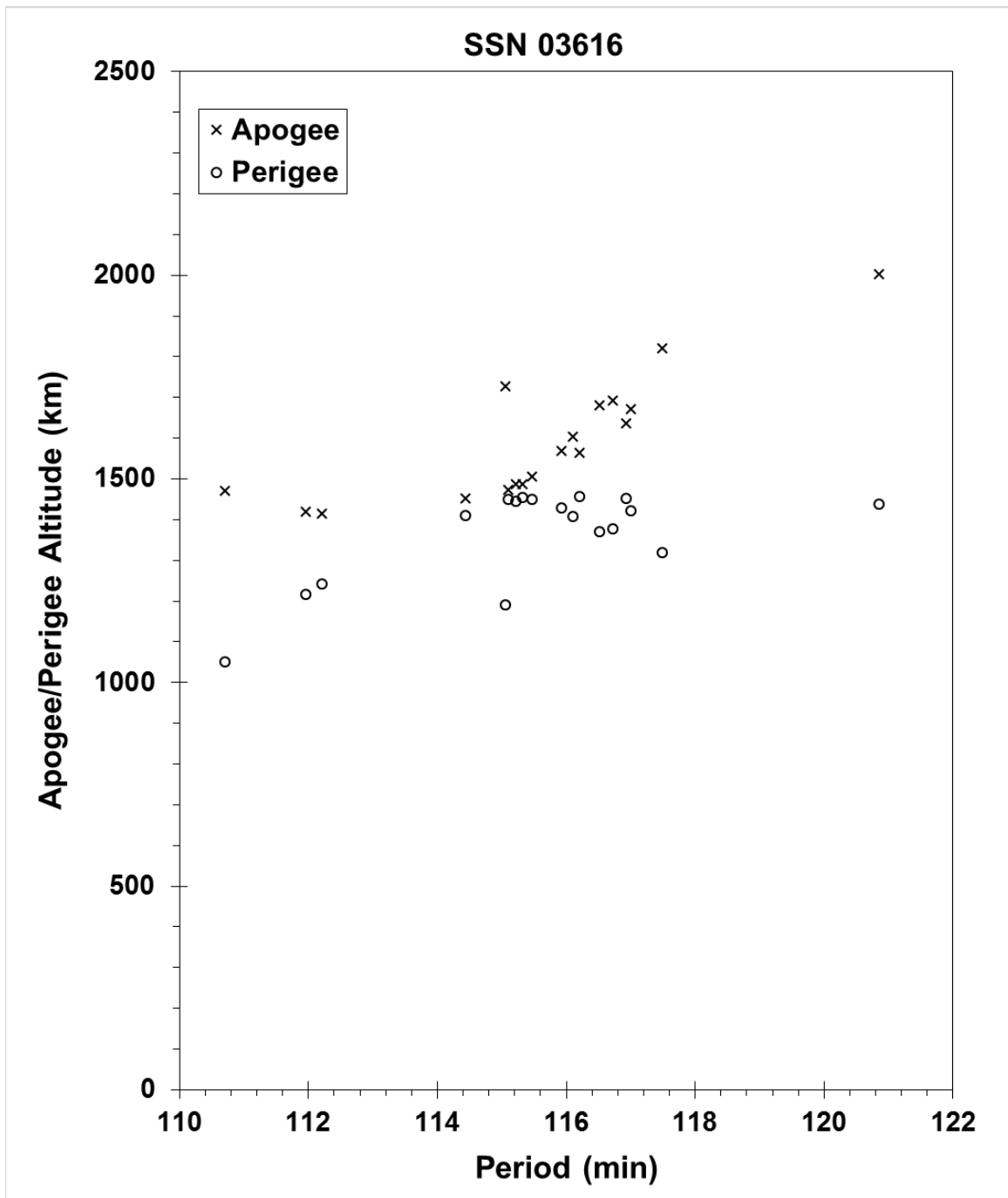
MAXIMUM ΔP : 5.7 min
MAXIMUM ΔI : 0.8 deg

COMMENTS

The cause of this event, occurring approximately 4.9 years after launch, is assessed to be propulsion-related. This event is recognized, as of this edition, as the first breakup of a Delta 1 rocket body, predating the previously acknowledged first Delta event, the NOAA 3 R/B [q.v.], by approximately 43 days.

REFERENCE DOCUMENT

“First Delta Rocket Body Fragmentation Discovered”, *The Orbital Debris Quarterly News*, NASA JSC, August 2017.
Available online at <https://www.orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/odqnv21i3.pdf>.



**ESSA 8 R/B debris cloud of 17 fragments as reconstructed from the US SSN database.
These debris entered the public catalog approximately 43.4 years after the event.**

SATELLITE DATA

TYPE: Transtage Rocket Body
OWNER: US
LAUNCH DATE: 9 Feb 1969
DRY MASS (KG): 1,874
MAIN BODY: 3.048 m diameter x ~5 m long right circular cylinder
MAJOR APPENDAGES: none
ATTITUDE CONTROL: inactive 3-axis
ENERGY SOURCES: on-board propellants; pressure vessels; chemical batteries

EVENT DATA

DATE: 28 Feb 2018 LOCATION: 5.26S, 129.97E (asc)
TIME: 2101 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 37247.3 km

PRE-EVENT ELEMENTS

EPOCH: 18059.72033524 MEAN ANOMALY: 119.8108
RIGHT ASCENSION: 300.9448 MEAN MOTION: 0.97608930
INCLINATION: 6.2739 MEAN MOTION DOT/2: -0.00000274
ECCENTRICITY: 0.0164003 MEAN MOTION DOT DOT/6: 0.00000
ARG. OF PERIGEE: 128.1168 BSTAR: 0.00000

DEBRIS CLOUD DATA

MAXIMUM ΔP : 134 min
MAXIMUM ΔI : 0.9 deg

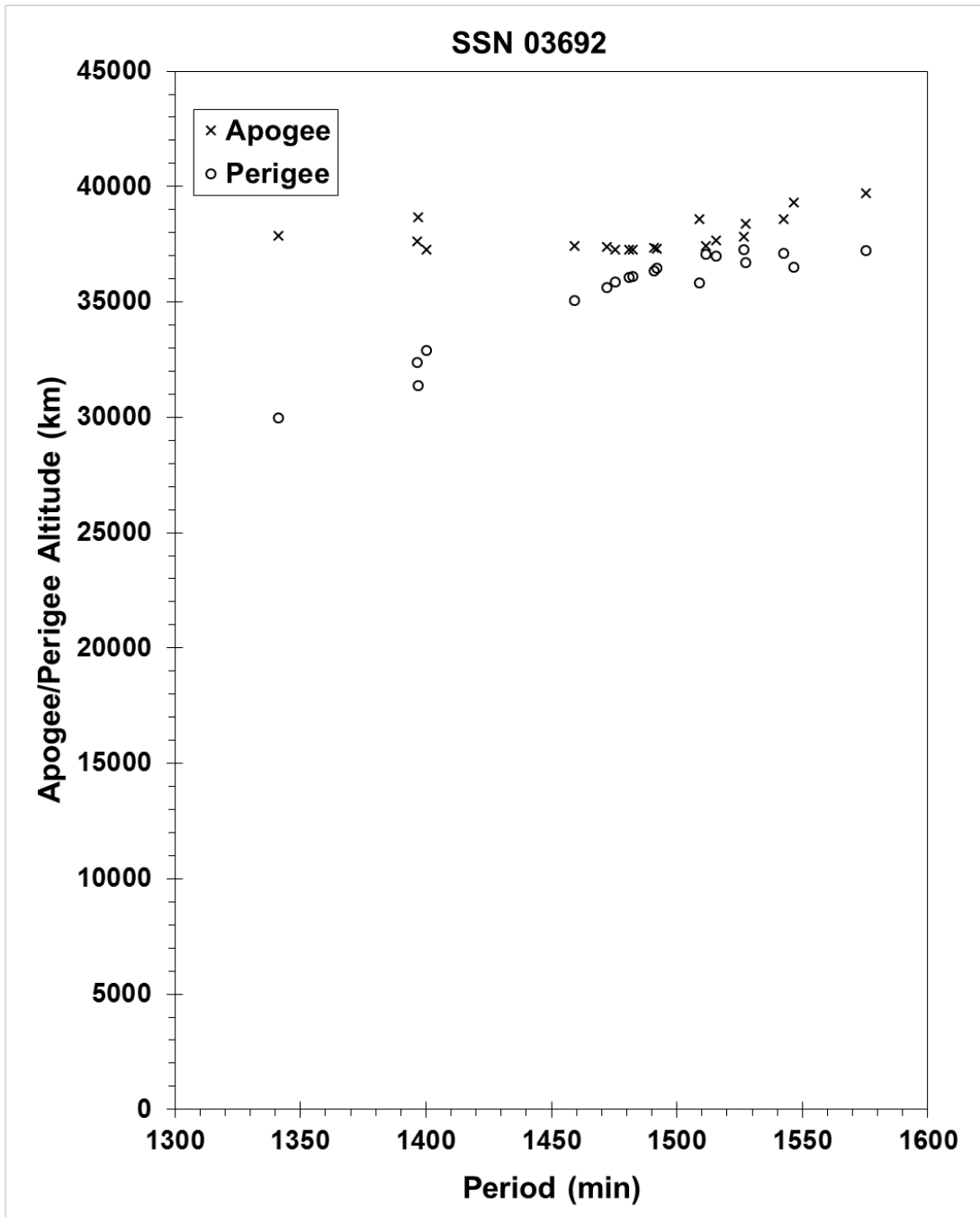
COMMENTS

The cause of this event, occurring approximately 49 years after launch, is likely due to on-board stored energy sources. This is the fourth breakup of a Titan Transtage rocket body, and the second Transtage fragmentation near geosynchronous orbit. Due to difficulties in cataloging highly elliptical and deep-space orbit many more fragments may be resident on-orbit than those cataloged.

REFERENCE DOCUMENTS

“Fragmentation of Titan Transtage in GEO”, *The Orbital Debris Quarterly News*, NASA JSC, May 2018.
Available online at <https://www.orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/odqnv22i2.pdf>.

Cowardin, H., P. Anz-Meador, and J. Reyes, “Characterizing GEO Titan IIIC Transtage Fragmentations using Ground-based and Telescopic Measurements”. In *Proceedings of the AMOS 2017 Technical Conference*. Available online at <https://amostech.com/TechnicalPapers/2017/Orbital-Debris/Cowardin.pdf>



Transtage R/B debris cloud of parent body and 18 fragments 4 months after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Vostok Second Stage
OWNER: CIS
LAUNCH DATE: 26.52 Mar 1969
DRY MASS (KG): 1440
MAIN BODY: Cylinder; 2.6 m diameter by 3.8 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: Unknown

EVENT DATA

DATE: 28 Mar 1969
TIME: 1845 GMT
ALTITUDE: 555 km
LOCATION: 59N, 91E (dsc)
ASSESSED CAUSE: Unknown

PRE-EVENT ELEMENTS

EPOCH: 69087.21308063
RIGHT ASCENSION: 33.3926
INCLINATION: 81.1687
ECCENTRICITY: .0276787
ARG. OF PERIGEE: 184.7318
MEAN ANOMALY: 175.1148
MEAN MOTION: 14.71400174
MEAN MOTION DOT/2: .0
MEAN MOTION DOT DOT/6: .0
BSTAR: .0

DEBRIS CLOUD DATA

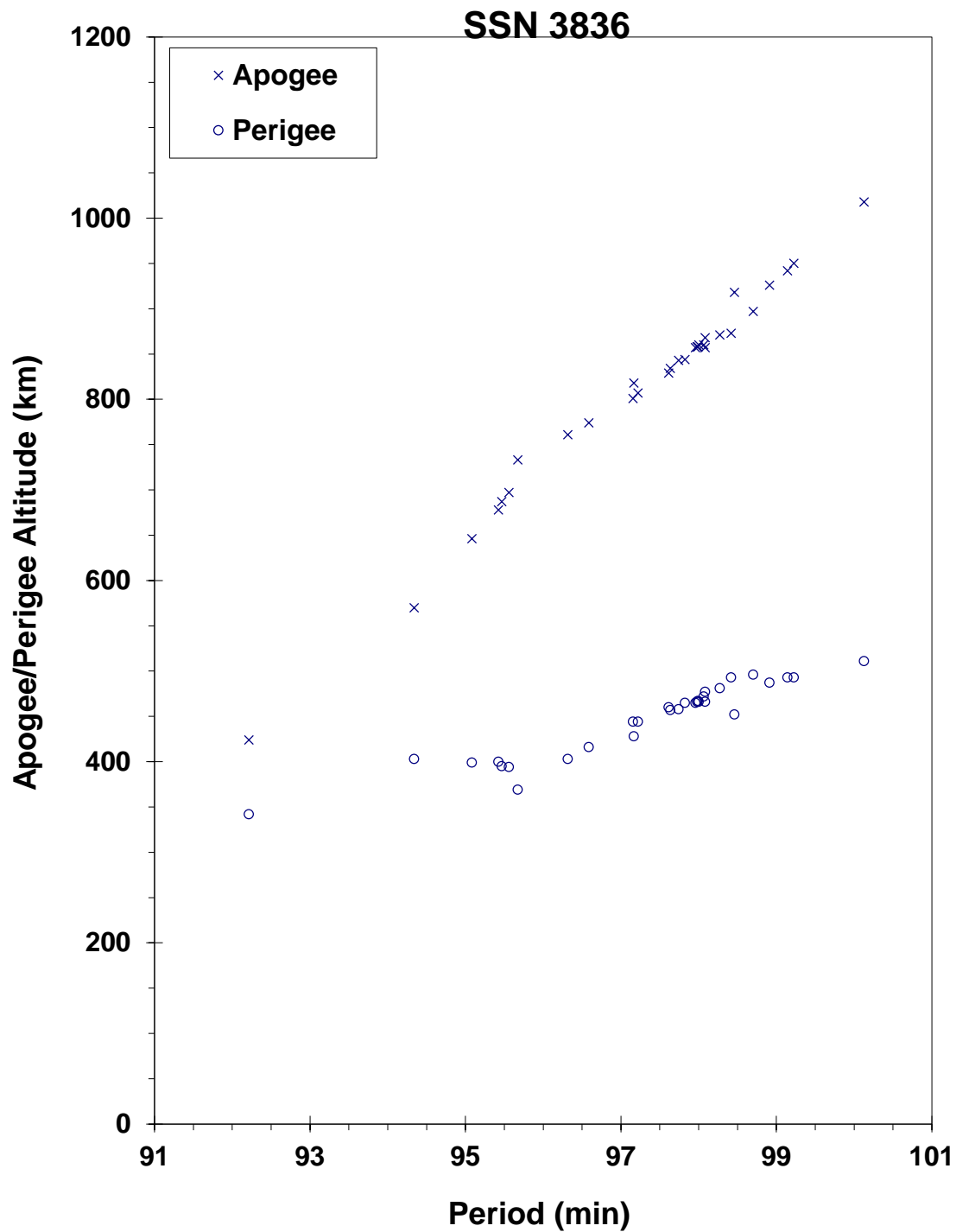
MAXIMUM ΔP : 2.4 min
MAXIMUM ΔI : 0.5 deg

COMMENTS

The vehicle successfully deployed the Meteor 1-1 payload into the desired orbit. An object believed to be the rocket body was found on 27 March in an orbit (1) of 565 km by 755 km, similar to earlier missions of the Vostok second stage. Early on 28 March an object was found in an orbit (2) of 460 km by 850 km with elements as indicated above. Analysis indicates that a transition from orbit (1) to orbit (2) was possible during the latter part of 27 March. Debris analysis clearly indicates that the orbit of the parent satellite had to be similar to orbit (2). Radar cross-section data supports the belief that the post-event object in the center of the debris cloud is the rocket body. No object was found in orbit (1) after the event.

REFERENCE DOCUMENT

History of the Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Meteor 1-1 R/B debris cloud of 31 fragments 2 months after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: TE 364-4 (STAR 37E motor)
OWNER: US
LAUNCH DATE: 26.09 Jul 1969
DRY MASS (KG): 1100 (70 without solid propellants)
MAIN BODY: Sphere-nozzle; 1.0 m by 1.8 m
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|-------------|-----------------|----------------|
| DATE: | 26 Jul 1969 | LOCATION: | 0N, 333E (dsc) |
| TIME: | 0228 GMT | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | 270 km | | |

POST-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 69208.17261261 | MEAN ANOMALY: | 166.4542 |
| RIGHT ASCENSION: | 130.0186 | MEAN MOTION: | 9.78100102 |
| INCLINATION: | 30.3692 | MEAN MOTION DOT/2: | .00000270 |
| ECCENTRICITY: | .2800849 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 187.9970 | BSTAR: | .0 |

DEBRIS CLOUD DATA

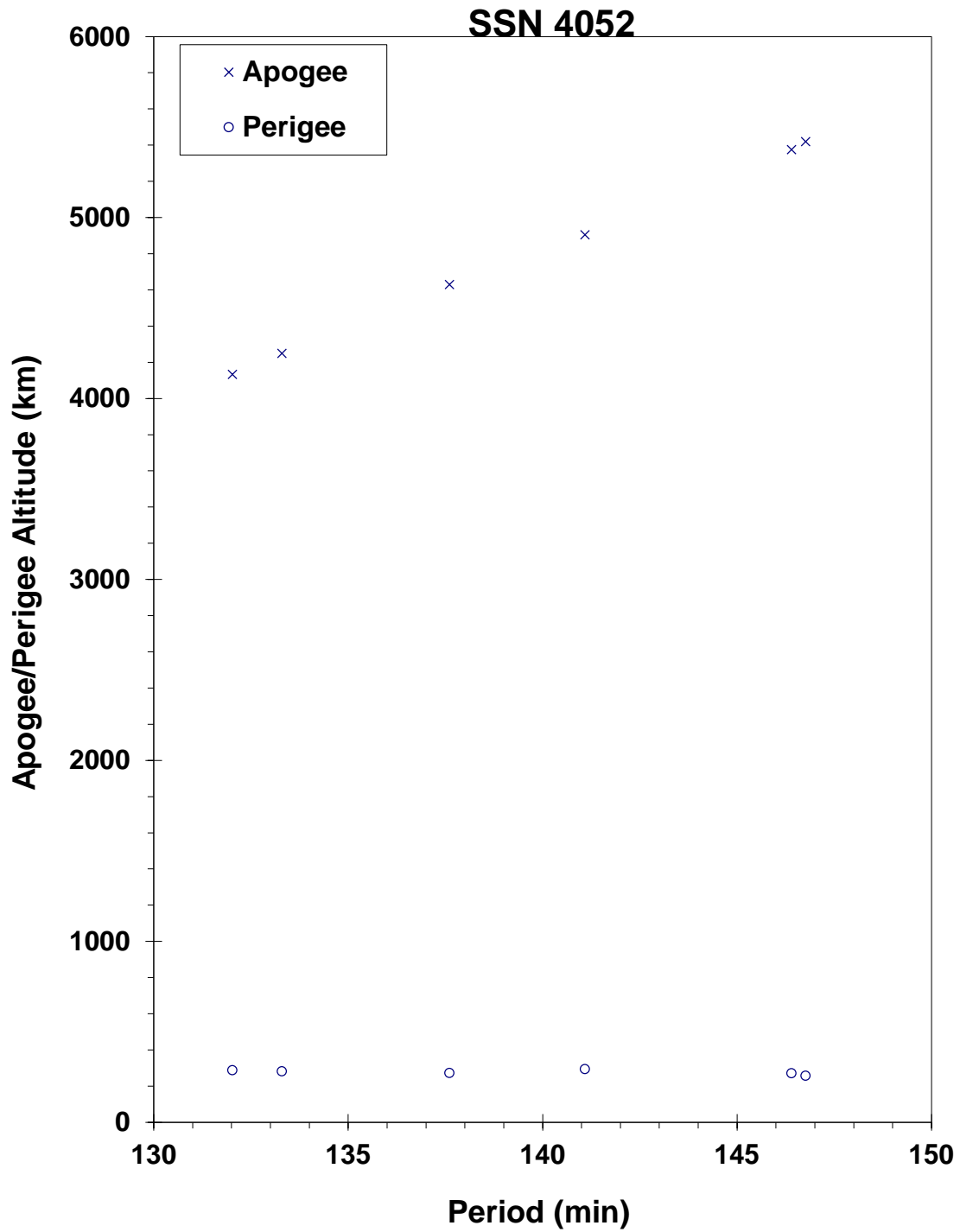
MAXIMUM ΔP : Unknown
MAXIMUM ΔI : 1.2 deg

COMMENTS

This solid-propellant upper stage failed soon after ignition, following a normal launch. The cause of the failure is assessed to be a possible rupture of the motor casing or nozzle. See similar failures of two PAM-D upper stages in 1984. Elements above are first developed for the rocket body about 1 day after the event. Rocket body may later have been cross-tagged with satellite 4053. Validity of debris identification and cataloging after 1969 is suspect.

REFERENCE DOCUMENT

TRW Space Log, Winter 1969-70 edition, Vol. 9, No. 4, W.A. Donop, ed., TRW Systems Group, Redondo Beach, pp. 34-36.



Intelsat 3 F-5 R/B debris cloud of 6 fragments 10 days after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Agena D Stage
OWNER: US
LAUNCH DATE: 30.57 Sep 1969
DRY MASS (KG): 600
MAIN BODY: Cylinder; 1.5 m diameter by 7.1 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: Unknown

EVENT DATA

DATE: 4 Oct 1969
TIME: 1553 GMT
ALTITUDE: 920 km
LOCATION: 54N, 178E (dsc)
ASSESSED CAUSE: Unknown

POST-EVENT ELEMENTS

EPOCH: 69295.54249482
RIGHT ASCENSION: 243.5157
INCLINATION: 69.9611
ECCENTRICITY: .0117819
ARG. OF PERIGEE: 87.4011
MEAN ANOMALY: 274.0514
MEAN MOTION: 13.68701087
MEAN MOTION DOT/2: .00000064
MEAN MOTION DOT DOT/6: .0
BSTAR: .0

DEBRIS CLOUD DATA

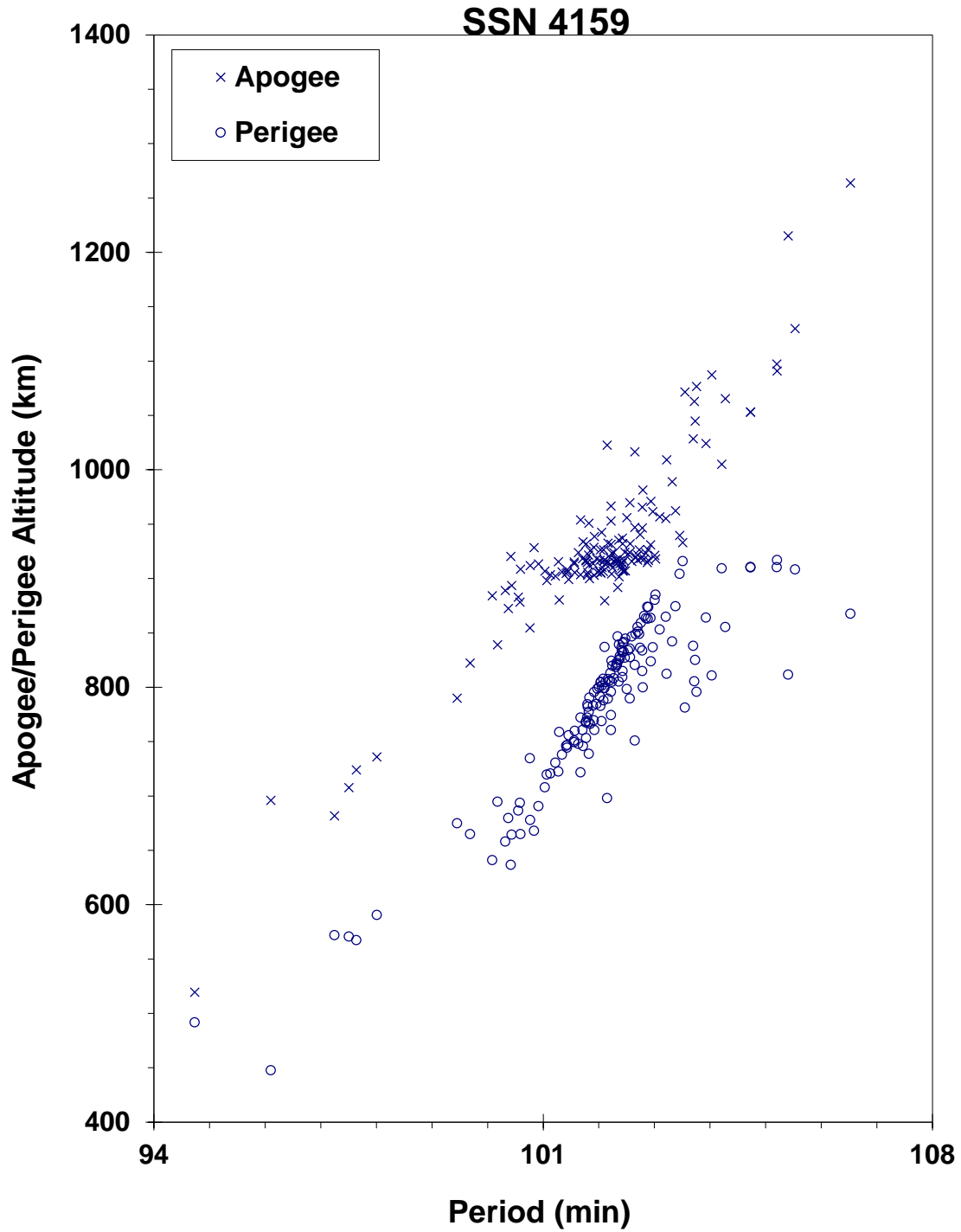
MAXIMUM ΔP : 3.1 min
MAXIMUM ΔI : 1.0 deg

COMMENTS

This was the first of two Agena D stages to fragment in a span of only 12 months. The vehicle delivered ten payloads to an orbit of about 905 km by 940 km. Four days later, before the rocket body had been cataloged, a large fragmentation occurred. What appeared to be the largest piece of the rocket body was found in the orbit described by the elements above almost 3 weeks after the event. See 1967-53 as a reference to an earlier mission of this type. Both missions were sponsored by DOD and public information is limited.

REFERENCE DOCUMENT

"Fragmentations of Asteroids and Artificial Satellites in Orbit", W. Wiesel, Icarus, Vol. 34, 1978, pp. 99-116.



**OPS 7613 R/B debris cloud (excluding 10 payloads) of 152 fragments 8 months after the event.
The largest fragment was found in an eccentric orbit with an orbital period of more than 105
min and is presumed to be the rocket body remnant.**

SATELLITE DATA

TYPE: Agena D Stage
OWNER: US
LAUNCH DATE: 8.35 Apr 1970
DRY MASS (KG): 600
MAIN BODY: Cylinder; 1.5 m diameter by 7.1 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: Unknown

EVENT DATA

| | | | |
|-----------|-------------|-----------------|-----------------|
| DATE: | 17 Oct 1970 | LOCATION: | 50S, 142E (asc) |
| TIME: | 0317 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 1075 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 70289.33183878 | MEAN ANOMALY: | 141.3434 |
| RIGHT ASCENSION: | 203.5235 | MEAN MOTION: | 13.49254887 |
| INCLINATION: | 99.8780 | MEAN MOTION DOT/2: | .0 |
| ECCENTRICITY: | .0016616 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 218.6463 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 14.2 min
MAXIMUM ΔI : 0.8 deg

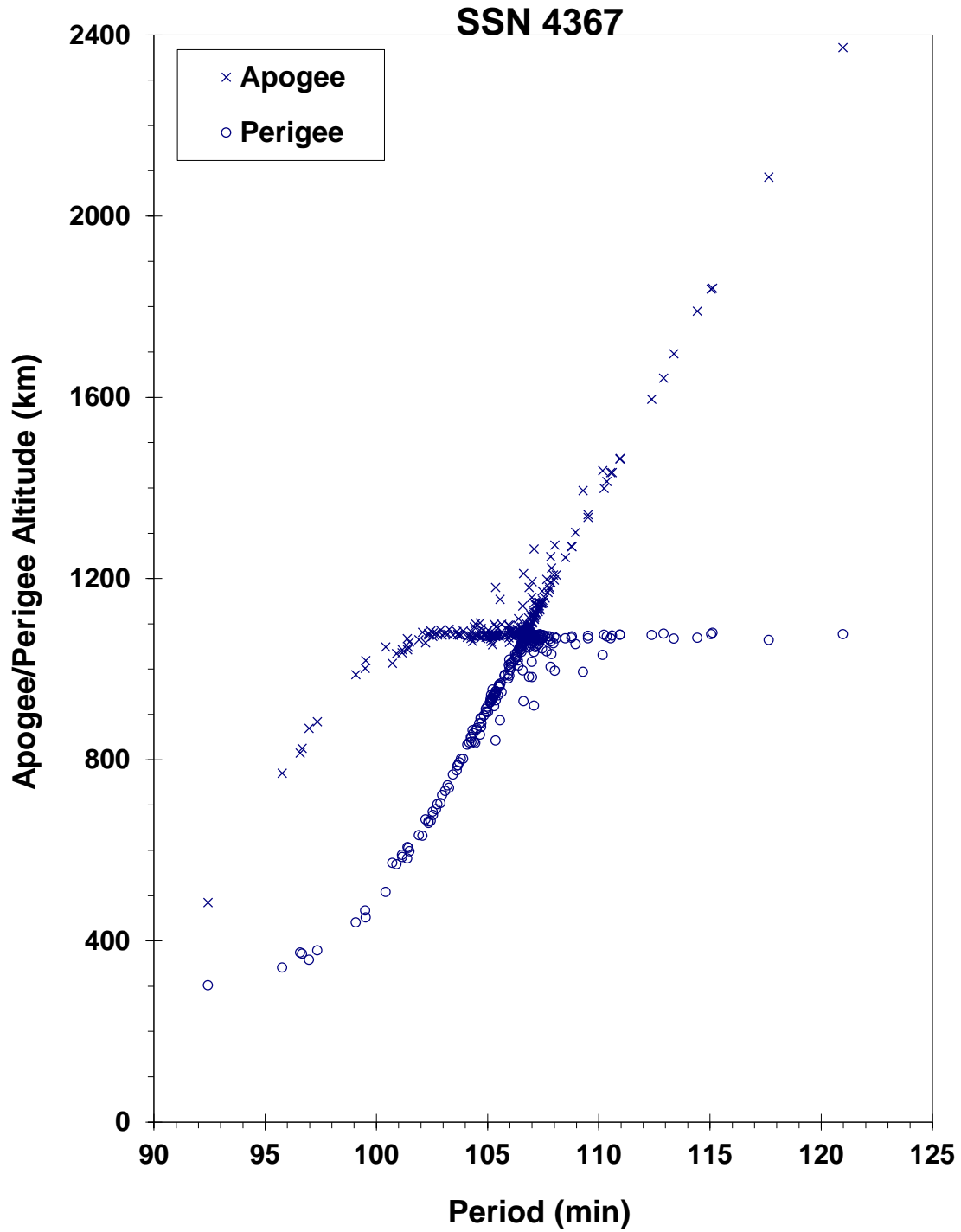
COMMENTS

This was the second Agena D stage to fragment in a span of only 12 months. The event occurred 6 months after the successful deployment of the Nimbus 4 payload. Twice in 1985, again in 1986, once in 1991, and twice in 1995 Nimbus 4 R/B debris spawned a few additional fragments, accounting for an additional 16 new debris objects between the 6 sub-events.

REFERENCE DOCUMENTS

"Fragmentations of Asteroids and Artificial Satellites in Orbit", W. Wiesel, *Icarus*, Vol. 34, 1978, pp. 99-116.

"Analysis of the Nimbus 4 Rocket Body Breakup and Subsequent Debris Anomalies", N.L. Johnson, Kaman Sciences Corporation, February 1992.



Nimbus 4 R/B debris cloud of 246 fragments 8 months after the event as reconstructed from the US SSN database. Some lower period fragments already exhibit the effects of natural decay.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 23.18 Oct 1970
DRY MASS (KG): 1400
MAIN BODY: Irregular; 1.8 m by 4.2 m
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

DATE: 23 Oct 1970
TIME: 1513 GMT
ALTITUDE: 1195 km
LOCATION: 22S, 217E (asc)
ASSESSED CAUSE: Deliberate

PRE-EVENT ELEMENTS

EPOCH: 70296.40542099
RIGHT ASCENSION: 129.1049
INCLINATION: 62.9380
ECCENTRICITY: .1039489
ARG. OF PERIGEE: 60.4933
MEAN ANOMALY: 309.5623
MEAN MOTION: 12.82808179
MEAN MOTION DOT/2: .00019973
MEAN MOTION DOT DOT/6: .0
BSTAR: .0

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

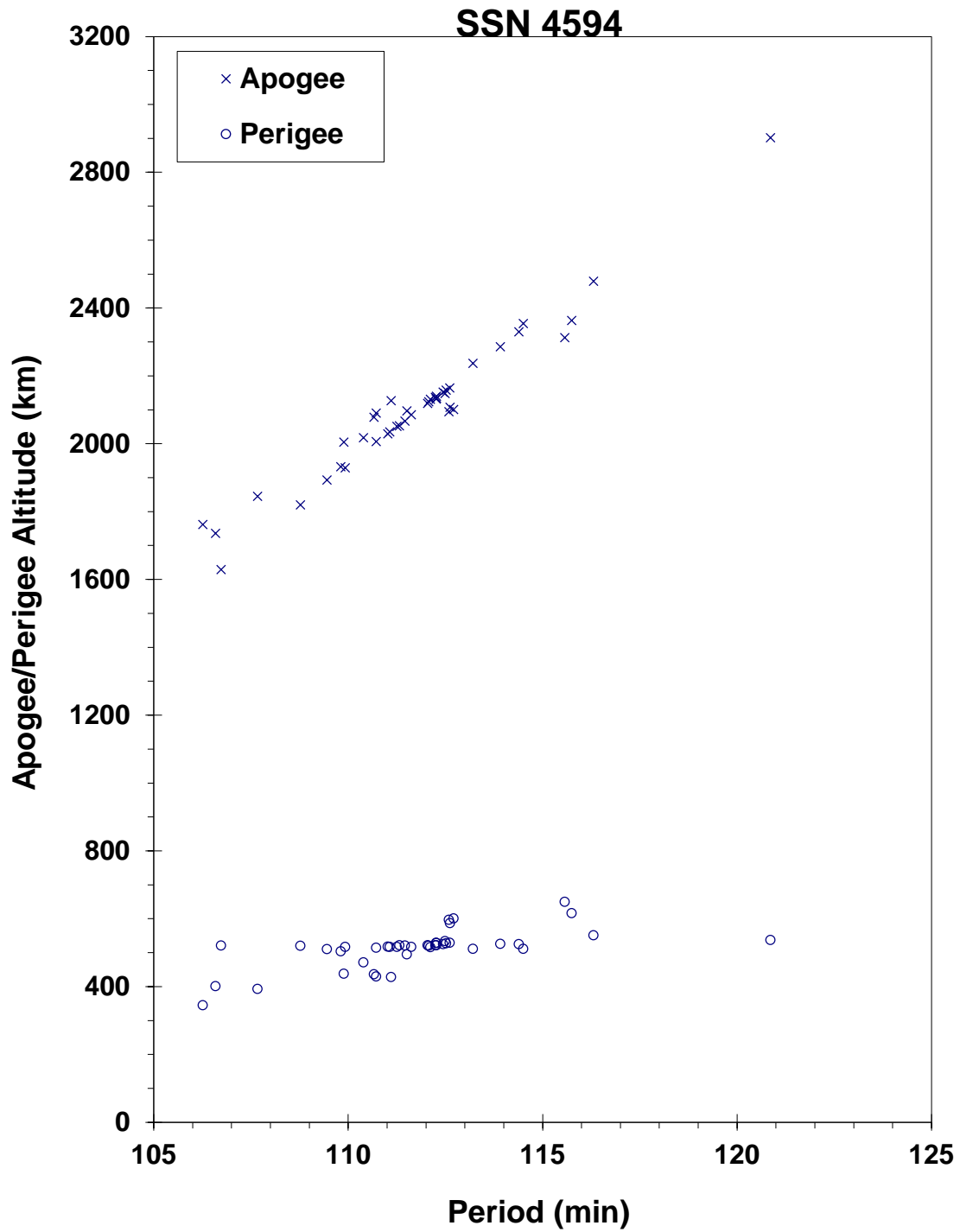
COMMENTS

Cosmos 374 was launched on a two-revolution rendezvous with Cosmos 373. After a close approach, Cosmos 374 continued on before its warhead was intentionally fired. Cosmos 374 was part of test series begun with Cosmos 249. Considerable cross-cataloging of Cosmos 374 and Cosmos 375 debris occurred; therefore, ΔP and ΔI are not calculated.

REFERENCE DOCUMENTS

"Artificial Satellite Break-Ups (Part 2): Soviet Anti-Satellite Program", N.L. Johnson, Journal of the British Interplanetary Society, August 1983, pp. 357-362.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 374 official debris cloud of 43 fragments 5 months after the event as reconstructed from the US SSN database. All fragments were cataloged after the Cosmos 375 fragmentation, and some contamination exists.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 30.09 Oct 1970
DRY MASS (KG): 1400
MAIN BODY: Irregular; 1.8 m by 4.2 m
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

DATE: 30 Oct 1970
TIME: 0600 GMT
ALTITUDE: 535 km
LOCATION: 54N, 23E (asc)
ASSESSED CAUSE: Deliberate

POST-EVENT ELEMENTS

EPOCH: 70306.81102869
RIGHT ASCENSION: 96.4080
INCLINATION: 62.8057
ECCENTRICITY: .1022289
ARG. OF PERIGEE: 56.0864
MEAN ANOMALY: 313.3102
MEAN MOTION: 12.87482205
MEAN MOTION DOT/2: .00009999
MEAN MOTION DOT DOT/6: .0
BSTAR: .0

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

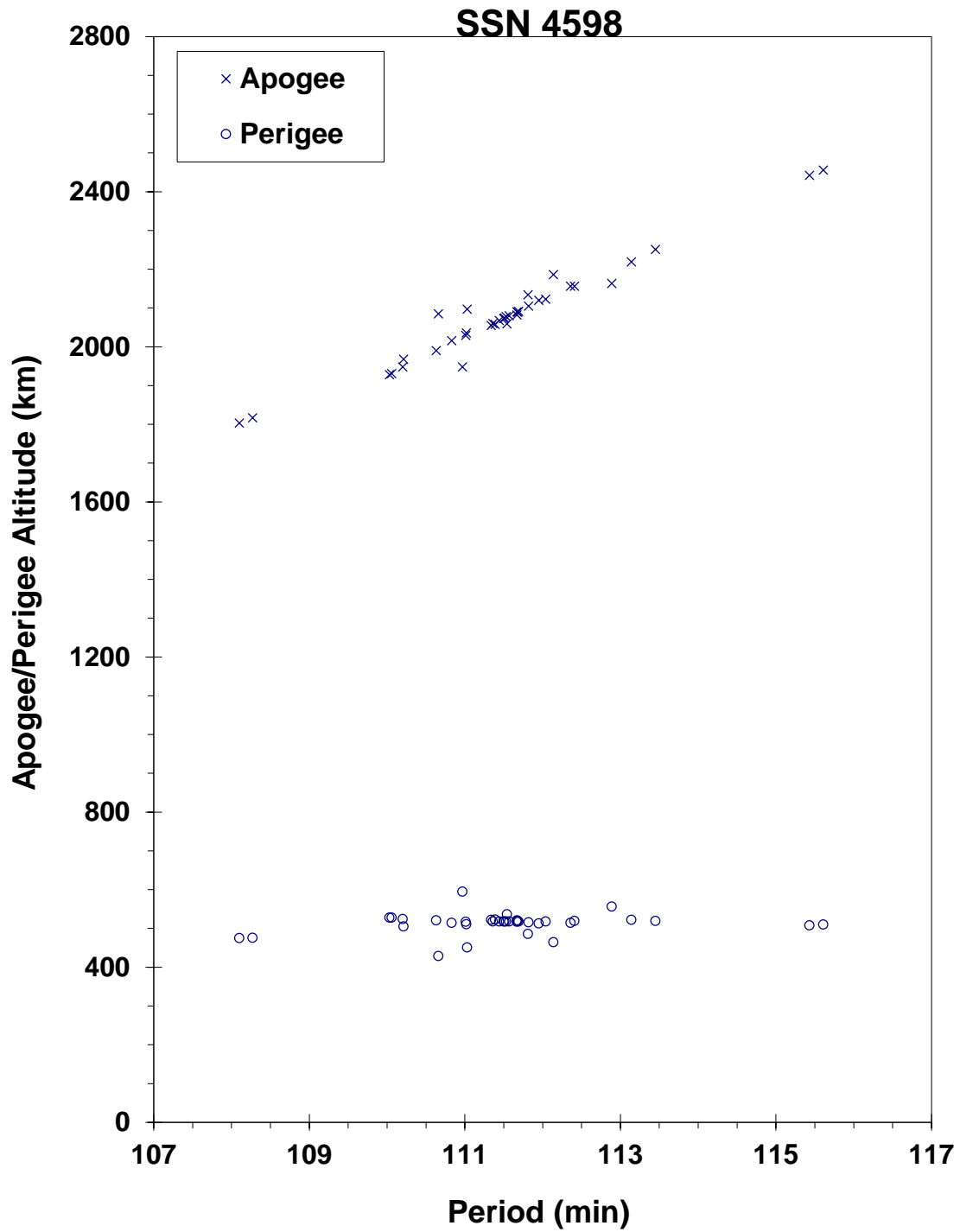
COMMENTS

Cosmos 375 was launched on a two-revolution rendezvous with Cosmos 373. The fragmentation occurred in the vicinity of Cosmos 373. Cosmos 375 was part of test series begun with Cosmos 249. Elements above are first reliable ones for orbit after final maneuver that took place immediately before fragmentation. Considerable cross-cataloging of Cosmos 374 and Cosmos 375 debris occurred; therefore, ΔP and ΔI are not calculated.

REFERENCE DOCUMENTS

"Artificial Satellite Break-Ups (Part 2): Soviet Anti-Satellite Program", N.L. Johnson, Journal of the British Interplanetary Society, August 1983, pp. 357-362.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 375 debris cloud of 38 fragments about 4 months after the event as reconstructed from the US SSN database. Some contamination exists with Cosmos 374 debris.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 25.47 Feb 1971
DRY MASS (KG): 1400
MAIN BODY: Irregular; 1.8 m by 4.2 m
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

DATE: 25 Feb 1971
TIME: 1431 GMT
ALTITUDE: 585 km
LOCATION: 54N, 21E (asc)
ASSESSED CAUSE: Deliberate

POST-EVENT ELEMENTS

EPOCH: 71057.77590281
RIGHT ASCENSION: 352.8670
INCLINATION: 65.7618
ECCENTRICITY: .1046189
ARG. OF PERIGEE: 50.3064
MEAN ANOMALY: 318.5528
MEAN MOTION: 12.68709606
MEAN MOTION DOT/2: .00013192
MEAN MOTION DOT DOT/6: .0
BSTAR: .0

DEBRIS CLOUD DATA

MAXIMUM ΔP : 2.8 min
MAXIMUM ΔI : 1.2 deg

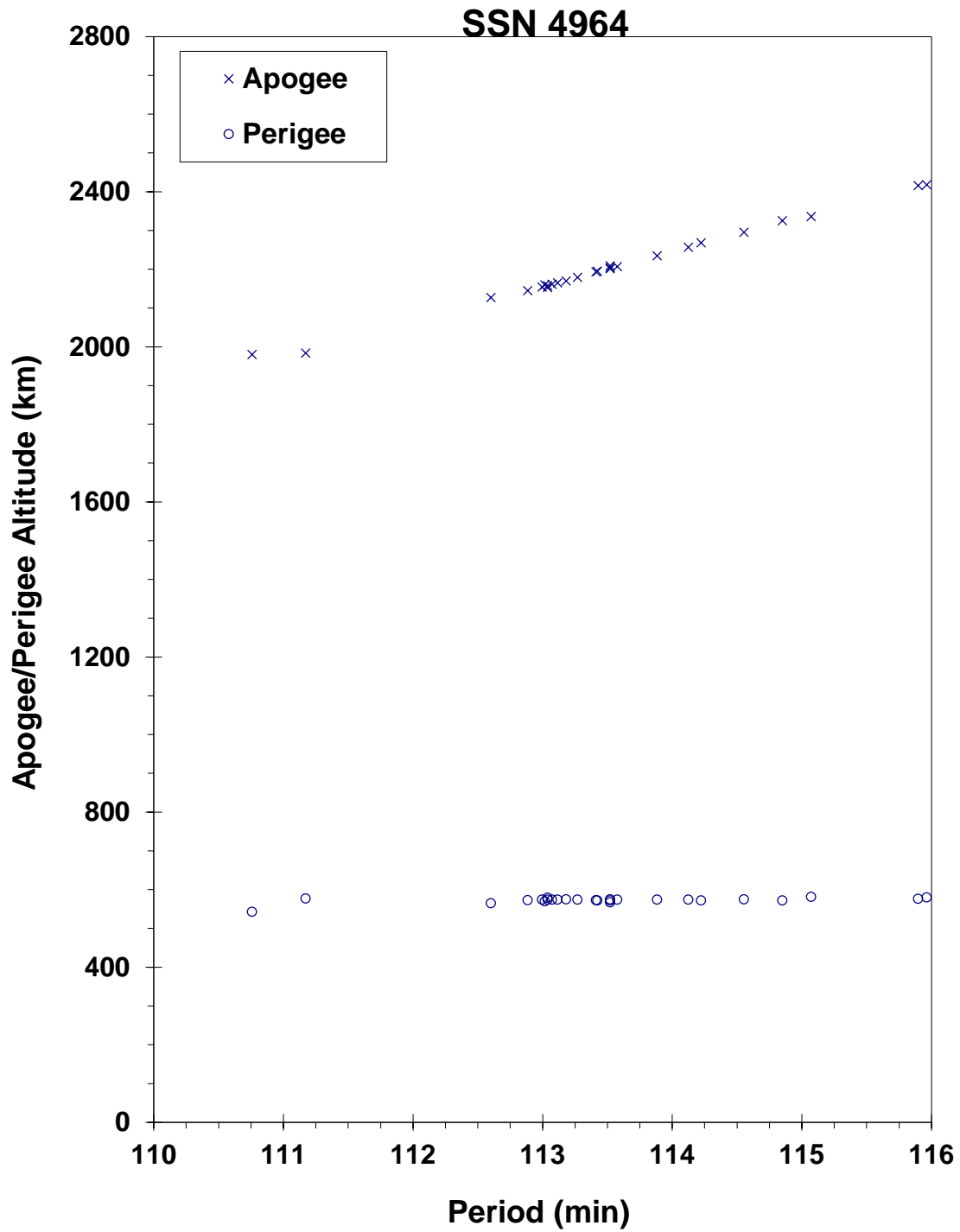
COMMENTS

Cosmos 397 was launched on a two-revolution rendezvous with Cosmos 394. The fragmentation occurred in the vicinity of Cosmos 394. Cosmos 397 was part of the test series begun with Cosmos 249. Elements above are first available for orbit after final maneuver that took place immediately before fragmentation.

REFERENCE DOCUMENTS

"Artificial Satellite Break-Ups (Part 2): Soviet Anti-Satellite Program", N.L. Johnson, Journal of the British Interplanetary Society, August 1983, pp. 357-362.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 397 cataloged debris cloud of 26 fragments about 7 weeks after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 3.55 Dec 1971
DRY MASS (KG): 1400
MAIN BODY: Irregular; 1.8 m by 4.2 m
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

DATE: 3 Dec 1971
TIME: 1651 GMT
ALTITUDE: 230 km
LOCATION: 51N, 7E (asc)
ASSESSED CAUSE: Deliberate

POST-EVENT ELEMENTS

EPOCH: 71339.01001769
RIGHT ASCENSION: 294.0999
INCLINATION: 65.7483
ECCENTRICITY: .1062360
ARG. OF PERIGEE: 53.3215
MEAN ANOMALY: 316.0762
MEAN MOTION: 13.65823046
MEAN MOTION DOT/2: .00001349
MEAN MOTION DOT DOT/6: .0
BSTAR: .0

DEBRIS CLOUD DATA

MAXIMUM ΔP : 3.6 min
MAXIMUM ΔI : 0.7 deg

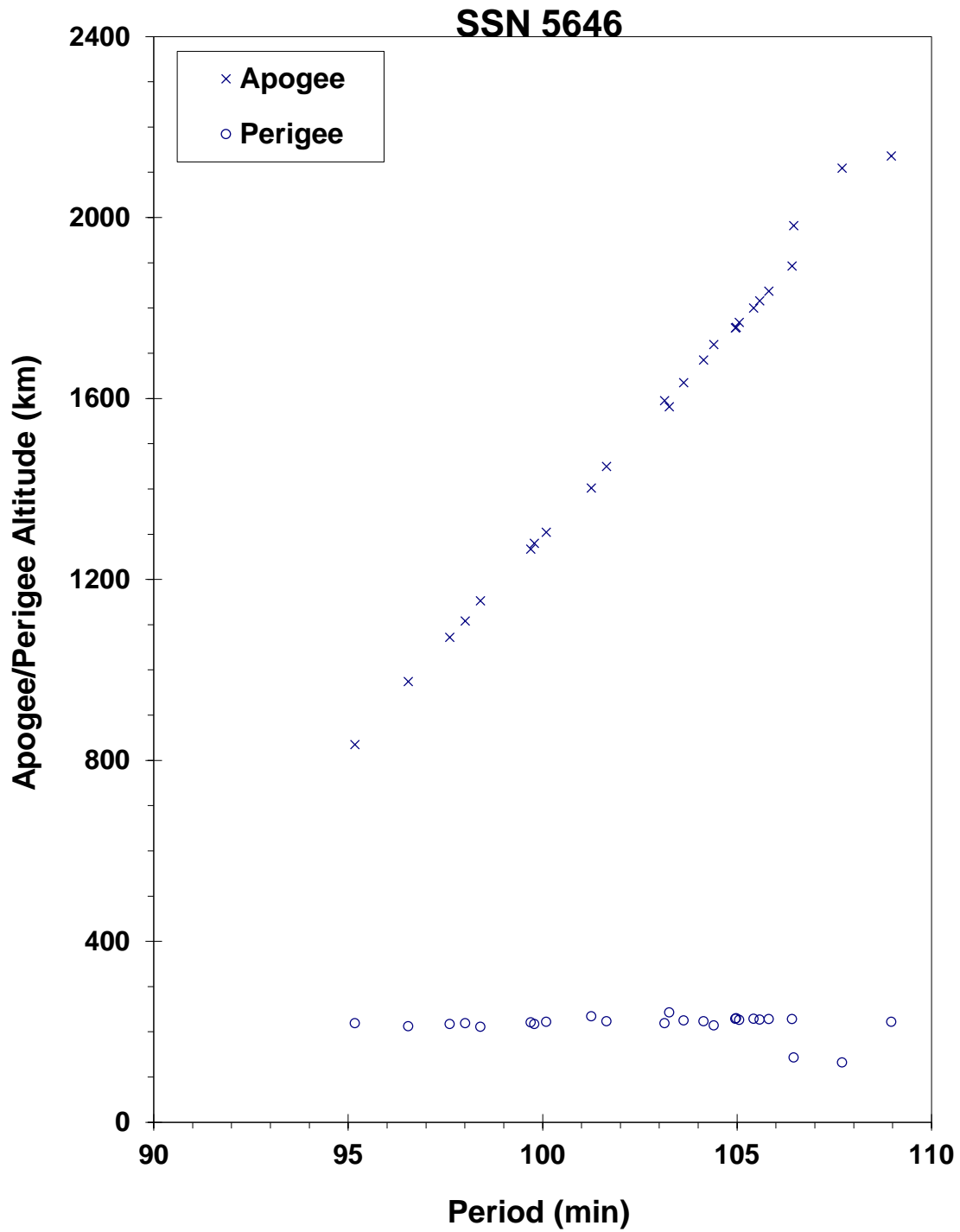
COMMENTS

Cosmos 462 was launched on a two-revolution rendezvous with Cosmos 459. The fragmentation occurred in the vicinity of Cosmos 459. Cosmos 462 was part of test series begun with Cosmos 249. Elements above are first available for orbit after final maneuver that took place immediately before fragmentation.

REFERENCE DOCUMENTS

"Artificial Satellite Break-Ups (Part 2): Soviet Anti-Satellite Program", N.L. Johnson, Journal of the British Interplanetary Society, August 1983, pp. 357-362.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 462 debris cloud of 25 cataloged fragments within 1 week of the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Delta Second Stage (900)
OWNER: US
LAUNCH DATE: 23.75 Jul 1972
DRY MASS (KG): 800
MAIN BODY: Cylinder-nozzle; 1.4 m diameter by 6.3 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: On-board propellants, range safety device

EVENT DATA

DATE: 22 May 1975 LOCATION: 34S, 46E (asc)
TIME: 1827 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 730 km

PRE-EVENT ELEMENTS

EPOCH: 75142.56642671 MEAN ANOMALY: 323.2981
RIGHT ASCENSION: 196.3353 MEAN MOTION: 14.36209995
INCLINATION: 98.3439 MEAN MOTION DOT/2: .00000060
ECCENTRICITY: .0193108 MEAN MOTION DOT DOT/6: .0
ARG. OF PERIGEE: 38.1650 BSTAR: .000027579

DEBRIS CLOUD DATA

MAXIMUM ΔP : 9.3 min
MAXIMUM ΔI : 1.0 deg

COMMENTS

This was the second Delta Second Stage to experience a severe fragmentation. The event occurred 34 months after the successful deployment of the Landsat 1 payload. Cause of the explosion is assessed to be related to the nearly 150 kg of residual propellants and characteristics of the sun-synchronous orbit.

REFERENCE DOCUMENTS

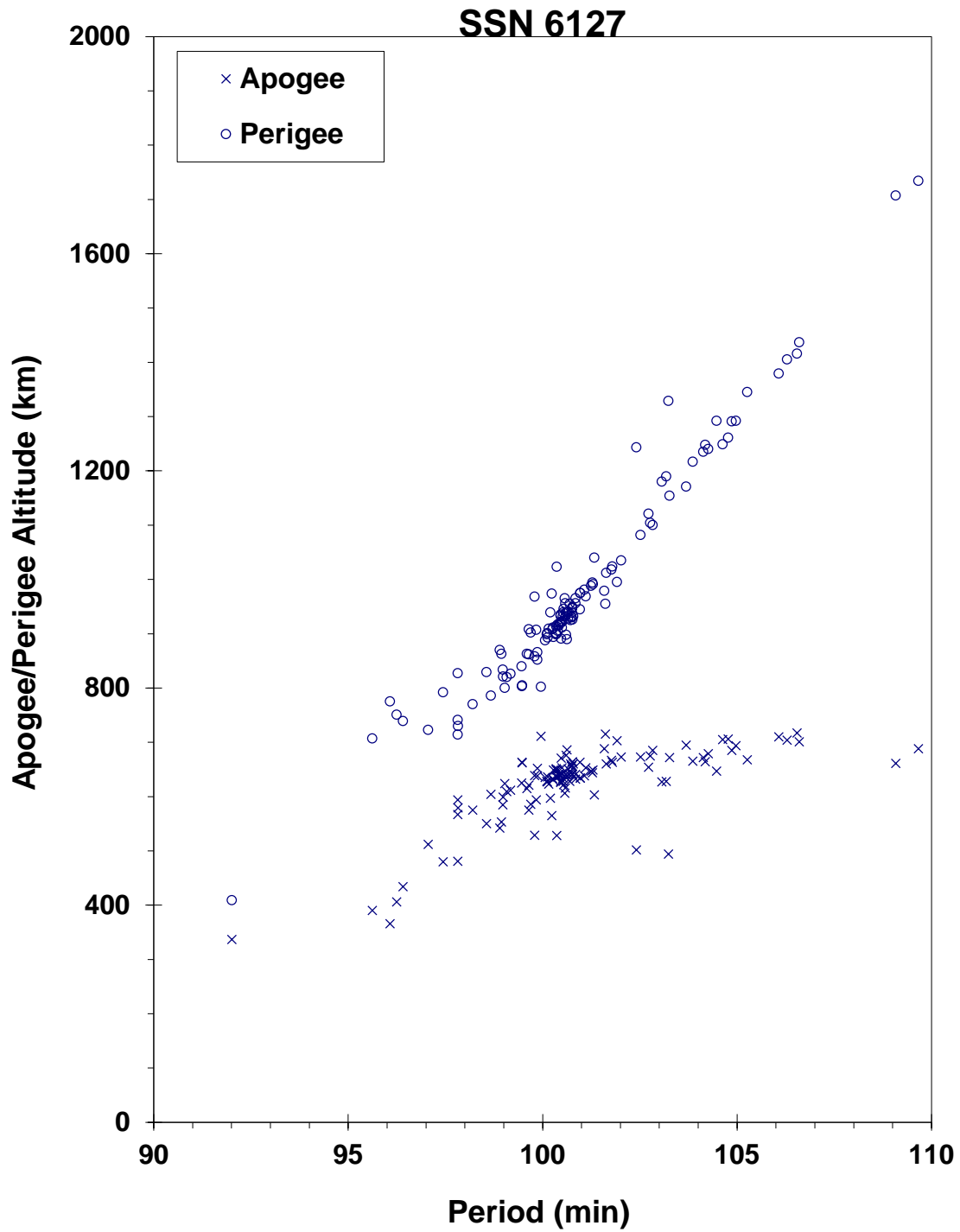
Dynamics of Satellite Disintegration, R. Dasenbrock, B. Kaufman, and W. Heard, NRL Report 7954, Naval Research Laboratory, Washington, 30 January 1976.

"Fragmentations of Asteroids and Artificial Satellites in Orbit", W. Wiesel, Icarus, Vol. 34, 1978, pp. 99-116.

Explosion of Satellite 10704 and other Delta Second Stage Rockets, J.R. Gabbard, Technical Memorandum 81-5, DCS Plans, Hdqtrs NORAD/ADCOM, Colorado Springs, May 1981.

Investigation of Delta Second Stage On-Orbit Explosions, C.S. Gumpel, Report MDC-H0047, McDonnell Douglas Astronautics Company - West, Huntington Beach, April 1982.

A Later Look at Delta Second Stage On-Orbit Explosions, J.R. Gabbard, Technical Report CS85-BMDSC-00-24, Teledyne Brown Engineering, Colorado Springs, March 1985.



Landsat 1 R/B debris cloud of 133 fragments 4 months after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Proton Third Stage
OWNER: CIS
LAUNCH DATE: 3.38 Apr 1973
DRY MASS (KG): 4000
MAIN BODY: Cylinder; 4.0 m diameter by 12.0 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: On-board propellants

EVENT DATA

DATE: 3 Apr 1973 LOCATION: 45N, 290E (dsc)
TIME: 2236 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 225 km

PRE-EVENT ELEMENTS

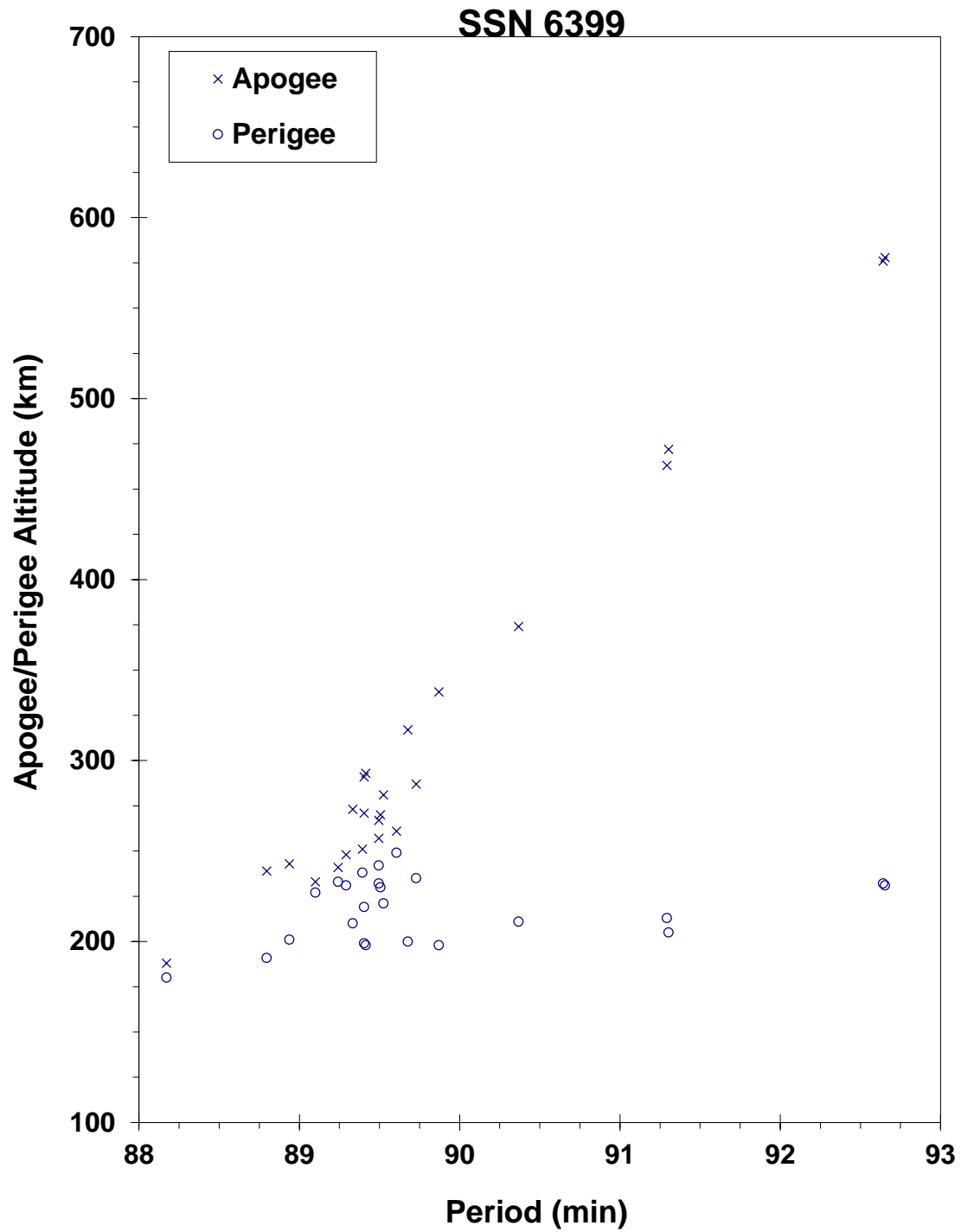
EPOCH: 73093.61404736 MEAN ANOMALY: 357.9254
RIGHT ASCENSION: 334.5652 MEAN MOTION: 16.20127597
INCLINATION: 51.4798 MEAN MOTION DOT/2: .00508885
ECCENTRICITY: .0037670 MEAN MOTION DOT DOT/6: .0
ARG. OF PERIGEE: 2.1878 BSTAR: .0

DEBRIS CLOUD DATA

MAXIMUM ΔP : 3.8 min
MAXIMUM ΔI : 0.5 deg

COMMENTS

This is the only known fragmentation of the Proton third stage. The event occurred less than 14 hours after reaching orbit. The event was apparently unrelated to the later payload malfunction. NAVSPASUR counted at least 95 objects shortly after the event, but most reentered before being officially cataloged. Information uncovered by Mr. Nicholas Johnson during an information exchange with Russian officials in the Spring of 1993 revealed that residual propellants resulted in an over-pressurization of the rocket body, causing this fragmentation. After this event, the Russians reported that the Proton third stage has been vented to avoid future events of this nature.



Salyut 2 R/B debris cloud of 25 fragments as reconstructed from the US SSN database. Most elements were developed within 2 days of the event.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 19.38 Apr 1973
DRY MASS (KG): 6300
MAIN BODY: Sphere-cylinder; 2.8 m diameter by 6.5 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

DATE: 6 May 1973
TIME: 0724 GMT
ALTITUDE: 310 km
LOCATION: 71S, 215E (asc)
ASSESSED CAUSE: Deliberate

PRE-EVENT ELEMENTS

EPOCH: 73125.63953480
RIGHT ASCENSION: 305.5573
INCLINATION: 72.8514
ECCENTRICITY: .0137599
ARG. OF PERIGEE: 22.9846
MEAN ANOMALY: 337.7411
MEAN MOTION: 16.05578988
MEAN MOTION DOT/2: .00433078
MEAN MOTION DOT DOT/6: .00010923
BSTAR: .0

DEBRIS CLOUD DATA

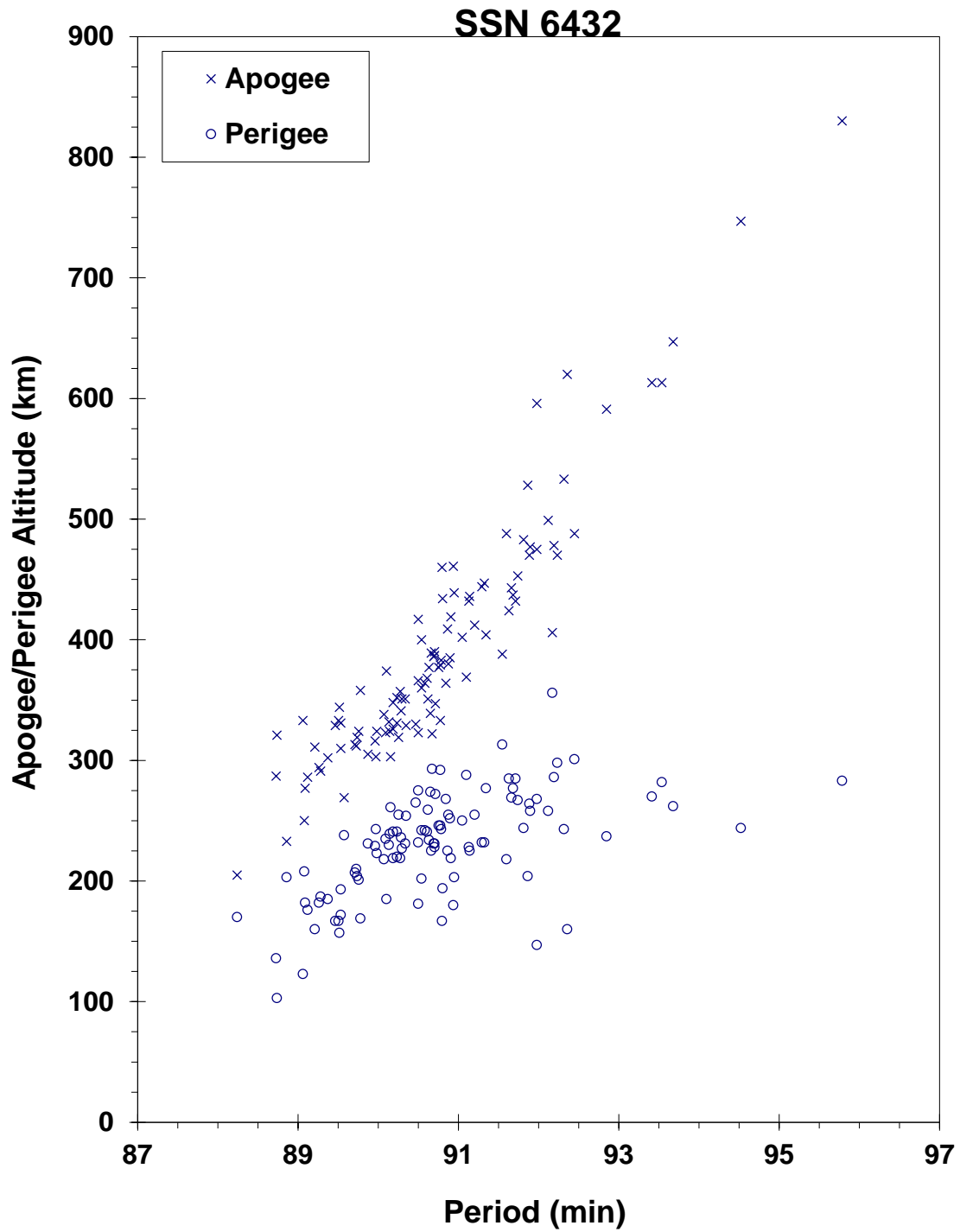
MAXIMUM ΔP : 6.0 min
MAXIMUM ΔI : 1.3 deg

COMMENTS

Spacecraft was destroyed after a malfunction prevented controlled reentry and landing in the Soviet Union. A total of 88 fragments were cataloged without elements.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 554 debris cloud of 107 fragments using initial elements as developed over several weeks. Some decay effects are present. Source is the US SSN database.

SATELLITE DATA

TYPE: Delta Second Stage (300)
OWNER: US
LAUNCH DATE: 6.71 Nov 1973
DRY MASS (KG): 840
MAIN BODY: Cylinder-nozzle; 1.4 m diameter by 6.3 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: On-board propellants, range safety device

EVENT DATA

DATE: 28 Dec 1973 LOCATION: 37S, 181E (asc)
TIME: 0904 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 1515 km

PRE-EVENT ELEMENTS

EPOCH: 73359.56303028 MEAN ANOMALY: 202.2816
RIGHT ASCENSION: 41.7242 MEAN MOTION: 12.40088347
INCLINATION: 102.0500 MEAN MOTION DOT/2: .00000577
ECCENTRICITY: .0005689 MEAN MOTION DOT DOT/6: .000000056523
ARG. OF PERIGEE: 157.8450 BSTAR: .0

DEBRIS CLOUD DATA

MAXIMUM ΔP : 10.4 min
MAXIMUM ΔI : 1.4 deg

COMMENTS

This was the first of seven Delta Second Stages to experience severe fragmentations between 1973 and 1981. Six of the seven stages were left in mid-morning, sun-synchronous orbits with residual propellants. Fragmentations occurred from 2-35 months after launch. The seventh stage exploded within hours of launch on a geosynchronous mission. The assessed cause in all cases is a propellant-induced explosion. Depletion burns to remove residual propellants were initiated in 1981, and no vented Delta Second Stages have fragmented since. In the case of the NOAA 3 R/B, fragmentation took place nearly 2 months after successful deployment of the NOAA 3 payload. Approximately 130 kg of propellants were left on board.

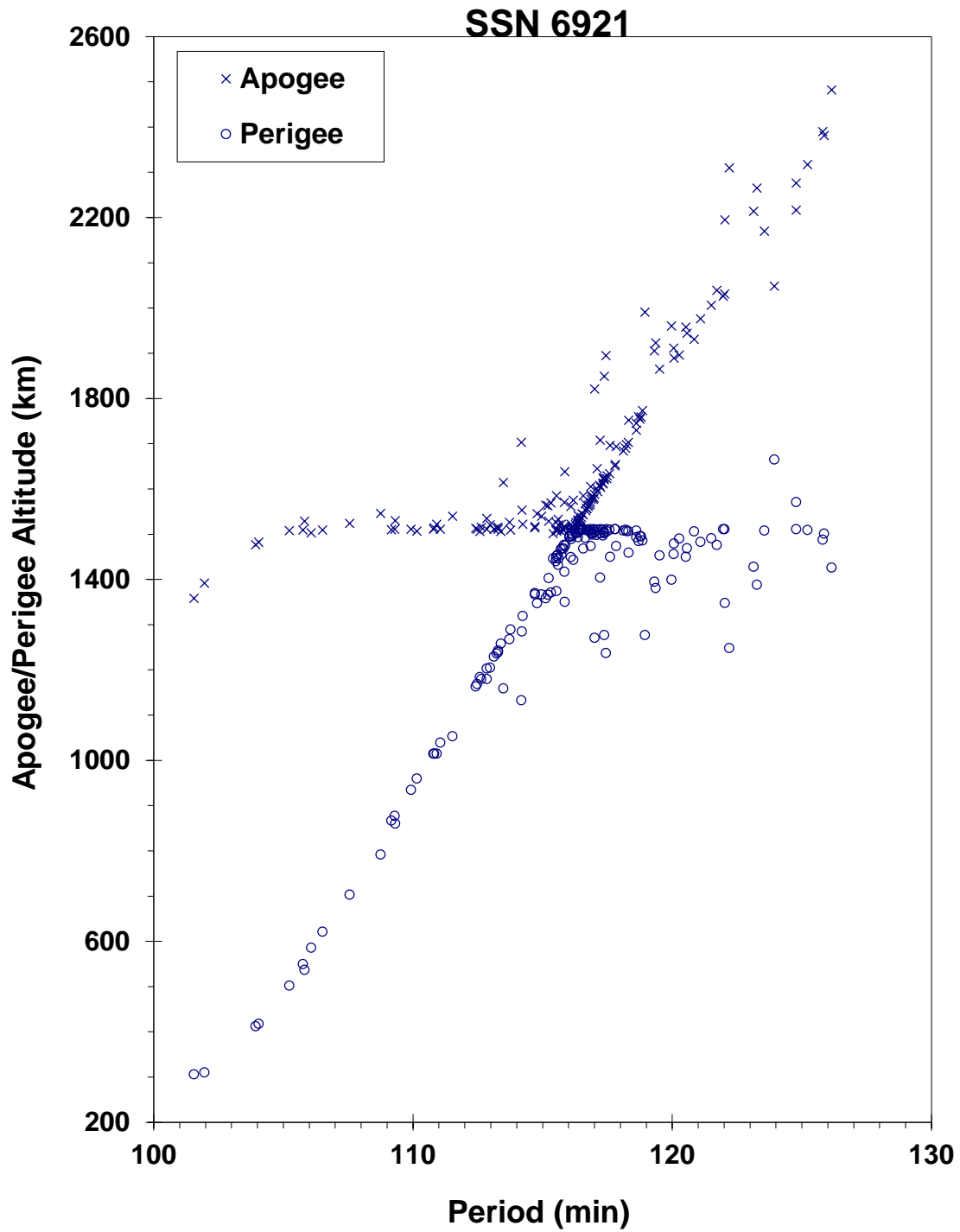
REFERENCE DOCUMENTS

Dynamics of Satellite Disintegration, R. Dasenbrock, B. Kaufman, and W. Heard, NRL Report 7954, Naval Research Laboratory, Washington, 30 January 1976.

Explosion of Satellite 10704 and other Delta Second Stage Rockets, J.R. Gabbard, Technical Memorandum 81-5, DCS Plans, Hdqtrs NORAD/ADCOM, Colorado Springs, May 1981.

Investigation of Delta Second Stage On-Orbit Explosions, C.S. Gumpel, Report MDC-H0047, McDonnell Douglas Astronautics Company - West, Huntington Beach, April 1982.

A Later Look at Delta Second Stage On-Orbit Explosions, J.R. Gabbard, Technical Report CS85-BMDSC-00-24, Teledyne Brown Engineering, Colorado Springs, March 1985.



NOAA 3 R/B debris cloud of 160 fragments 4 months after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Rocket Body
OWNER: US
LAUNCH DATE: 16.33 Mar 1974
DRY MASS (KG): 50
MAIN BODY: Box; 1 m long by 1 m wide by 1 m high
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of event
ENERGY SOURCES: None at time of event

EVENT DATA

DATE: 17 Jan 2005
TIME: 0214 GMT
ALTITUDE: 885 km
LOCATION: 80.6S, 306.8E
ASSESSED CAUSE: Accidental Collision

PRE-EVENT ELEMENTS

EPOCH: 05016.54972523
RIGHT ASCENSION: 350.2846
INCLINATION: 99.0928
ECCENTRICITY: .0066248
ARG. OF PERIGEE: 104.6813
MEAN ANOMALY: 256.1717
MEAN MOTION: 14.24162249
MEAN MOTION DOT/2: 0.00000028
MEAN MOTION DOT DOT/6: 0.0
BSTAR: 0.000031607

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

The THOR 2A upper stage collided with a piece of fragmentation debris (1999-57CV, SSN# 26207) from the March 2000 explosion of the third stage of a Chinese CZ-4 launch vehicle (1999-57C, SSN# 25942). This was the third historical collision of cataloged objects; the collision occurred over Antarctica as both object were near the southernmost point in their respective retrograde orbits. The relative velocity of the collision was just under 6 km/s. 1999-57CV is believed to be relatively small, with a radar cross-section of 600 cm². The collision produced less than 10 cataloged debris. Because the upper stage had been in orbit for over 30 years, it is believed there was no on-board propellant remaining at the time of the event.

REFERENCE DOCUMENT

“Accidental Collisions of Cataloged Satellites Identified”, The Orbital Debris Quarterly News, NASA JSC, April 2005.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv9i2.pdf>.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Delta Second Stage (2310)
OWNER: US
LAUNCH DATE: 15.72 Nov 1974
DRY MASS (KG): 840
MAIN BODY: Cylinder-nozzle; 1.4 m diameter by 5.8 m length
MAJOR APPENDAGES: Mini-skirt; 2.4m by 0.3 m
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: On-board propellants, range safety device

EVENT DATA

DATE: 20 Aug 1975 LOCATION: 52S, 278E (dsc)
TIME: 1307 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 1465 km

PRE-EVENT ELEMENTS

EPOCH: 75231.53619619 MEAN ANOMALY: 309.0001
RIGHT ASCENSION: 277.2201 MEAN MOTION: 12.52826370
INCLINATION: 101.6940 MEAN MOTION DOT/2: .00000083
ECCENTRICITY: .0009694 MEAN MOTION DOT DOT/6: .0
ARG. OF PERIGEE: 51.1891 BSTAR: .0

DEBRIS CLOUD DATA

MAXIMUM ΔP : 15.7 min
MAXIMUM ΔI : 1.8 deg

COMMENTS

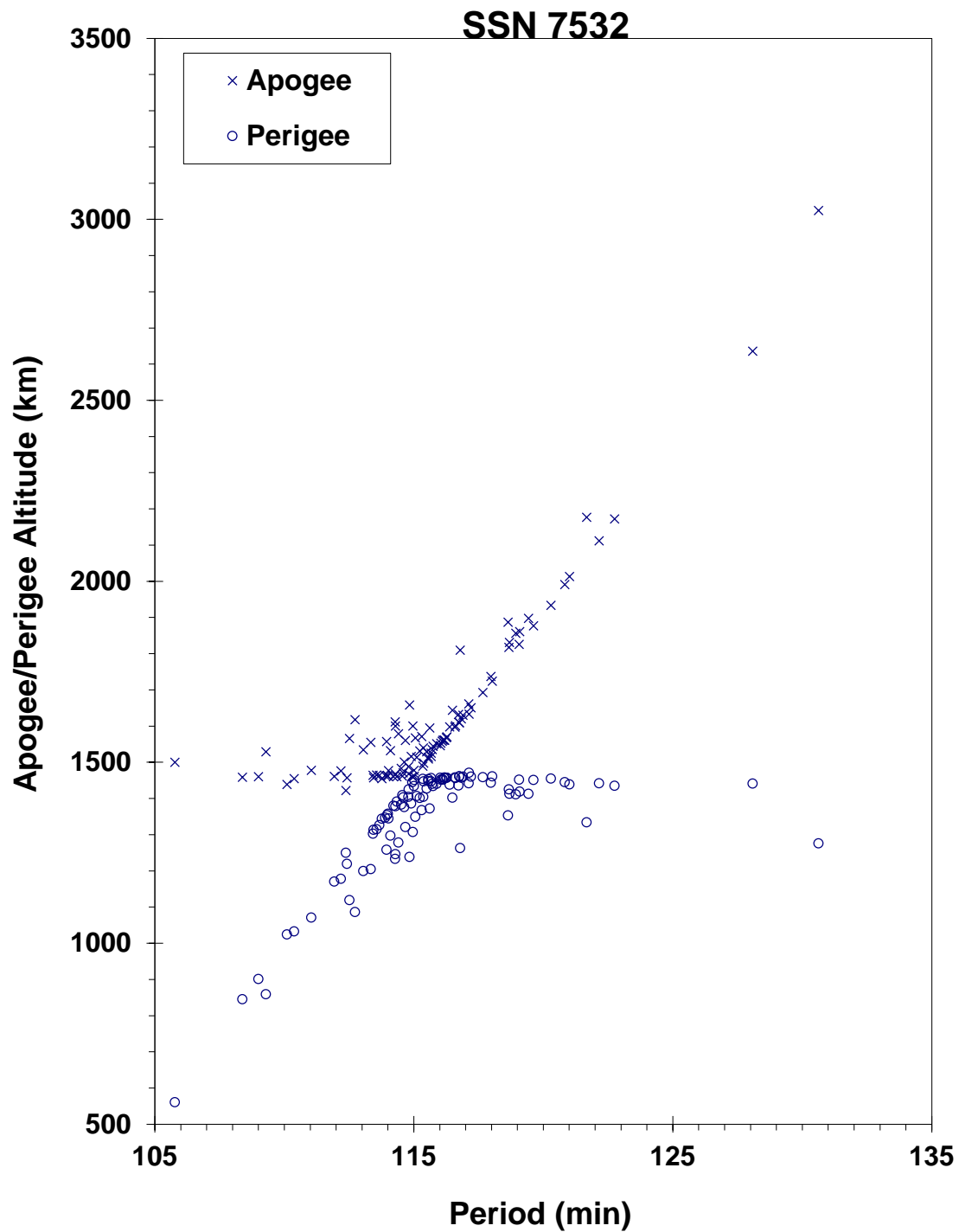
This was the third Delta Second Stage to experience a severe fragmentation. The event occurred 9 months after the successful deployment of the NOAA 4 payload. Cause of the explosion is assessed to be related to the estimated more than 200 kg of residual propellants and characteristics of the sun-synchronous orbit. A fragment from this event (satellite number 8138) may have generated six or more additional pieces in September 1981.

REFERENCE DOCUMENTS

Explosion of Satellite 10704 and other Delta Second Stage Rockets, J.R. Gabbard, Technical Memorandum 81-5, DCS Plans, Hdqtrs NORAD/ADCOM, Colorado Springs, May 1981.

Investigation of Delta Second Stage On-Orbit Explosions, C.S. Gumpel, Report MDC-H0047, McDonnell Douglas Astronautics Company - West, Huntington Beach, April 1982.

A Later Look at Delta Second Stage On-Orbit Explosions, J.R. Gabbard, Technical Report CS85-BMDSC-00-24, Teledyne Brown Engineering, Colorado Springs, March 1985.



NOAA 4 R/B debris cloud of 101 fragments 6 months after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 24.46 Dec 1974
DRY MASS (KG): 3000
MAIN BODY: Cylinder; 1.3 m diameter by 17 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge (?)

EVENT DATA (1)

| | | | |
|-----------|-------------|-----------------|-----------------|
| DATE: | 17 Apr 1975 | LOCATION: | 01N, 278E (dsc) |
| TIME: | 2148 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 440 km | | |

PRE-EVENT ELEMENTS (1)

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 75107.81173798 | MEAN ANOMALY: | 71.8460 |
| RIGHT ASCENSION: | 271.0743 | MEAN MOTION: | 15.44155646 |
| INCLINATION: | 65.0355 | MEAN MOTION DOT/2: | .00007106 |
| ECCENTRICITY: | .0014224 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 288.1084 | BSTAR: | .0 |

EVENT DATA (2)

| | | | |
|-----------|------------|-----------------|-----------------|
| DATE: | 2 Aug 1975 | LOCATION: | 02S, 258E (dsc) |
| TIME: | 1623 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 435 km | | |

PRE-EVENT ELEMENTS (2)

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 75214.45597981 | MEAN ANOMALY: | 68.4232 |
| RIGHT ASCENSION: | 274.3453 | MEAN MOTION: | 15.46205523 |
| INCLINATION: | 65.0458 | MEAN MOTION DOT/2: | .00001715 |
| ECCENTRICITY: | .0020980 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 291.4623 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 3.5 min*
MAXIMUM ΔI : 0.9 deg*

*Based on NRL analysis

COMMENTS

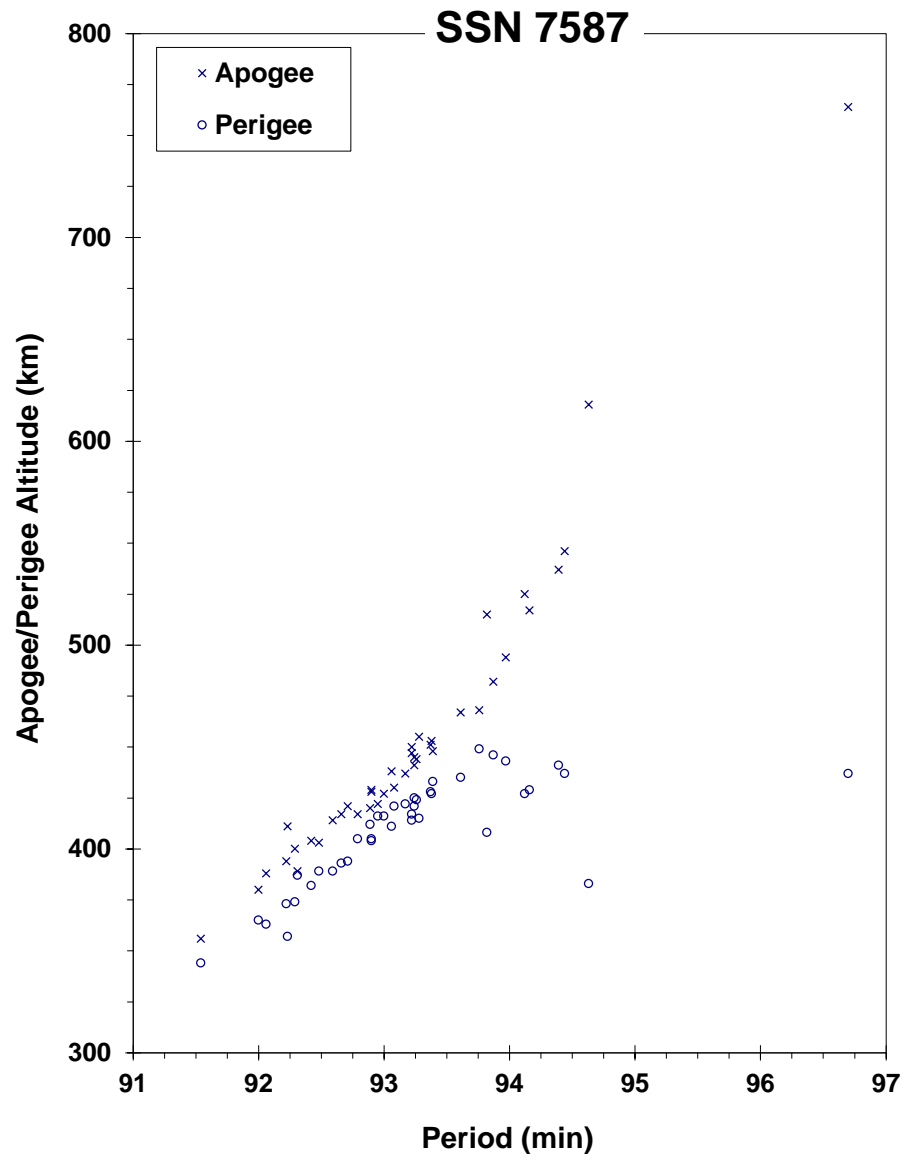
Cosmos 699 was the first of a new type spacecraft. Many members of this class have experienced breakups. Beginning in 1988 old spacecraft have been commanded to lower perigee at end of life, resulting in an accelerated natural decay with fewer fragmentations. For several spacecraft, two distinct events have been detected and observational data suggest that the spacecraft remain essentially intact after each event. In most cases, breakups occur after spacecraft has ceased orbit maintenance and entered natural decay. Debris are sometimes highly unidirectional. In the case of Cosmos 699, the spacecraft had been in a regime of natural decay for 1 month at the time of the event.

REFERENCE DOCUMENTS

An Analysis of the Breakup of Satellite 1974-103A (Cosmos 699), W. B. Heard, NRL Report 7991, Naval Research Laboratory, Washington, 23 April 1976.

"Artificial Satellite Break-Ups (Part 1): Soviet Ocean Surveillance Satellites", N. L. Johnson, Journal of the British Interplanetary Society, February 1983, pp. 51-58.

History of the Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 699 debris cloud of 41 fragments after the first breakup event as reconstructed from radar observations following the first breakup event.

SATELLITE DATA

TYPE: Delta Second Stage (2910)
OWNER: US
LAUNCH DATE: 22.75 Jan 1975
DRY MASS (KG): 840
MAIN BODY: Cylinder-nozzle; 1.4 m diameter by 5.8 m length
MAJOR APPENDAGES: Mini-skirt; 2.4 m by 0.2 m
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: On-board propellants, range safety device

EVENT DATA (1)

| | | | |
|-----------|------------|-----------------|------------|
| DATE: | 9 Feb 1976 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS (1)

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 76040.08509016 | MEAN ANOMALY: | 189.3492 |
| RIGHT ASCENSION: | 60.2329 | MEAN MOTION: | 14.19373945 |
| INCLINATION: | 97.7751 | MEAN MOTION DOT/2: | .0 |
| ECCENTRICITY: | .0120730 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 170.9843 | BSTAR: | .0 |

EVENT DATA (2)

| | | | |
|-----------|-------------|-----------------|----------------|
| DATE: | 19 Jun 1976 | LOCATION: | 7N, 344E (dsc) |
| TIME: | 0659 GMT | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | 750 km | | |

PRE-EVENT ELEMENTS (2)

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 76170.97576375 | MEAN ANOMALY: | 217.2433 |
| RIGHT ASCENSION: | 175.3897 | MEAN MOTION: | 14.19574919 |
| INCLINATION: | 97.7497 | MEAN MOTION DOT/2: | .0 |
| ECCENTRICITY: | .0115288 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 143.6594 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 5.6 min
MAXIMUM ΔI : 2.3 deg

COMMENTS

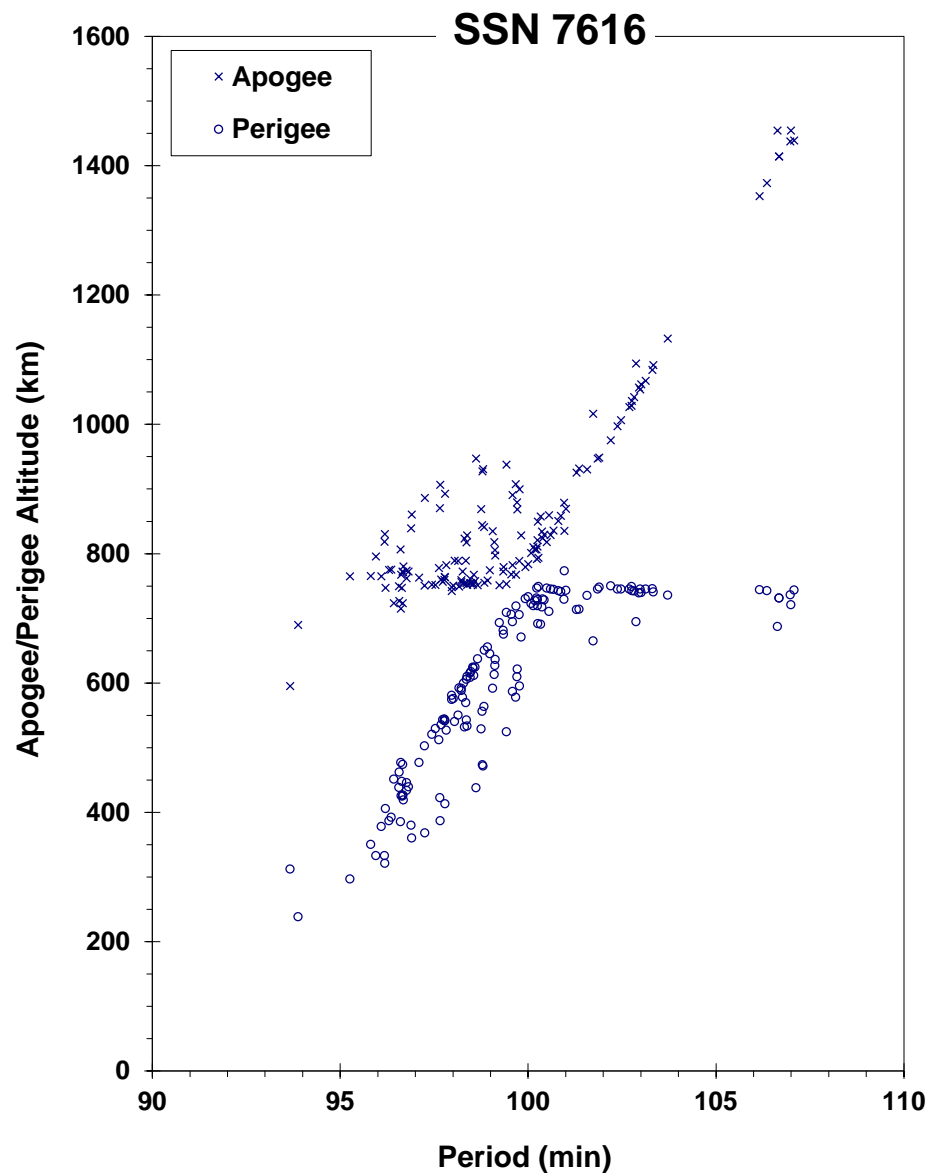
This was the fourth Delta Second Stage to experience a severe fragmentation. The first event occurred almost 13 months after the successful deployment of the Landsat 2 payload. Only 14 fragments were cataloged after the first event and all possessed orbital period changes of less than 0.6 min. Four months later a much larger fragmentation occurred. The cause of the second event is assessed to be related to the estimated 150 kg of residual propellants on board and characteristics of the sun-synchronous orbit.

REFERENCE DOCUMENTS

Explosion of Satellite 10704 and other Delta Second Stage Rockets, J.R. Gabbard, Technical Memorandum 81-5, DCS Plans, Hdqtrs NORAD/ADCOM, Colorado Springs, May 1981.

Investigation of Delta Second Stage On-Orbit Explosions, C.S. Gumpel, Report MDC-H0047, McDonnell Douglas Astronautics Company - West, Huntington Beach, April 1982.

A Later Look at Delta Second Stage On-Orbit Explosions, J.R. Gabbard, Technical Report CS85-BMDSC-00-24, Teledyne Brown Engineering, Colorado Springs, March 1985.



Landsat 2 R/B debris cloud of 147 fragments about 6 weeks after the second event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Delta Second Stage (2910)
OWNER: US
LAUNCH DATE: 12.34 Jun 1975
DRY MASS (KG): 840
MAIN BODY: Cylinder-nozzle; 1.4 m diameter by 5.8 m length
MAJOR APPENDAGES: Mini-skirt; 2.4 m by 0.3 m
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: On-board propellants, range safety device

EVENT DATA

| | | | |
|-----------|------------|-----------------|-----------------|
| DATE: | 1 May 1991 | LOCATION: | 66N, 322E (asc) |
| TIME: | 0856 GMT | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | 1090 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 91112.56709963 | MEAN ANOMALY: | 211.7525 |
| RIGHT ASCENSION: | 329.2109 | MEAN MOTION: | 13.43007146 |
| INCLINATION: | 99.5801 | MEAN MOTION DOT/2: | .00000050 |
| ECCENTRICITY: | .0006217 | MEAN MOTION DOT DOT/6: | 0.0 |
| ARG. OF PERIGEE: | 148.3989 | BSTAR: | .0055458 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 27.4 min*
MAXIMUM ΔI : 2.4 min*

*Based on uncataloged debris data

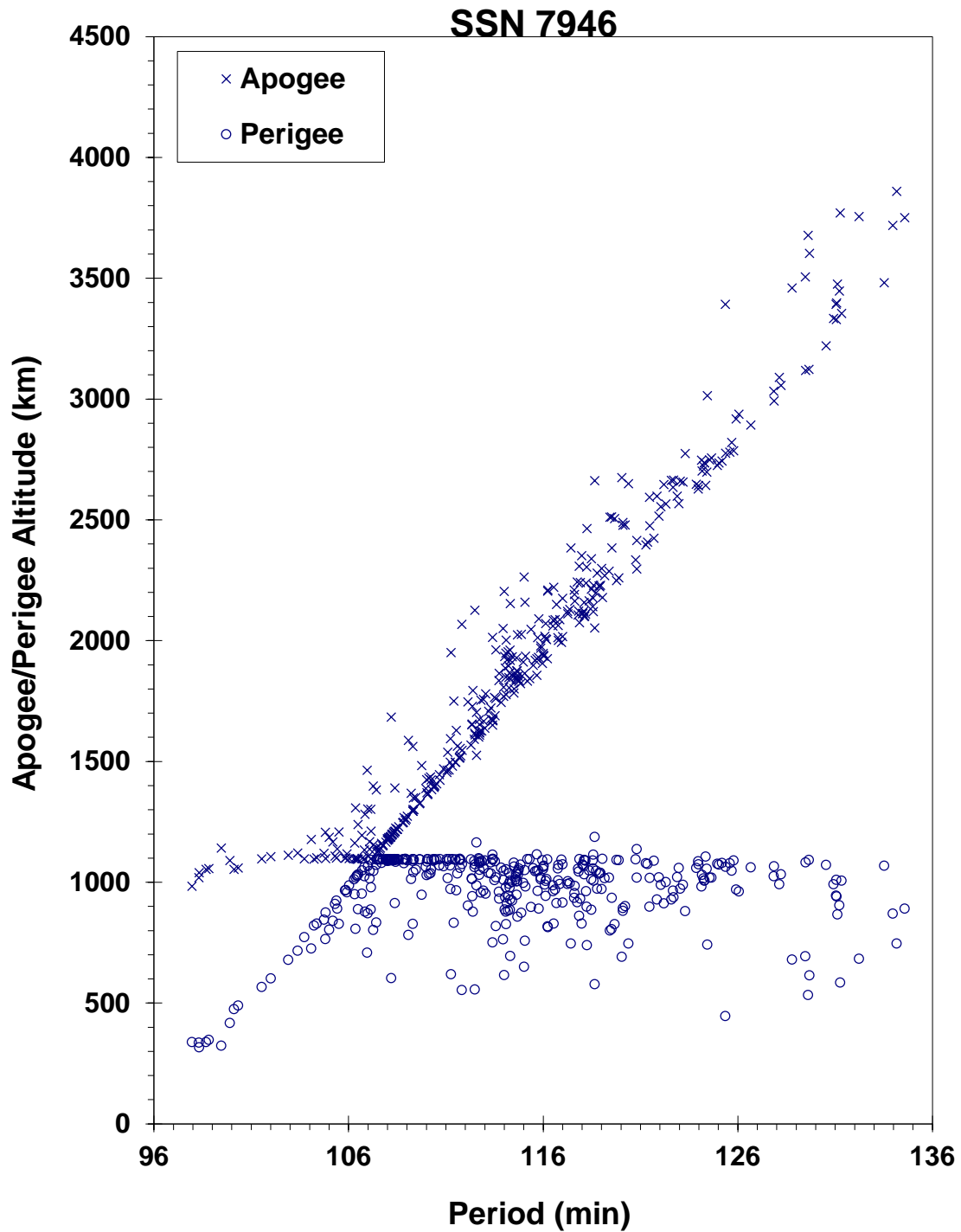
COMMENTS

This was the eighth Delta Second Stage to experience a severe fragmentation. The event occurred nearly 191 months after the successful deployment of the Nimbus 6 payload. Cause of the explosion is assessed to be related to the estimated 245 kg of residual propellants on board and characteristics of the sun-synchronous orbit.

REFERENCE DOCUMENTS

The Fragmentation of the Nimbus 6 Rocket Body, D. J. Nauer and N. L. Johnson, Technical Report CS91-TR-JSC-017, Teledyne Brown Engineering, Colorado Springs, Colorado, November 1991.

Nimbus 6 Delta Upper Stage Rocket Body Breakup Report, E. L. Jenkins and H. V. Reynolds, Naval Space Surveillance Center, Dahlgren, Virginia, 1991.



Nimbus 6 R/B debris cloud of 386 identified fragments within 1 week after the event as reconstructed from Naval Space Surveillance System database. This diagram is taken from the first cited reference.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 5.62 Sep 1975
DRY MASS (KG): 5700
MAIN BODY: Cone-cylinder; 2.7 m diameter by 6.3 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

| | | | |
|-----------|------------|-----------------|-----------------|
| DATE: | 6 Sep 1975 | LOCATION: | 32N, 293E (asc) |
| TIME: | 1906 GMT | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | 185 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 75249.72782895 | MEAN ANOMALY: | 294.2107 |
| RIGHT ASCENSION: | 189.2795 | MEAN MOTION: | 16.09422927 |
| INCLINATION: | 67.1445 | MEAN MOTION DOT/2: | .00430774 |
| ECCENTRICITY: | .0113994 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 67.1020 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

Spacecraft was apparently destroyed after a malfunction prevented controlled reentry and landing in the Soviet Union. Most debris reentered before being officially cataloged. All but three official fragments were cataloged without elements.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 29.46 Oct 1975
DRY MASS (KG): 3000
MAIN BODY: Cylinder; 1.3 m diameter by 17 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge (?)

EVENT DATA

| | | | |
|-----------|-------------|-----------------|---------------|
| DATE: | 25 Jan 1976 | LOCATION: | 53N, 7E (asc) |
| TIME: | 1400 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 440 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 76025.37753295 | MEAN ANOMALY: | 88.9272 |
| RIGHT ASCENSION: | 303.6319 | MEAN MOTION: | 15.43461781 |
| INCLINATION: | 65.0177 | MEAN MOTION DOT/2: | .00000373 |
| ECCENTRICITY: | .0009065 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 271.0782 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 1.6 min
MAXIMUM ΔI : 0.4 deg

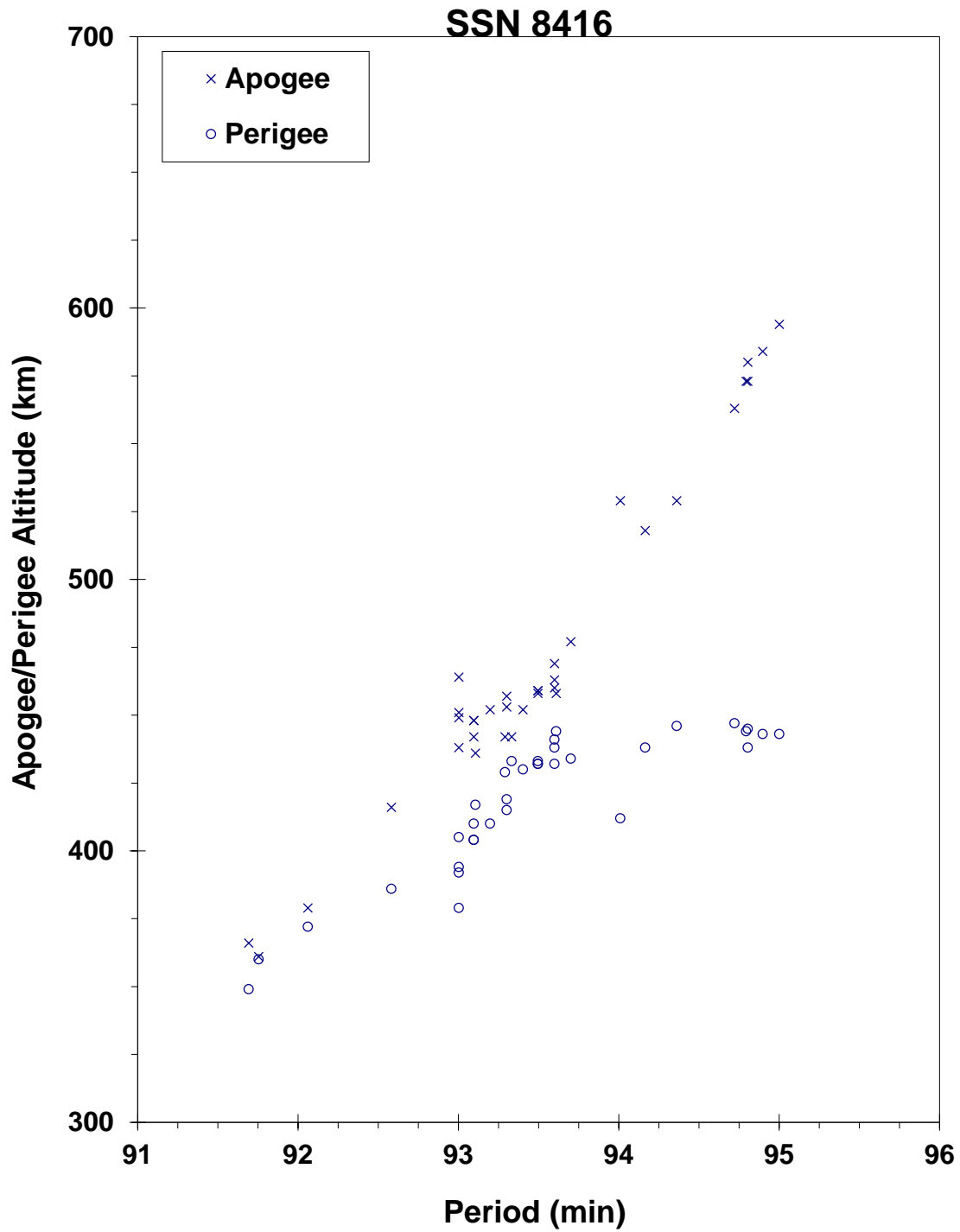
COMMENTS

Cosmos 777 was the second spacecraft of the Cosmos 699-type to experience a fragmentation. It is the only one to breakup before terminating its precise orbit maintenance pattern and entering a regime of natural decay. A second event may have occurred about 90 minutes after the event cited above.

REFERENCE DOCUMENTS

"Artificial Satellite Break-Ups (Part 1): Soviet Ocean Surveillance Satellites", N. L. Johnson, Journal of the British Interplanetary Society, February 1983, pp. 51-58.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 777 debris cloud of 35 fragments about 10 days after the event as reconstructed from the US SSN database. Some drag effects are already evident.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 2.44 Jul 1976
DRY MASS (KG): 3000
MAIN BODY: Cylinder; 1.3 m diameter by 17 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge (?)

EVENT DATA

| | | | |
|-----------|-------------|-----------------|----------------|
| DATE: | 17 May 1977 | LOCATION: | 9S, 284E (dsc) |
| TIME: | 1018 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 430 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 77136.94211102 | MEAN ANOMALY: | 73.5502 |
| RIGHT ASCENSION: | 131.3837 | MEAN MOTION: | 15.45822335 |
| INCLINATION: | 65.0556 | MEAN MOTION DOT/2: | .00007521 |
| ECCENTRICITY: | .0021270 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 286.3253 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 8.0 min*
MAXIMUM ΔI : 1.1 deg*

*Based on uncataloged debris data

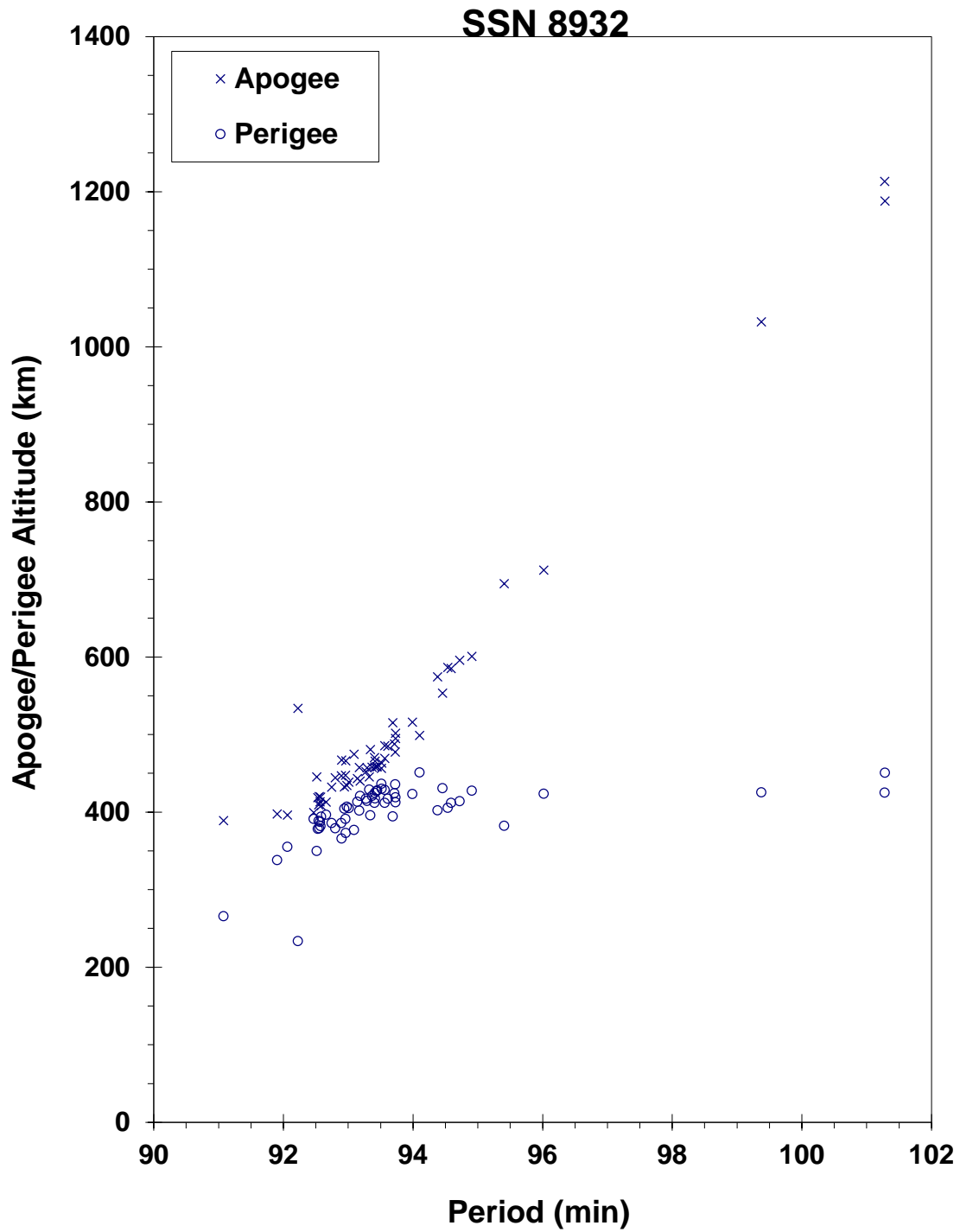
COMMENTS

Cosmos 838 was the third spacecraft of the Cosmos 699-type to experience a fragmentation. Spacecraft had been in a regime of natural decay for 6 months prior to the event. Many debris reentered before being officially cataloged.

REFERENCE DOCUMENTS

"Artificial Satellite Break-Ups (Part 1): Soviet Ocean Surveillance Satellites", N. L. Johnson, Journal of the British Interplanetary Society, February 1983, pp. 51-58.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 838 debris cloud of 59 fragments about 1 week after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 8.88 Jul 1976
DRY MASS (KG): 650
MAIN BODY: Polyhedron; 1.4 m by 1.4 m
MAJOR APPENDAGES: Solar panels, gravity-gradient boom (?)
ATTITUDE CONTROL: Gravity gradient (?)
ENERGY SOURCES: Battery

EVENT DATA

DATE: 29 Sep 1977
TIME: 0717 GMT
ALTITUDE: 1910 km
LOCATION: 33S, 162E (dsc)
ASSESSED CAUSE: Battery

PRE-EVENT ELEMENTS

EPOCH: 77270.46732078
RIGHT ASCENSION: 85.9347
INCLINATION: 65.8538
ECCENTRICITY: .0706585
ARG. OF PERIGEE: 351.1444
MEAN ANOMALY: 7.6996
MEAN MOTION: 12.32137908
MEAN MOTION DOT/2: .00000367
MEAN MOTION DOT DOT/6: .0
BSTAR: .0

DEBRIS CLOUD DATA

MAXIMUM ΔP : 2.7 min
MAXIMUM ΔI : 0.3 deg

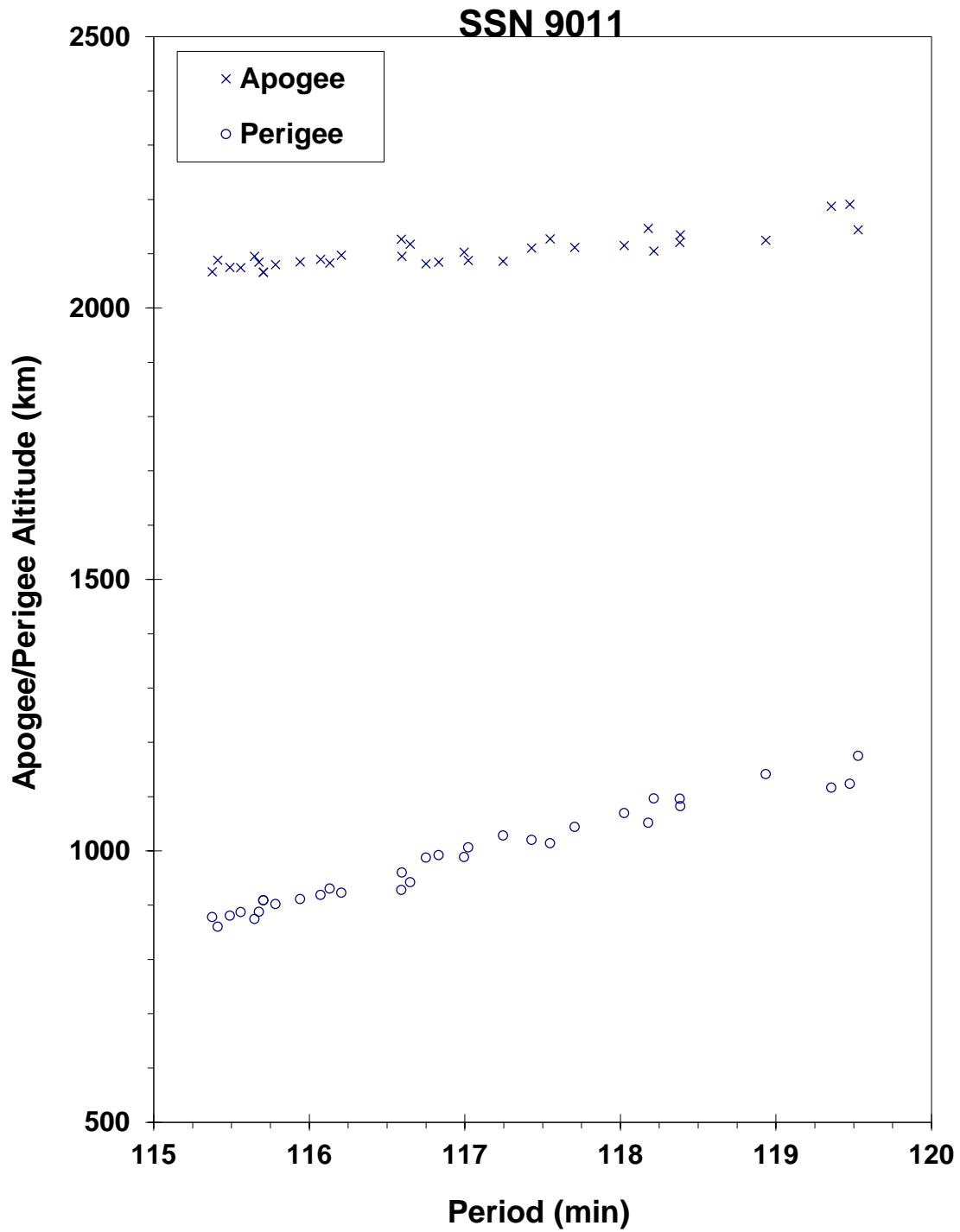
COMMENTS

Cosmos 839 was the first of three satellites of the same class to experience unexplained fragmentations. These satellites are used in conjunction with the Cosmos 249-type spacecraft, which are deliberately fragmented; but the cause of the Cosmos 839-type events appears to be unrelated since they occur more than 1 year after tests with Cosmos 249-type spacecraft. In the case of Cosmos 839, 14 months elapsed between its test with a Cosmos 249-type spacecraft and its fragmentation. Russian officials have determined that battery malfunctions were the causes of these events.

REFERENCE DOCUMENTS

"Artificial Satellite Break-Ups (Part 2): Soviet Anti-Satellite Program", N.L. Johnson, Journal of the British Interplanetary Society, August 1983, pp. 357-362.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 839 debris cloud of 33 fragments about 5 weeks after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 22.66 Jul 1976
DRY MASS (KG): 5700
MAIN BODY: Cone-cylinder; 2.7 m diameter by 6.3 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge (?)

EVENT DATA

| | | | |
|-----------|-------------|-----------------|-----------------|
| DATE: | 25 Jul 1976 | LOCATION: | 49N, 100E (dsc) |
| TIME: | 1718 GMT | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | 210 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 76207.45032150 | MEAN ANOMALY: | 291.2246 |
| RIGHT ASCENSION: | 152.6930 | MEAN MOTION: | 16.04433196 |
| INCLINATION: | 67.1467 | MEAN MOTION DOT/2: | .00313532 |
| ECCENTRICITY: | .0136374 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 70.3553 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

Spacecraft was apparently destroyed after a malfunction prevented controlled reentry and landing in the Soviet Union. No elements were cataloged on any of the official debris. Most fragments reentered rapidly.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Delta Second Stage (2310)
OWNER: US
LAUNCH DATE: 29.71 Jul 1976
DRY MASS (KG): 840
MAIN BODY: Cylinder-nozzle; 1.4 m diameter by 5.8 m length
MAJOR APPENDAGES: Mini-skirt; 2.4 m by 0.3 m
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: On-board propellants, range safety device

EVENT DATA

DATE: 24 Dec 1977 LOCATION: 40S, 146E (asc)
TIME: 1133 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 1510 km

PRE-EVENT ELEMENTS

EPOCH: 77354.53228225 MEAN ANOMALY: 330.8663
RIGHT ASCENSION: 38.5560 MEAN MOTION: 12.38394892
INCLINATION: 102.0192 MEAN MOTION DOT/2: .0
ECCENTRICITY: .0010085 MEAN MOTION DOT DOT/6: .0
ARG. OF PERIGEE: 29.2920 BSTAR: .0

DEBRIS CLOUD DATA

MAXIMUM ΔP : 12.5 min
MAXIMUM ΔI : 3.0 deg

COMMENTS

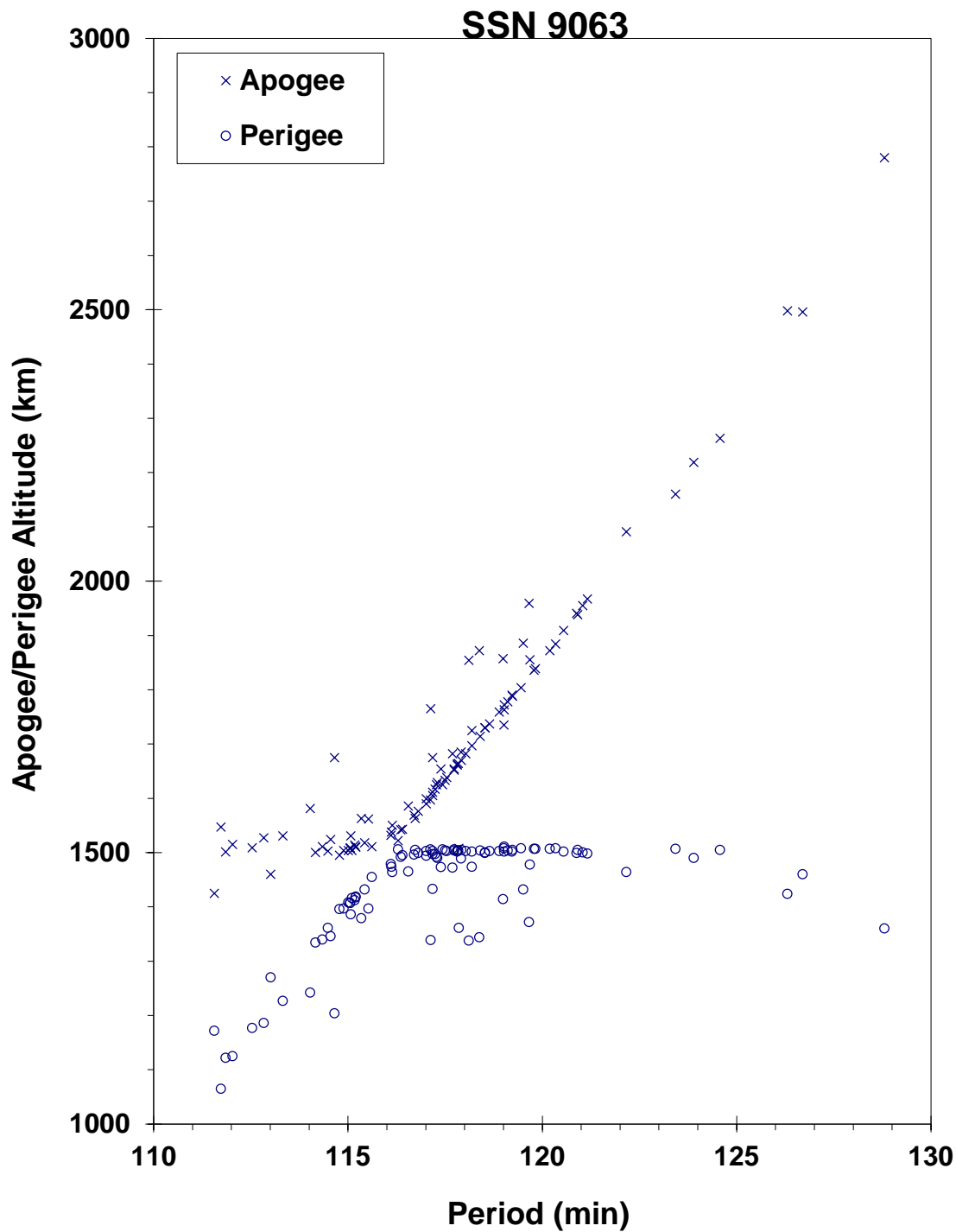
This was the sixth Delta Second Stage to experience a severe fragmentation. The event occurred 17 months after the successful deployment of the NOAA 5 payload. Cause of the explosion is assessed to be related to the estimated 250 kg of residual propellants on board and characteristics of the sun-synchronous orbit.

REFERENCE DOCUMENTS

Explosion of Satellite 10704 and other Delta Second Stage Rockets, J.R. Gabbard, Technical Memorandum 81-5, DCS Plans, Hdqtrs NORAD/ADCOM, Colorado Springs, May 1981.

Investigation of Delta Second Stage On-Orbit Explosions, C.S. Gumpel, Report MDC-H0047, McDonnell Douglas Astronautics Company - West, Huntington Beach, April 1982.

A Later Look at Delta Second Stage On-Orbit Explosions, J.R. Gabbard, Technical Report CS85-BMDSC-00-24, Teledyne Brown Engineering, Colorado Springs, March 1985.



NOAA 5 R/B debris cloud of 98 fragments about 4 months after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 22.38 Oct 1976
DRY MASS (KG): 1250
MAIN BODY: Cylinder; 1.7 m diameter by 2 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive device

EVENT DATA

DATE: 15 Mar 1977 LOCATION: 39N, 114E (asc)
TIME: 1256 GMT ASSESSED CAUSE: Deliberate
ALTITUDE: 5375 km

PRE-EVENT ELEMENTS

EPOCH: 77066.03986408 MEAN ANOMALY: 4.4196
RIGHT ASCENSION: 98.8078 MEAN MOTION: 2.00311741
INCLINATION: 63.1553 MEAN MOTION DOT/2: .0
ECCENTRICITY: .7312859 MEAN MOTION DOT DOT/6: .0
ARG. OF PERIGEE: 318.6653 BSTAR: .0

DEBRIS CLOUD DATA

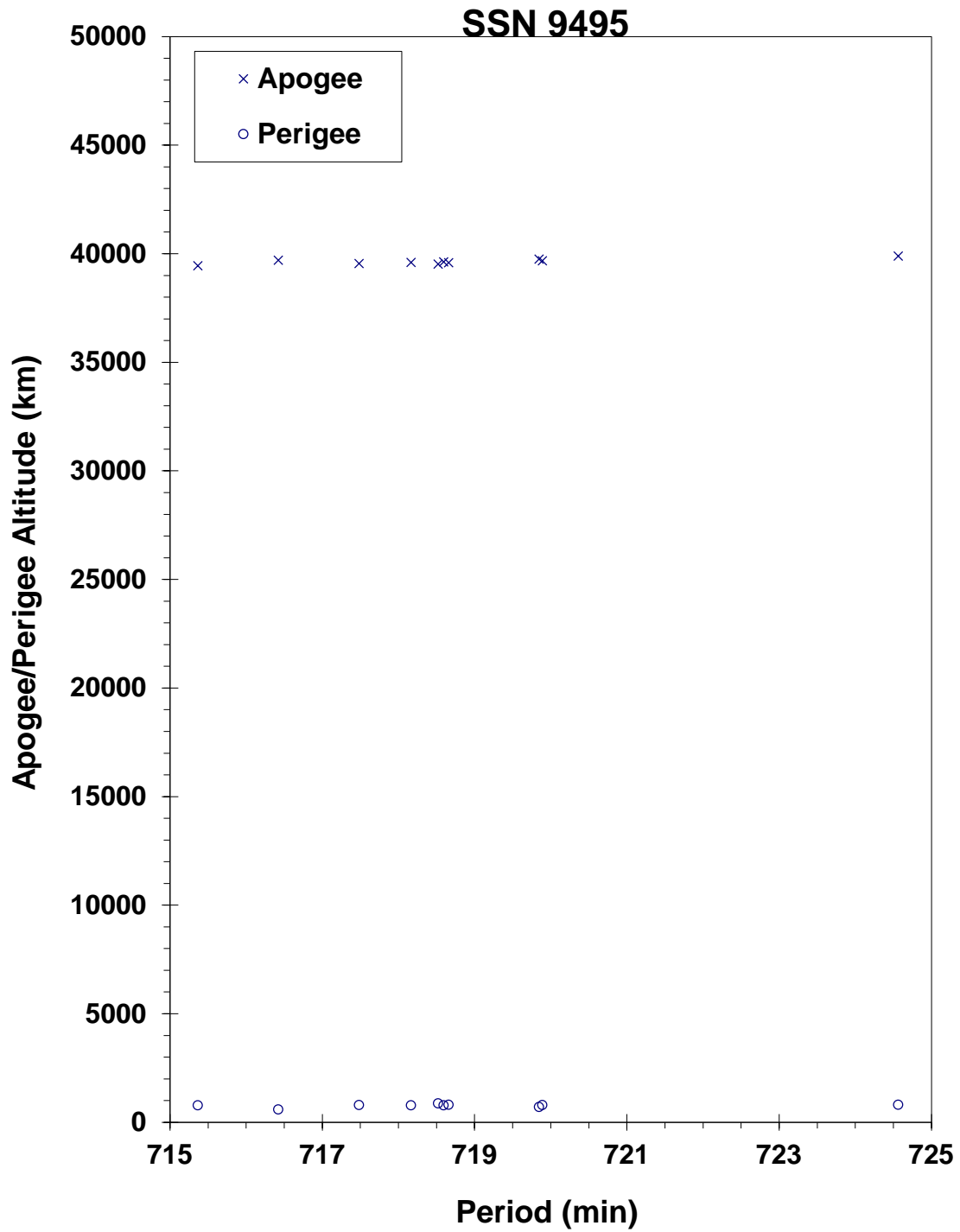
MAXIMUM ΔP : 5.7 min
MAXIMUM ΔI : 0.4 deg

COMMENTS

Cosmos 862 was the first of a new class of operational satellites in highly elliptical, semi-synchronous orbits that experienced a total of 16 fragmentations during the period 1977-1986. Due to the nature of these orbits, which result in high altitudes over the Northern Hemisphere where most surveillance sensors are located, debris detection and tracking is extremely difficult. Only the largest fragments can be seen. Cosmos 862 spacecraft were equipped with self-destruct packages in the event that spacecraft control was lost; this was the cause of breakups until the explosives were removed after Cosmos 1481.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 862 debris cloud of 10 cataloged fragments 2 weeks after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 9.84 Dec 1976
DRY MASS (KG): 650
MAIN BODY: Polyhedron; 1.4 m by 1.4 m
MAJOR APPENDAGES: Solar panels, gravity-gradient boom (?)
ATTITUDE CONTROL: Gravity gradient (?)
ENERGY SOURCES: Battery

EVENT DATA

DATE: 27 Nov 1978
TIME: 1703 GMT
ALTITUDE: 560 km
LOCATION: 65S, 306E (dsc)
ASSESSED CAUSE: Battery

PRE-EVENT ELEMENTS

EPOCH: 78331.59395829
RIGHT ASCENSION: 11.0317
INCLINATION: 65.8440
ECCENTRICITY: .0050108
ARG. OF PERIGEE: 304.0553
MEAN ANOMALY: 55.5772
MEAN MOTION: 14.93841919
MEAN MOTION DOT/2: .00000004
MEAN MOTION DOT DOT/6: .0
BSTAR: .0

DEBRIS CLOUD DATA

MAXIMUM ΔP : 1.3 min*
MAXIMUM ΔI : 0.0 deg*

*Based on uncataloged debris data

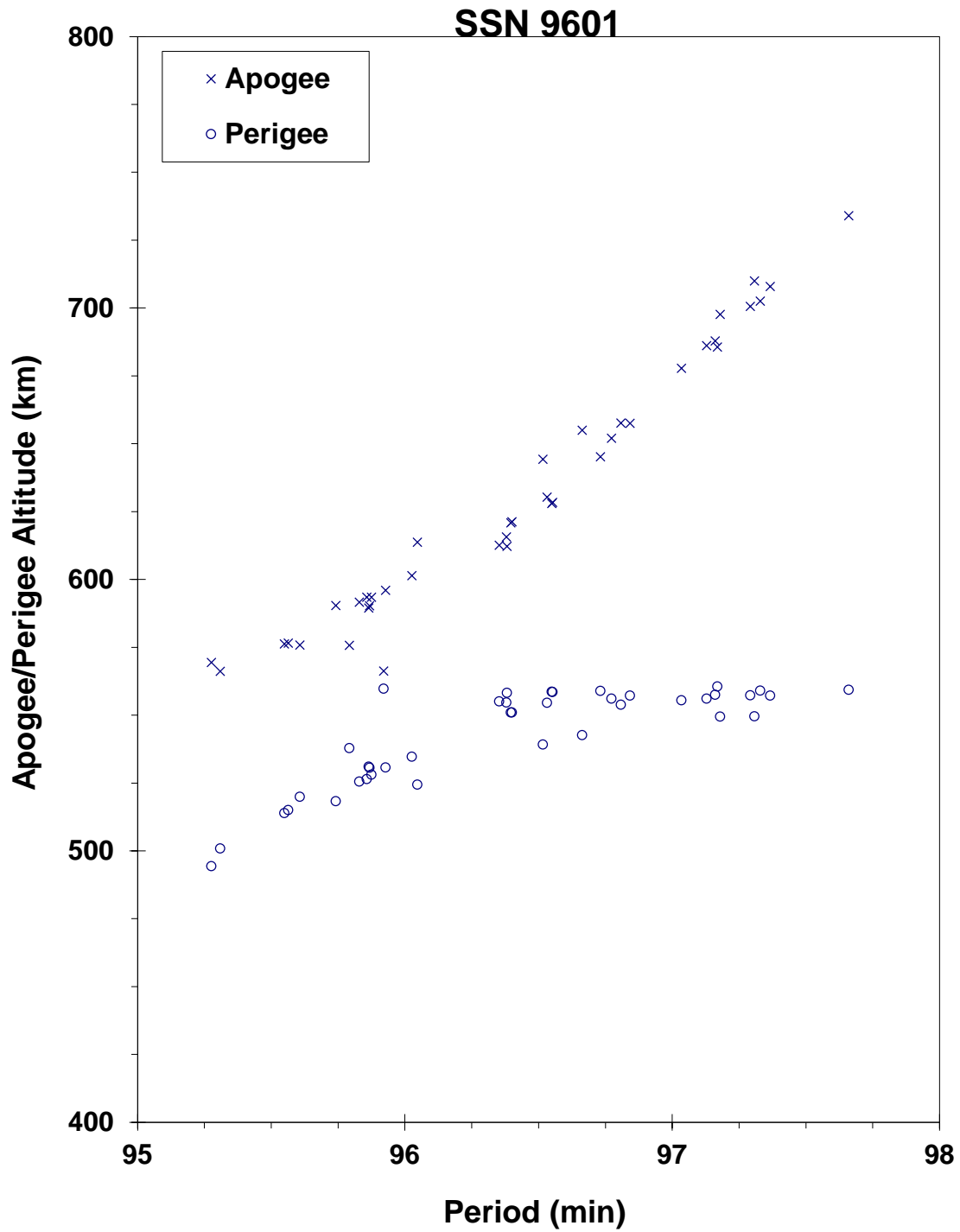
COMMENTS

Cosmos 880 was the second spacecraft of the Cosmos 839-type to experience a fragmentation. Although these satellites are used in conjunction with the Cosmos 249-type spacecraft that are deliberately fragmented, the cause of the Cosmos 839-type events appears to be unrelated. In the case of Cosmos 880, 23 months elapsed since its test with a Cosmos 249-type spacecraft. Russian officials have determined that battery malfunctions were the cause of these events.

REFERENCE DOCUMENTS

"Artificial Satellite Break-Ups (Part 2): Soviet Anti-Satellite Program", N.L. Johnson, Journal of the British Interplanetary Society, August 1983, pp. 357-362.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 880 debris cloud of 40 fragments 2 days after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 17.40 Dec 1976
DRY MASS (KG): 6300
MAIN BODY: Sphere-cylinder; 2.4 m diameter by 6.5 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 29 Dec 1976 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 76362.45360574 | MEAN ANOMALY: | 302.3648 |
| RIGHT ASCENSION: | 227.6719 | MEAN MOTION: | 16.11011505 |
| INCLINATION: | 65.0214 | MEAN MOTION DOT/2: | 0.00147448 |
| ECCENTRICITY: | 0.0113306 | MEAN MOTION DOT DOT/6: | 0 |
| ARG. OF PERIGEE: | 58.8529 | BSTAR: | 0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

Spacecraft was destroyed after a malfunction prevented reentry and landing in the Soviet Union. Event identified by Russian officials during investigation cited below.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 27.53 Dec 1976
DRY MASS (KG): 1400
MAIN BODY: Irregular; 1.8 m by 4.2 m
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

DATE: 27 Dec 1976
TIME: 1840 GMT
ALTITUDE: 2090 km
LOCATION: 65S, 210E (asc)
ASSESSED CAUSE: Deliberate

POST-EVENT ELEMENTS

EPOCH: 76362.79720829
RIGHT ASCENSION: 306.5669
INCLINATION: 65.8434
ECCENTRICITY: .1087102
ARG. OF PERIGEE: 57.0236
MEAN ANOMALY: 313.0540
MEAN MOTION: 12.54457816
MEAN MOTION DOT/2: .00004000
MEAN MOTION DOT DOT/6: .0
BSTAR: .0

DEBRIS CLOUD DATA

MAXIMUM ΔP : 4.3 min
MAXIMUM ΔI : 0.2 deg

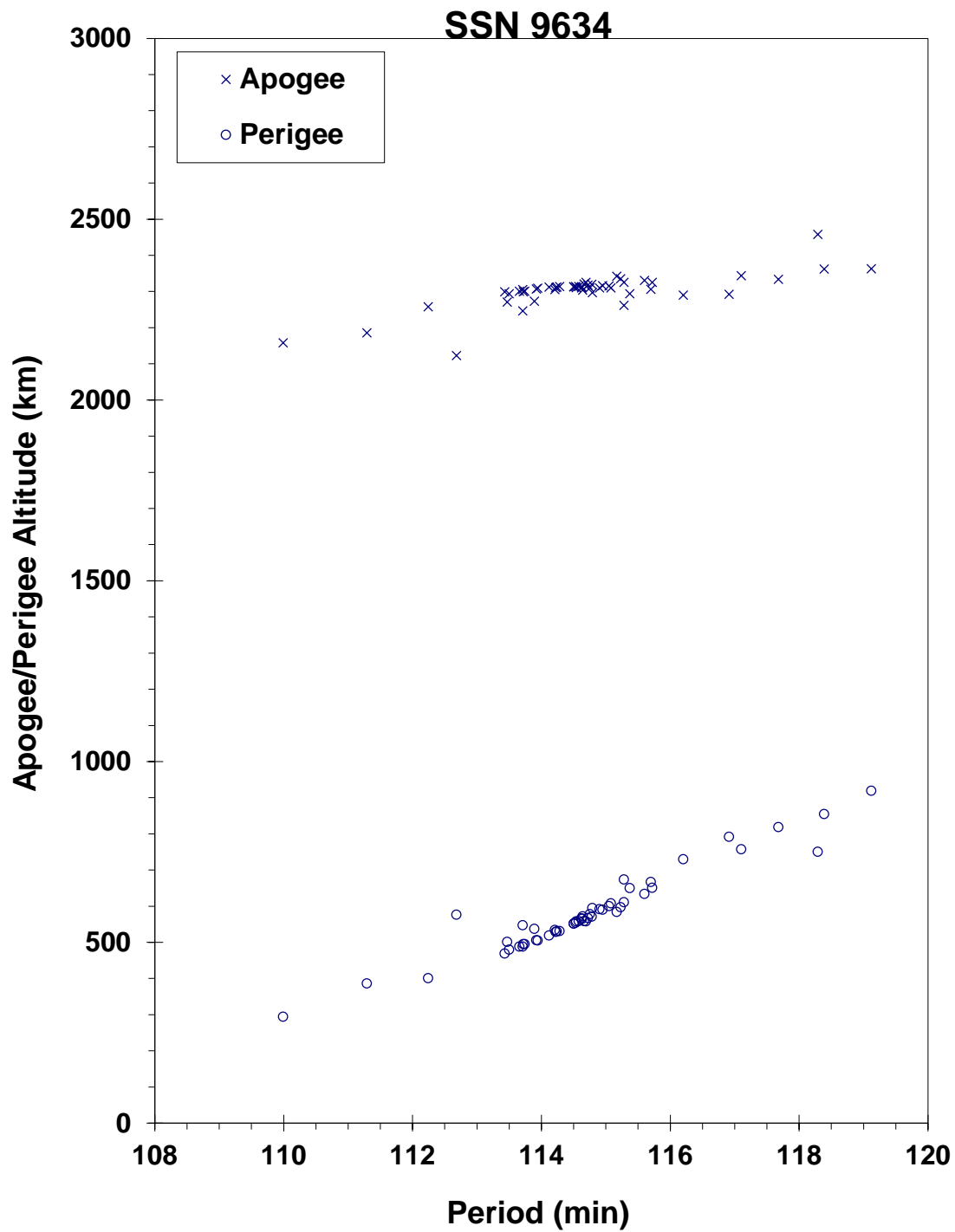
COMMENTS

Cosmos 886 was launched on a two-revolution rendezvous with Cosmos 880. After a close approach, Cosmos 886 continued on before its warhead was intentionally fired. Cosmos 886 was part of test series begun with Cosmos 249. The elements above are the first available after the final maneuver of Cosmos 886 but represent the revolution immediately after the event.

REFERENCE DOCUMENTS

"Artificial Satellite Break-Ups (Part 2): Soviet Anti-Satellite Program", N.L. Johnson, Journal of the British Interplanetary Society, August 1983, pp. 357-362.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 886 debris cloud of 53 fragments 5 months after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 11.07 Apr 1977
DRY MASS (KG): 1250
MAIN BODY: Cylinder; 1.7 m diameter by 2 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

| | | | |
|-----------|------------|-----------------|------------|
| DATE: | 8 Jun 1978 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 78156.86414074 | MEAN ANOMALY: | 5.0496 |
| RIGHT ASCENSION: | 115.5660 | MEAN MOTION: | 2.00599850 |
| INCLINATION: | 63.1514 | MEAN MOTION DOT/2: | .0 |
| ECCENTRICITY: | .7100107 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 319.7397 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 2.6 min*
MAXIMUM ΔI : 0.5 deg*

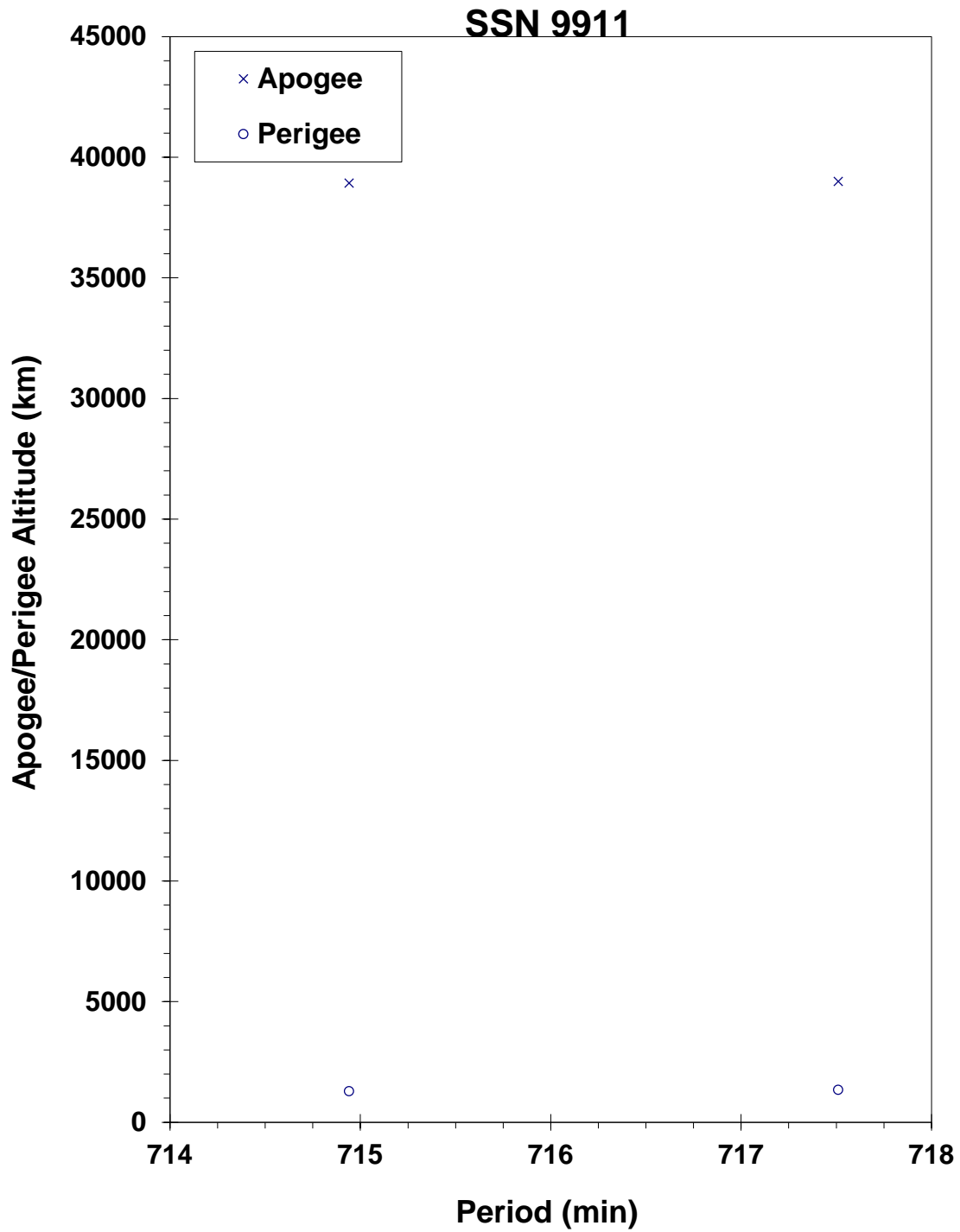
*See Comments

COMMENTS

Cosmos 903 was another spacecraft of the Cosmos 862-type to experience a fragmentation. One new fragment was cataloged within a week of the event. The ΔP and ΔI values above are based on the lower period (717.5 min) orbit of Cosmos 903 after the event.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 903 and a single piece of debris 3 weeks after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 16.08 Jun 1977
DRY MASS (KG): 1250
MAIN BODY: Irregular; 1.7 m by 2 m
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

DATE: 30 March 1979
TIME: 1545 GMT
ALTITUDE: 3280 km
LOCATION: 63S, 0E (dsc)
ASSESSED CAUSE: Deliberate

PRE-EVENT ELEMENTS

EPOCH: 79089.17562851
RIGHT ASCENSION: 156.1576
INCLINATION: 62.9498
ECCENTRICITY: .6980052
ARG. OF PERIGEE: 322.3289
MEAN ANOMALY: 5.2297
MEAN MOTION: 2.00553521
MEAN MOTION DOT/2: .0
MEAN MOTION DOT DOT/6: .0
BSTAR: .0

DEBRIS CLOUD DATA

MAXIMUM ΔP : 22.6 min*
MAXIMUM ΔI : 0.6 deg*

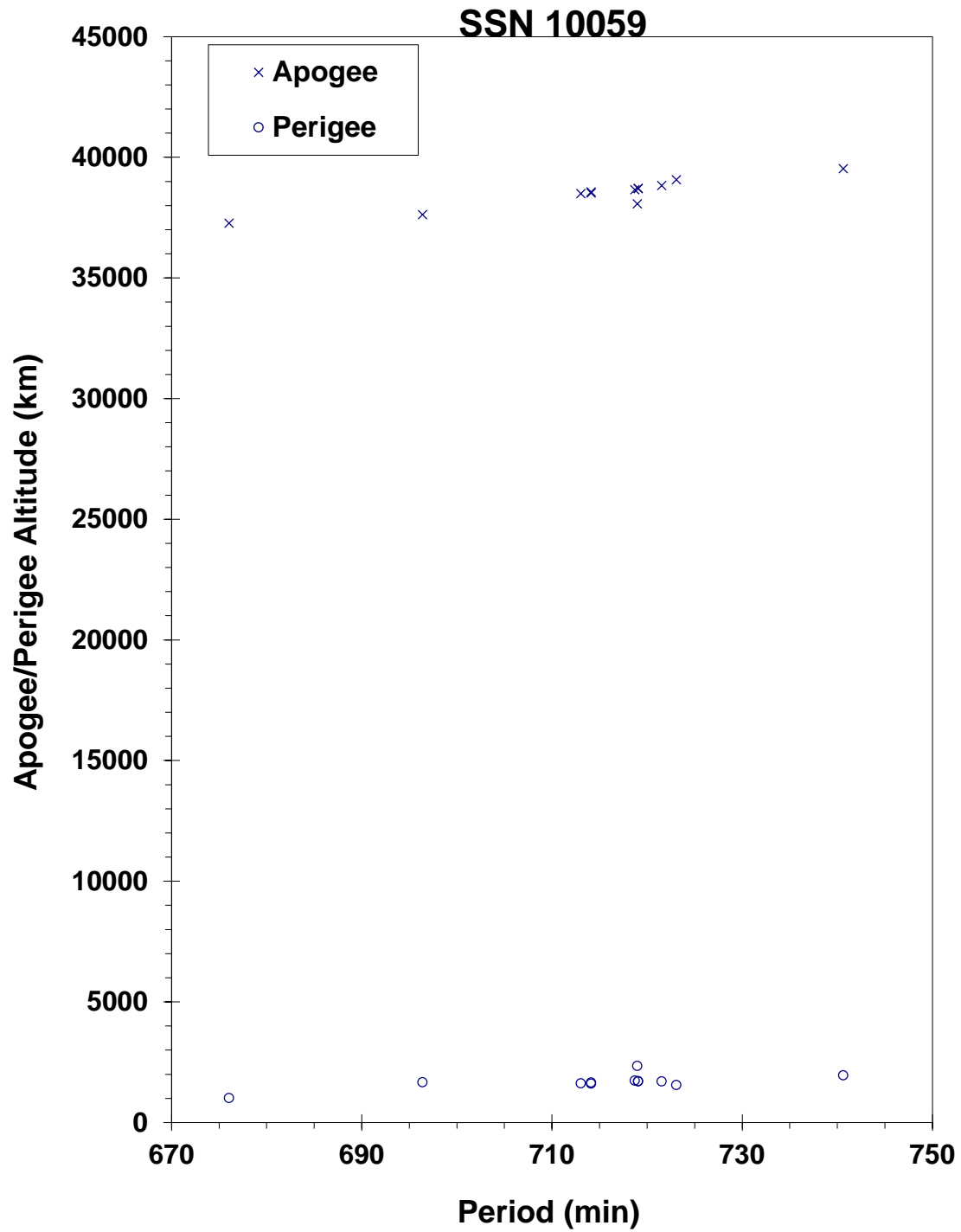
*Based on uncataloged debris data

COMMENTS

Cosmos 917 was another spacecraft of the Cosmos 862-type to experience a fragmentation.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 917 debris cloud of 12 fragments about 3 weeks after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Delta Second Stage (2914)
OWNER: US
LAUNCH DATE: 14.44 Jul 1977
DRY MASS (KG): 900
MAIN BODY: Cylinder-nozzle; 1.4 m diameter by 5.8 m length
MAJOR APPENDAGES: Mini-skirt; 2.4 m by 0.3 m
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: On-board propellants, range safety device

EVENT DATA

DATE: 14 Jul 1977
TIME: 1612 GMT
ALTITUDE: 1450 km
LOCATION: 14N, 249E (dsc)
ASSESSED CAUSE: Propulsion

POST-EVENT ELEMENTS

EPOCH: 77197.57445278
RIGHT ASCENSION: 262.0317
INCLINATION: 29.0493
ECCENTRICITY: .0973469
ARG. OF PERIGEE: 66.7255
MEAN ANOMALY: 303.2693
MEAN MOTION: 12.95114397
MEAN MOTION DOT/2: .00007335
MEAN MOTION DOT DOT/6: .0
BSTAR: .0

DEBRIS CLOUD DATA

MAXIMUM ΔP : 9.7 min*
MAXIMUM ΔI : 3.0 deg*

*Based on uncataloged debris data

COMMENTS

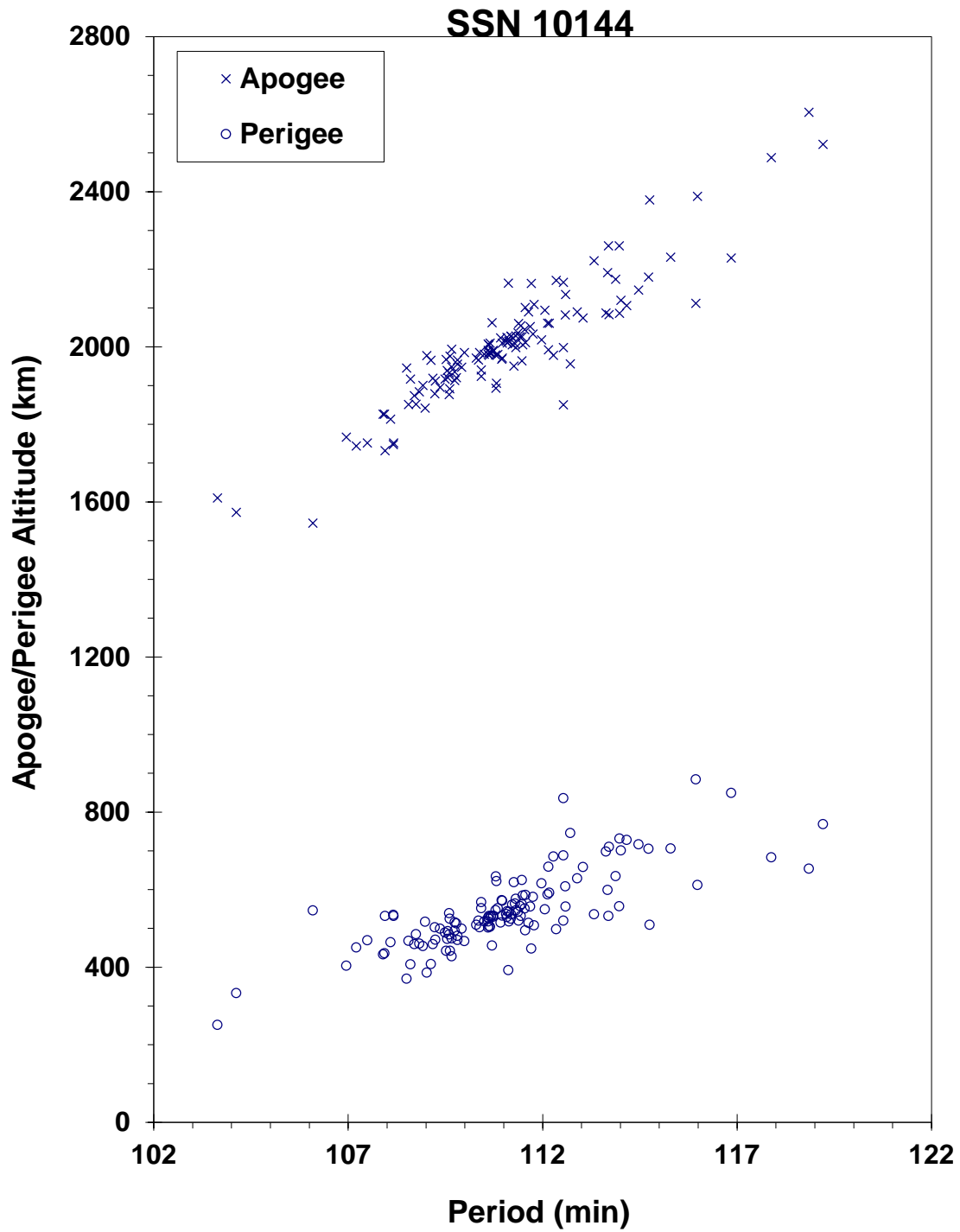
This was the fifth Delta Second Stage to experience a severe fragmentation. It is also the only one that was not in a sun-synchronous orbit, which had performed a depletion burn, and which fragmented on the day of launch. This rocket body did perform its mission successfully, carrying the third stage and the payload into a low Earth orbit. The energy for the breakup is assessed to have been the 40 kg of propellants (mainly oxidizer) remaining after the depletion burn. The elements above are the first available after the depletion burn although also after the event.

REFERENCE DOCUMENTS

Explosion of Satellite 10704 and other Delta Second Stage Rockets, J.R. Gabbard, Technical Memorandum 81-5, DCS Plans, Hdqtrs NORAD/ADCOM, Colorado Springs, May 1981.

Investigation of Delta Second Stage On-Orbit Explosions, C.S. Gumpel, Report MDC-H0047, McDonnell Douglas Astronautics Company - West, Huntington Beach, April 1982.

A Later Look at Delta Second Stage On-Orbit Explosions, J.R. Gabbard, Technical Report CS85-BMDSC-00-24, Teledyne Brown Engineering, Colorado Springs, March 1985.



Himawari 1 R/B debris cloud of 132 fragments 5 months after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 20.20 Jul 1977
DRY MASS (KG): 1250
MAIN BODY: Cylinder; 1.7 m diameter by 2 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 24 Oct 1977 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 77289.02131186 | MEAN ANOMALY: | 4.2624 |
| RIGHT ASCENSION: | 305.6648 | MEAN MOTION: | 2.00651833 |
| INCLINATION: | 62.9440 | MEAN MOTION DOT/2: | .0 |
| ECCENTRICITY: | .7341055 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 318.8771 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 5.3 min*
MAXIMUM ΔI : 0.7 deg*

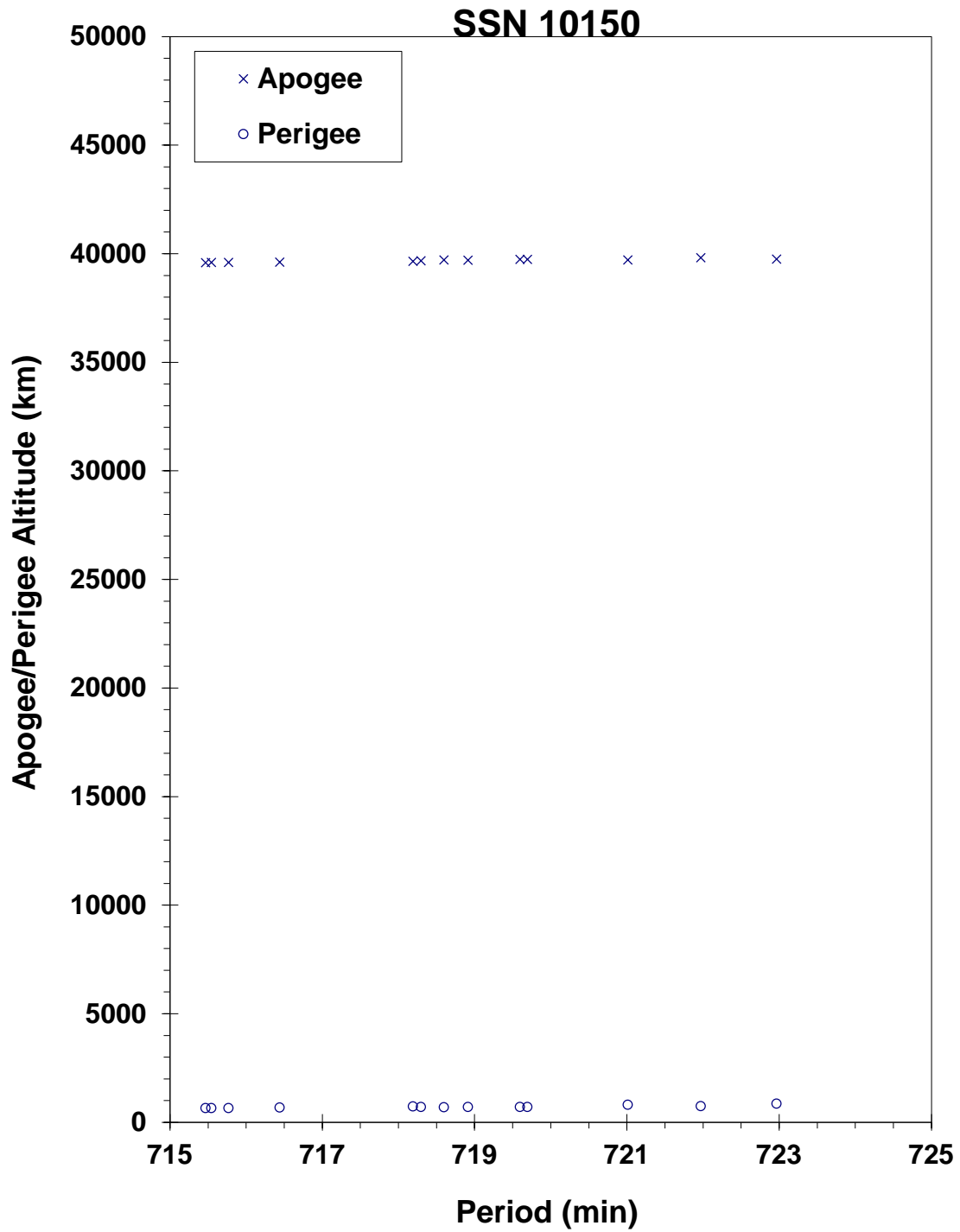
*Based on uncataloged debris data

COMMENTS

Cosmos 931 was another spacecraft of the Cosmos 862-type to experience a fragmentation. Debris were not officially cataloged until 4 years after the event.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 931 debris cloud of 13 fragments 2 weeks after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 20.73 Sept 1977
DRY MASS (KG): 1750
MAIN BODY: Cone; 2 m by 4 m
MAJOR APPENDAGES: Plate + 2 solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, Battery

EVENT DATA

| | | | |
|-----------|-------------|-----------------|-------------|
| DATE: | 23 Jun 1978 | LOCATION: | 0.0N, 98.7E |
| TIME: | Unknown | ASSESSED CAUSE: | Battery |
| ALTITUDE: | 35790 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 88166.03647595 | MEAN ANOMALY: | 78.3897 |
| RIGHT ASCENSION: | 78.3897 | MEAN MOTION: | 1.00252588 |
| INCLINATION: | 0.1137 | MEAN MOTION DOT/2: | .0 |
| ECCENTRICITY | .0001436 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 325.2771 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

This event was revealed by Russian officials in an orbital debris meeting in February 1992 in Moscow. This is the first known geostationary orbit fragmentation and was not detected by the Space Surveillance Network (SSN). Russian photographs originally linked to the breakup were later determined to have been misidentified.

REFERENCE DOCUMENTS

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

“Breakup in Review-Two GEO Breakups”, Orbital Debris Monitor, April 1992, p 35-36.

Insufficient data to construct a Gabbard diagram.

COSMOS 970

1977-121A

10531

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 21.44 Dec 1977
DRY MASS (KG): 1400
MAIN BODY: Cylinder; 2 m diameter by 4 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

| | | | |
|-----------|-------------|-----------------|-----------------|
| DATE: | 21 Dec 1977 | LOCATION: | 38S, 274E (asc) |
| TIME: | 1710 GMT | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | 1135 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 77355.65049149 | MEAN ANOMALY: | 245.5638 |
| RIGHT ASCENSION: | 282.1792 | MEAN MOTION: | 13.58084598 |
| INCLINATION: | 65.8467 | MEAN MOTION DOT/2: | .00023007 |
| ECCENTRICITY: | .0129854 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 116.3098 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 4.7 min
MAXIMUM ΔI : 1.1 deg

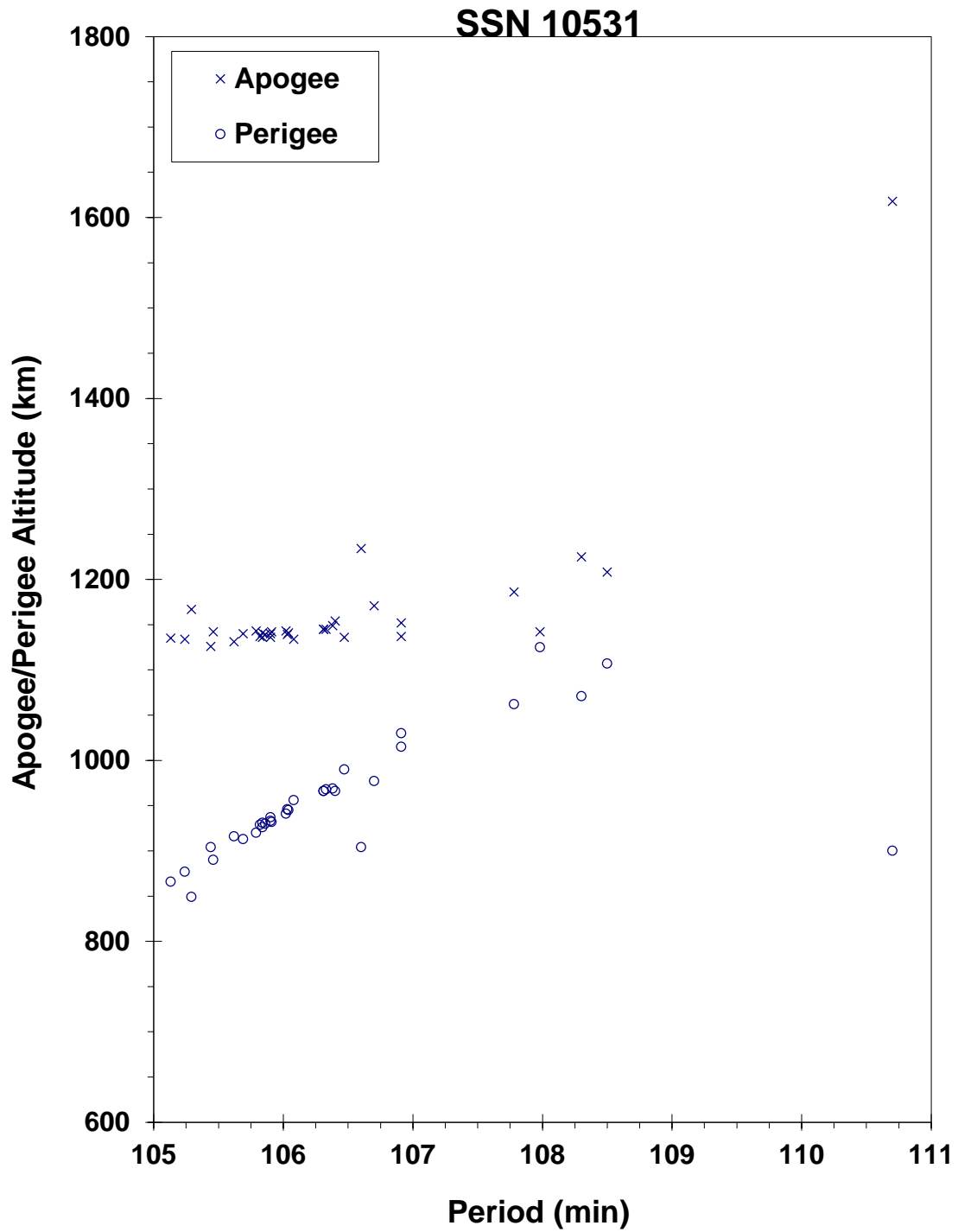
COMMENTS

Cosmos 970 was launched on a two-revolution rendezvous with Cosmos 967. After a close approach, Cosmos 970 continued on before its warhead was intentionally fired. Cosmos 970 was part of test series begun with Cosmos 249.

REFERENCE DOCUMENTS

"Artificial Satellite Break-Ups (Part 2): Soviet Anti-Satellite Program", N. L. Johnson, Journal of the British Interplanetary Society, August 1983, p. 357-362.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 970 debris cloud of 34 fragments about 5 months after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Delta Second Stage (2910)
OWNER: US
LAUNCH DATE: 5.75 Mar 1978
DRY MASS (KG): 900
MAIN BODY: Cylinder-nozzle; 1.4 m diameter by 5.8 m length
MAJOR APPENDAGES: Mini-skirt; 2.4 m by 0.3 m
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: On-board propellants, range safety device

EVENT DATA

DATE: 27 Jan 1981 LOCATION: 80S, 301E (asc)
TIME: 0432 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 910 km

PRE-EVENT ELEMENTS

EPOCH: 81026.99107090 MEAN ANOMALY: 147.0549
RIGHT ASCENSION: 68.7927 MEAN MOTION: 13.96108433
INCLINATION: 98.8485 MEAN MOTION DOT/2: .00000434
ECCENTRICITY: .0006255 MEAN MOTION DOT DOT/6: .0
ARG. OF PERIGEE: 212.9842 BSTAR: .00032708

DEBRIS CLOUD DATA

MAXIMUM ΔP : 9.1 min
MAXIMUM ΔI : 0.5 deg

COMMENTS

This was the seventh Delta Second Stage to experience a severe fragmentation. The event occurred nearly 35 months after the successful deployment of the Landsat 3 payload. Cause of the explosion is assessed to be related to the estimated 100 kg of residual propellants on board and characteristics of the sun-synchronous orbit.

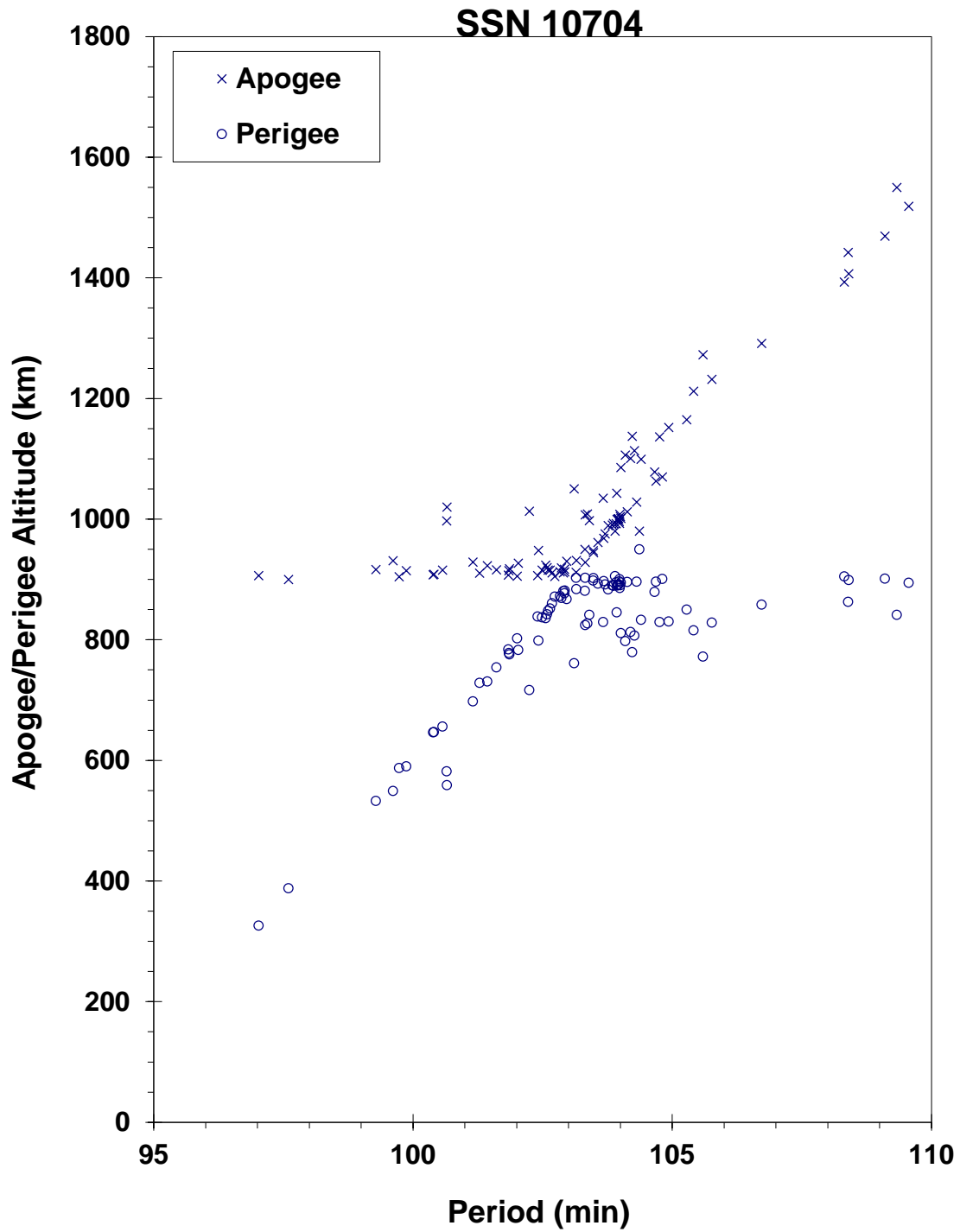
REFERENCE DOCUMENTS

Explosion of Satellite 10704 and other Delta Second Stage Rockets, J.R. Gabbard, Technical Memorandum 81-5, DCS Plans, Hdqtrs NORAD/ADCOM, Colorado Springs, May 1981.

Analysis of PARCS Recorded Data on the Breakup of Satellite 10704 on 27 January 1981, S.F. Hoffman and P.P. Shinkunas, Technical Report MSB82-ADC-0138, Teledyne Brown Engineering, Huntsville, February 1982.

Investigation of Delta Second Stage On-Orbit Explosions, C.S. Gumpel, Report MDC-H0047, McDonnell Douglas Astronautics Company - West, Huntington Beach, April 1982.

A Later Look at Delta Second Stage On-Orbit Explosions, J.R. Gabbard, Technical Report CS85-BMDSC-00-24, Teledyne Brown Engineering, Colorado Springs, March 1985.



Landsat 3 R/B debris cloud of 90 identified fragments 4 days after the event as reconstructed from the US SSN database.

SATELLITE DATA

| | |
|-------------------|--|
| TYPE: | Payload |
| OWNER: | CIS |
| LAUNCH DATE: | 6.13 Sep 1978 |
| DRY MASS (KG): | 1250 |
| MAIN BODY: | Cylinder; 1.7 m diameter by 2 m length |
| MAJOR APPENDAGES: | Solar panels (?) |
| ATTITUDE CONTROL: | Active, 3-axis |
| ENERGY SOURCES: | On-board propellants, explosive charge |

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 10 Oct 1978 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 78277.19859350 | MEAN ANOMALY: | 4.9827 |
| RIGHT ASCENSION: | 336.7676 | MEAN MOTION: | 2.00213289 |
| INCLINATION: | 62.8388 | MEAN MOTION DOT/2: | .0 |
| ECCENTRICITY: | .7350882 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 318.4262 | BSTAR: | .0 |

DEBRIS CLOUD DATA

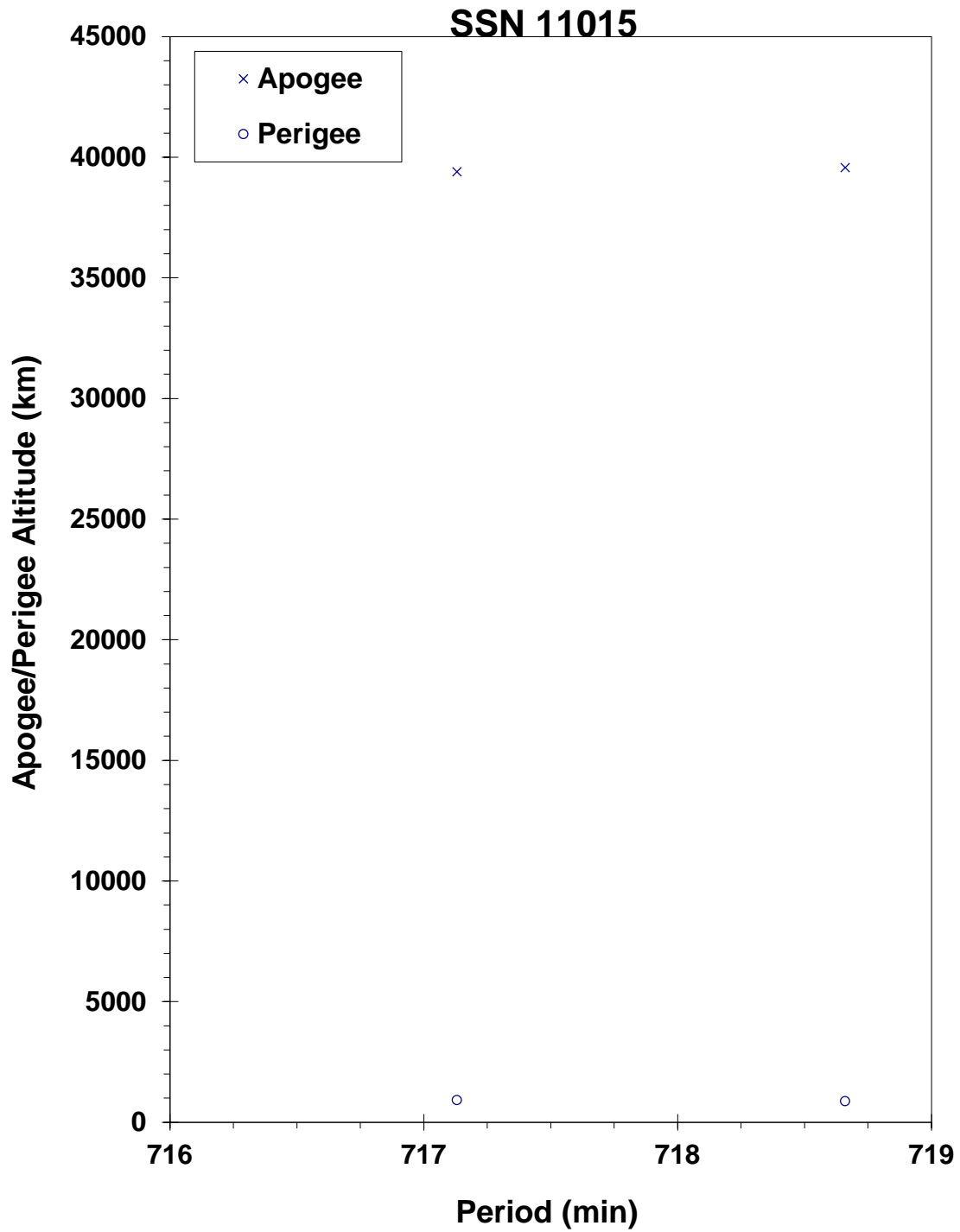
| | |
|----------------------|---------|
| MAXIMUM ΔP : | Unknown |
| MAXIMUM ΔI : | Unknown |

COMMENTS

Cosmos 1030 was another spacecraft of the Cosmos 862-type to experience a fragmentation. After entering a Molniya-type transfer orbit on 6 September, Cosmos 1030 maneuvered about 14 September to enter an operational orbit. Elements on the first identifiable fragment did not appear until a year after the event. Official cataloging of debris did not begin until 3 years after the event.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1030 and a single debris fragment 1 year after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Delta Second Stage (2910)
OWNER: US
LAUNCH DATE: 24.34 Oct 1978
DRY MASS (KG): 900
MAIN BODY: Cylinder-nozzle; 2.4 m diameter by 8 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: On-board propellants, range safety device

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 26 Dec 1981 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|--------------|
| EPOCH: | 81360.19972720 | MEAN ANOMALY: | 311.8261 |
| RIGHT ASCENSION: | 277.7553 | MEAN MOTION: | 13.85390161 |
| INCLINATION: | 99.3003 | MEAN MOTION DOT/2: | .000000425 |
| ECCENTRICITY: | .0010821 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 48.3801 | BSTAR: | .00004426123 |

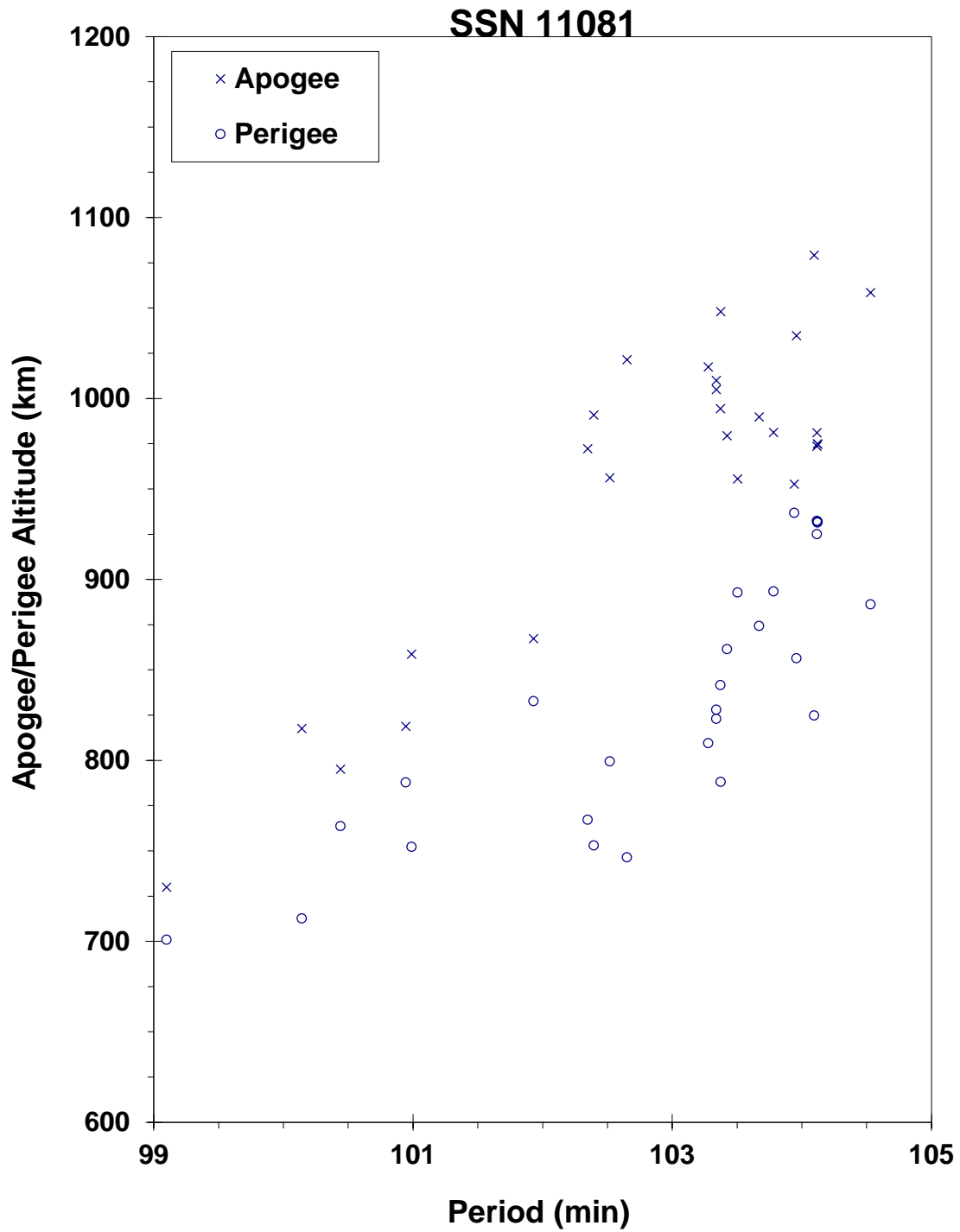
DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : 0.6 deg*

*Based on uncataloged debris data

COMMENTS

Nimbus 7 R/B is designated Cameo in US Space Command Satellite Catalog in reference to scientific piggy-back payload attached to the Delta second stage. This satellite experienced an anomalous event prior to and after the event cited above (See Section 3). Most fragments decayed very rapidly, preventing an accurate assessment of the event and its resulting debris cloud. No new objects were cataloged as a result of this event. The event apparently occurred prior to 0700 GMT.



The Nimbus 7 R/B debris cloud remnant of 27 fragments a few days after the event as reconstructed from the US SSN database. Most fragments have already experienced considerable drag effects.

SATELLITE DATA

TYPE: Tsyklon Third Stage
OWNER: CIS
LAUNCH DATE: 26.29 Oct 1978
DRY MASS (KG): 1360
MAIN BODY: Cone-cylinder; 2.1 m diameter by 3.3 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: Unknown

EVENT DATA

DATE: 9 May 1988
TIME: 1218 GMT
ALTITUDE: 1705 km
LOCATION: 29S, 126E (dsc)
ASSESSED CAUSE: Propulsion

PRE-EVENT ELEMENTS

EPOCH: 88121.02005933
RIGHT ASCENSION: 359.3059
INCLINATION: 82.5543
ECCENTRICITY: .0011463
ARG. OF PERIGEE: 81.1553
MEAN ANOMALY: 279.0818
MEAN MOTION: 11.97080974
MEAN MOTION DOT/2: .000000208
MEAN MOTION DOT DOT/6: .0
BSTAR: .0

DEBRIS CLOUD DATA

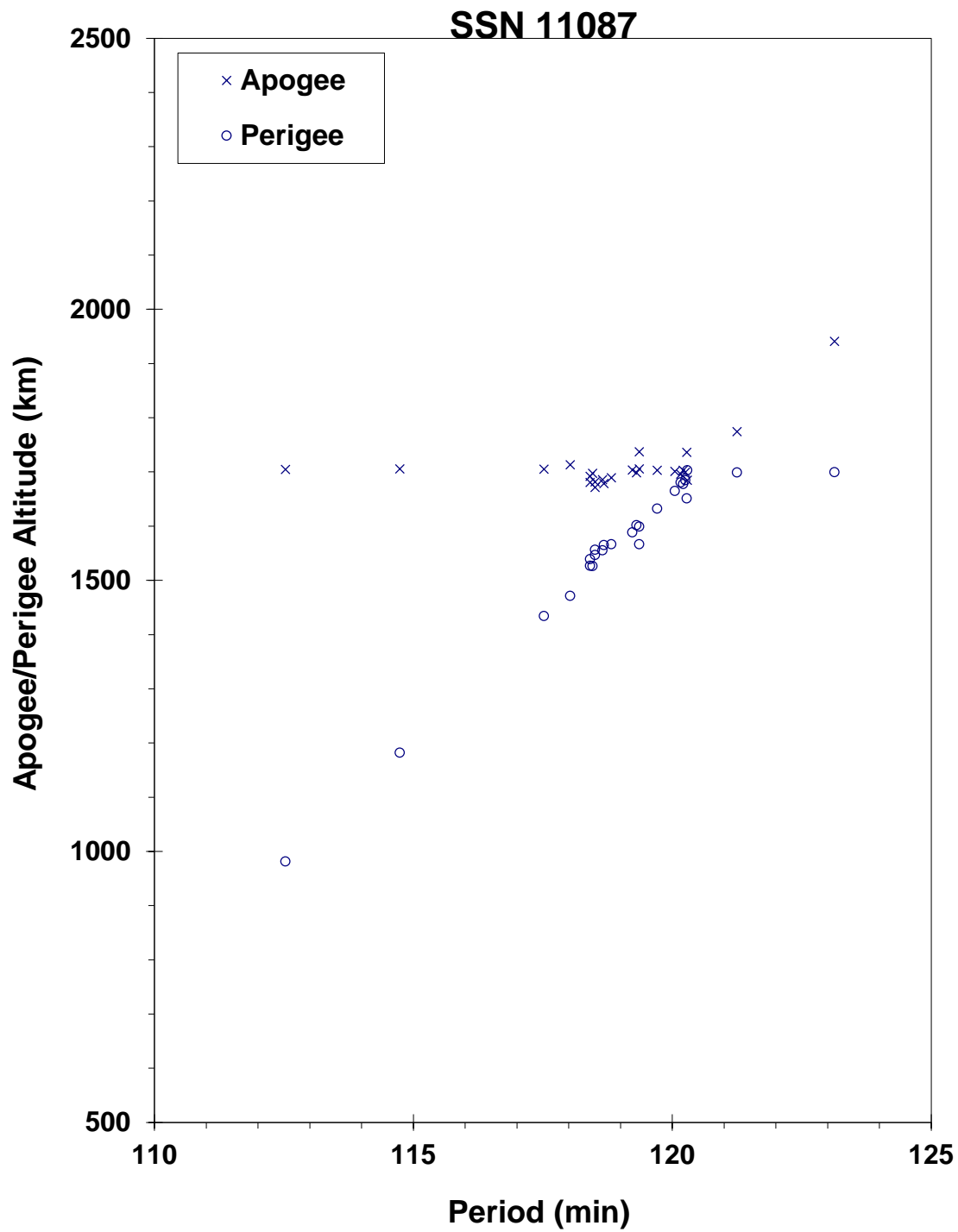
MAXIMUM ΔP : 7.8 min
MAXIMUM ΔI : 0.9 deg

COMMENTS

This flight, which successfully carried three separate payloads, was the fifth orbital mission of the Tsyklon third stage. Propellants used were N_2O_4 and UDMH. Nearly 10 years elapsed from launch to breakup. A second Tsyklon third stage breakup after a similar length of time; see 1987-068B.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1045 R/B debris cloud of 25 fragments as determined 1 week after the event.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 24.35 Feb 1979
DRY MASS (KG): 850
MAIN BODY: Cylinder; 2.1 m diameter by 1.3 m length
MAJOR APPENDAGES: 1 solar panel
ATTITUDE CONTROL: Spin-stabilized
ENERGY SOURCES: None

EVENT DATA

DATE: 13 Sep 1985 LOCATION: 35N, 234E (asc)
TIME: 2043 GMT ASSESSED CAUSE: Deliberate
ALTITUDE: 525 km

PRE-EVENT ELEMENTS

EPOCH: 85256.72413718 MEAN ANOMALY: 260.9644
RIGHT ASCENSION: 182.5017 MEAN MOTION: 15.11755304
INCLINATION: 97.6346 MEAN MOTION DOT/2: .00000616
ECCENTRICITY: .0022038 MEAN MOTION DOT DOT/6: .0
ARG. OF PERIGEE: 99.4081 BSTAR: .000037918

DEBRIS CLOUD DATA

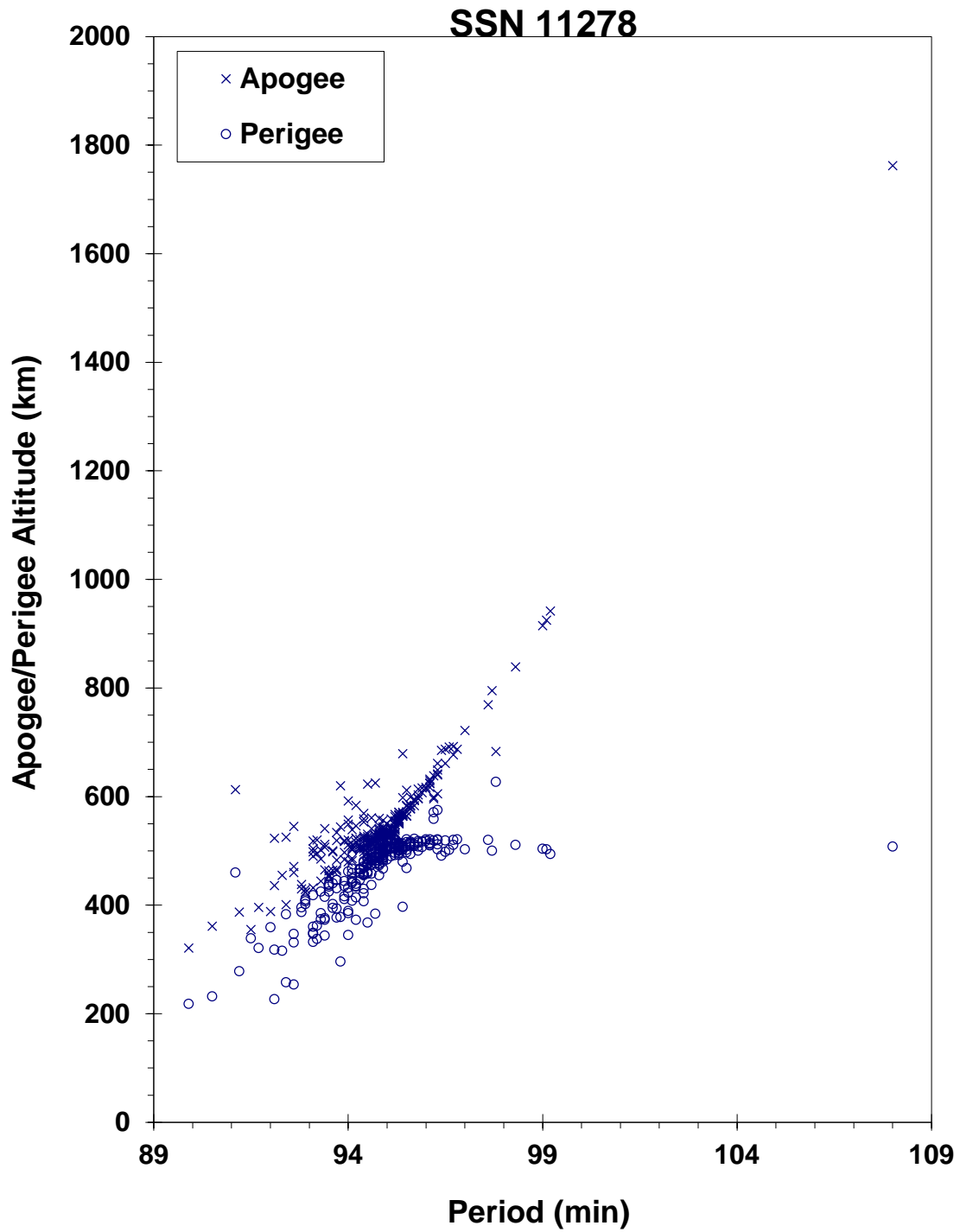
MAXIMUM ΔP : 12.7 min
MAXIMUM ΔI : 1.4 deg

COMMENTS

P-78 was impacted by a sub-orbital object at high velocity as part of a planned test.

REFERENCE DOCUMENT

Postmortem of a Hypervelocity Impact: Summary, R. L. Kling, Technical Report CS86-LKD-001, Teledyne Brown Engineering, Colorado Springs, September 1986.



P-78 debris cloud of 267 fragments seen 11 hours after the event by the US SSN PARCS radar.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 18.50 Apr 1979
DRY MASS (KG): 3000
MAIN BODY: Cylinder; 1.3 m diameter by 17 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge (?)

EVENT DATA

| | | | |
|-----------|-------------|-----------------|-----------------|
| DATE: | 17 Sep 1979 | LOCATION: | 53S, 336E (dsc) |
| TIME: | 1039 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 385 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 79260.33615661 | MEAN ANOMALY: | 61.9566 |
| RIGHT ASCENSION: | 271.8638 | MEAN MOTION: | 15.58096051 |
| INCLINATION: | 65.0398 | MEAN MOTION DOT/2: | .00102640 |
| ECCENTRICITY: | .0016936 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 297.9871 | BSTAR: | .0013492 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 7.1 min*
MAXIMUM ΔI : 0.3 deg*

*Based on uncataloged debris data

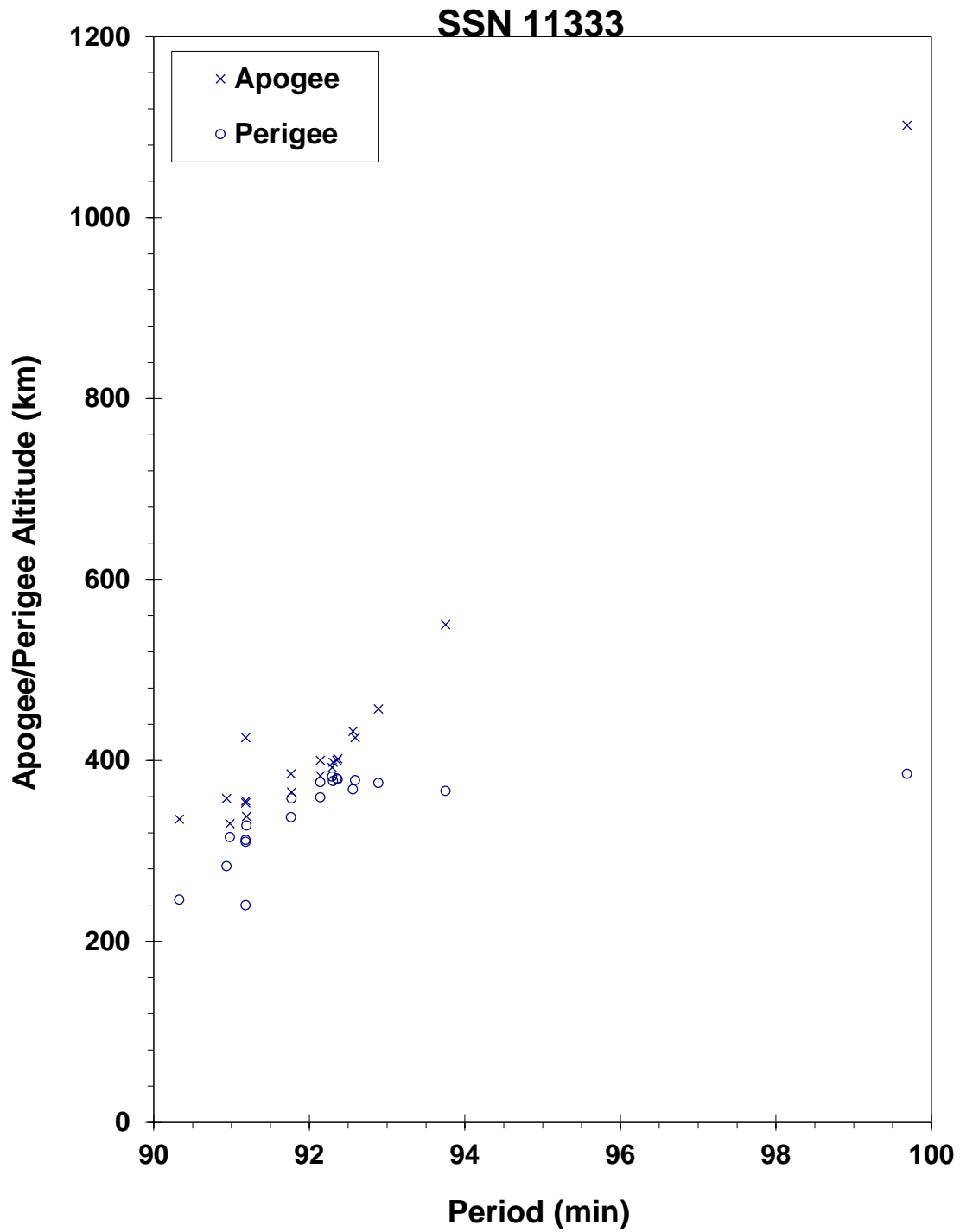
COMMENTS

Cosmos 1094 was the fourth spacecraft of the Cosmos 699-type to experience a fragmentation. Spacecraft had been in a regime of natural decay for 4 months prior to the event. All new debris decayed before being officially cataloged.

REFERENCE DOCUMENTS

"Artificial Satellite Break-Ups (Part 1): Soviet Ocean Surveillance Satellites", N. L. Johnson, Journal of the British Interplanetary Society, February 1983, pp. 51-58.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1094 debris cloud of 20 fragments within 1 week of the event as reconstructed from the US SSN database.

SATELLITE DATA

| | |
|-------------------|--|
| TYPE: | Payload |
| OWNER: | CIS |
| LAUNCH DATE: | 27.76 Jun 1979 |
| DRY MASS (KG): | 1250 |
| MAIN BODY: | Cylinder; 1.7 m diameter by 2 m length |
| MAJOR APPENDAGES: | Solar panels |
| ATTITUDE CONTROL: | Active, 3-axis |
| ENERGY SOURCES: | On-board propellants, explosive charge |

EVENT DATA

| | | | |
|-----------|--------------|-----------------|------------|
| DATE: | Mid-Feb 1980 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 80048.26161234 | MEAN ANOMALY: | 5.0375 |
| RIGHT ASCENSION: | 104.4713 | MEAN MOTION: | 2.00453352 |
| INCLINATION: | 63.3495 | MEAN MOTION DOT/2: | .0 |
| ECCENTRICITY: | .7238911 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 318.4445 | BSTAR: | .0 |

DEBRIS CLOUD DATA

| | |
|----------------------|----------|
| MAXIMUM ΔP : | 3.5 min* |
| MAXIMUM ΔI : | 0.2 deg* |

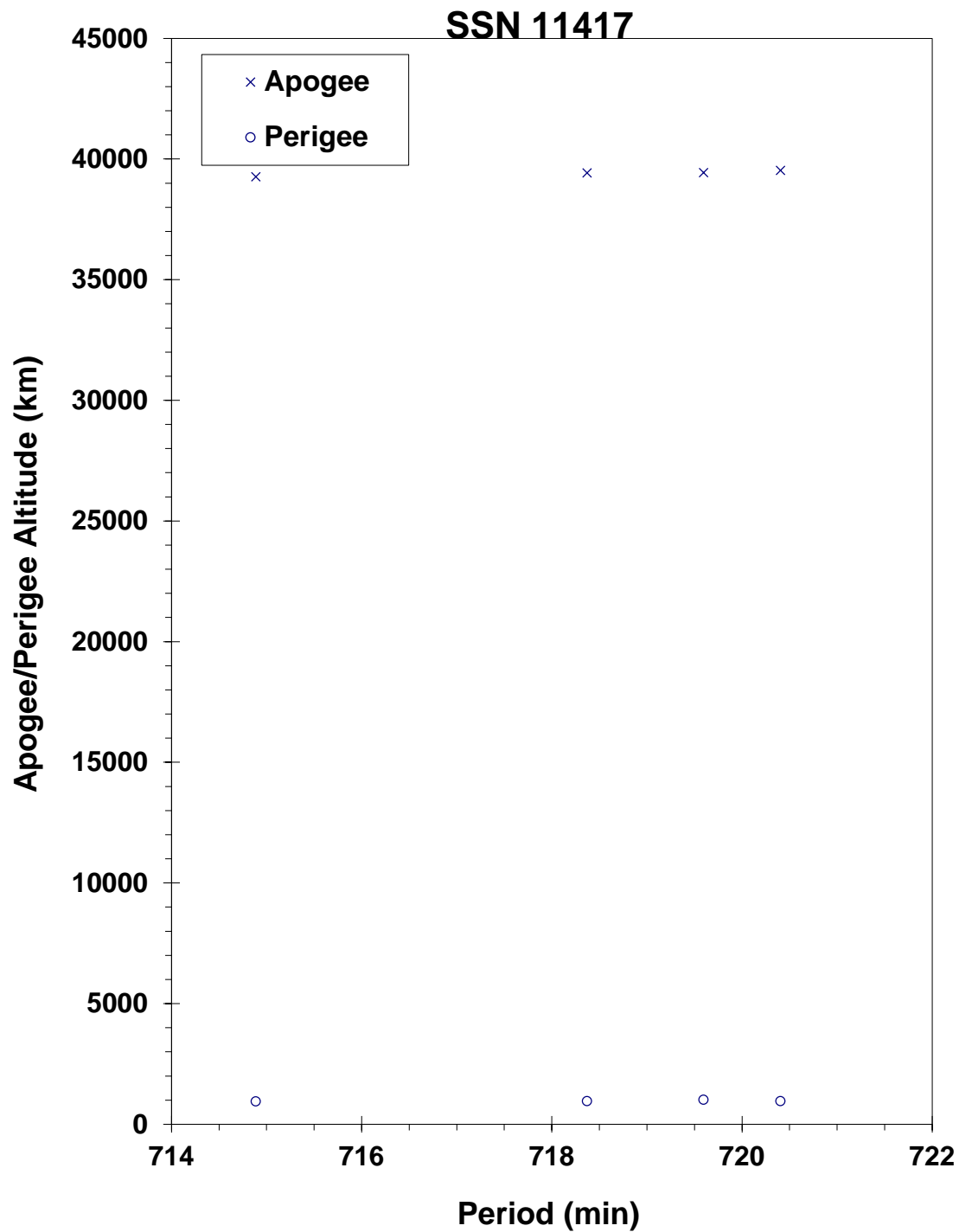
*Based on uncataloged debris data

COMMENTS

Cosmos 1109 was another spacecraft of the Cosmos 862-type to experience a fragmentation. Cosmos 1109 maneuvered into an operational orbit about 19 July. The payload was "lost" after 17 February 1980 and three pieces of debris were soon found that could be traced back to that period.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1109 and three fragments in February 1980 as reconstructed
from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 28.01 Aug 1979
DRY MASS (KG): 1250
MAIN BODY: Cylinder; 1.7 m diameter by 2 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

DATE: 9 Sep 1979
TIME: 0230 GMT
ALTITUDE: 8375 km
LOCATION: 52N, 304E (asc)
ASSESSED CAUSE: Deliberate

PRE-EVENT ELEMENTS

EPOCH: 79249.09448656
RIGHT ASCENSION: 288.1742
INCLINATION: 63.0212
ECCENTRICITY: .7383335
ARG. OF PERIGEE: 318.3799
MEAN ANOMALY: 3.7678
MEAN MOTION: 2.00548359
MEAN MOTION DOT/2: .0
MEAN MOTION DOT DOT/6: .0
BSTAR: .0

DEBRIS CLOUD DATA

MAXIMUM ΔP : 4.0 min*
MAXIMUM ΔI : 0.1 deg*

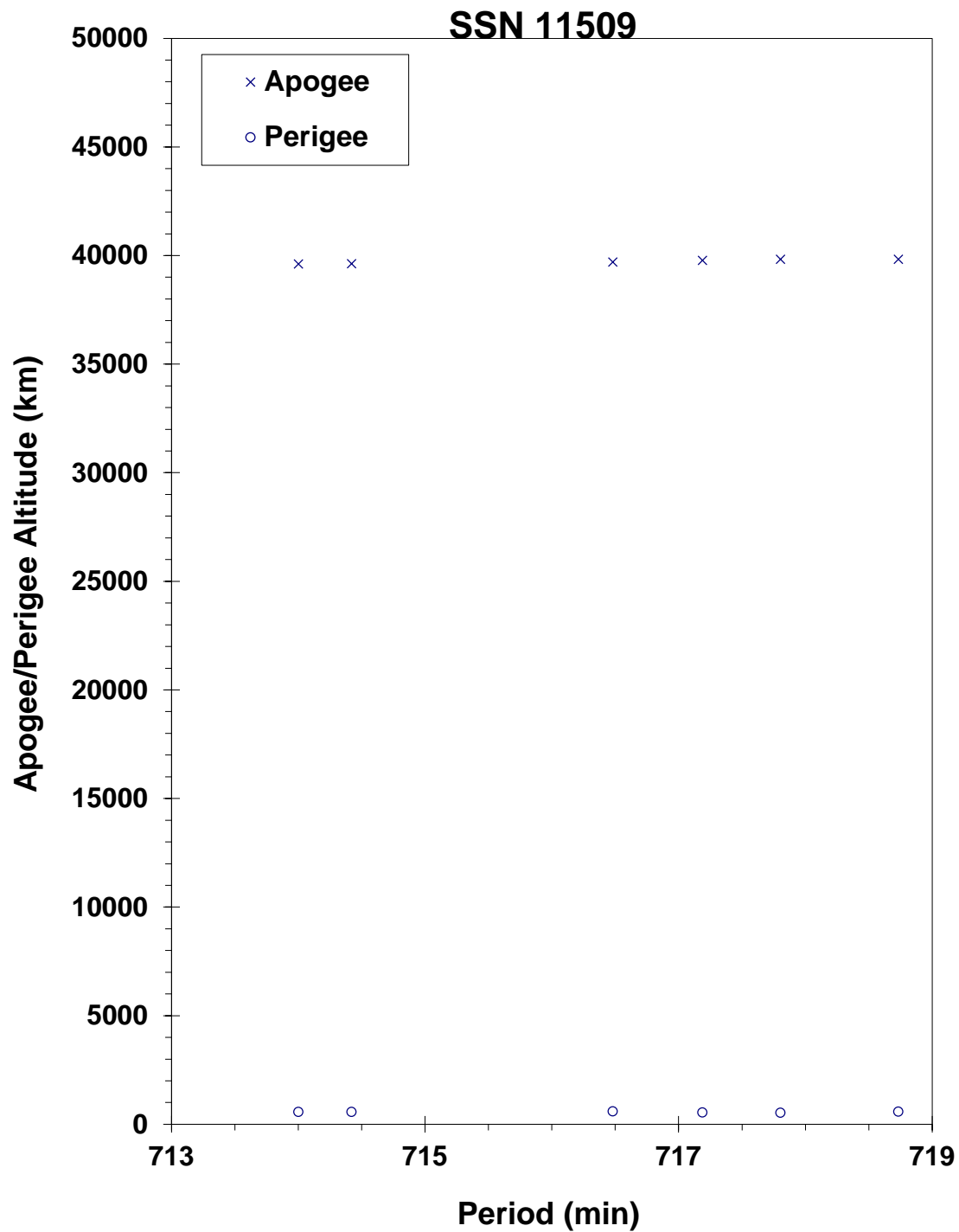
*Based on uncataloged debris data

COMMENTS

Cosmos 1124 was another spacecraft of the Cosmos 862-type to experience a fragmentation. After insertion into a Molniya-type transfer orbit on 28 August, Cosmos 1124's ascending node was allowed to drift until 3 September when a maneuver placed the spacecraft into an operational, semi-synchronous orbit. The fragmentation occurred 6 days later. The spacecraft never maneuvered again and soon drifted off station.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1124 debris cloud of 6 fragments about 1 week after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 7.07 Dec 1979
DRY MASS (KG): 463
MAIN BODY: Box; 1.2 m length by 1.62 m length by 1.17 m height
MAJOR APPENDAGES: 2 Solar Panels, antenna/feeds
ATTITUDE CONTROL: Active, 3 axis
ENERGY SOURCES: On-board solid AKM and liquid monopropellants

EVENT DATA

| | | | |
|-----------|---------------|-----------------|--------------------|
| DATE: | ~ 11 Dec 1979 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion (solid) |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 79345.24087748 | MEAN ANOMALY: | 174.6228 |
| RIGHT ASCENSION: | 276.7452 | MEAN MOTION: | 2.28199220 |
| INCLINATION: | 23.7341 | MEAN MOTION DOT/2: | .00044295 |
| ECCENTRICITY: | .7298759 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 181.2679 | BSTAR: | .000009999 |

POST-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 82327.71145422 | MEAN ANOMALY: | 349.7392 |
| RIGHT ASCENSION: | 164.2803 | MEAN MOTION: | 1.82544069 |
| INCLINATION: | 8.1767 | MEAN MOTION DOT/2: | .00000069 |
| ECCENTRICITY: | .4789421 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 31.9453 | BSTAR: | .0 |

DEBRIS CLOUD DATA

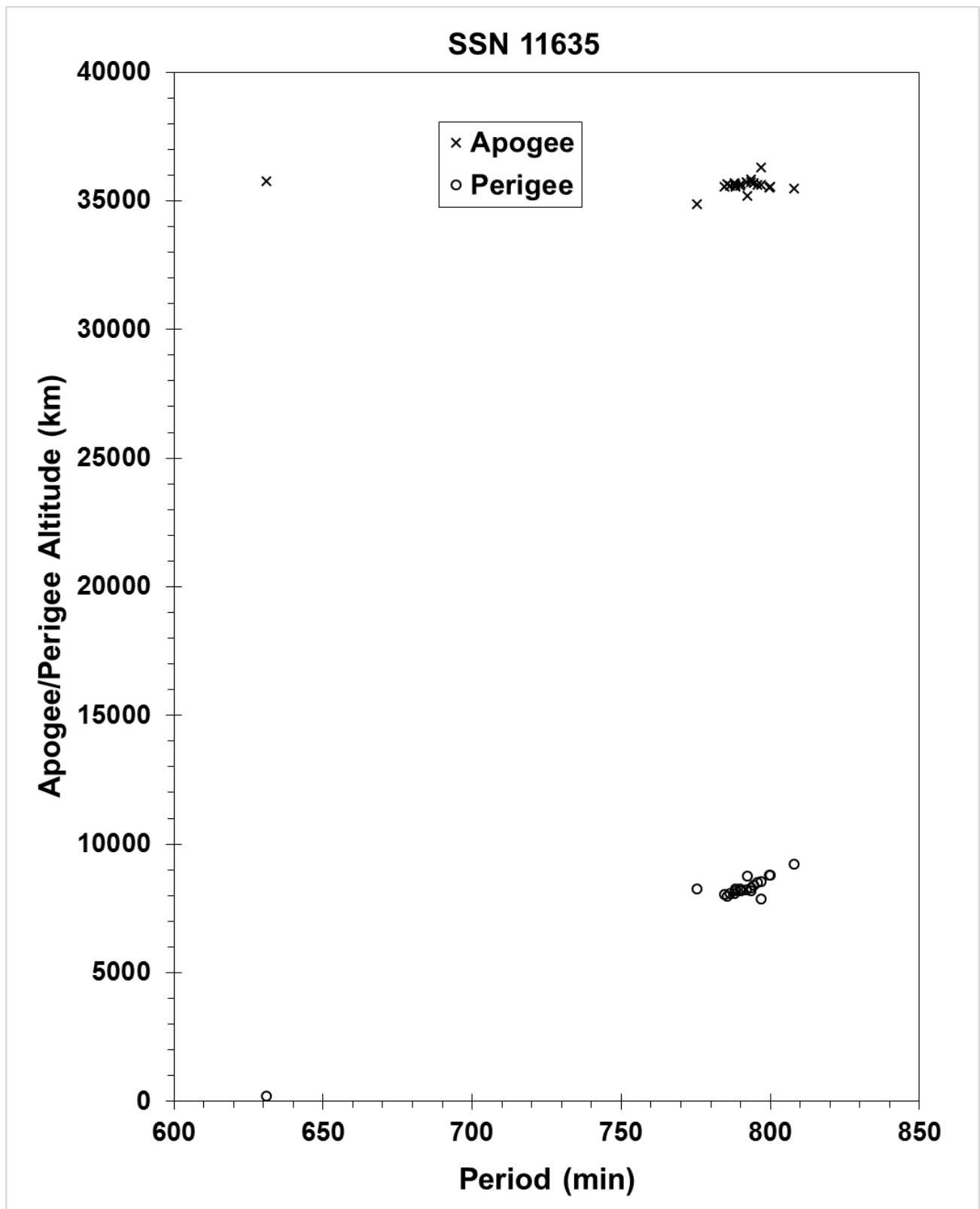
MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

Detected by software.

This mission was the third launch of the RCA SATCOM series. Communications with payload was lost upon firing of Aerojet SVM-7 solid apogee kick motor. Detection and tracking of debris has always been extremely difficult in part due to low inclination and highly elliptical orbit. Debris entered the SSN catalog in 2006 and after. The magnitude of the event and the total number of pieces created are unknown.

REFERENCE DOCUMENT



SATCOM 3 debris cloud of 22 fragments (including SATCOM 3) over 26 years after the event as reconstructed from the US SSN database.

CAT R/B

1979-104B

11659

SATELLITE DATA

TYPE: Ariane 1 Third Stage
OWNER: ESA
LAUNCH DATE: 24.72 Dec 1979
DRY MASS (KG): 1400
MAIN BODY: Cylinder; 2.6 m diameter by 10.3 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: On-board propellants, range safety device

EVENT DATA

| | | | |
|-----------|----------|-----------------|------------|
| DATE: | Apr 1980 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 80088.55565320 | MEAN ANOMALY: | 17.6019 |
| RIGHT ASCENSION: | 101.5521 | MEAN MOTION: | 2.48253031 |
| INCLINATION: | 17.9092 | MEAN MOTION DOT/2: | .001764977 |
| ECCENTRICITY: | .7152375 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 264.7858 | BSTAR: | .001078542 |

DEBRIS CLOUD DATA

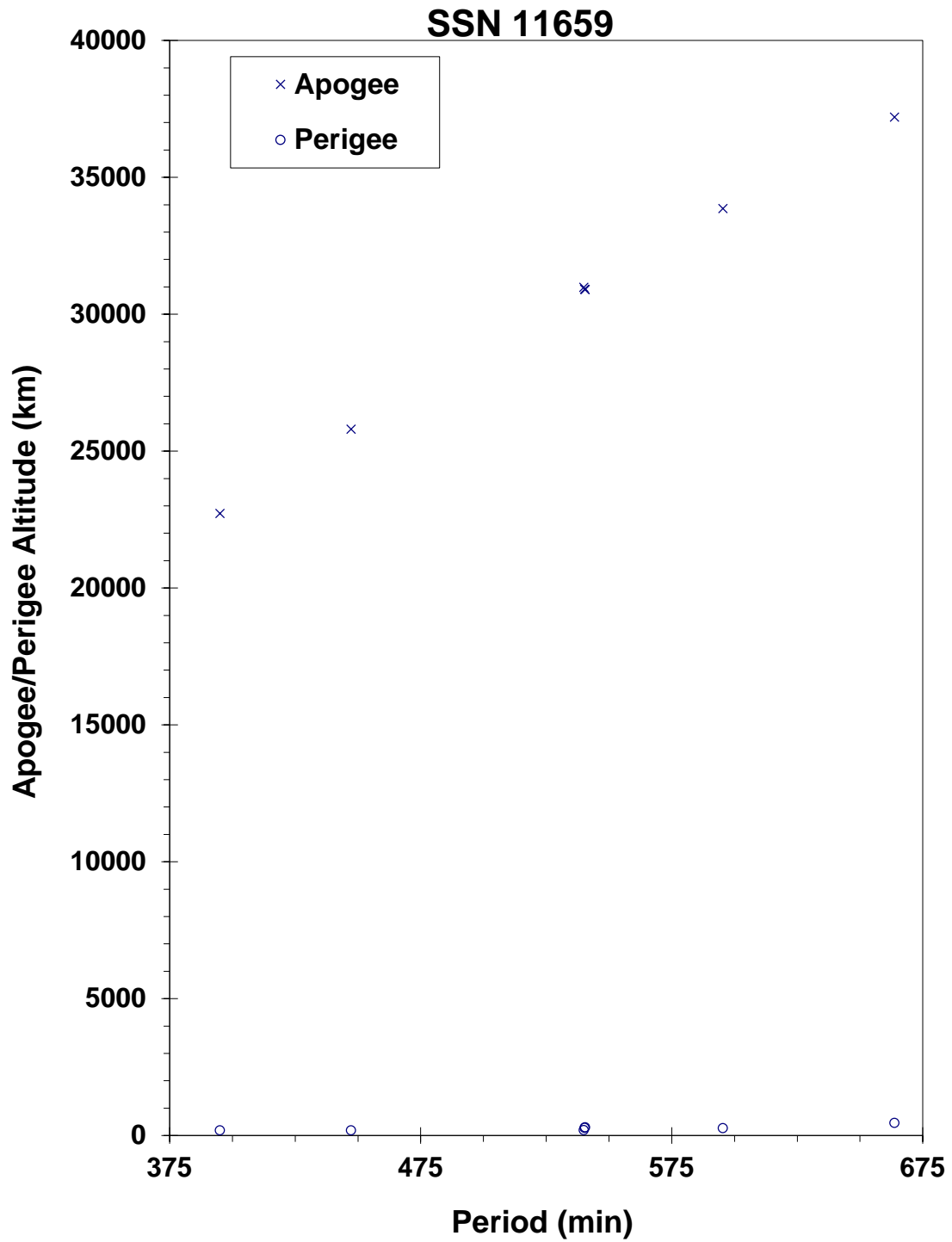
MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

This mission was the inaugural flight of the Ariane 1 launch vehicle. Payload and R/B were apparently cross-tagged until mid-January 1980. Detection and tracking of debris has always been extremely difficult in part due to low inclination and highly elliptical orbit. Debris data were first developed in the second half of April, and calculations suggest the fragmentation occurred during the first week of April. The magnitude of the event and the total number of pieces created are unknown. Many debris had high decay rates.

REFERENCE DOCUMENT

A Preliminary Analysis of the Fragmentation of the Spot 1 Ariane Third Stage, N. L. Johnson, Technical Report CS87-LKD-003, Teledyne Brown Engineering, Colorado Springs, March 1987.



CAT R/B debris cloud of 7 fragments about 8 weeks after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 14.44 Mar 1980
DRY MASS (KG): 3000
MAIN BODY: Cylinder; 1.3 m diameter by 17 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge (?)

EVENT DATA

DATE: 15 Jul 1981 LOCATION: 10N, 106E (asc)
TIME: 0921 GMT ASSESSED CAUSE: Unknown
ALTITUDE: 430 km

PRE-EVENT ELEMENTS

EPOCH: 81196.19449955 MEAN ANOMALY: 110.8351
RIGHT ASCENSION: 174.9184 MEAN MOTION: 15.54665775
INCLINATION: 65.0101 MEAN MOTION DOT/2: .00025375
ECCENTRICITY: .0068471 MEAN MOTION DOT DOT/6: .0
ARG. OF PERIGEE: 248.6139 BSTAR: .00034595

DEBRIS CLOUD DATA

MAXIMUM ΔP : 1.0 min*
MAXIMUM ΔI : 0.5 deg*

*Based on uncataloged debris data

COMMENTS

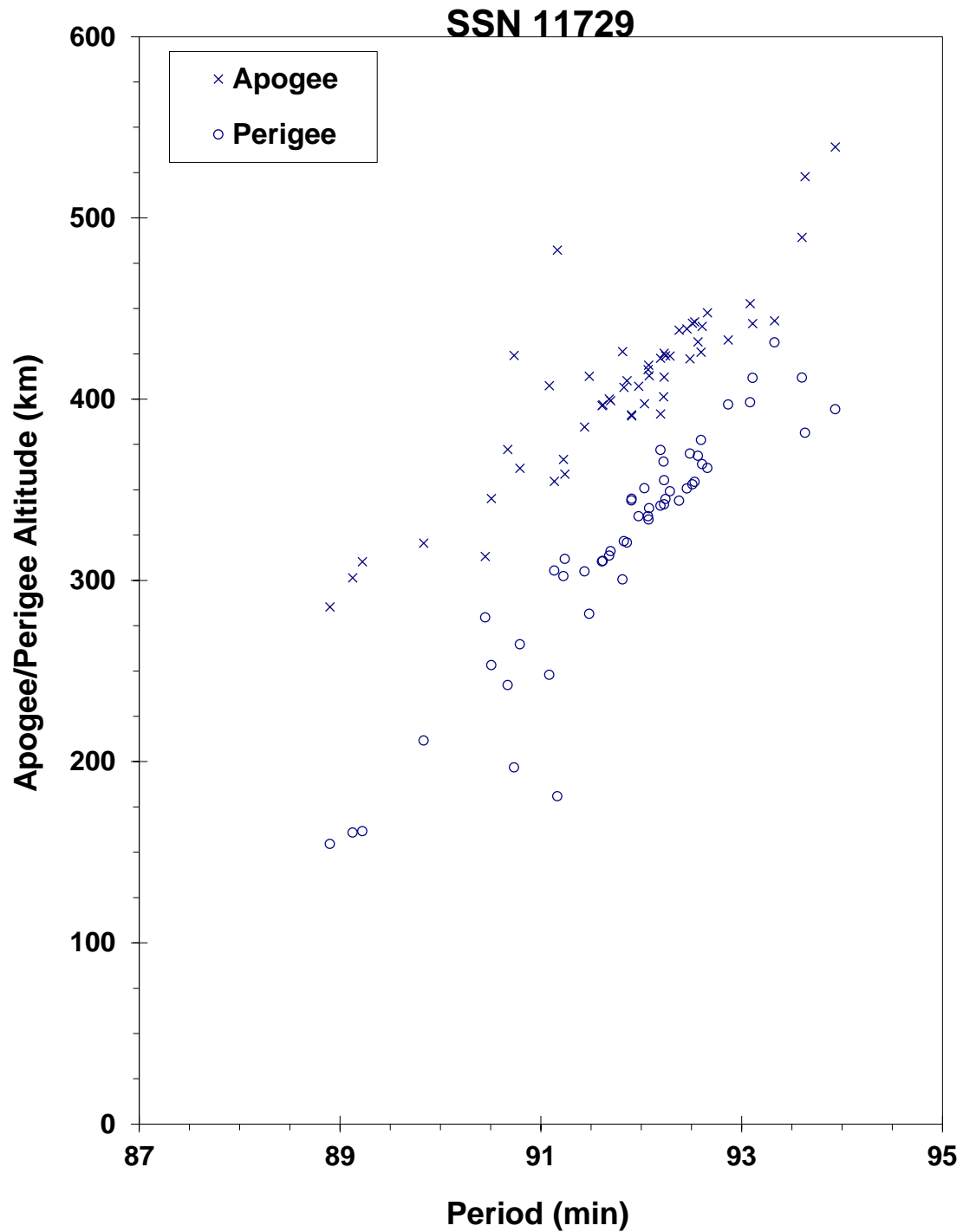
Cosmos 1167 was the fifth spacecraft of the Cosmos 699-type to experience a fragmentation. The spacecraft had been in a regime of natural decay for 3 months prior to the event. Most debris reentered before being officially cataloged.

REFERENCE DOCUMENTS

The Fragmentations of USSR Satellites 11729 and 12504 (U), J. R. Gabbard and P. M. Landry, Technical Memorandum 82-S-03, DCS/Plans, Hdqtrs NORAD/ADCOM, Colorado Springs, August 1982 (Secret).

"Artificial Satellite Break-Ups (Part 1): Soviet Ocean Surveillance Satellites", N. L. Johnson, Journal of the British Interplanetary Society, February 1983, pp. 51-58.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1167 debris cloud remnant of 53 fragments about 2 weeks after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 18.04 Apr 1980
DRY MASS (KG): 1400
MAIN BODY: Cylinder; 2 m diameter by 4 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

| | | | |
|-----------|-------------|-----------------|-----------------|
| DATE: | 18 Apr 1980 | LOCATION: | 47N, 322E (asc) |
| TIME: | 0726 GMT | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | 1625 km | | |

POST-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 80109.51771250 | MEAN ANOMALY: | 102.2095 |
| RIGHT ASCENSION: | 250.9679 | MEAN MOTION: | 13.64414319 |
| INCLINATION: | 66.1153 | MEAN MOTION DOT/2: | .0 |
| ECCENTRICITY: | .0865337 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 248.5294 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 5.4 min
MAXIMUM ΔI : 0.6 deg

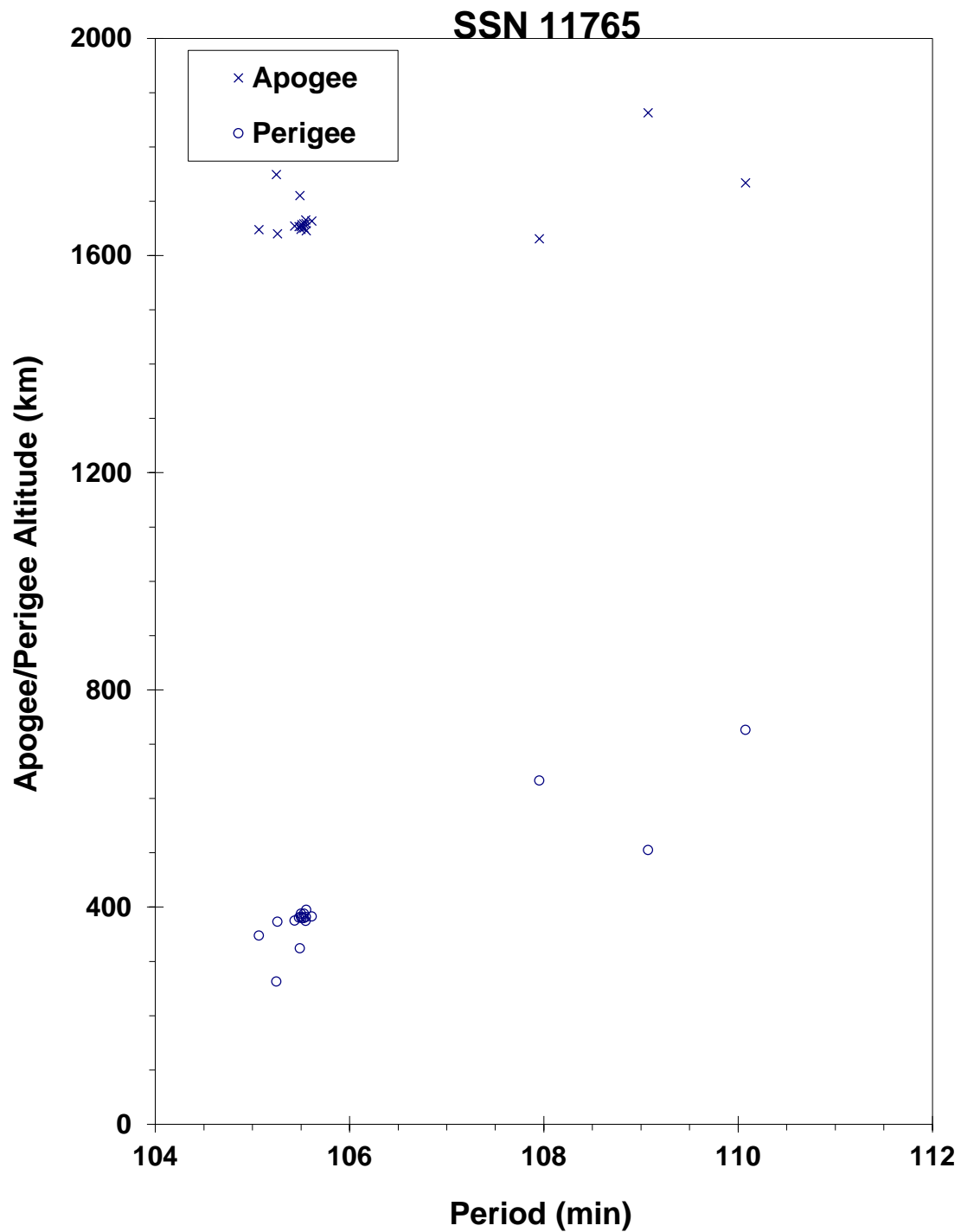
COMMENTS

Cosmos 1174 was launched on a two-revolution rendezvous with Cosmos 1171. After a close approach, Cosmos 1174 performed a final maneuver shortly before its warhead was intentionally fired. Elements above are first data available after the final maneuver but also following the fragmentation. Cosmos 1174 was part of test series begun with Cosmos 249.

REFERENCE DOCUMENTS

"Artificial Satellite Break-Ups (Part 2): Soviet Anti-Satellite Program", N.L. Johnson, Journal of the British Interplanetary Society, August 1983, pp. 357-362.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1174 debris cloud of 18 identified fragments about 10 days after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 14.87 June 1980
DRY MASS (KG): 1250
MAIN BODY: Cylinder; 1.7 m diameter by 2 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive device

EVENT DATA

| | | | |
|-----------|----------------|-----------------|------------|
| DATE: | 26 August 1980 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 80238.74437670 | MEAN ANOMALY: | 4.7027 |
| RIGHT ASCENSION: | 155.4878 | MEAN MOTION: | 2.00554276 |
| INCLINATION: | 62.9033 | MEAN MOTION DOT/2: | .00000217 |
| ECCENTRICITY: | .7321456 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 318.3182 | BSTAR: | .0 |

DEBRIS CLOUD DATA

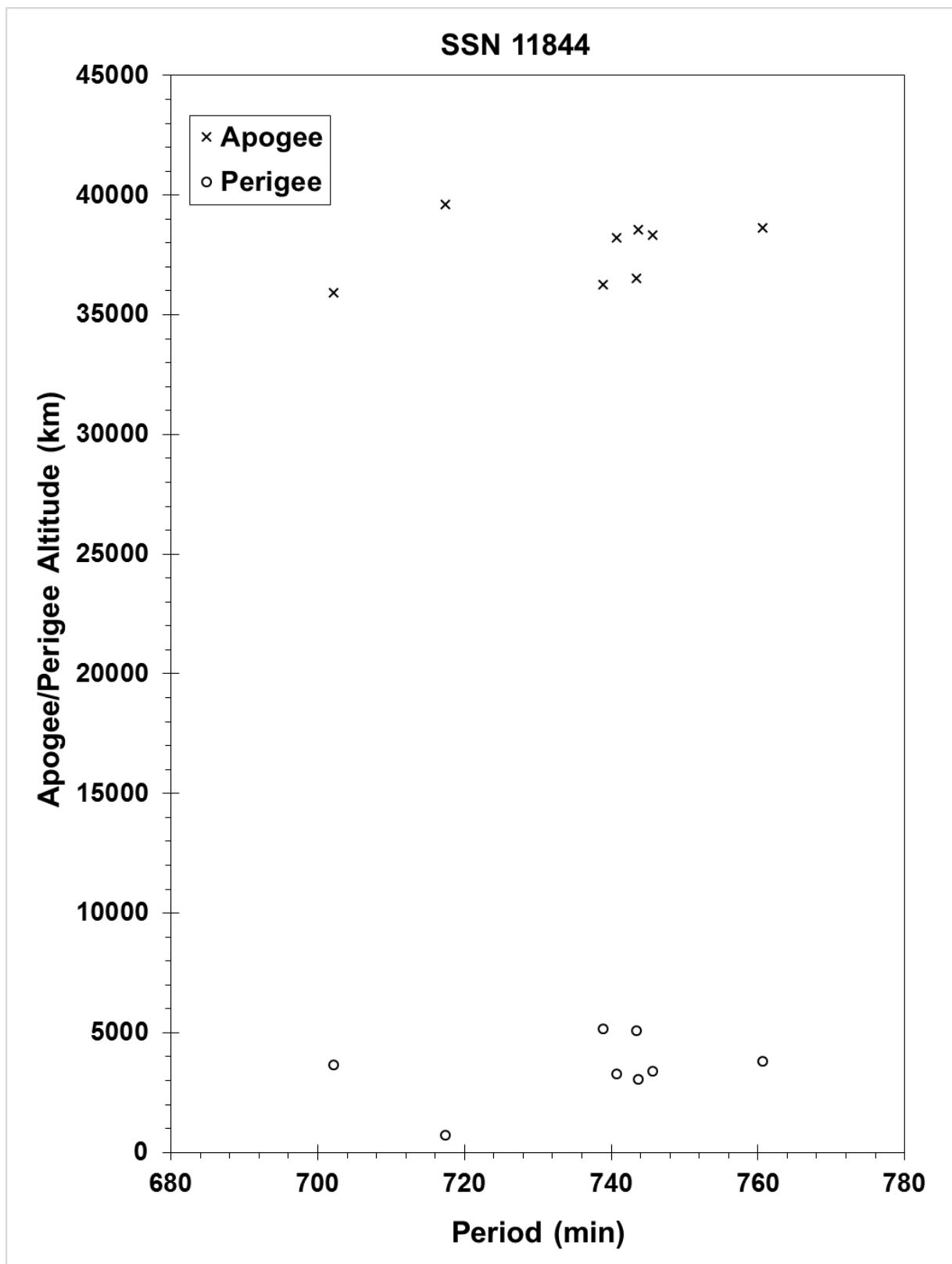
MAXIMUM ΔP : 42.6 min
MAXIMUM ΔI : 7.1 deg

COMMENTS

Cosmos 1188 was a member of the Cosmos 862 class of operational satellites in highly elliptical, semi-synchronous orbits that experienced a total of 16 fragmentations during the period 1977-1986. Due to the nature of these orbits, which result in high altitudes over the Northern Hemisphere where most surveillance sensors are located, debris detection and tracking is extremely difficult. Only the largest fragments can be seen. Cosmos 862 class spacecraft were equipped with self-destruct packages in the event that spacecraft control was lost; this was the cause of breakups until the explosives were removed after Cosmos 1481.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1188 debris cloud; fragments cataloged up to 23 years after the event as reconstructed from the US SSN database.

COSMOS 1191

1980-057A

11871

SATELLITE DATA

| | |
|-------------------|--|
| TYPE: | Payload |
| OWNER: | CIS |
| LAUNCH DATE: | 2.04 Jul 1980 |
| DRY MASS (KG): | 1250 |
| MAIN BODY: | Cylinder; 1.7 m diameter by 2 m length |
| MAJOR APPENDAGES: | Solar panels |
| ATTITUDE CONTROL: | Active, 3-axis |
| ENERGY SOURCES: | On-board propellants, explosive charge |

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 14 May 1981 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 81133.07322634 | MEAN ANOMALY: | 5.1166 |
| RIGHT ASCENSION: | 198.5704 | MEAN MOTION: | 2.00555560 |
| INCLINATION: | 62.6448 | MEAN MOTION DOT/2: | .00001257 |
| ECCENTRICITY: | .7180863 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 319.4330 | BSTAR: | .0 |

DEBRIS CLOUD DATA

| | |
|----------------------|----------|
| MAXIMUM ΔP : | 6.0 min* |
| MAXIMUM ΔI : | 0.1 deg* |

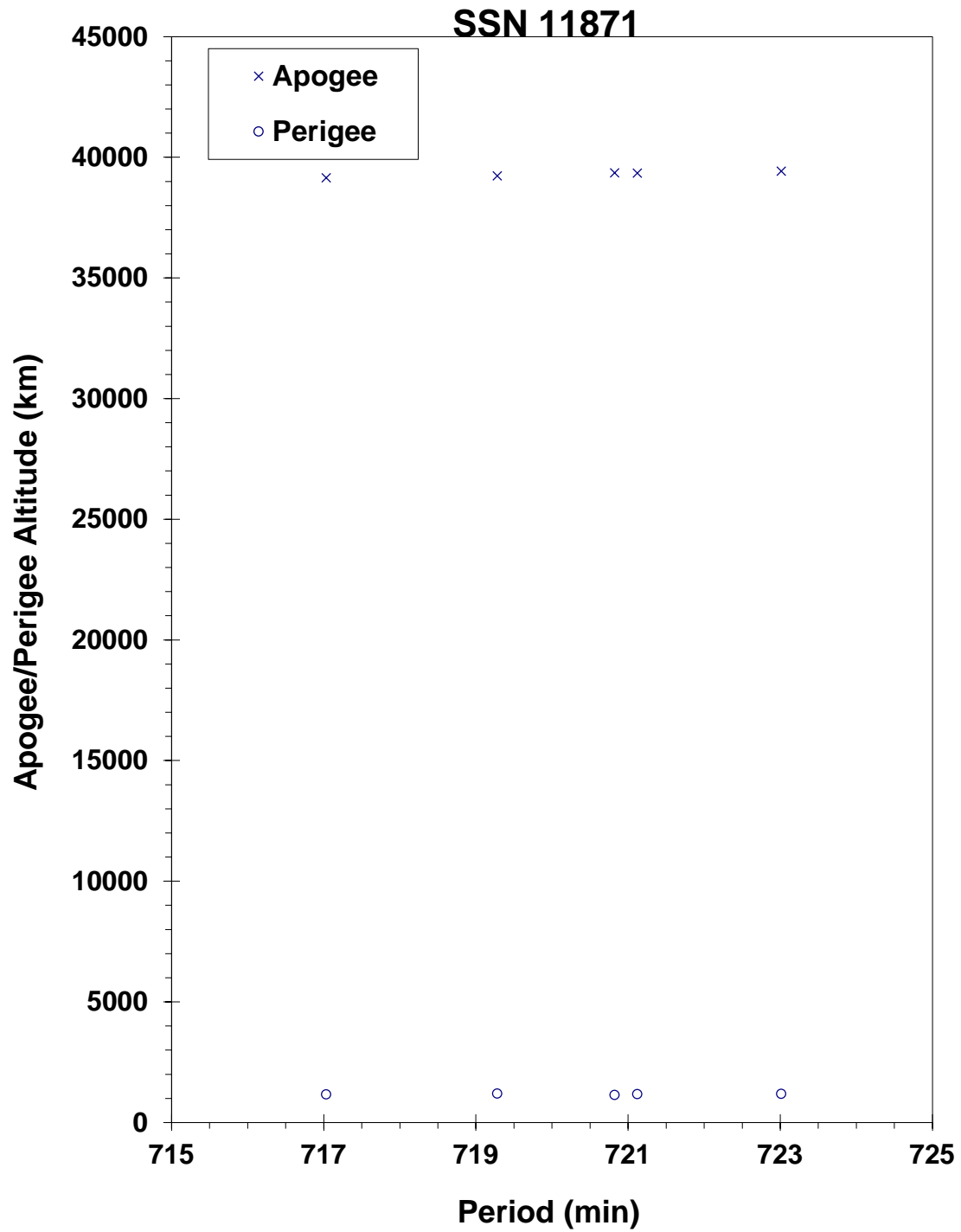
*Based on uncataloged debris data

COMMENTS

Cosmos 1191 was another spacecraft of the Cosmos 862-type to experience a fragmentation. The first debris elements were developed for 25 May.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1191 debris cloud of 5 identified fragments 1 month after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 24.46 Oct 1980
DRY MASS (KG): 1250
MAIN BODY: Cylinder; 1.7 m diameter by 2 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 12 Feb 1983 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 83042.34036514 | MEAN ANOMALY: | 6.0502 |
| RIGHT ASCENSION: | 36.1600 | MEAN MOTION: | 2.00587025 |
| INCLINATION: | 65.2478 | MEAN MOTION DOT/2: | 0.00001154 |
| ECCENTRICITY: | 0.7021051 | MEAN MOTION DOT DOT/6: | 0 |
| ARG. OF PERIGEE: | 314.5975 | BSTAR: | 0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

Cosmos 1217 was another member of the Cosmos 862-type to experience a fragmentation.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentation-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 4.63 Nov 1980
DRY MASS (KG): 3000
MAIN BODY: Cylinder; 1.3 m diameter by 17 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge (?)

EVENT DATA (1)

| | | | |
|-----------|-------------|-----------------|-----------------|
| DATE: | 20 Jun 1982 | LOCATION: | 10S, 332E (dsc) |
| TIME: | 1818 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 875 km | | |

PRE-EVENT ELEMENTS (1)

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 82171.72558670 | MEAN ANOMALY: | 0.2166 |
| RIGHT ASCENSION: | 330.3811 | MEAN MOTION: | 14.49658466 |
| INCLINATION: | 65.0033 | MEAN MOTION DOT/2: | .00000066 |
| ECCENTRICITY: | .0219432 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 357.8883 | BSTAR: | .000025640 |

EVENT DATA (2)

| | | | |
|-----------|-------------|-----------------|-----------------|
| DATE: | 25 Aug 1982 | LOCATION: | 65S, 238E (dsc) |
| TIME: | 1231 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 665 km | | |

PRE-EVENT ELEMENTS (2)

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 82230.91714195 | MEAN ANOMALY: | 22.7965 |
| RIGHT ASCENSION: | 159.4489 | MEAN MOTION: | 14.49745561 |
| INCLINATION: | 65.0025 | MEAN MOTION DOT/2: | .0 |
| ECCENTRICITY: | .0225583 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 336.3217 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 3.4 min*
MAXIMUM ΔI : 1.8 deg*

*Based on uncataloged debris data

COMMENTS

Cosmos 1220 was the seventh spacecraft of the Cosmos 699-type to experience a fragmentation. The spacecraft had been in a natural decay regime for more than 14 months at the time of the first event. A total of 47 fragments had been officially cataloged by the time of the second event that occurred 2 months later. See similar dual events happening in the summer of 1982 with Cosmos 1306 and Cosmos 1260.

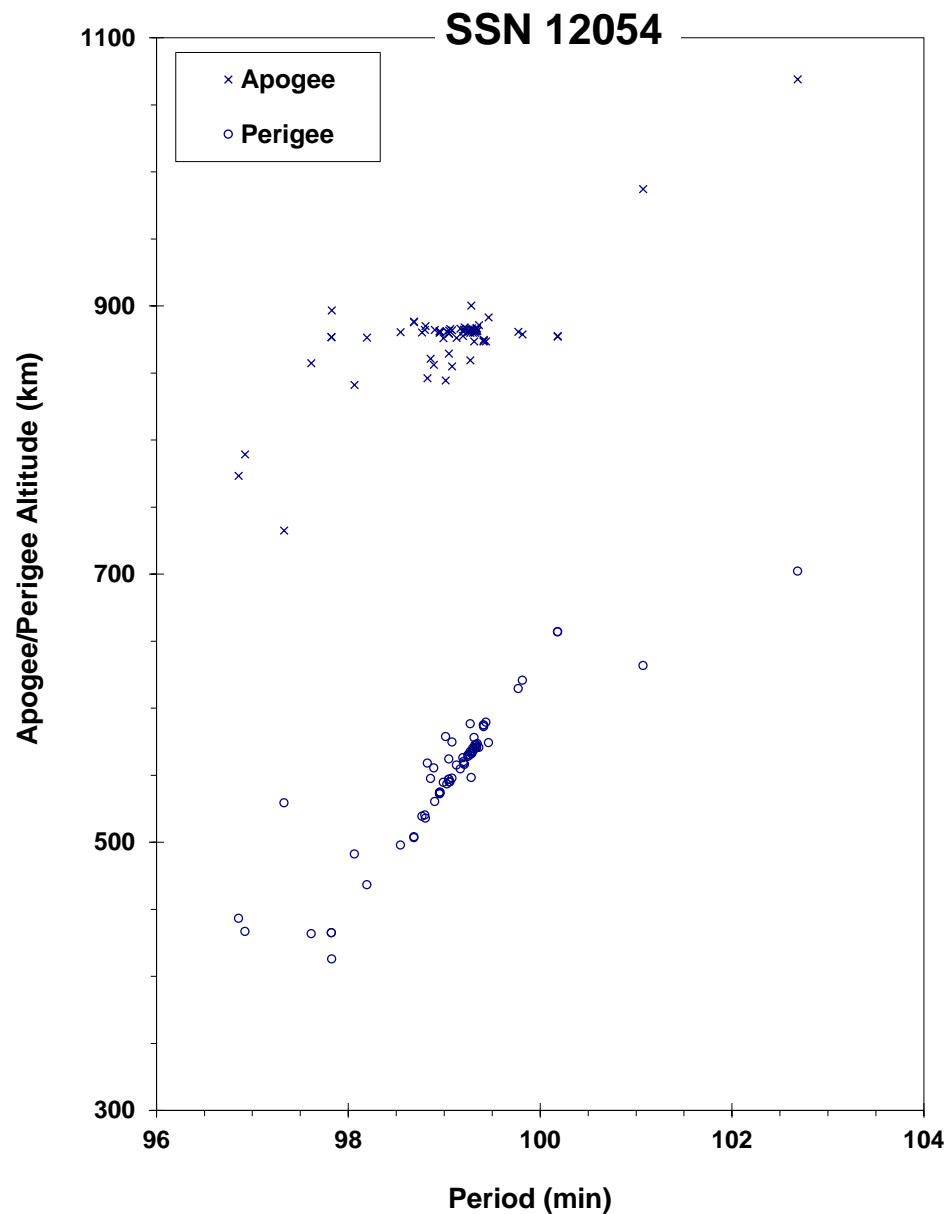
REFERENCE DOCUMENTS

Analysis of PARCS Recorded Data on the Breakup of Satellite 12054, J.W. Rider, Technical Report MSB83-ADC-0162, Teledyne Brown Engineering, Huntsville, January 1983.

Analysis of Cosmos 1220 and Cosmos 1306 Fragments (U), D. Fennessy, Report AH-23, FTD/OLAI, Cheyenne Mountain, Colorado, 12 January 1983 (Secret).

"Artificial Satellite Break-Ups (Part 1): Soviet Ocean Surveillance Satellites", N. L. Johnson, Journal of the British Interplanetary Society, February 1983, pp. 51-58.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1220 debris cloud of 72 fragments about 1 week after the first event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 19.41 Feb 1981
DRY MASS (KG): 1250
MAIN BODY: Cylinder; 1.7 m diameter by 2 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 20 Oct 1981 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 81293.17083627 | MEAN ANOMALY: | 5.0298 |
| RIGHT ASCENSION: | 214.2278 | MEAN MOTION: | 2.00570861 |
| INCLINATION: | 62.9685 | MEAN MOTION DOT/2: | .0 |
| ECCENTRICITY: | .7233048 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 318.2473 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 2.7 min*
MAXIMUM ΔI : 0.4 deg*

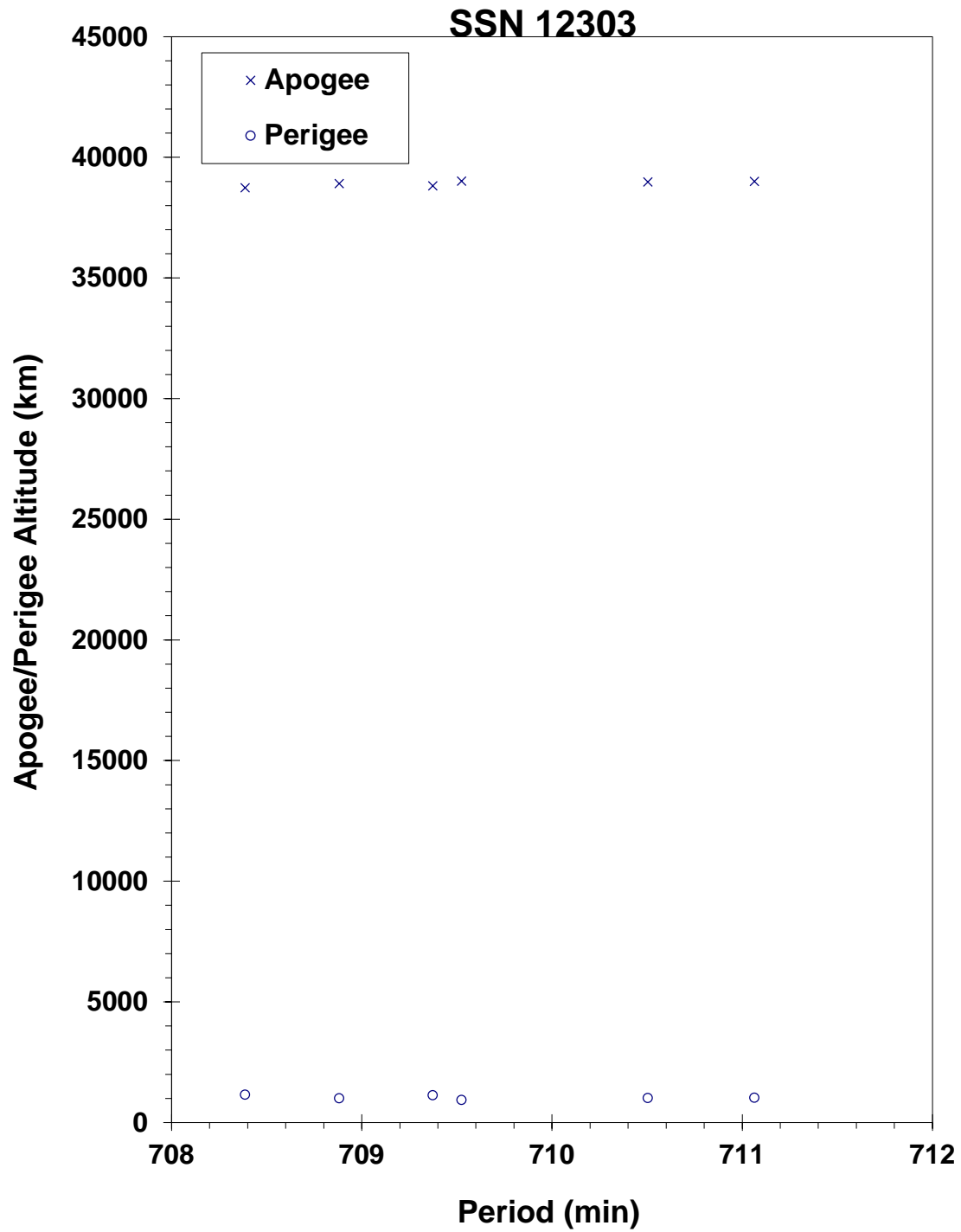
*See comments below

COMMENTS

Cosmos 1247 was another spacecraft of the Cosmos 862-type to experience a fragmentation. Cosmos 1247 appears to have completed the first burn of a 2-phase maneuver sequence on the event date, followed by debris generation. The ΔP and ΔI values above are based on the post-maneuver, 711-minute orbit of 12303 rather than the pre-maneuver, 718-minute orbit cited above.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1247 debris cloud of 6 fragments about 6 weeks after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 20.99+ Mar 1981
DRY MASS (KG): 3000
MAIN BODY: Cylinder; 1.3 m diameter by 17 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge (?)

EVENT DATA (1)

| | | | |
|-----------|------------|-----------------|----------------|
| DATE: | 8 May 1982 | LOCATION: | 40N, 62E (asc) |
| TIME: | 0444 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 555 km | | |

PRE-EVENT ELEMENTS (1)

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 82127.98788154 | MEAN ANOMALY: | 28.1726 |
| RIGHT ASCENSION: | 337.2406 | MEAN MOTION: | 14.88799005 |
| INCLINATION: | 65.0246 | MEAN MOTION DOT/2: | .00003980 |
| ECCENTRICITY: | .0214690 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 330.7493 | BSTAR: | .00028791 |

EVENT DATA (2)

| | | | |
|-----------|-------------|-----------------|-----------------|
| DATE: | 10 Aug 1982 | LOCATION: | 51N, 238E (dsc) |
| TIME: | 2335 GMT | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | 750 km | | |

PRE-EVENT ELEMENTS (2)

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 82222.89259484 | MEAN ANOMALY: | 62.7628 |
| RIGHT ASCENSION: | 45.7388 | MEAN MOTION: | 14.89366232 |
| INCLINATION: | 65.0248 | MEAN MOTION DOT/2: | .00004369 |
| ECCENTRICITY: | .0219155 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 295.0884 | BSTAR: | .00030390 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 5.2 min
MAXIMUM ΔI : 1.0 deg

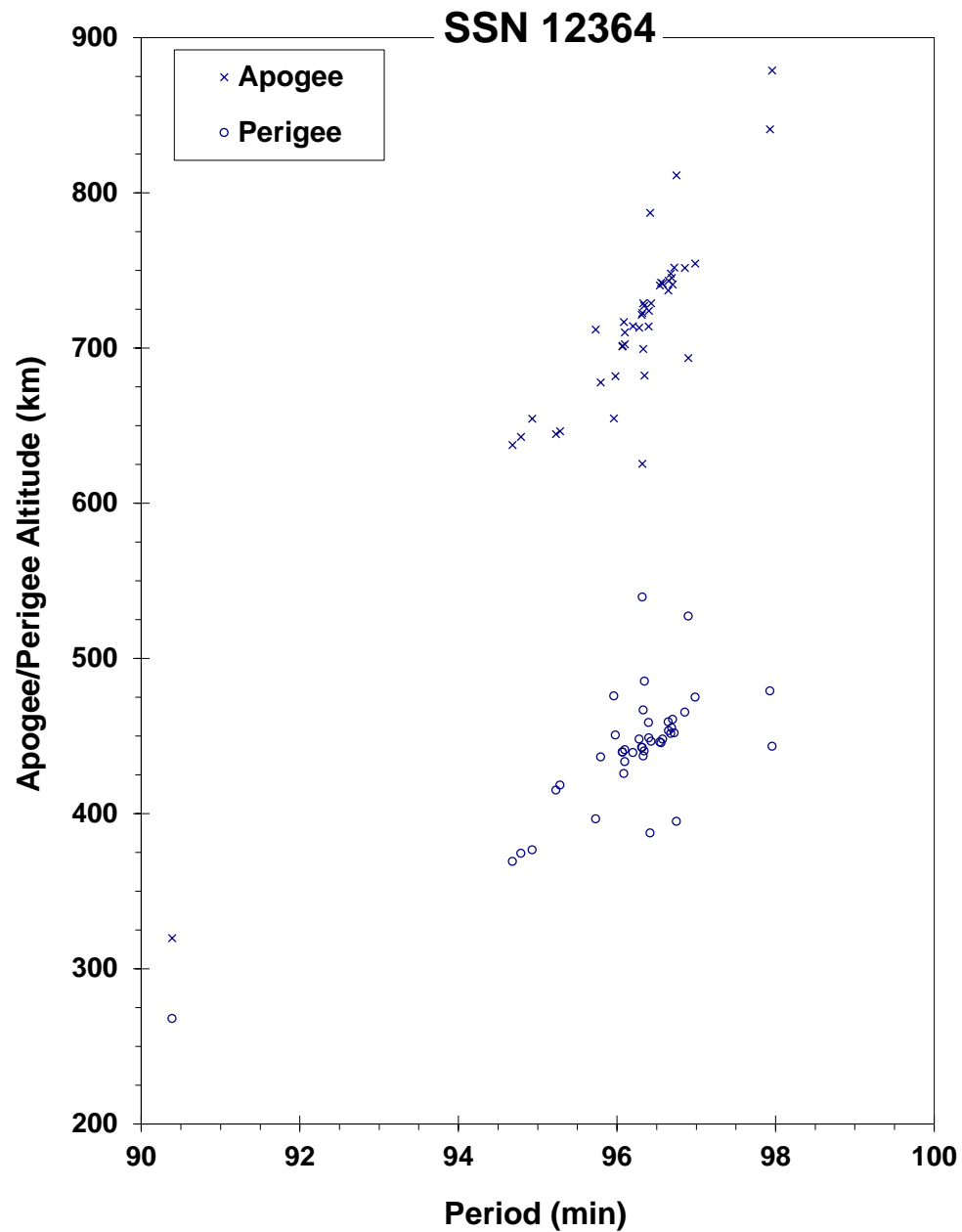
COMMENTS

Cosmos 1260 was the sixth spacecraft of the Cosmos 699-type to experience a fragmentation. The spacecraft had been in a regime of natural decay for 8 months before the first event. After the event the main remnant became satellite 13183, which then fragmented 3 months later. A total of 40 new fragments were officially cataloged prior to the second event. See also Cosmos 1220 and Cosmos 1306 for similar dual fragmentations of Cosmos 699-type spacecraft during this period.

REFERENCE DOCUMENTS

"Artificial Satellite Break-Ups (Part 1): Soviet Ocean Surveillance Satellites", N. L. Johnson, Journal of the British Interplanetary Society, February 1983, pp. 51-58.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



**Cosmos 1260 debris cloud of 43 fragments 3 weeks after the first event
from the US SSN database.**

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 31.40 Mar 1981
DRY MASS (KG): 1250
MAIN BODY: Cylinder; 1.7 m diameter by 2 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

| | | | |
|-----------|--------------|-----------------|------------|
| DATE: | Apr-May 1981 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 81095.90157023 | MEAN ANOMALY: | 4.6715 |
| RIGHT ASCENSION: | 282.6240 | MEAN MOTION: | 2.00494188 |
| INCLINATION: | 63.0386 | MEAN MOTION DOT/2: | .0 |
| ECCENTRICITY: | .7369210 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 316.4347 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 2.3 min*
MAXIMUM ΔI : 0.3 deg*

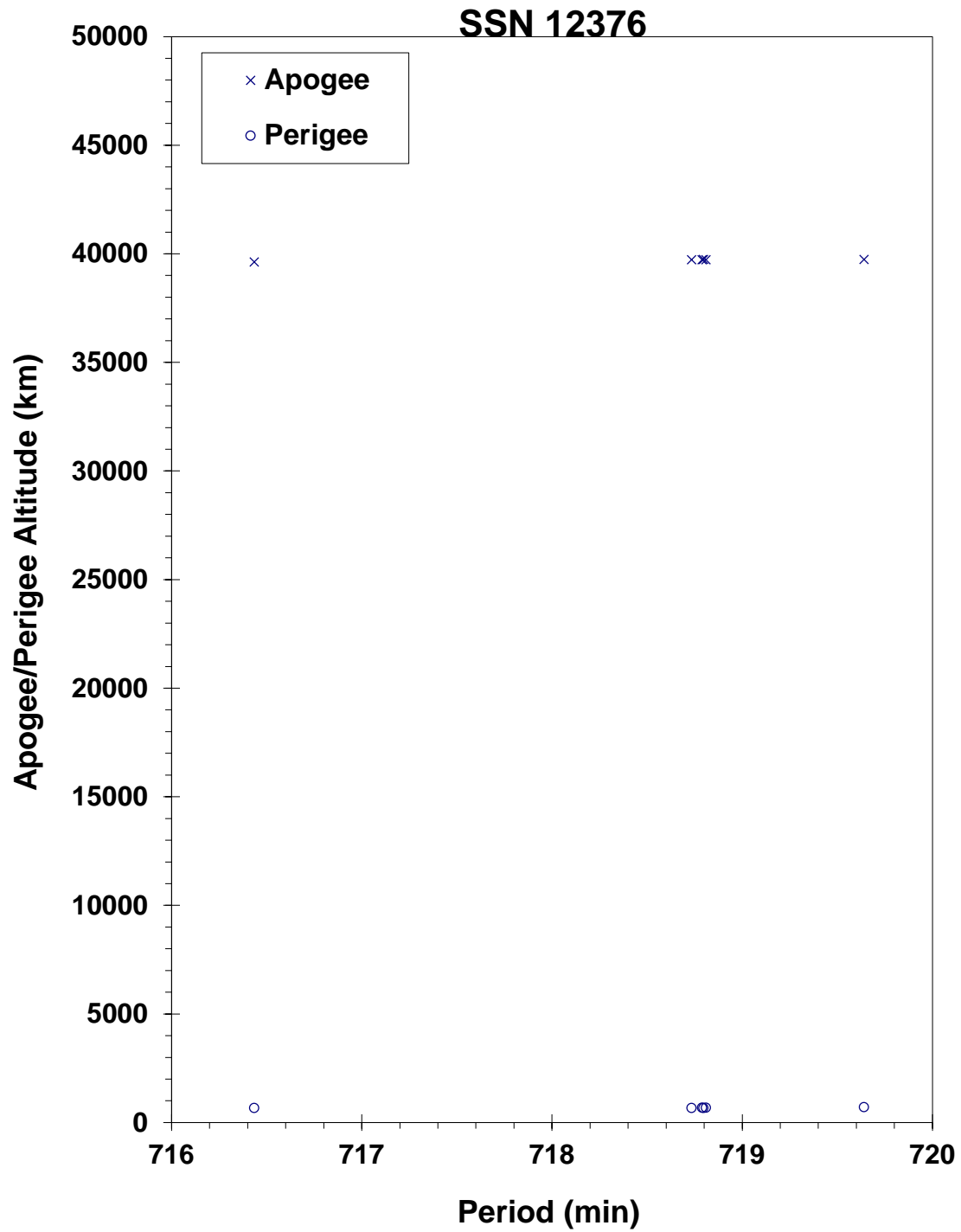
*Based on uncataloged debris data

COMMENTS

Cosmos 1261 was another spacecraft of the Cosmos 862-type to experience a fragmentation. The spacecraft attempted to maneuver from its transfer orbit to an operational orbit 3 days after launch. The maneuver appears to have been unsuccessful, and the spacecraft never became groundtrack-stabilized. Some debris appeared immediately after the maneuver, while additional debris were discovered in mid-May. More than one event may have occurred. The element set above is the first available after the unsuccessful maneuver.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1261 debris cloud of 6 fragments about 8 weeks after (initial) event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 4.66 Jun 1981
DRY MASS (KG): 800
MAIN BODY: Cylinder; 2.4 m diameter by 2 m length
MAJOR APPENDAGES: Gravity-gradient boom
ATTITUDE CONTROL: Gravity gradient
ENERGY SOURCES: Unknown

EVENT DATA

DATE: 24 Jul 1981 LOCATION: 68N, 197E (asc)
TIME: 2351 GMT ASSESSED CAUSE: Battery
ALTITUDE: 980 km

PRE-EVENT ELEMENTS

EPOCH: 81205.39693092 MEAN ANOMALY: 221.3567
RIGHT ASCENSION: 119.8245 MEAN MOTION: 13.73455672
INCLINATION: 82.9633 MEAN MOTION DOT/2: .000000580
ECCENTRICITY: .0036415 MEAN MOTION DOT DOT/6: .0
ARG. OF PERIGEE: 139.0334 BSTAR: .00004538900

DEBRIS CLOUD DATA

MAXIMUM ΔP : 4.9 min
MAXIMUM ΔI : 0.4 deg

COMMENTS

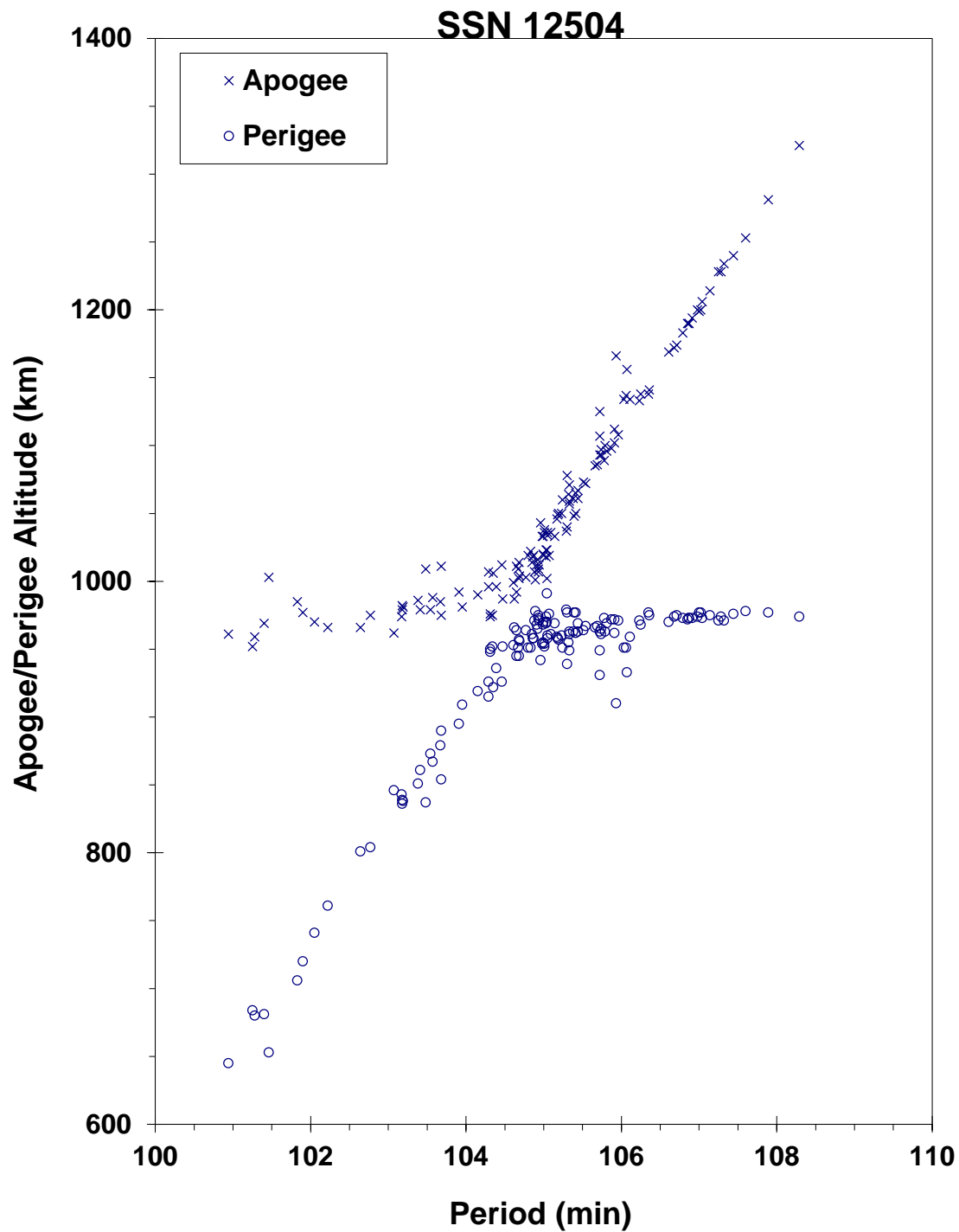
Cosmos 1275 is the only member of its class, Parus, to explosively fragment; the Cosmos 1934 event was a collision with tracked, mission-related debris. The satellite was only 50 days old at the time of the event. During the February 1992 Space Debris Conference in Moscow, Russian analysts discussed independent studies about the probable cause of the breakup. Later, the official Russian assessment asserted that a battery malfunction was the likely culprit.

REFERENCE DOCUMENTS

The Fragmentations of USSR Satellites 11729 and 12504 (U), J.R. Gabbard and P.M. Landry, Technical Memorandum 82-S-03, DCS/Plans, Hdqtrs NORAD/ADCOM, Colorado Springs, August 1982 (Secret).

Determining the Cause of a Satellite Breakup: A Case Study of the Kosmos 1275 Breakup, D.S. McKnight, IAA-87-573, 38th Congress of the International Astronautical Federation, Brighton, England, October 1987.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1275 debris cloud of 136 identified fragments 1 week after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 19.81 Jun 1981
DRY MASS (KG): 1250
MAIN BODY: Cylinder; 1.7 m diameter by 2 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

| | | | |
|-----------|----------------|-----------------|------------|
| DATE: | Early Dec 1986 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 86334.22199701 | MEAN ANOMALY: | 12.7886 |
| RIGHT ASCENSION: | 288.0814 | MEAN MOTION: | 2.00618298 |
| INCLINATION: | 67.1073 | MEAN MOTION DOT/2: | .0 |
| ECCENTRICITY: | .6594262 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 291.9890 | BSTAR: | .0 |

DEBRIS CLOUD DATA

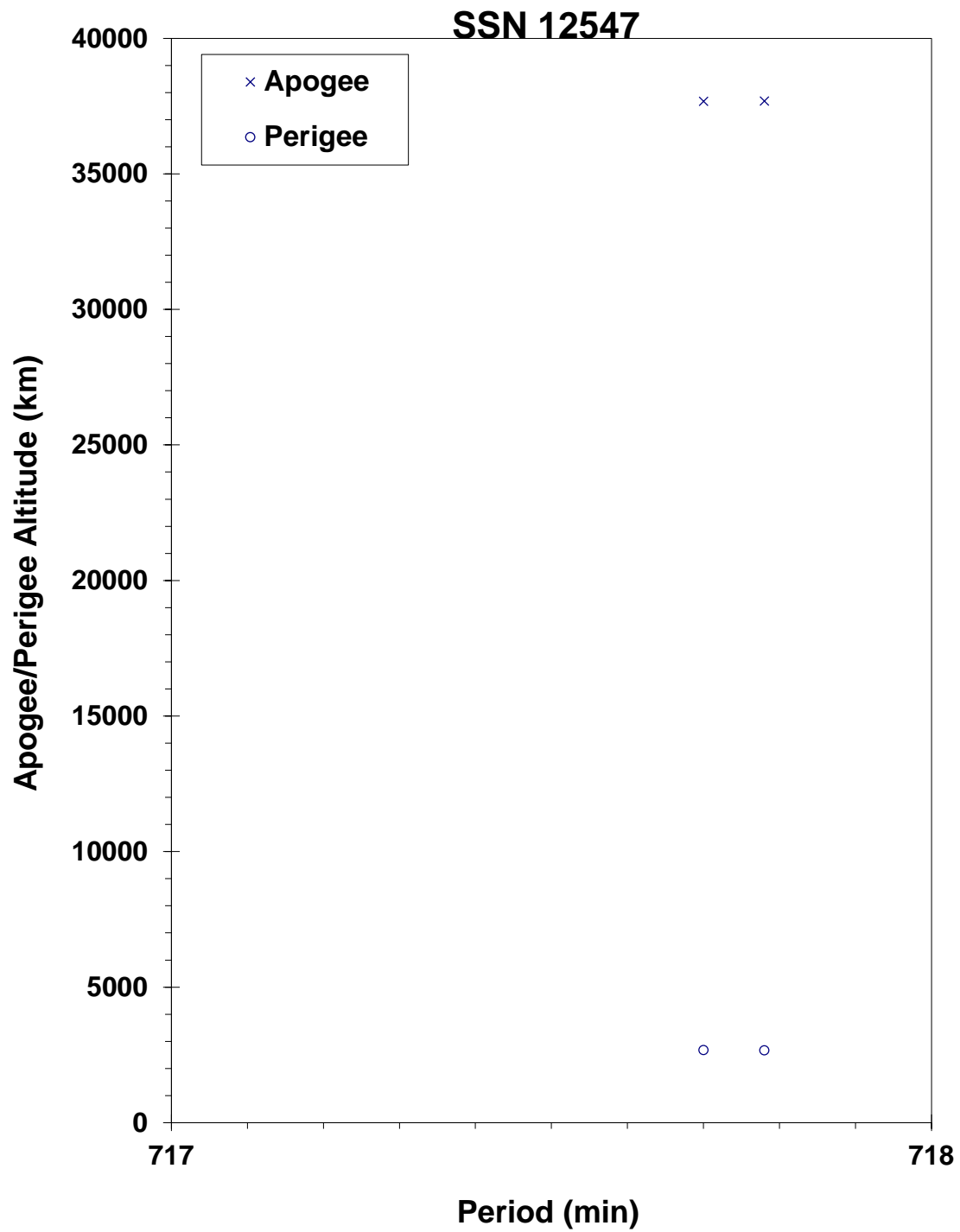
MAXIMUM ΔP : 0.1 min
MAXIMUM ΔI : 0.0 deg

COMMENTS

Cosmos 1278 was another spacecraft of the Cosmos 862-type to experience a fragmentation. Spacecraft had apparently been inactive since early 1984. Additional fragments may exist, but surveillance for small objects in this high eccentricity orbit is difficult.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1278 and additional fragment in mid-December 1986. Elements from the US SSN as published by the NASA Goddard Space Flight Center.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 4.01 Aug 1981
DRY MASS (KG): 1250
MAIN BODY: Cylinder; 1.7 m diameter by 2 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 21 Nov 1981 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 81324.16708257 | MEAN ANOMALY: | 4.8196 |
| RIGHT ASCENSION: | 249.5852 | MEAN MOTION: | 1.98014597 |
| INCLINATION: | 63.1086 | MEAN MOTION DOT/2: | .00000781 |
| ECCENTRICITY: | .7350717 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 317.0022 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 8.6 min*
MAXIMUM ΔI : 0.2 deg*

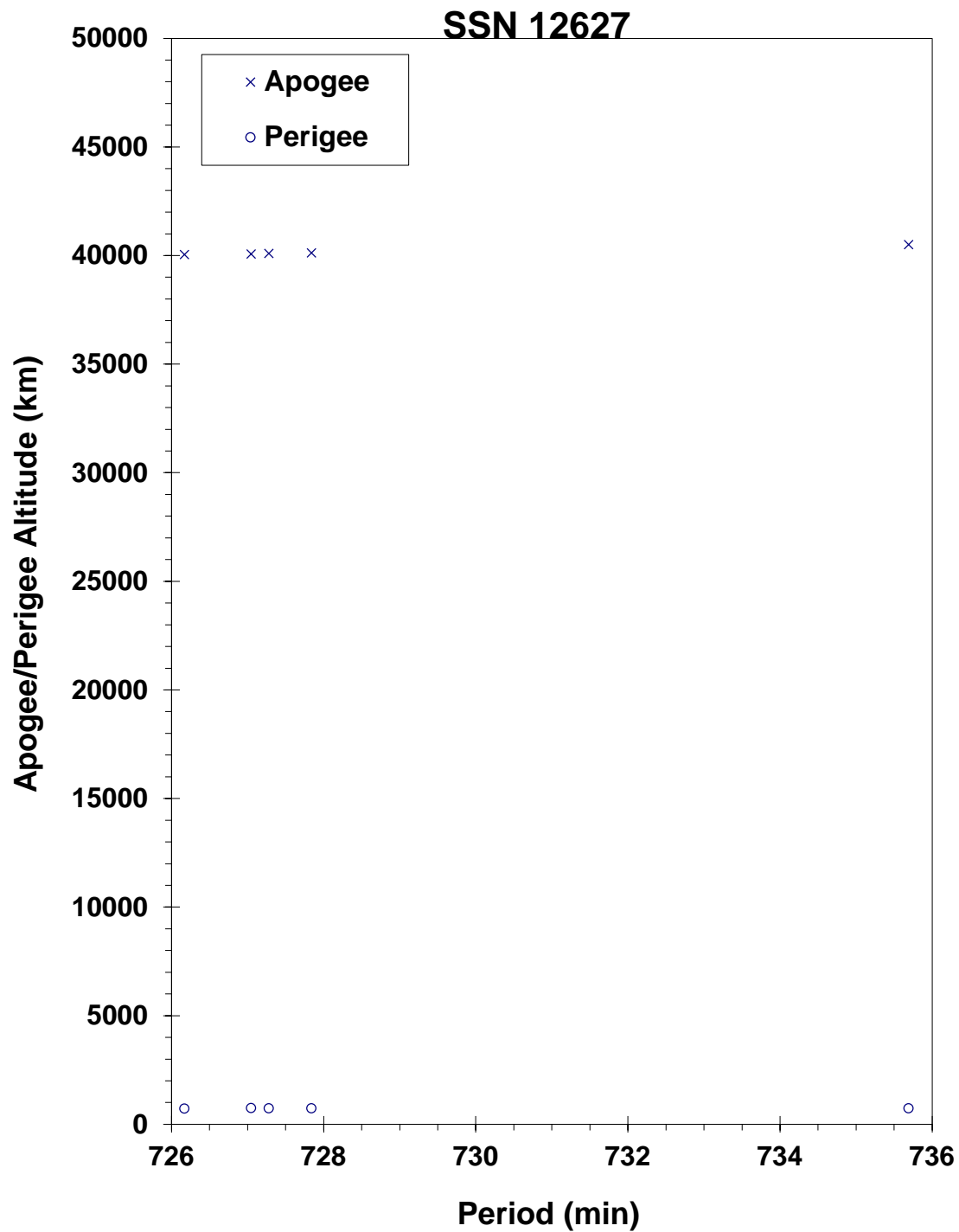
*Based on uncataloged debris data

COMMENTS

Cosmos 1285 was another spacecraft of the Cosmos 862-type to experience a fragmentation. Spacecraft was placed in a temporary transfer orbit on the day of launch by its launch vehicle but never maneuvered to an operational orbit, suggesting an early fatal spacecraft malfunction. Event occurred 3.5 months after the launch.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1285 debris cloud of 5 fragments less than 1 week after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 4.35 Aug 1981
DRY MASS (KG): 3000
MAIN BODY: Cylinder; 1.3 m diameter by 17 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge (?)

EVENT DATA

DATE: 29 Sep 1982 LOCATION: 51N, 80E (asc)
TIME: 0520 GMT ASSESSED CAUSE: Unknown
ALTITUDE: 325 km

PRE-EVENT ELEMENTS

EPOCH: 82272.21193719 MEAN ANOMALY: 92.4681
RIGHT ASCENSION: 132.9736 MEAN MOTION: 15.86141247
INCLINATION: 65.0071 MEAN MOTION DOT/2: .00400345
ECCENTRICITY: .0017215 MEAN MOTION DOT DOT/6: .0
ARG. OF PERIGEE: 267.4145 BSTAR: .0015199

DEBRIS CLOUD DATA

MAXIMUM ΔP : 0.9 min*
MAXIMUM ΔI : 0.2 deg*

*Based on uncataloged debris data

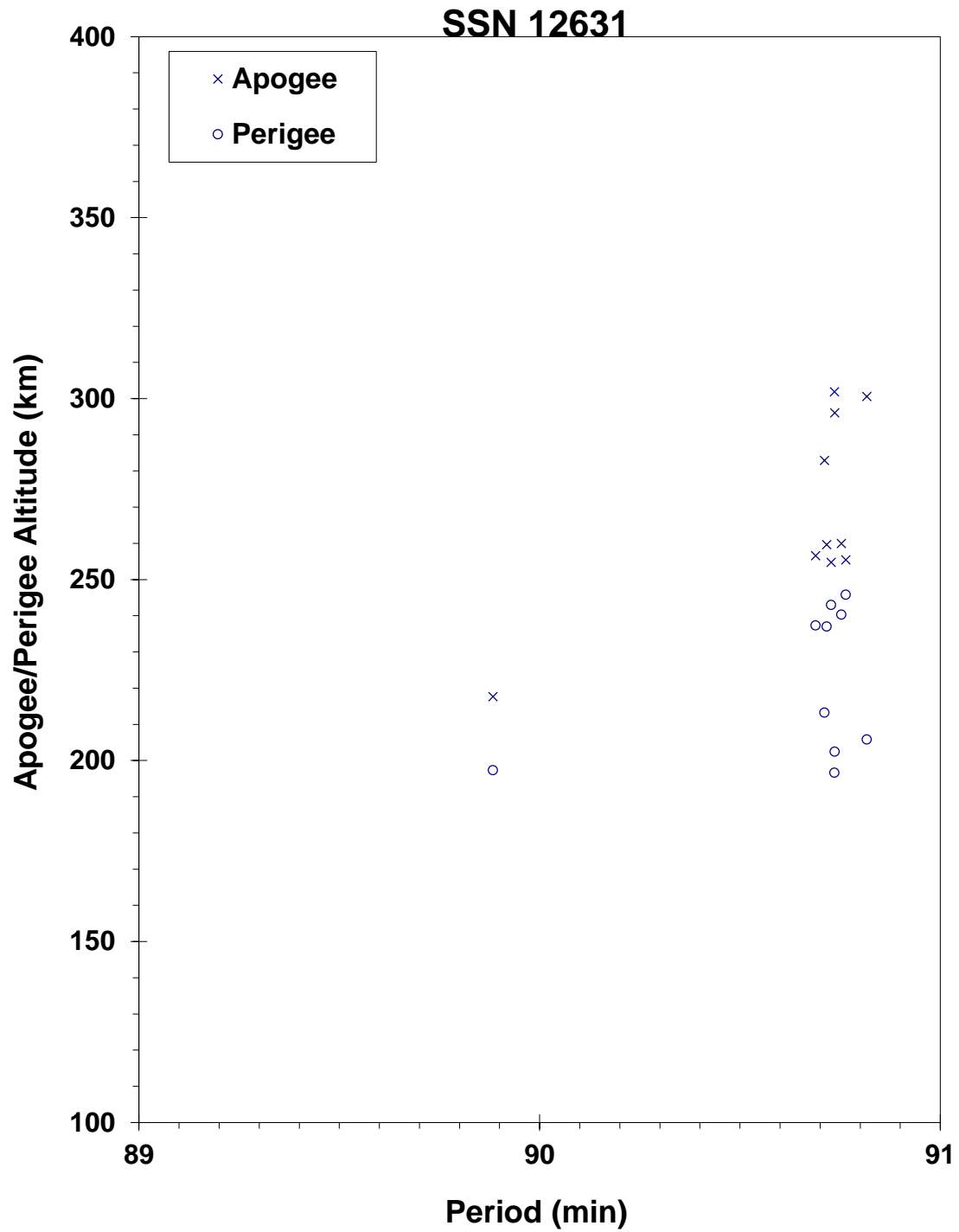
COMMENTS

Cosmos 1286 was the ninth spacecraft of the Cosmos 699-type to experience a fragmentation. The spacecraft had been in a regime of natural decay for more than 6 months at the time of the event. The low altitude and high drag conditions made determination of the precise breakup time uncertain. The breakup or a precursor event may have occurred earlier on 29 September 1982. Most fragments decayed before being officially cataloged.

REFERENCE DOCUMENTS

"Artificial Satellite Break-Ups (Part 1): Soviet Ocean Surveillance Satellites", N. L. Johnson, Journal of the British Interplanetary Society, February 1983, pp. 51-58.

History of Soviet/Russian Satellite Fragmentation-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1286 debris cloud of 10 fragments 1 day after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Molniya Final Stage
OWNER: CIS
LAUNCH DATE: 11.36 Sep 1981
DRY MASS (KG): 1100
MAIN BODY: Cylinder; 2.7 m diameter by 3 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 11 Sep 1981 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

POST-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 81258.60717998 | MEAN ANOMALY: | 26.9249 |
| RIGHT ASCENSION: | 68.6245 | MEAN MOTION: | 5.48678032 |
| INCLINATION: | 62.8166 | MEAN MOTION DOT/2: | .0 |
| ECCENTRICITY: | .4855644 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 286.6972 | BSTAR: | .0 |

DEBRIS CLOUD DATA

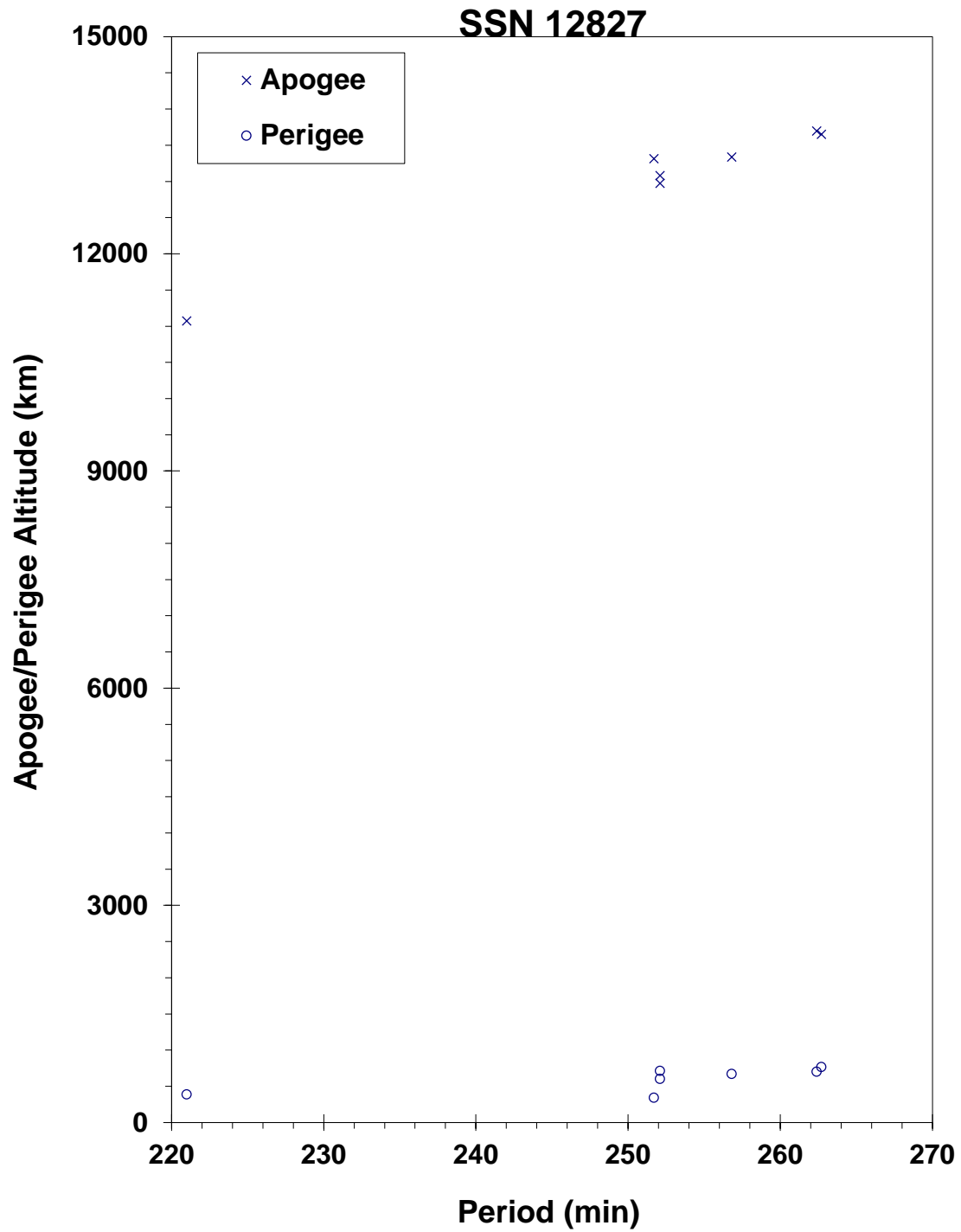
MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

Cosmos 1305 R/B malfunctioned about 1 hour after launch during a maneuver from a LEO parking orbit to a Molniya-type orbit. The maneuver was initiated at approximately 0937 GMT near 58S, 245E (asc) at an altitude of 600 km. Apogee was raised to less than 14,000 km. Debris tracking after the event was limited, preventing an accurate assessment of magnitude of the event. First debris officially cataloged in June 1983. Debris generation is assumed to have occurred during or immediately after the unsuccessful maneuver. The element set above is for the rocket body after burn termination.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1305 R/B debris cloud of 7 fragments about 2 years after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 14.85 Sep 1981
DRY MASS (KG): 3000
MAIN BODY: Cylinder; 1.3 m diameter by 17 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge (?)

EVENT DATA (1)

| | | | |
|-----------|-------------|-----------------|----------------|
| DATE: | 12 Jul 1982 | LOCATION: | 65S, 40E (asc) |
| TIME: | 2325 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 380 km | | |

PRE-EVENT ELEMENTS (1)

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 82193.22052182 | MEAN ANOMALY: | 72.7640 |
| RIGHT ASCENSION: | 43.8843 | MEAN MOTION: | 15.58171668 |
| INCLINATION: | 64.9399 | MEAN MOTION DOT/2: | .00042116 |
| ECCENTRICITY: | .0019953 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 287.2390 | BSTAR: | .00055055 |

EVENT DATA (2)

| | | | |
|-----------|-------------|-----------------|-----------------|
| DATE: | 18 Sep 1982 | LOCATION: | 32N, 293E (asc) |
| TIME: | 1702 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 370 km | | |

PRE-EVENT ELEMENTS (2)

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 82260.17037940 | MEAN ANOMALY: | 44.8033 |
| RIGHT ASCENSION: | 173.7764 | MEAN MOTION: | 15.65882738 |
| INCLINATION: | 64.9408 | MEAN MOTION DOT/2: | .00076164 |
| ECCENTRICITY: | .0002181 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 315.2578 | BSTAR: | .00073994 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 2.1 min*
MAXIMUM ΔI : 0.2 deg

*Based on uncataloged debris data

COMMENTS

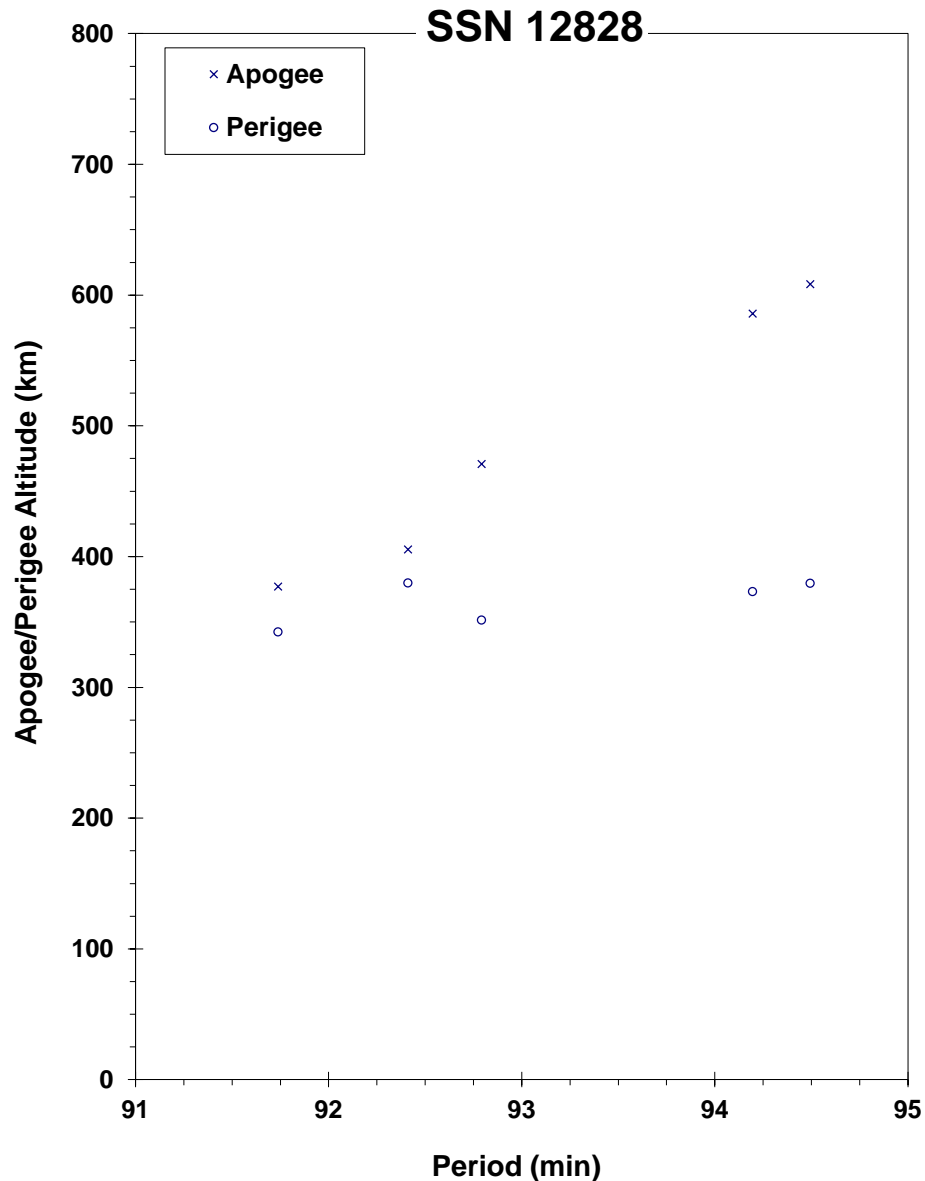
Cosmos 1306 was the eighth spacecraft of the Cosmos 699-type to experience a fragmentation. The first event occurred 5 months after the spacecraft had entered a regime of natural decay. After the event the main remnant was tagged as satellite 13369, while a piece of debris tagged as 12828 decayed on 16 July 1982. Only 5 new fragments were officially cataloged prior to the second event when satellite 13369 experienced a fragmentation. Three long-lived fragments cataloged with 1981-89 (13393, 13404, and 14837) were actually part of the breakup of 1980-89, another Cosmos 699-type satellite. Most Cosmos 1306 debris reentered quickly and elements were developed for only a few fragments.

REFERENCE DOCUMENTS

Analysis of Cosmos 1220 and Cosmos 1306 Fragments (U), D. Fennessy, Report AH-23, FTD/OLAI, Cheyenne Mountain, Colorado, 12 January 1983 (Secret)

"Artificial Satellite Break-Ups (Part 1): Soviet Ocean Surveillance Satellites", N. L. Johnson, Journal of the British Interplanetary Society, February 1983, pp. 51-58.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1306 debris cloud of 5 identified fragments 1 day after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 31.95 Oct 1981
DRY MASS (KG): 1250
MAIN BODY: Cylinder; 1.7 m diameter by 2 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive

EVENT DATA

| | | | |
|-----------|----------------|-----------------|------------|
| DATE: | 25-28 Jan 1984 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 84024.46309667 | MEAN ANOMALY: | 4.4900 |
| RIGHT ASCENSION: | 219.5352 | MEAN MOTION: | 2.00535027 |
| INCLINATION: | 62.8286 | MEAN MOTION DOT/2: | .0 |
| ECCENTRICITY: | .7103977 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 324.1891 | BSTAR: | .0 |

DEBRIS CLOUD DATA

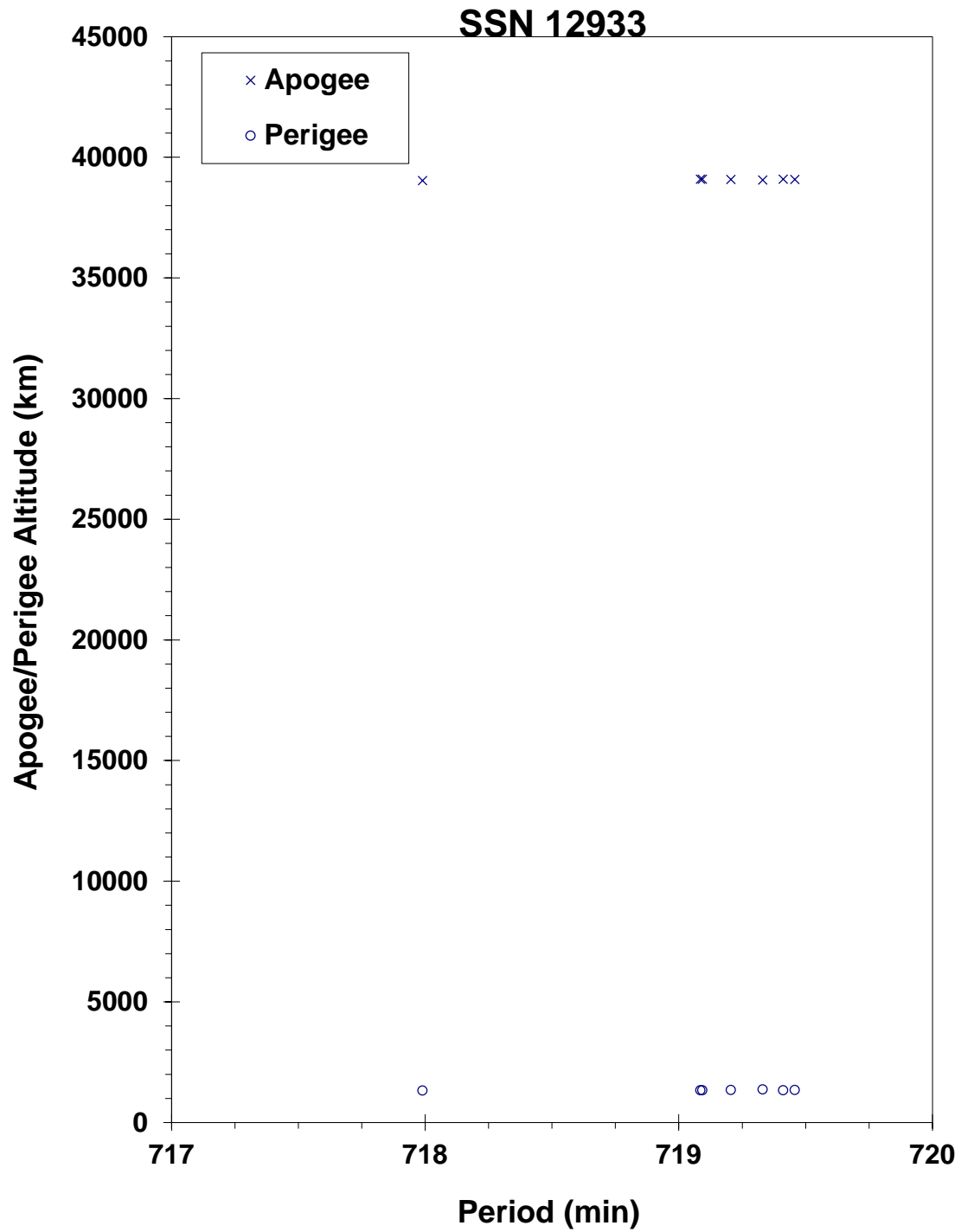
MAXIMUM ΔP : 1.8 min
MAXIMUM ΔI : 0.3 deg

COMMENTS

Cosmos 1317 was another spacecraft of the Cosmos 862-type to experience a fragmentation

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1317 debris cloud of 7 fragments about 2 weeks after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 7.57 April 1982
DRY MASS (KG): 1250
MAIN BODY: Cylinder; 1.7 m diameter by 2 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive device

EVENT DATA

| | | | |
|-----------|------------------|-----------------|---------------------|
| DATE: | 2 September 1984 | LOCATION: | 39.5N, 121.3E (asc) |
| TIME: | 0957 GMT | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | 5020 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 84246.40302854 | MEAN ANOMALY: | 3.8405 |
| RIGHT ASCENSION: | 227.2447 | MEAN MOTION: | 2.00449045 |
| INCLINATION: | 62.8163 | MEAN MOTION DOT/2: | .00000970 |
| ECCENTRICITY: | .7154927 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 326.7879 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 25.5 min
MAXIMUM ΔI : 7.3 deg

COMMENTS

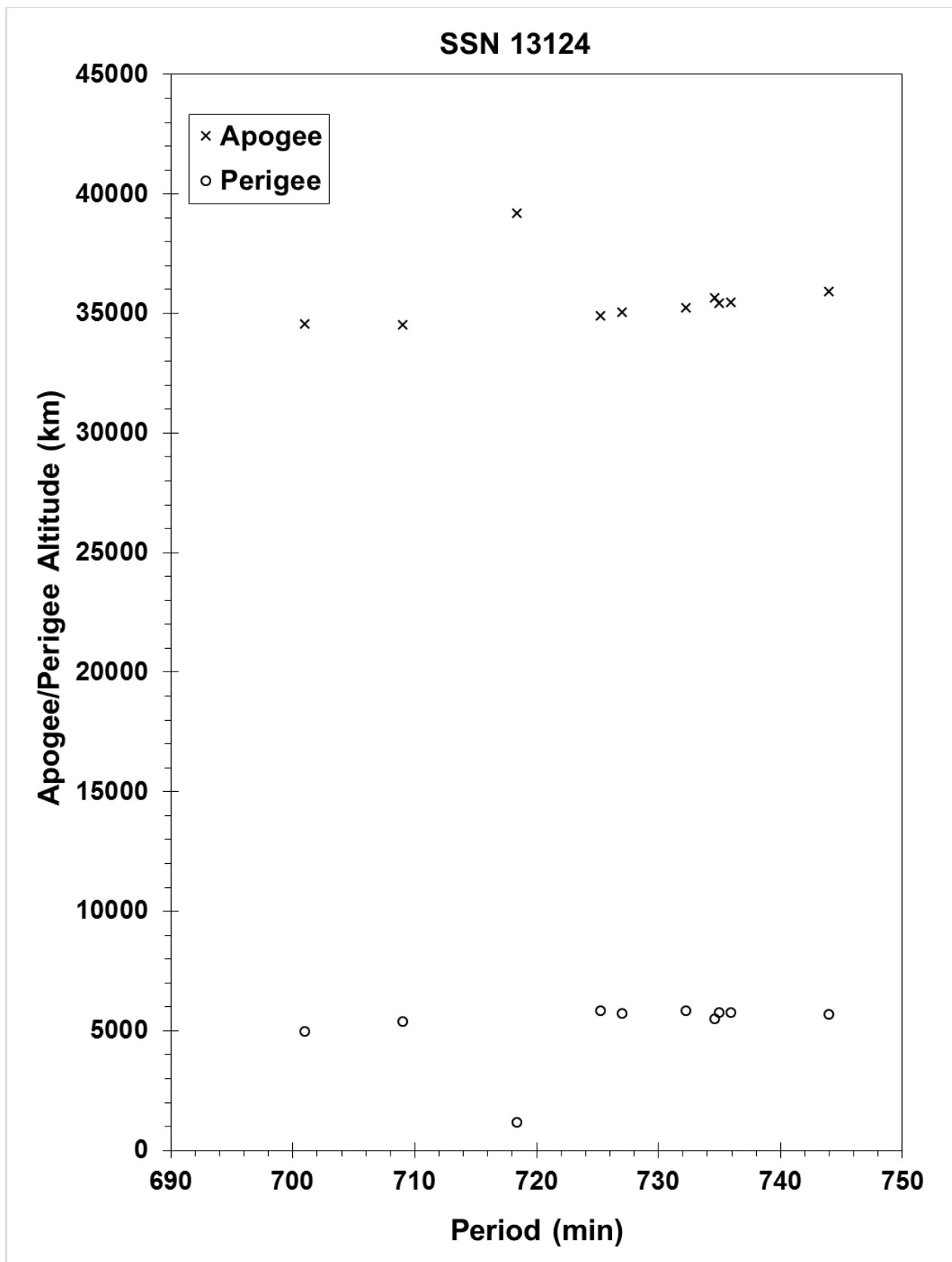
Cosmos 1348 was a member of the Cosmos 862 class of operational satellites in highly elliptical, semi-synchronous orbits that experienced a total of 16 fragmentations during the period 1977-1986. Due to the nature of these orbits, which result in high altitudes over the Northern Hemisphere where most surveillance sensors are located, debris detection and tracking is extremely difficult. Only the largest fragments can be seen. Cosmos 862 class spacecraft were equipped with self-destruct packages in the event that spacecraft control was lost; this was the cause of breakups until the explosives were removed after Cosmos 1481.

Podvig reported an estimate end of life for Cosmos 1348 as 22 July 1984. If correct, this could imply a breakup mechanism different than other members of the Cosmos 862 class. An examination of the mean motion history of Cosmos 1348 indicated, however, that it was under active control until the event date of 2 September 1984, implying that the cause attributed to the fragmentation of other class members was similarly responsible for this spacecraft's fragmentation.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson *et al.*, Kaman Sciences Corporation, October 1995.

Podvig, P., "History and the Current Status of the Russian Early-Warning System". *Science and Global Security* 10 (2002): 21-60.



Cosmos 1348 debris cloud; fragments cataloged over 23 years after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 29.41 Apr 1982
DRY MASS (KG): 3000
MAIN BODY: Cylinder; 1.3 m diameter by 17 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge (?)

EVENT DATA (1)

| | | | |
|-----------|------------|-----------------|-----------------|
| DATE: | 8 Aug 1983 | LOCATION: | 32S, 310E (asc) |
| TIME: | 2331 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 365 km | | |

PRE-EVENT ELEMENTS (1)

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 83220.21851552 | MEAN ANOMALY: | 66.8795 |
| RIGHT ASCENSION: | 279.4096 | MEAN MOTION: | 15.63233551 |
| INCLINATION: | 65.0504 | MEAN MOTION DOT/2: | .00048258 |
| ECCENTRICITY: | .0024043 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 292.8515 | BSTAR: | .00051620 |

EVENT DATA (2)

| | | | |
|-----------|------------|-----------------|----------------|
| DATE: | 1 Feb 1984 | LOCATION: | 4S, 200E (asc) |
| TIME: | 0322 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 320 km | | |

PRE-EVENT ELEMENTS (2)

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 84031.38369465 | MEAN ANOMALY: | 81.7159 |
| RIGHT ASCENSION: | 25.3553 | MEAN MOTION: | 15.84652631 |
| INCLINATION: | 65.0404 | MEAN MOTION DOT/2: | .00119378 |
| ECCENTRICITY: | .0017572 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 278.1110 | BSTAR: | .00050318 |

EVENT DATA (3)

| | | | |
|-----------|-----------------|-----------------|---------|
| DATE: | 20 Feb 1984 | LOCATION: | Unknown |
| TIME: | Before 0340 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS (3)

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 84050.69015256 | MEAN ANOMALY: | 105.8772 |
| RIGHT ASCENSION: | 316.3115 | MEAN MOTION: | 15.97914042 |
| INCLINATION: | 65.0338 | MEAN MOTION DOT/2: | .00430956 |
| ECCENTRICITY: | .0014134 | MEAN MOTION DOT DOT/6: | .000083799 |
| ARG. OF PERIGEE: | 254.0517 | BSTAR: | .00093344 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 36.8 min*
MAXIMUM ΔI : 2.3 deg*

*Based on uncataloged debris data (Event 1)

COMMENTS

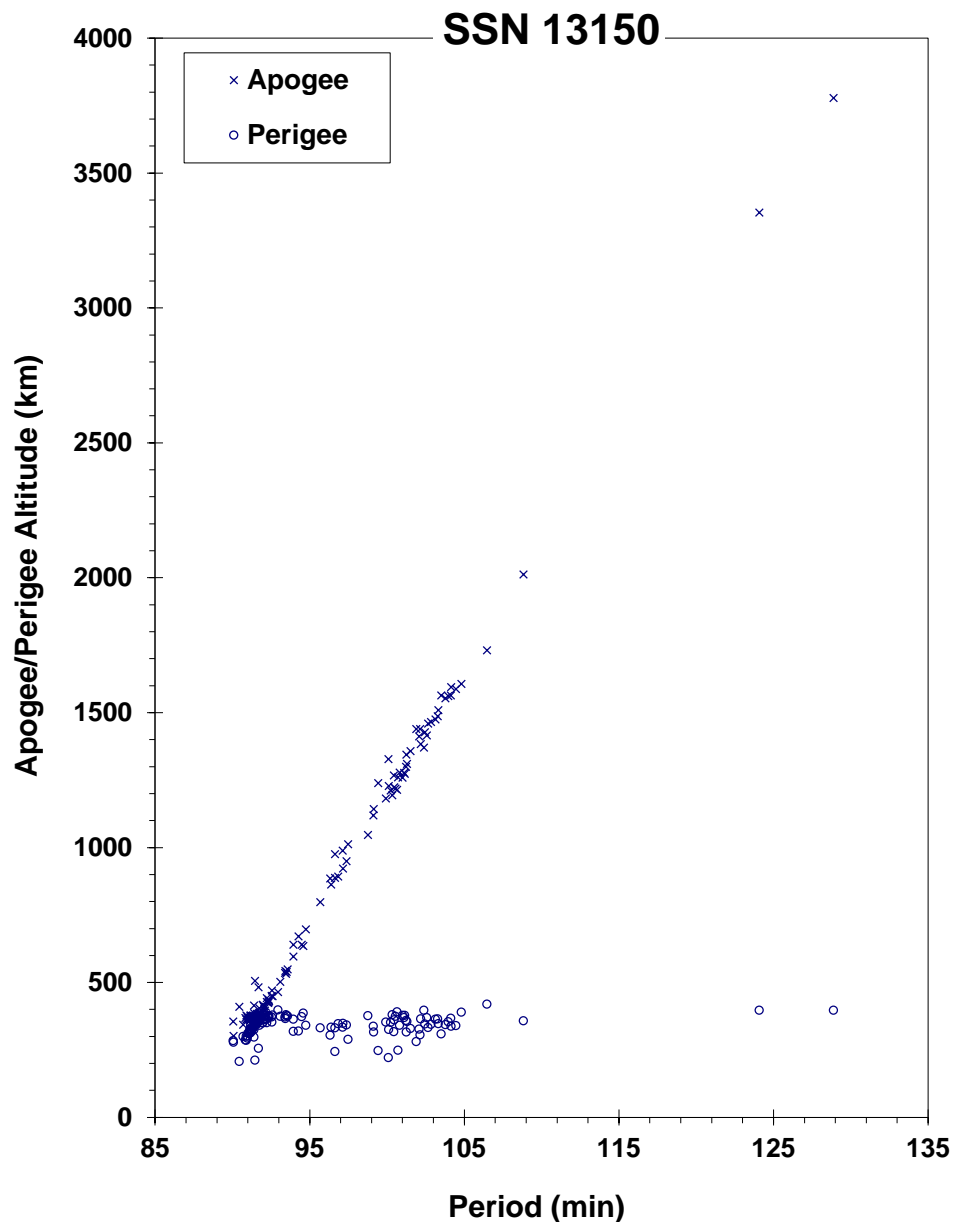
Cosmos 1355 was the tenth spacecraft of the Cosmos 699-type to experience a fragmentation. The spacecraft had been in a regime of natural decay for 6 months prior to the first event. Twenty-one fragments were cataloged following the first event, and the main body became satellite 14275. This object spawned at least seven more fragments on 1 February. The parent was then retagged to the original 13150 satellite number. The third event resulted in the development of 13 new fragment element sets, but none were cataloged and the low altitude prevented an estimate of a precise breakup location.

REFERENCE DOCUMENTS

"Artificial Satellite Break-Ups (Part 1): Soviet Ocean Surveillance Satellites", N. L. Johnson, Journal of the British Interplanetary Society, February 1983, pp. 51-58.

Analysis of the Fragmentation of Kosmos 1355, N. L. Johnson, Technical Report CS84-SPACECMD-28, Teledyne Brown Engineering, Colorado Springs, January 1985.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1355 debris cloud of 150 fragments about 7 hours after the first event in August 1983 as seen by the US SSN PARCS radar. Figure from the cited reference.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 6.72 Jun 1982
DRY MASS (KG): 650
MAIN BODY: Polyhedron; 1.4 m by 1.4 m
MAJOR APPENDAGES: Solar panels, gravity-gradient boom (?)
ATTITUDE CONTROL: Gravity gradient (?)
ENERGY SOURCES: Battery

EVENT DATA

DATE: 21 Oct 1985
TIME: 0353 GMT
ALTITUDE: 995 km
LOCATION: 66N, 351E (asc)
ASSESSED CAUSE: Battery

PRE-EVENT ELEMENTS

EPOCH: 85293.85195210
RIGHT ASCENSION: 350.2805
INCLINATION: 65.8390
ECCENTRICITY: .0005355
ARG. OF PERIGEE: 26.5667
MEAN ANOMALY: 333.5602
MEAN MOTION: 13.71079597
MEAN MOTION DOT/2: .00000158
MEAN MOTION DOT DOT/6: .0
BSTAR: .00023894

DEBRIS CLOUD DATA

MAXIMUM ΔP : 2.3 min*
MAXIMUM ΔI : 0.1 deg*

*Based on uncataloged debris data

COMMENTS

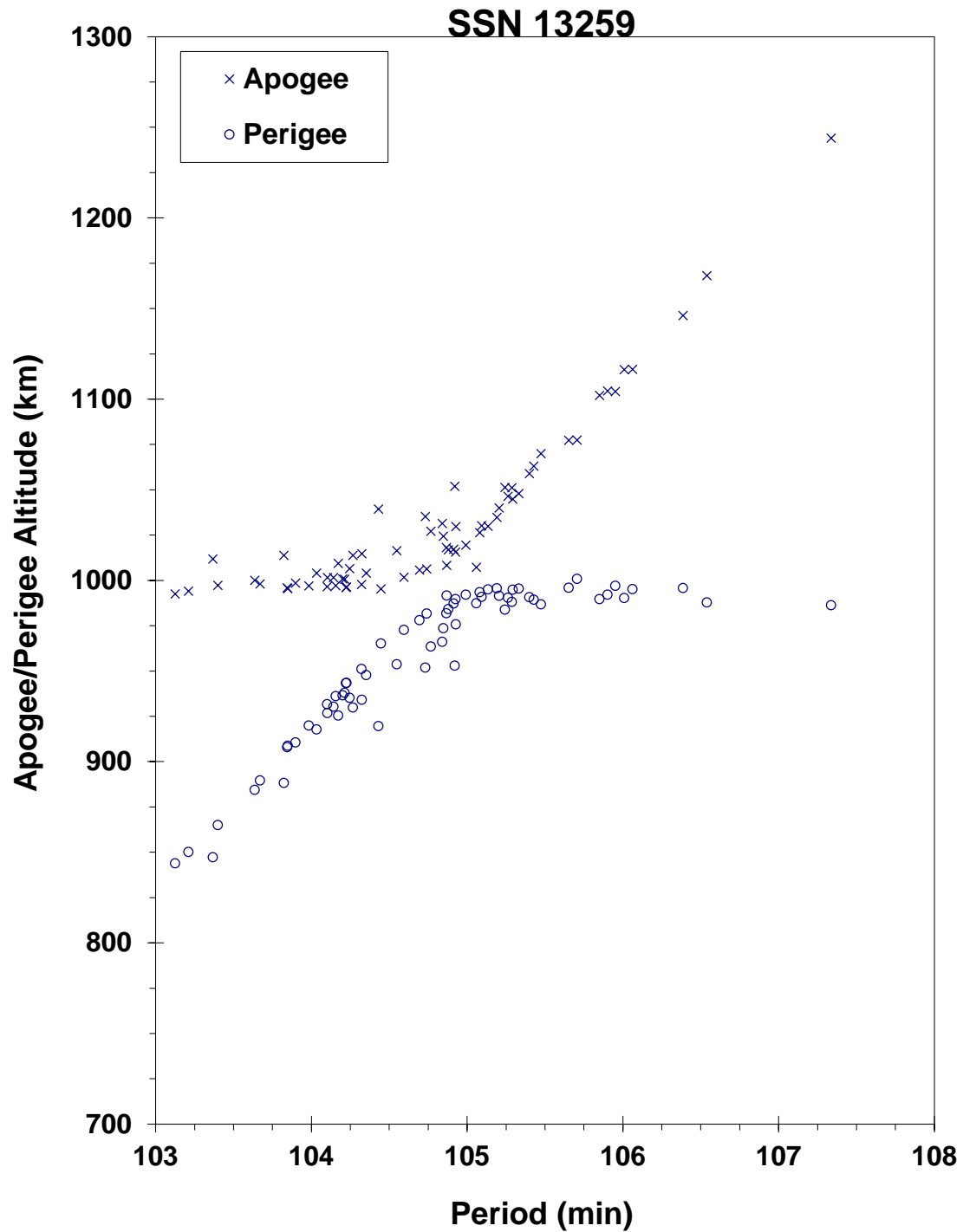
Cosmos 1375 was the third spacecraft of the Cosmos 839-type to experience a fragmentation. Although these satellites are used in conjunction with the Cosmos 249-type spacecraft that are deliberately fragmented, the cause of Cosmos 839-type events appears to be unrelated. In the case of Cosmos 1375, 40 months elapsed since its test with a Cosmos 249-type spacecraft.

REFERENCE DOCUMENTS

"Artificial Satellite Break-Ups (Part 2): Soviet Anti-Satellite Program", N.L. Johnson, Journal of the British Interplanetary Society, August 1983, pp. 357-362.

Analysis of the Kosmos 1375 Fragmentation, J. M. Koskella and R. L. Kling, Technical Report CS86-USASDC-0006, Teledyne Brown Engineering, Colorado Springs, March 1986.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1375 debris cloud of 68 fragments seen a few hours after the event by the US SSN PARCS radar.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 4.74 Sep 1982
DRY MASS (KG): 3000
MAIN BODY: Cylinder; 1.3 m diameter by 17 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge (?)

EVENT DATA

DATE: 20 Dec 1983 LOCATION: 25S, 45E (dsc)
TIME: 1215 GMT ASSESSED CAUSE: Unknown
ALTITUDE: 330 km

PRE-EVENT ELEMENTS

EPOCH: 83354.22079767 MEAN ANOMALY: 42.0375
RIGHT ASCENSION: 126.1259 MEAN MOTION: 15.81899265
INCLINATION: 65.0055 MEAN MOTION DOT/2: .00186341
ECCENTRICITY: .0020774 MEAN MOTION DOT DOT/6: .0
ARG. OF PERIGEE: 318.0927 BSTAR: .00088277

DEBRIS CLOUD DATA

MAXIMUM ΔP : 7.3 min*
MAXIMUM ΔI : 2.0 deg*

*Based on uncataloged debris data

COMMENTS

Cosmos 1405 was the eleventh spacecraft of the Cosmos 699-type to experience a fragmentation. Spacecraft had been in natural decay for 12 months prior to the event. Most debris reentered before being officially cataloged.

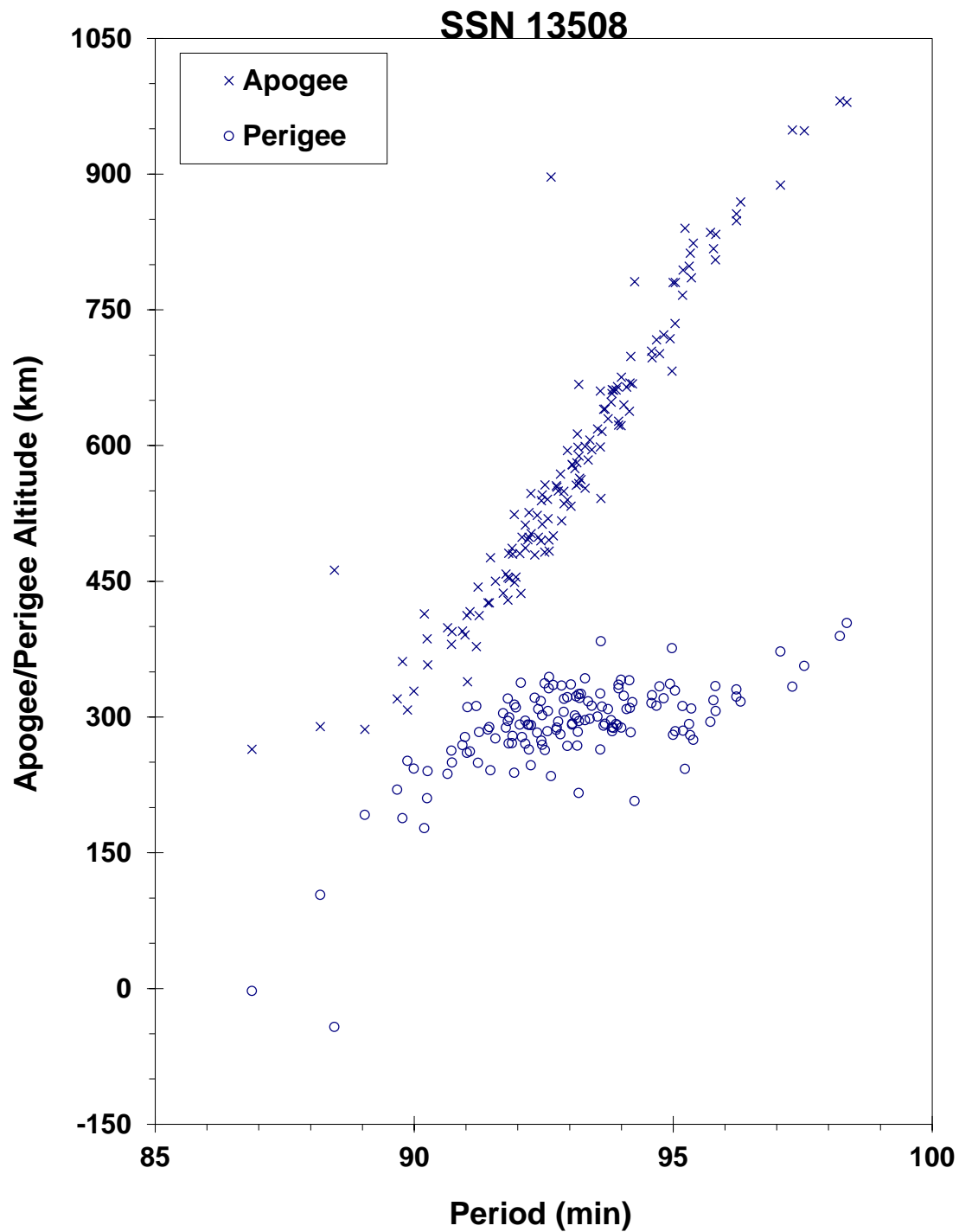
REFERENCE DOCUMENTS

"Artificial Satellite Break-Ups (Part 1): Soviet Ocean Surveillance Satellites", N. L. Johnson, Journal of the British Interplanetary Society, February 1983, pp. 51-58.

Separation of Objects from Cosmos 1405, F.T. Lipp, NAVSPASUR Technical Note 1-84, Naval Space Surveillance System, Dahlgren, 2 April 1984.

Analysis of the Fragmentation of Kosmos 1405, N.L. Johnson, Technical Report CS84-SPACECMD-10, Teledyne Brown Engineering, Colorado Springs, September 1984.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1405 debris cloud of 143 fragments 1 hour after the event as seen
by the US SSN PARCS radar.

SATELLITE DATA

TYPE: Molniya Final Stage
OWNER: CIS
LAUNCH DATE: 8.58 Dec 1982
DRY MASS (KG): 1100
MAIN BODY: Cylinder; 2.7 m diameter by 3 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|------------|-----------------|-----------------|
| DATE: | 8 Dec 1982 | LOCATION: | 62S, 302E (asc) |
| TIME: | 1448 GMT | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | 400 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 82342.56790507 | MEAN ANOMALY: | 305.2204 |
| RIGHT ASCENSION: | 316.3789 | MEAN MOTION: | 15.79849844 |
| INCLINATION: | 62.9496 | MEAN MOTION DOT/2: | .0 |
| ECCENTRICITY: | .0143321 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 56.2493 | BSTAR: | .0 |

DEBRIS CLOUD DATA

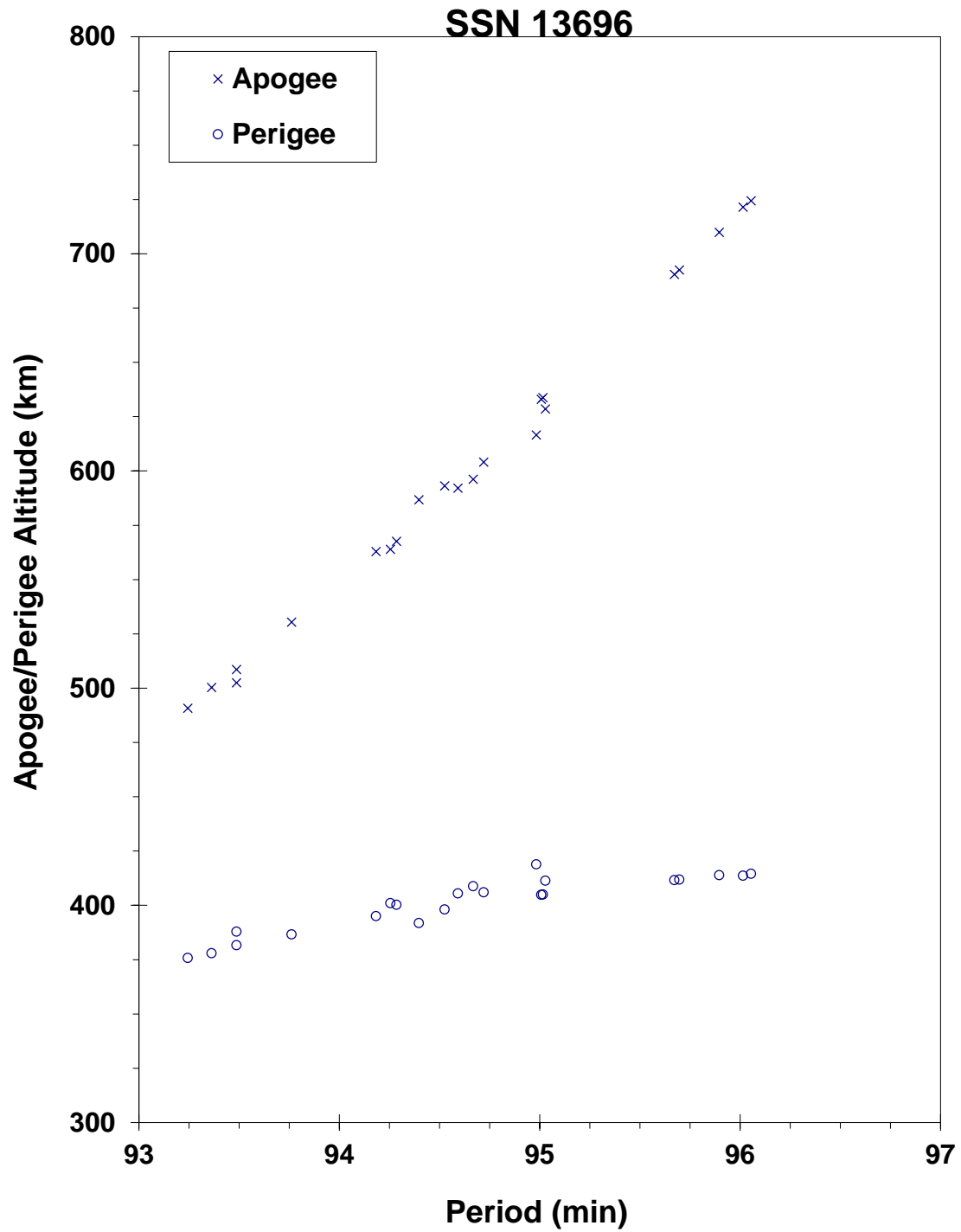
MAXIMUM ΔP : 4.9 min
MAXIMUM ΔI : 0.2 deg

COMMENTS

Fragmentation occurred at the time the Molniya final stage was fired to move the payload from a parking orbit to a Molniya-type transfer orbit. Pre-event elements are taken from satellite 13686 for first revolution parking orbit. A second fragmentation may have occurred on 9 December 1982.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1423 R/B debris cloud of 22 fragments soon after the event(s) as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 23.53 Mar 1983
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1.0 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

DATE: 3 Sep 1984 LOCATION: 12S, 352E (dsc)
TIME: 2023 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 400 km

PRE-EVENT ELEMENTS

EPOCH: 84247.05150886 MEAN ANOMALY: 106.3279
RIGHT ASCENSION: 94.4099 MEAN MOTION: 14.50264973
INCLINATION: 51.5306 MEAN MOTION DOT/2: .00079313
ECCENTRICITY: .0710960 MEAN MOTION DOT DOT/6: .0000075234
ARG. OF PERIGEE: 246.1573 BSTAR: .00035531

DEBRIS CLOUD DATA

MAXIMUM ΔP : 2.4 min*
MAXIMUM ΔI : 0.3 deg*

*Based on uncataloged debris data

COMMENTS

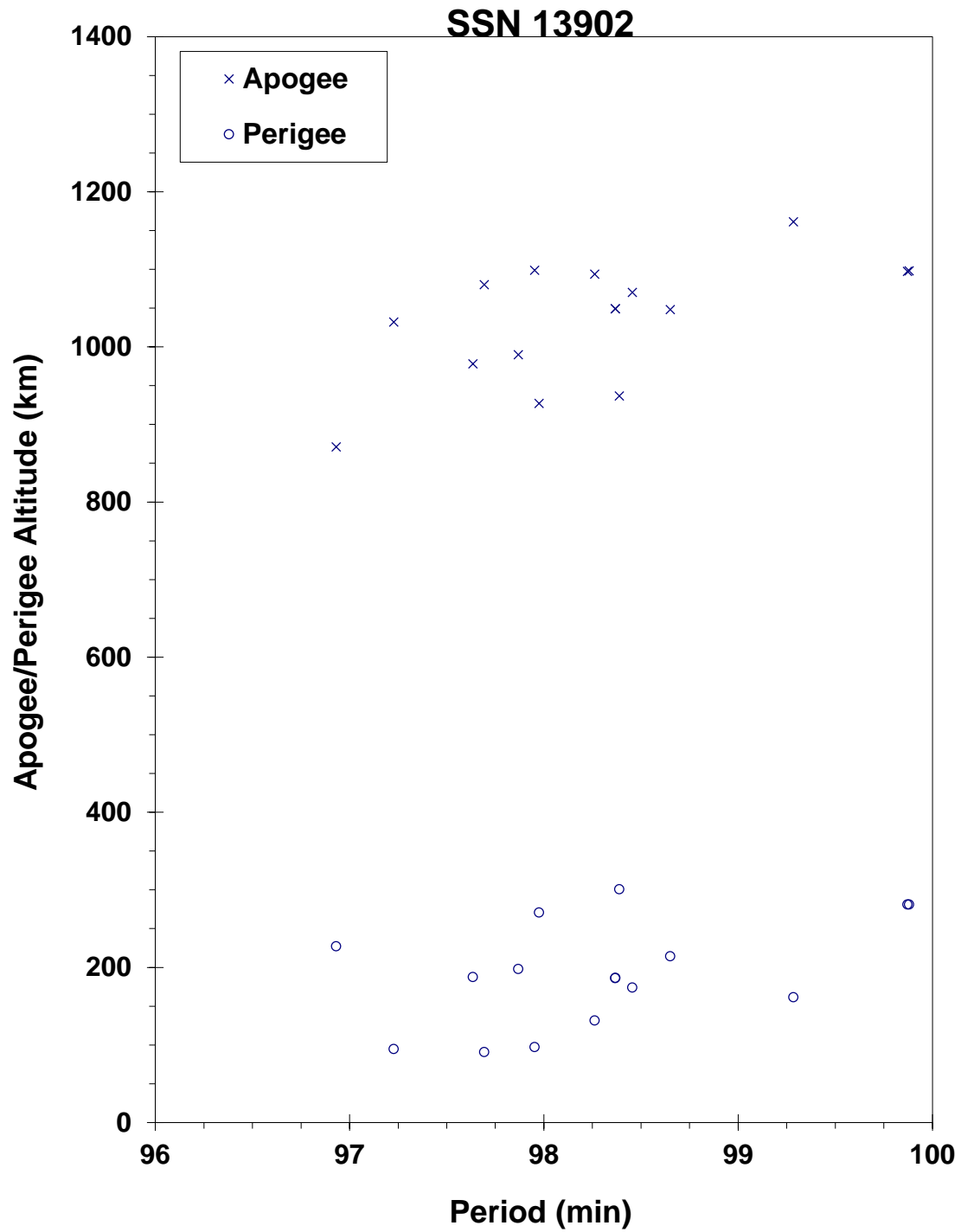
Parent satellite was one of two small engine units that are routinely released after the first burn of the Proton fourth stage. The nature of these objects was identified by Dr. Boris V. Cherniatiev, Deputy Constructor for the Energiya NPO, in October 1992. The cause of this fragmentation is assumed to be related to the residual hypergolic propellants on board and failure of the membrane separating the fuel and oxidizer. Element sets on 16 fragments were developed. None were officially cataloged. This was the first in a series of fragmentations of this object type.

REFERENCE DOCUMENTS

The Fragmentation of Proton Debris, D. J. Nauer, TBE Technical Report CS93-LKD-004, Teledyne Brown Engineering, Colorado Springs, 31 December 1992.

“Identification and Resolution of an Orbital Debris Problem with the Proton Launch Vehicle”, B.V. Cherniatiev et al, Proceedings of the First European Conference on Space Debris, April 1993.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Astron ullage motor debris cloud of 16 fragments as determined within a few days of the first event. Elements from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 28.66 Mar 1983
DRY MASS (KG): 1000
MAIN BODY: Cylinder-box; 1.9 m diameter by 7.5 m length
MAJOR APPENDAGES: 1 solar panel
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|-------------|-----------------|-----------------|
| DATE: | 30 Dec 1985 | LOCATION: | 68S, 300E (dsc) |
| TIME: | 1005 GMT | ASSESSED CAUSE: | Battery |
| ALTITUDE: | 825 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 85348.40460348 | MEAN ANOMALY: | 83.2801 |
| RIGHT ASCENSION: | 16.9717 | MEAN MOTION: | 14.22481975 |
| INCLINATION: | 98.6488 | MEAN MOTION DOT/2: | .00000037 |
| ECCENTRICITY: | .0015724 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 276.6589 | BSTAR: | .000025130 |

DEBRIS CLOUD DATA

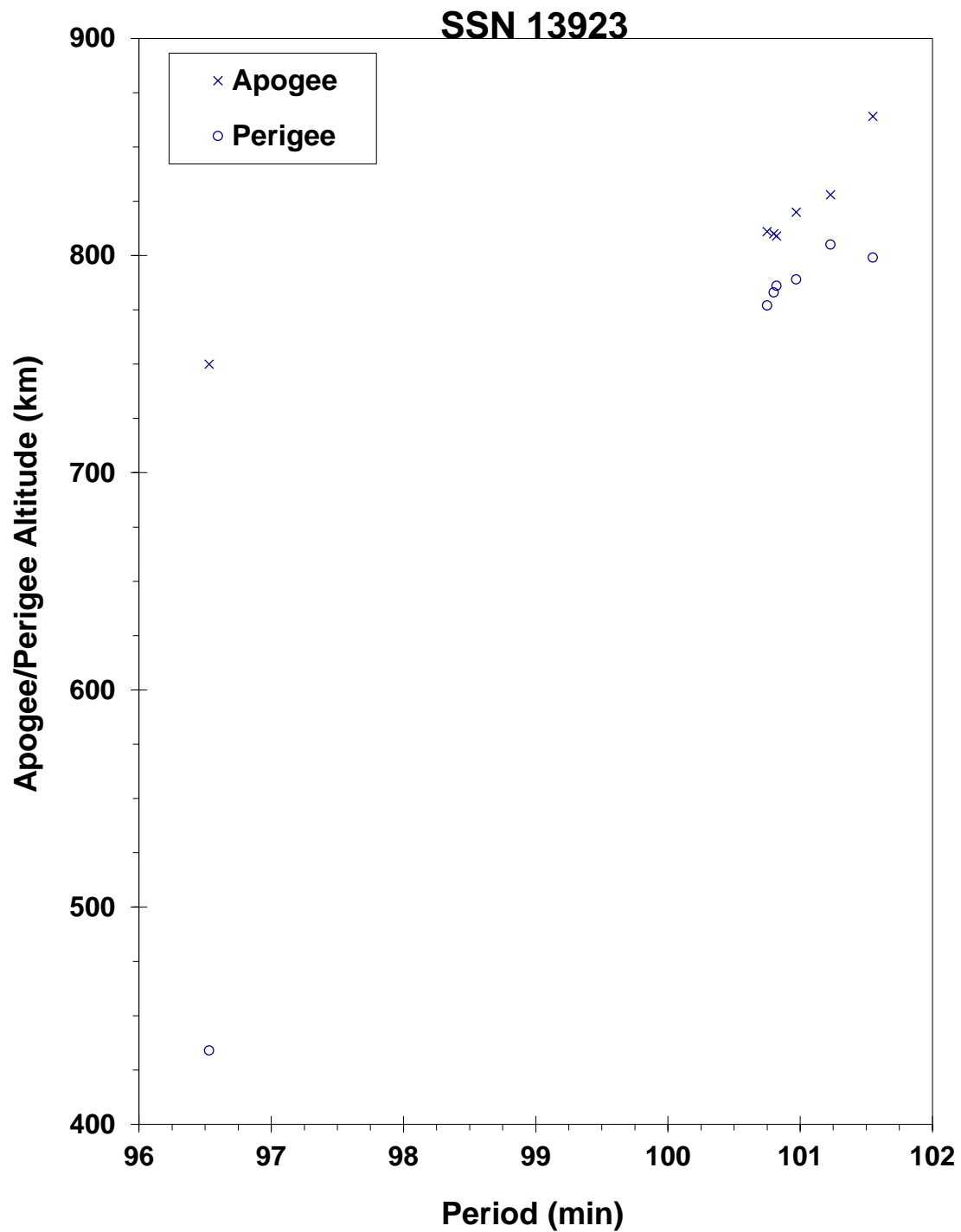
MAXIMUM ΔP : 4.7 min
MAXIMUM ΔI : 0.1 deg

COMMENTS

A malfunction on NOAA 8 caused a battery to overcharge, resulting in a minor explosion of the battery. The spacecraft was operational at the time of the event. Six new fragments were detected and cataloged. All decayed by February 1989, leaving the parent still in orbit.

REFERENCE DOCUMENT

"NOAA Turns Off Satellite Following Malfunction", Aviation Week and Space Technology, 13 January 1986, p. 21.



NOAA 8 debris cloud of 6 fragments plus the parent satellite 1 day after the event as reconstructed from the Naval Space Surveillance System database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 25.81 Apr 1983
DRY MASS (KG): 1250
MAIN BODY: Cylinder; 1.7 m diameter by 2 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 13 Aug 1983 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 83225.00107283 | MEAN ANOMALY: | 4.5332 |
| RIGHT ASCENSION: | 79.8630 | MEAN MOTION: | 2.00589678 |
| INCLINATION: | 63.3076 | MEAN MOTION DOT/2: | .0 |
| ECCENTRICITY: | .7324437 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 320.0041 | BSTAR: | .0068163 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 4.8 min*
MAXIMUM ΔI : 0.4 deg*

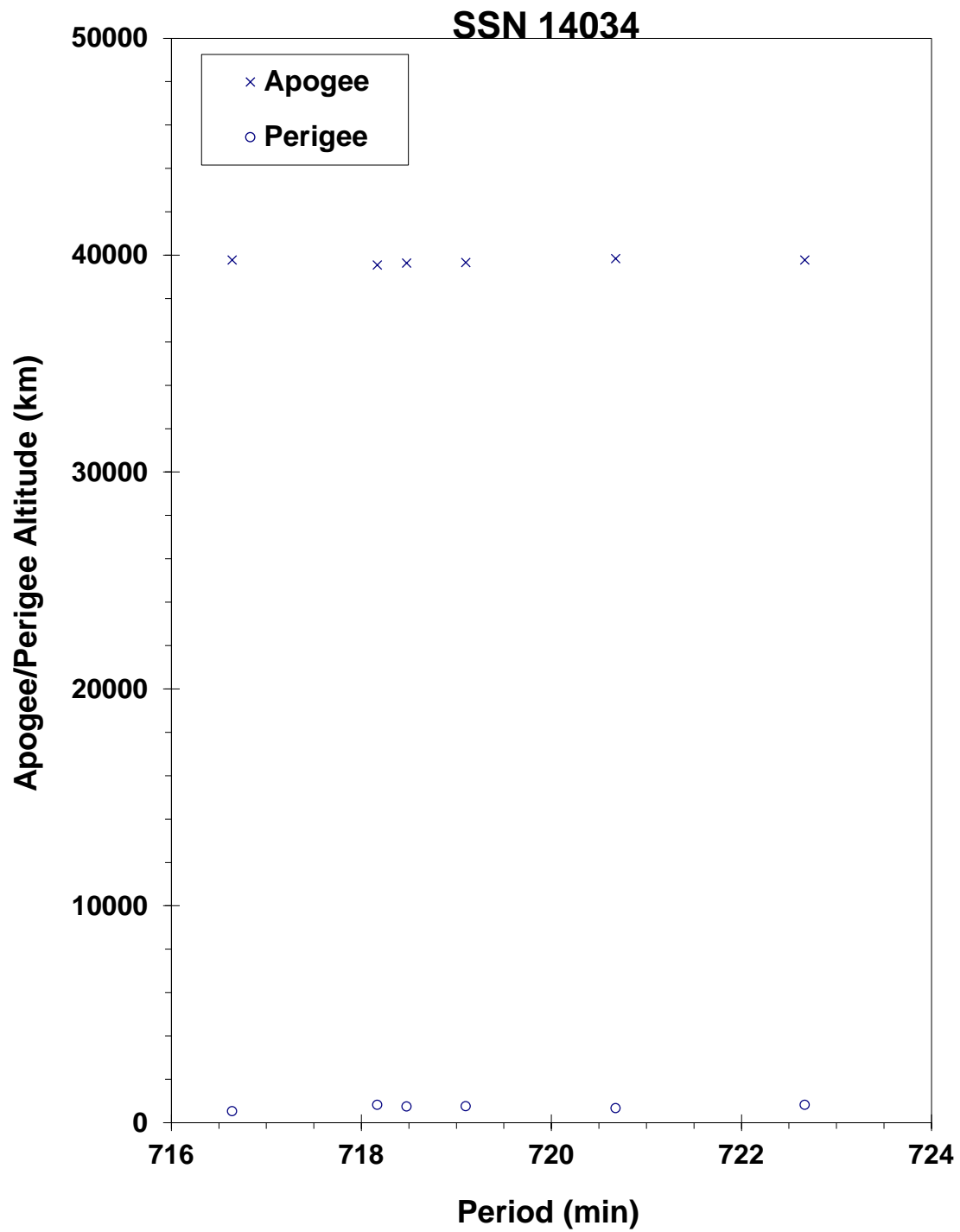
*Based on uncataloged debris data

COMMENTS

Cosmos 1456 was another spacecraft of the Cosmos 862-type to experience a fragmentation. The spacecraft may have been active at the time of the event, having last made a station-keeping maneuver on 22 June 1983. The next station-keeping maneuver should have occurred in the second half of August or early September 1983. The spacecraft began drifting off station immediately after the event and never recovered.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1456 debris cloud of 6 fragments less than 3 weeks after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 7.44 May 1983
DRY MASS (KG): 3000
MAIN BODY: Cylinder; 1.3 m diameter by 17 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge (?)

EVENT DATA (1)

| | | | |
|-----------|-------------|-----------------|----------------|
| DATE: | 11 Mar 1985 | LOCATION: | 4S, 196E (asc) |
| TIME: | 0940 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 750 km | | |

PRE-EVENT ELEMENTS (1)

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 85068.60956125 | MEAN ANOMALY: | 101.2285 |
| RIGHT ASCENSION: | 157.6403 | MEAN MOTION: | 14.49322542 |
| INCLINATION: | 65.0244 | MEAN MOTION DOT/2: | .00000357 |
| ECCENTRICITY: | .0224980 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 256.3703 | BSTAR: | .000080310 |

EVENT DATA (2)

| | | | |
|-----------|-------------|-----------------|----------------|
| DATE: | 13 May 1985 | LOCATION: | 10N, 82E (asc) |
| TIME: | 0133 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 845 km | | |

PRE-EVENT ELEMENTS (2)

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 85125.54047130 | MEAN ANOMALY: | 121.1528 |
| RIGHT ASCENSION: | 353.4544 | MEAN MOTION: | 14.49239036 |
| INCLINATION: | 65.0248 | MEAN MOTION DOT/2: | .0 |
| ECCENTRICITY: | .0222492 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 236.8082 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 5.9 min*
MAXIMUM ΔI : 1.0 deg*

*Based on uncataloged debris data

COMMENTS

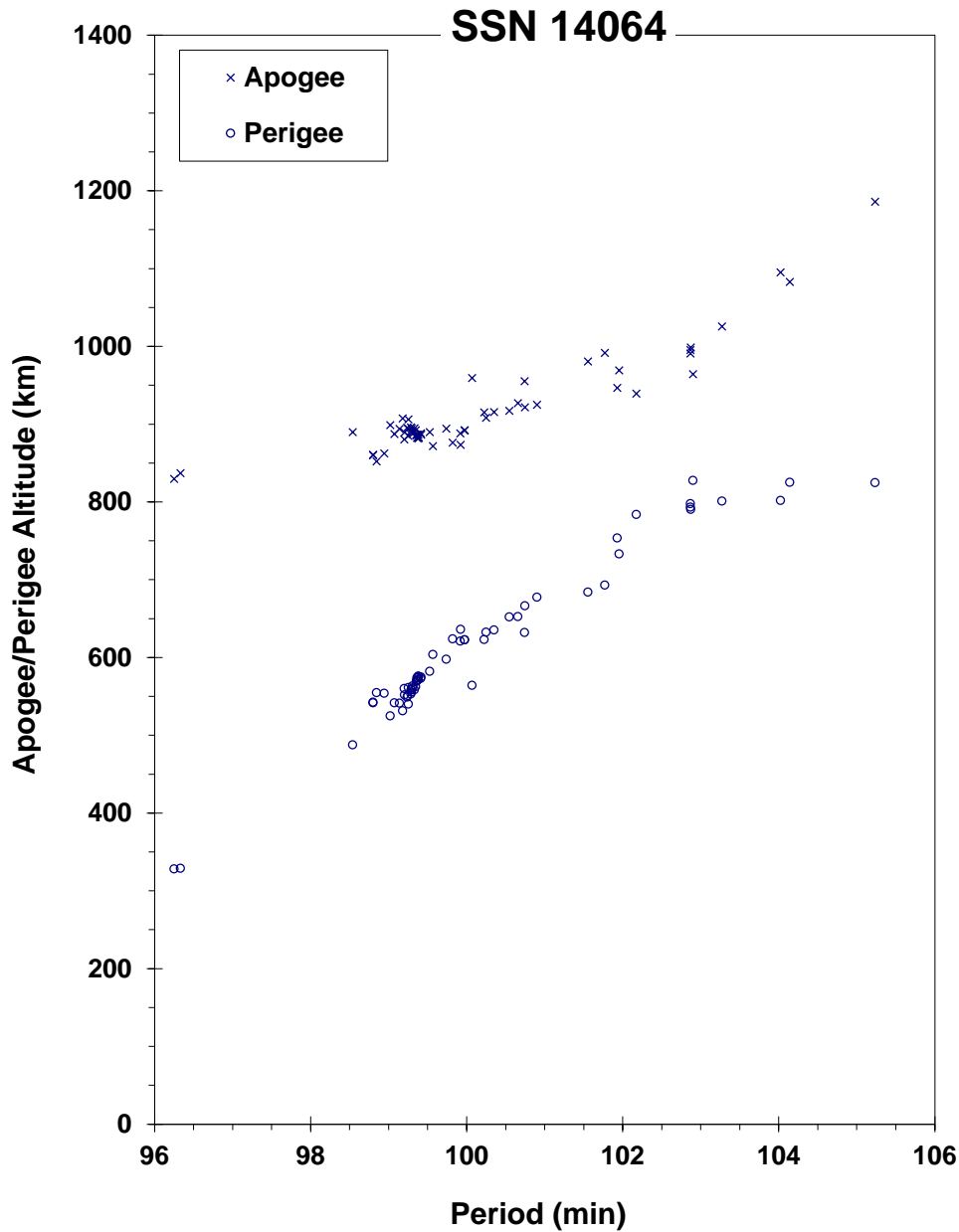
Cosmos 1461 was the twelfth spacecraft of the Cosmos 699-type to experience a fragmentation. Cosmos 1461 entered a natural decay regime more than 13 months prior to first event. After the first event as many as 20 fragments were detected but only six new objects were cataloged. The second event occurred 2 months later and produced considerably more debris. These events followed the pattern set by Cosmos 1220 and Cosmos 1260.

REFERENCE DOCUMENTS

"Artificial Satellite Break-Ups (Part 1): Soviet Ocean Surveillance Satellites", N. L. Johnson, Journal of the British Interplanetary Society, February 1983, pp. 51-58.

Analysis of the Fragmentation of Kosmos 1461, G.T. DeVere and N.L. Johnson, Technical Report CS85-BMDSC-0056, Teledyne Brown Engineering, Colorado Springs, September 1985.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1461 debris cloud remnant of 65 fragments 4 days after the second event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 8.80 Jul 1983
DRY MASS (KG): 1250
MAIN BODY: Cylinder; 1.7 m diameter by 2 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

| | | | |
|-----------|------------|-----------------|------------|
| DATE: | 9 Jul 1983 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 83189.85702098 | MEAN ANOMALY: | 4.6462 |
| RIGHT ASCENSION: | 166.3194 | MEAN MOTION: | 2.03523282 |
| INCLINATION: | 62.9394 | MEAN MOTION DOT/2: | .00000702 |
| ECCENTRICITY: | .7337681 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 317.9301 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 8.7 min*
MAXIMUM ΔI : 0.8 deg*

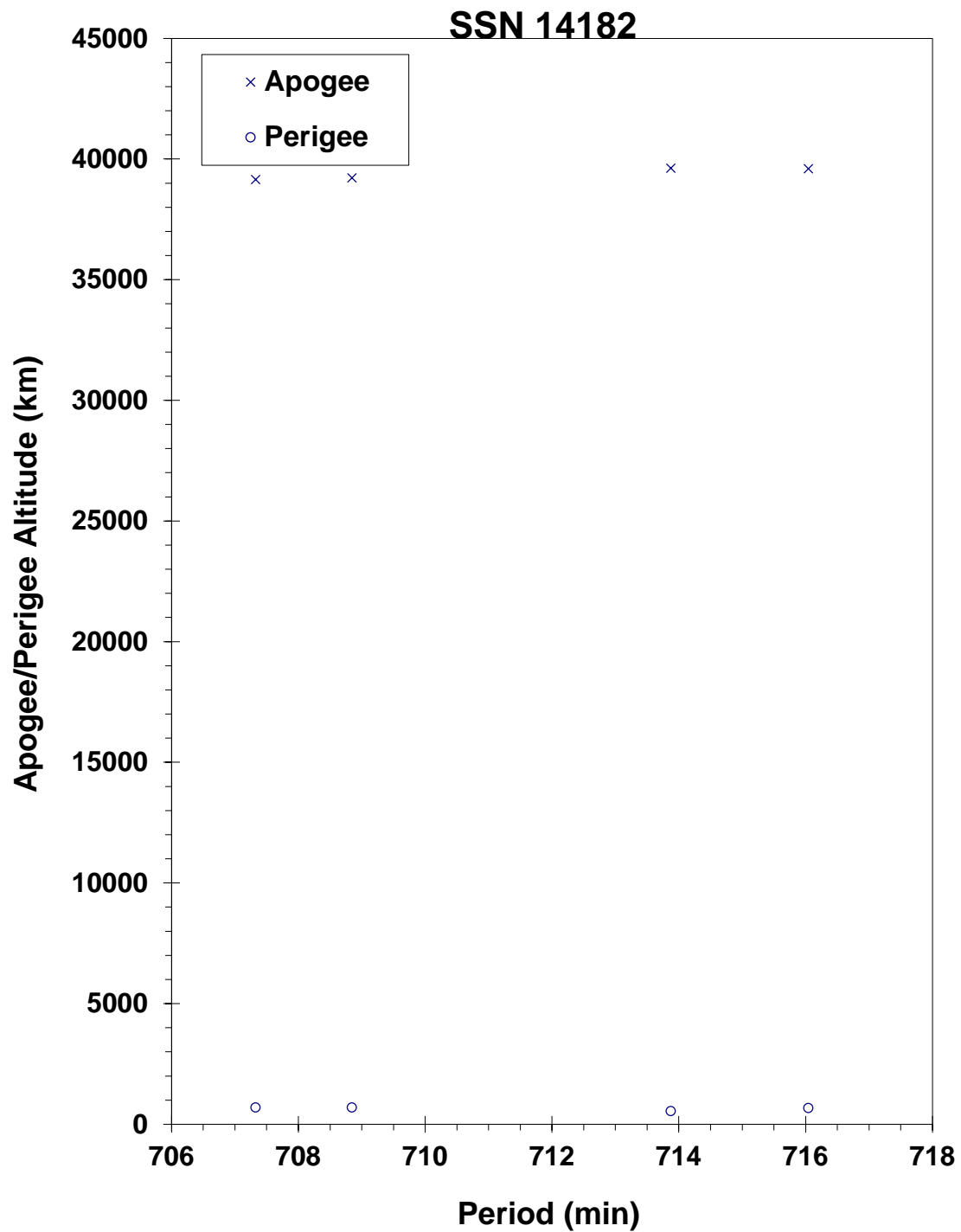
*Based on uncataloged debris data

COMMENTS

Cosmos 1481 was the twelfth spacecraft of the Cosmos 862-type to experience a fragmentation. The event apparently occurred within a day of launch. An expected orbital maneuver by Cosmos 1481 to move from its transfer orbit to an operational orbit about 3 days after launch was never performed.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1481 debris cloud of 4 objects 1 month after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 24.25 Jul 1983
DRY MASS (KG): 1800
MAIN BODY: Cylinder; 1.5 m diameter by 5.0 m length
MAJOR APPENDAGES: Solar panels, antenna
ATTITUDE CONTROL: Gravity gradient; momentum wheels
ENERGY SOURCES: Electrical system (?); pressurized vessels

EVENT DATA

DATE: 18 Oct 1993 LOCATION: 7S, 111E (asc)
TIME: 1204 GMT ASSESSED CAUSE: Unknown
ALTITUDE: 605 km

PRE-EVENT ELEMENTS

EPOCH: 93289.76777232 MEAN ANOMALY: 40.8047
RIGHT ASCENSION: 316.3082 MEAN MOTION: 14.98254133
INCLINATION: 97.5219 MEAN MOTION DOT/2: .00001299
ECCENTRICITY: .0033451 MEAN MOTION DOT DOT/6: .0
ARG. OF PERIGEE: 319.0655 BSTAR: .00011294

DEBRIS CLOUD DATA

MAXIMUM ΔP : 14.0 min *
MAXIMUM ΔI : 2.5 deg *

* Based on uncataloged debris data

COMMENTS

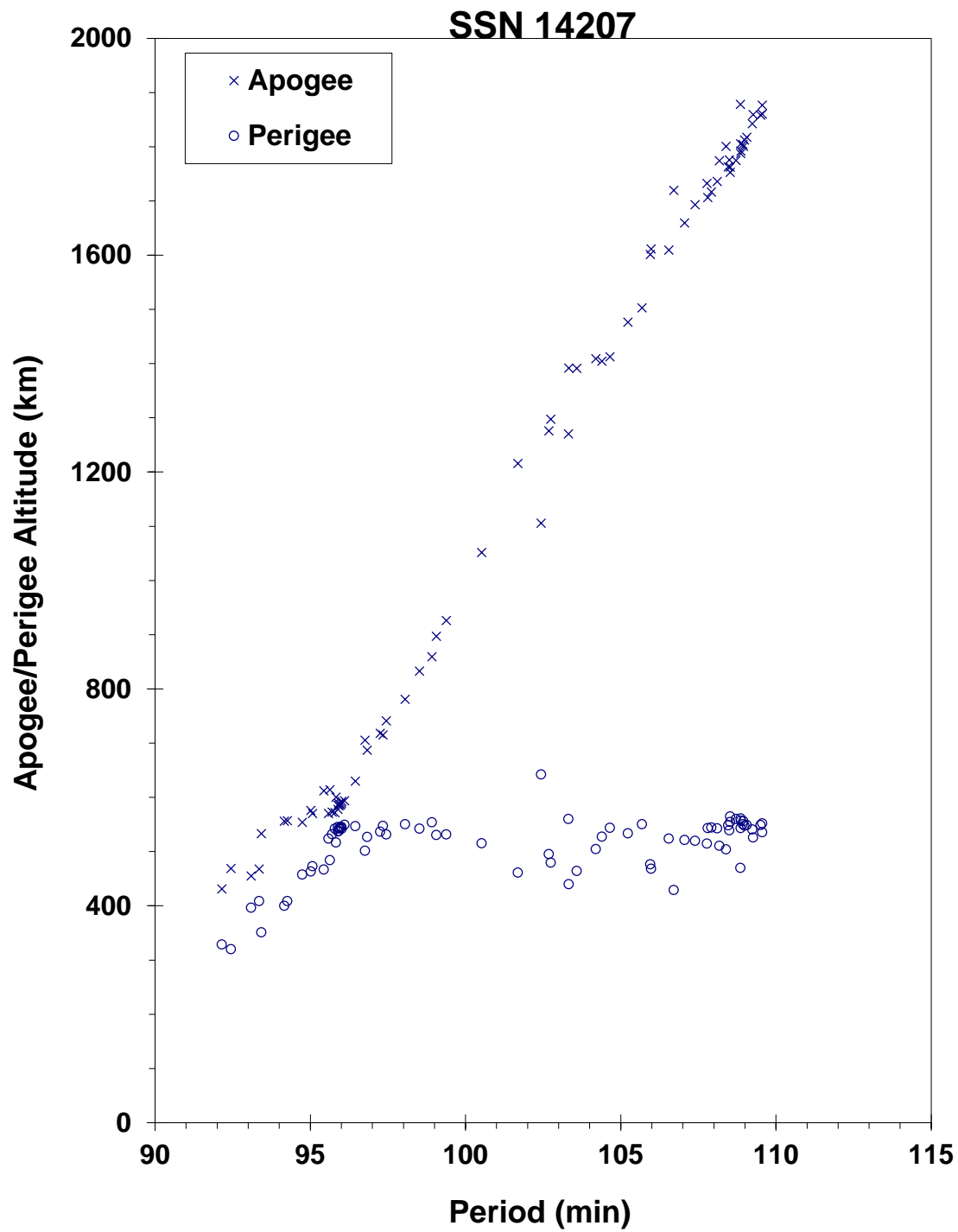
Cosmos 1484 was the third of four Resurs-O prototypes flown in sun-synchronous orbits, and the only one to fragment. This is the first sun-synchronous Russian satellite to ever fragment. The NAVSPOC generated 79 analyst satellites on this event.

REFERENCE DOCUMENTS

The Fragmentation of Cosmos 1484, D. J. Nauer, Technical Report CS94-LKD-003, Teledyne Brown Engineering, Colorado Springs, 17 November 1993.

The Soviet Year in Space, 1990, N. L. Johnson, Teledyne Brown Engineering, 1991.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1484 debris cloud of 79 fragments assembled by the NAVSPOC.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 29.04 Dec 1983
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

DATE: 4 Feb 1991 LOCATION: 28N, 106E (dsc)
TIME: 0312 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 18550 km

PRE-EVENT ELEMENTS

EPOCH: 91032.22560633 MEAN ANOMALY: 10.4843
RIGHT ASCENSION: 133.4557 MEAN MOTION: 4.30882556
INCLINATION: 51.9464 MEAN MOTION DOT/2: .00004140
ECCENTRICITY: .5787304 MEAN MOTION DOT DOT/6: .0
ARG. OF PERIGEE: 315.5487 BSTAR: .0018354

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

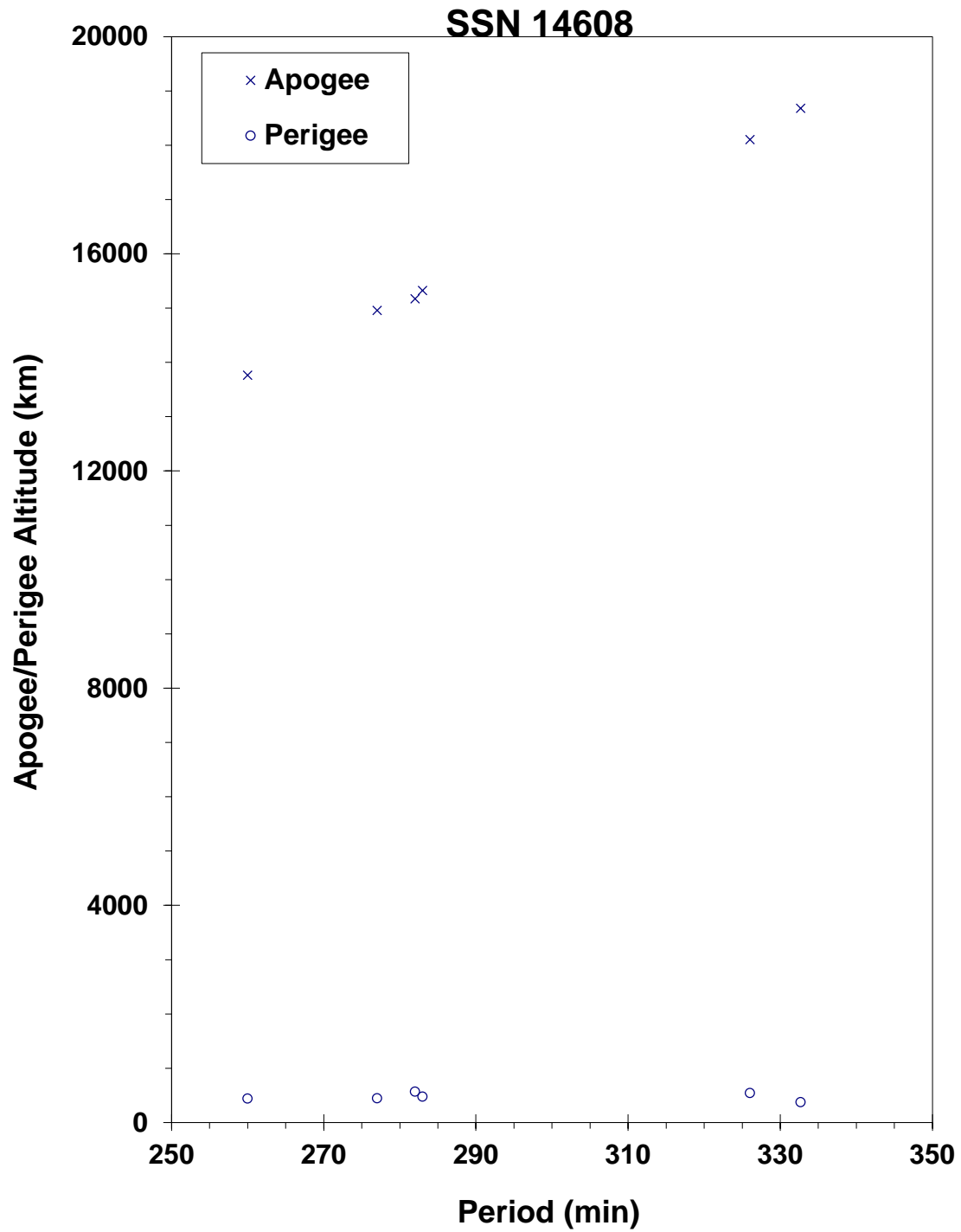
Parent satellite was one of two small engine units that are routinely released after the first burn of the Proton fourth stage. The nature of these objects was identified by Dr. Boris V. Chernlatiev, Deputy Constructor for the Energiya NPO, in October 1992. The cause of this fragmentation is assumed to be related to the residual hypergolic propellants on board and failure of the membrane separating the fuel and oxidizer. NAVSPASUR observed at least 12 fragments on the day of the event and approximately three dozen on 7 February. An element set was initially developed on only one new fragment. This was the third in a series of fragmentations of this object type.

REFERENCE DOCUMENTS

The Fragmentation of Proton Debris, D. J. Nauer, TBE Technical Report CS93-LKD-004, Teledyne Brown Engineering, Colorado Springs, 31 December 1992.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

"Identification and Resolution of an Orbital Debris Problem with the Proton Launch Vehicle", B. V. Cherniatiev et al, Proceedings of the First European Conference on Space Debris, April 1993.



Gabbard diagram of the five cataloged and single analyst satellite generated on the Cosmos 1519-21 debris cloud.

SATELLITE DATA

TYPE: PAM-D Upper Stage (STAR 48 motor)
OWNER: US
LAUNCH DATE: 3.54 Feb 1984
DRY MASS (KG): 2200
MAIN BODY: Sphere-nozzle; 1.2 m by 2.1 m
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Spin-stabilized
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|------------|-----------------|----------------|
| DATE: | 6 Feb 1984 | LOCATION: | 0N, 120E (asc) |
| TIME: | 1600 GMT | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | 280 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 84037.35377144 | MEAN ANOMALY: | 82.4657 |
| RIGHT ASCENSION: | 138.8370 | MEAN MOTION: | 15.97451864 |
| INCLINATION: | 28.4669 | MEAN MOTION DOT/2: | .00197501 |
| ECCENTRICITY: | .0006481 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 277.3659 | BSTAR: | .00040999 |

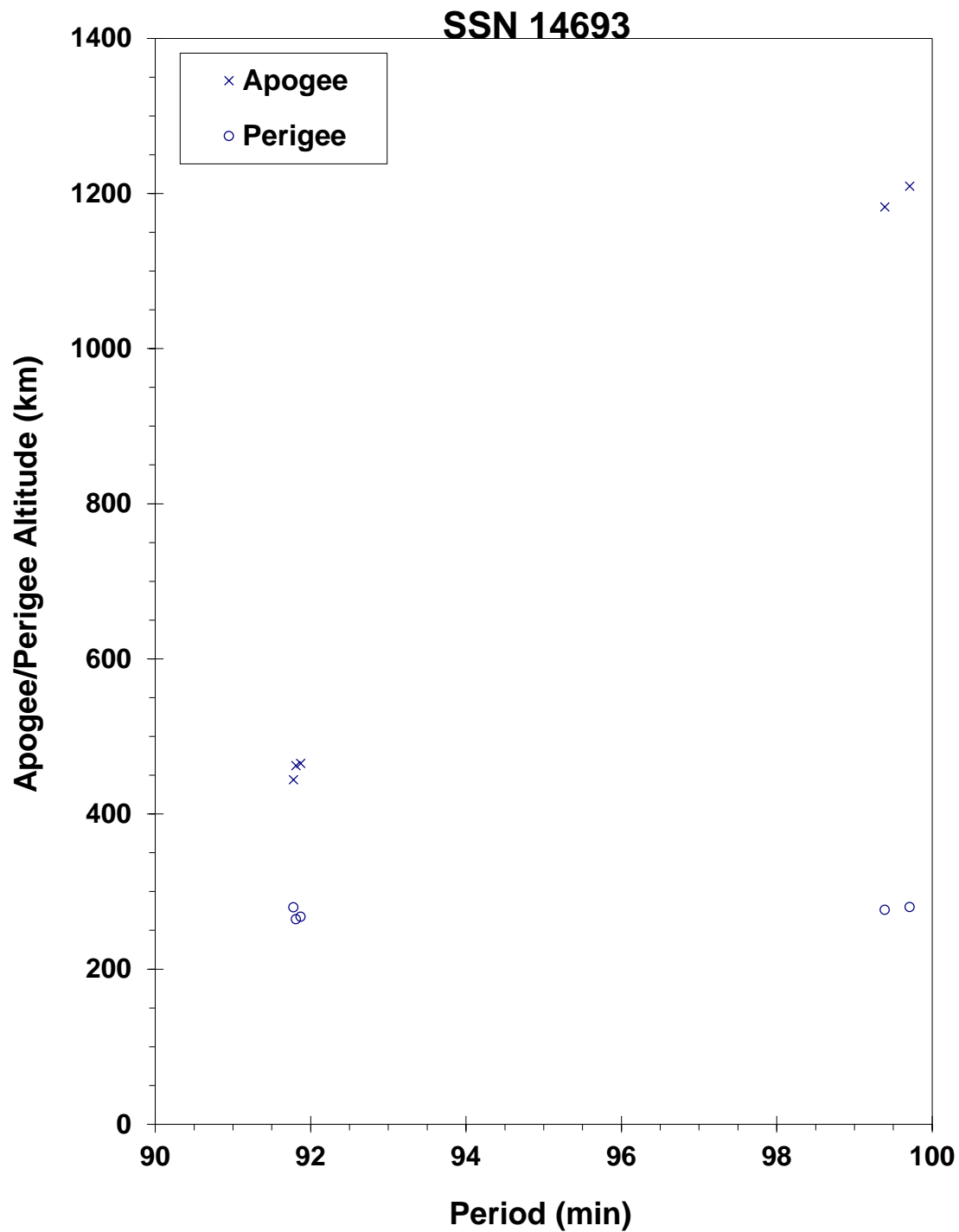
DEBRIS CLOUD DATA

MAXIMUM ΔP : 9.4 min*
MAXIMUM ΔI : 0.3 deg*

*Based on uncataloged debris data

COMMENTS

Palapa B2 and its PAM-D upper stage were deployed from the Space Shuttle Challenger at 1513 GMT, 6 February 1984. Ignition of the upper stage occurred on schedule at 1600 GMT, but the nozzle fragmented within 10 seconds. Without the nozzle the burn could not be sustained and a natural shutdown quickly followed. The PAM-D then separated from Palapa B2. The above elements are for the Shuttle prior to deployment. The Shuttle made a small posigrade evasive maneuver after deployment and before ignition of the PAM-D. See also Westar 6 R/B fragmentation.



Palapa B2 R/B debris cloud of 5 fragments about 3 days after the event as reconstructed from the US SSN database. The Palapa B2 R/B is the object with the second highest orbital period.

SATELLITE DATA

TYPE: PAM-D Upper Stage (STAR 48 motor)
OWNER: US
LAUNCH DATE: 3.54 Feb 1984
DRY MASS (KG): 2200
MAIN BODY: Sphere-nozzle; 1.2 m by 2.1 m
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Spin-stabilized
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|------------|-----------------|---------------|
| DATE: | 3 Feb 1984 | LOCATION: | 0N, 56E (asc) |
| TIME: | 2145 GMT | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | 305 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 84034.84362284 | MEAN ANOMALY: | 48.7355 |
| RIGHT ASCENSION: | 157.5848 | MEAN MOTION: | 15.88299499 |
| INCLINATION: | 28.4660 | MEAN MOTION DOT/2: | .00000250 |
| ECCENTRICITY: | .0006644 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 311.2683 | BSTAR: | .0 |

DEBRIS CLOUD DATA

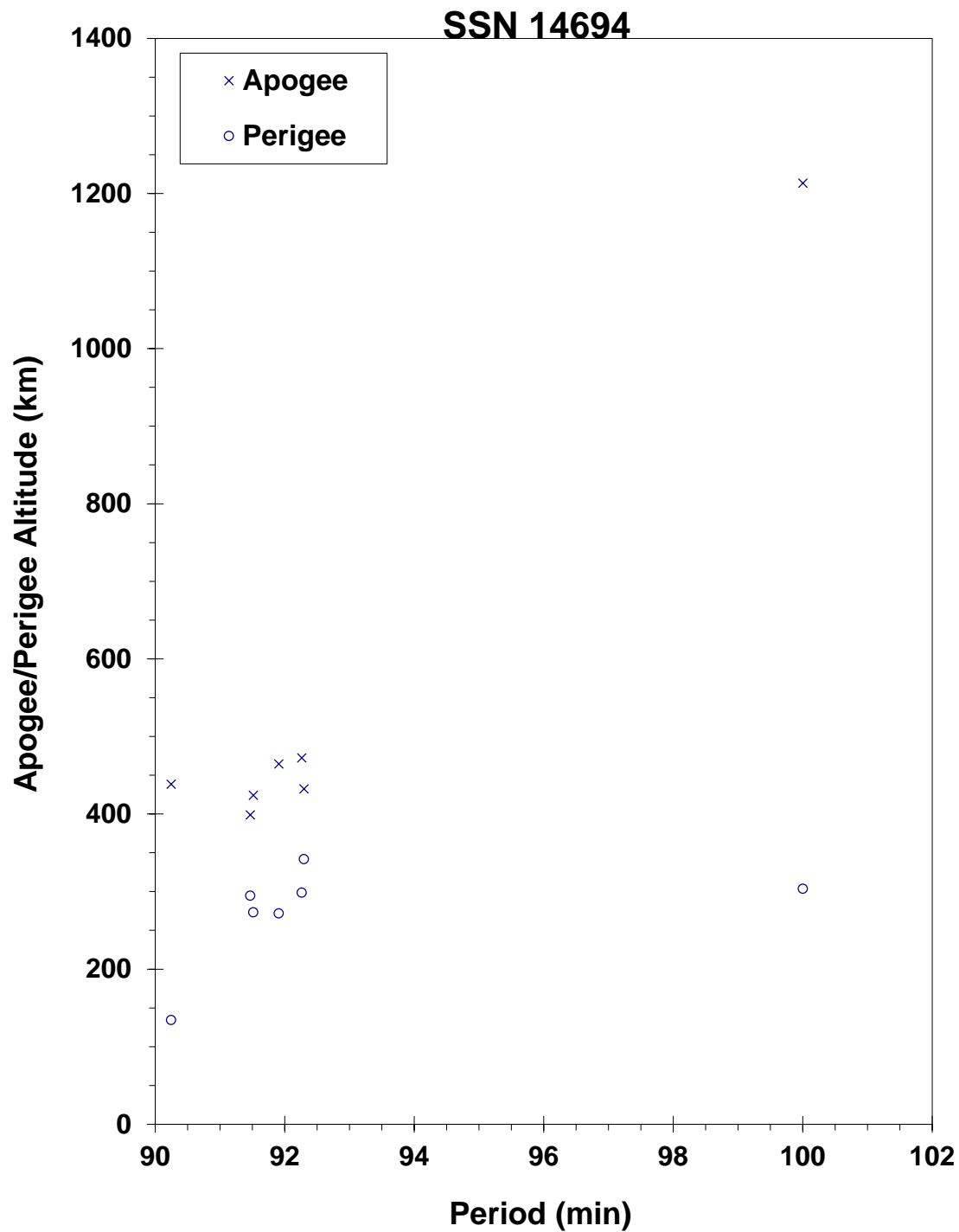
MAXIMUM ΔP : 9.7 min
MAXIMUM ΔI : 0.8 deg

COMMENTS

Westar 6 and its PAM-D upper stage were deployed from the Space Shuttle Challenger at 2100 GMT, 3 February 1984. Ignition of the upper stage occurred on schedule at 2145 GMT but the nozzle fragmented within 10 seconds. Without the nozzle the burn could not be sustained and a natural shutdown quickly followed. The PAM-D then separated from Westar 6. See also Palapa B2 R/B fragmentation.

REFERENCE DOCUMENT

Westar Failure. Technical Memorandum from N.L. Johnson, Teledyne Brown Engineering, to Preston Landry, NORAD/ADCOM/XPYS, Colorado Springs, 7 February 1984.



Westar 6 R/B debris cloud of 7 fragments less than 2 days after the event as reconstructed from the US SSN database. The Westar 6 R/B is the object in the high, 100-min orbit.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 7.95 Aug 1984
DRY MASS (KG): 3000
MAIN BODY: Cylinder; 1.3 m diameter by 17 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge (?)

EVENT DATA

DATE: 23 Feb 1986 LOCATION: 29N, 187E (asc)
TIME: 1850 GMT ASSESSED CAUSE: Unknown
ALTITUDE: 430 km

PRE-EVENT ELEMENTS

EPOCH: 86048.57631415 MEAN ANOMALY: 72.5463
RIGHT ASCENSION: 268.3025 MEAN MOTION: 15.47795866
INCLINATION: 65.0271 MEAN MOTION DOT/2: .00005888
ECCENTRICITY: .0022403 MEAN MOTION DOT DOT/6: .0
ARG. OF PERIGEE: 287.3230 BSTAR: .00011680

DEBRIS CLOUD DATA

MAXIMUM ΔP : 2.0 min
MAXIMUM ΔI : 0.4 deg

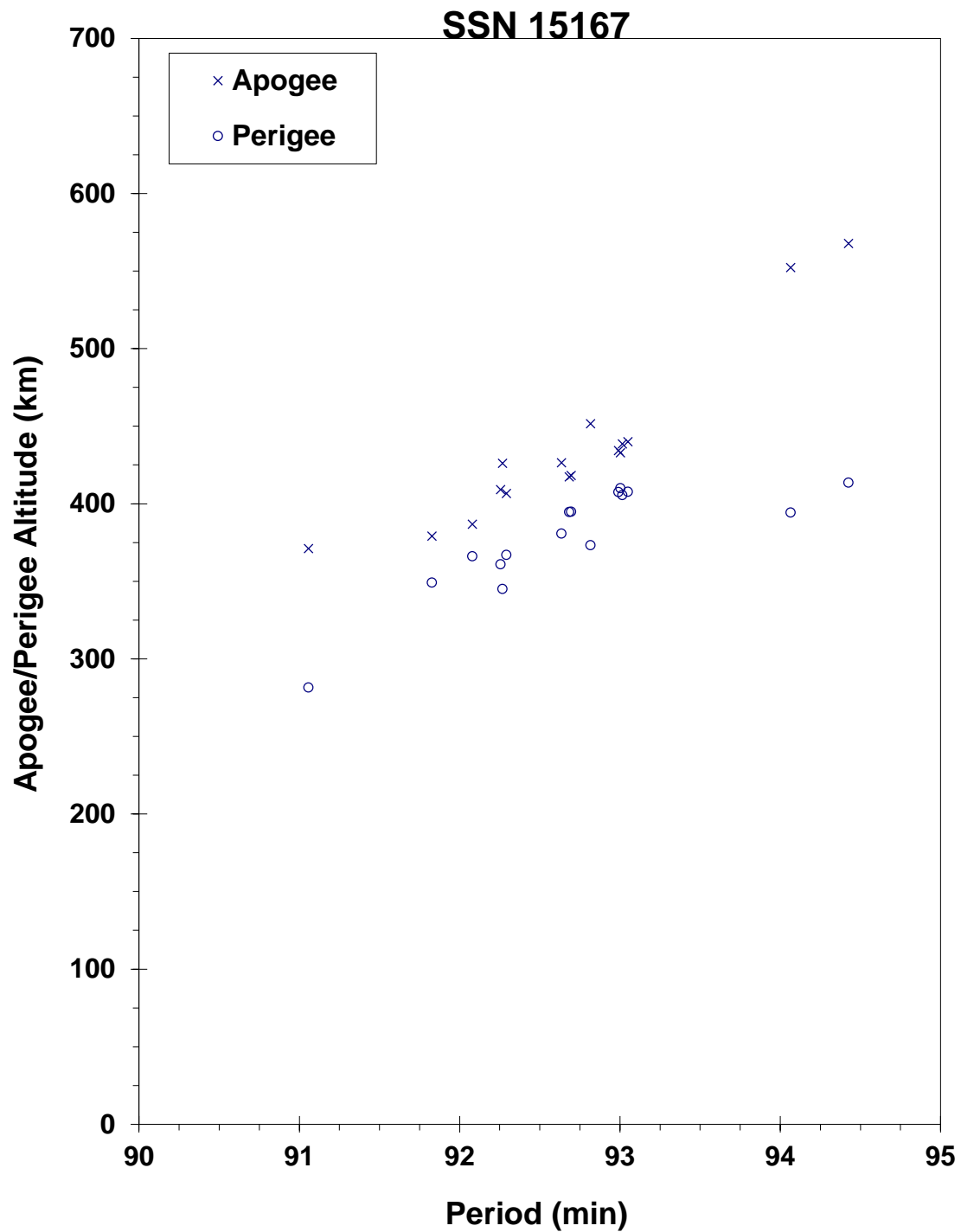
COMMENTS

Cosmos 1588 was the thirteenth spacecraft of the Cosmos 699-type to experience a fragmentation. Spacecraft had been in natural decay for 7 months prior to the event.

REFERENCE DOCUMENTS

"Artificial Satellite Break-Ups (Part 1): Soviet Ocean Surveillance Satellites", N. L. Johnson, Journal of the British Interplanetary Society, February 1983, pp. 51-58.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1588 cataloged debris cloud of 16 fragments 3 weeks after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 28.58 Sep 1984
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

DATE: 5 Sep 1992 LOCATION: 46.1S, 351.8E
TIME: 1451 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 835 km

PRE-EVENT ELEMENTS

EPOCH: 92249.36121283 MEAN ANOMALY: 6.5555
RIGHT ASCENSION: 353.4197 MEAN MOTION: 14.15474339
INCLINATION: 66.5712 MEAN MOTION DOT/2: -.00009086
ECCENTRICITY: .0007128 MEAN MOTION DOT DOT/6: .0
ARG. OF PERIGEE: 353.5641 BSTAR: -.004641

DEBRIS CLOUD DATA

MAXIMUM ΔP : 2.9 min
MAXIMUM ΔI : 0.5 deg

COMMENTS

Parent satellite was one of two small engine units that are routinely released after the first burn of the Proton fourth stage. The nature of these objects was identified by Dr. Boris V. Chernlatiev, Deputy Constructor for the Energiya NPO, in October 1992. The cause of this fragmentation is assumed to be related to the residual hypergolic propellants on board and failure of the membrane separating the fuel and oxidizer. NAVSPASUR has observed 62 objects associated with this breakup. This was the sixth in a series of fragmentations of this object type.

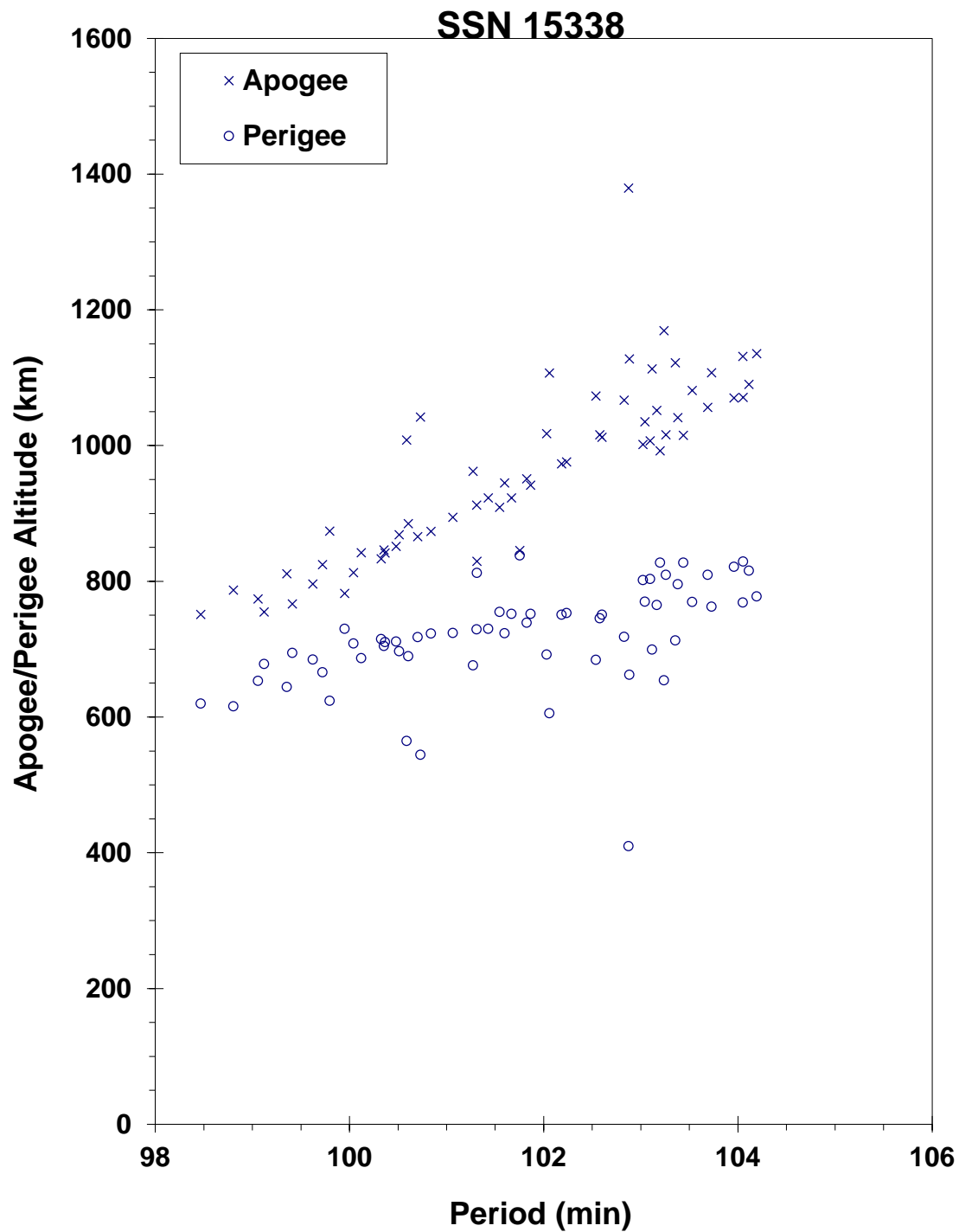
REFERENCE DOCUMENTS

Soviet Space Programs 1980-1985, Science and Technology Series, Volume 66, Nicholas L. Johnson, American Astronautical Society, Univelt, Inc., 1987.

The Fragmentation of Proton Debris, D. J. Nauer, TBE Technical Report CS93-LKD-004, Teledyne Brown Engineering, Colorado Springs, 31 December 1992.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

"Identification and Resolution of an Orbital Debris Problem with the Proton Launch Vehicle", B. V. Cherniatiev et al, Proceedings of the First European Conference on Space Debris, April 1993.



Naval Space Surveillance System tracked 62 objects within the Cosmos 1603 debris cloud, with 22 appearing in the Satellite Catalog.

SATELLITE DATA

TYPE: Ariane 3 Final Stage
OWNER: France
LAUNCH DATE: 10.05 Nov 1984
DRY MASS (KG): ~1100
MAIN BODY: Unknown
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 20 Nov 1984 | LOCATION: | Unknown |
| TIME: | 1425Z | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|-------------|------------------------|------------|
| EPOCH: | 84325.41617 | MEAN ANOMALY: | 146.5463 |
| RIGHT ASCENSION: | 236.1289 | MEAN MOTION: | 2.26087292 |
| INCLINATION: | 7.0293 | MEAN MOTION DOT/2: | .00001128 |
| ECCENTRICITY: | .7265710 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 187.8823 | BSTAR: | .0010954 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

This Ariane R/B fragmentation occurred to 10 days after launch but not detected until 2003. This is the first Ariane Rocket Body fragmentation that is attributed to France. Previous Ariane Stages were attributed to ESA.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 18.90 Apr 1985
DRY MASS (KG): 3000
MAIN BODY: Cylinder; 1.3 m diameter by 17 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge (?)

EVENT DATA

DATE: 20 Nov 1987 LOCATION: 65N, 300E (dsc)
TIME: 0131 GMT ASSESSED CAUSE: Unknown
ALTITUDE: 410 km

PRE-EVENT ELEMENTS

EPOCH: 87323.98216942 MEAN ANOMALY: 105.3951
RIGHT ASCENSION: 286.0367 MEAN MOTION: 15.56048984
INCLINATION: 65.0306 MEAN MOTION DOT/2: .00039428
ECCENTRICITY: .0018658 MEAN MOTION DOT DOT/6: .0
ARG. OF PERIGEE: 254.4728 BSTAR: .00055895

DEBRIS CLOUD DATA

MAXIMUM ΔP : 5.5 min*
MAXIMUM ΔI : 0.2 deg*

*Based on cataloged and uncataloged debris data

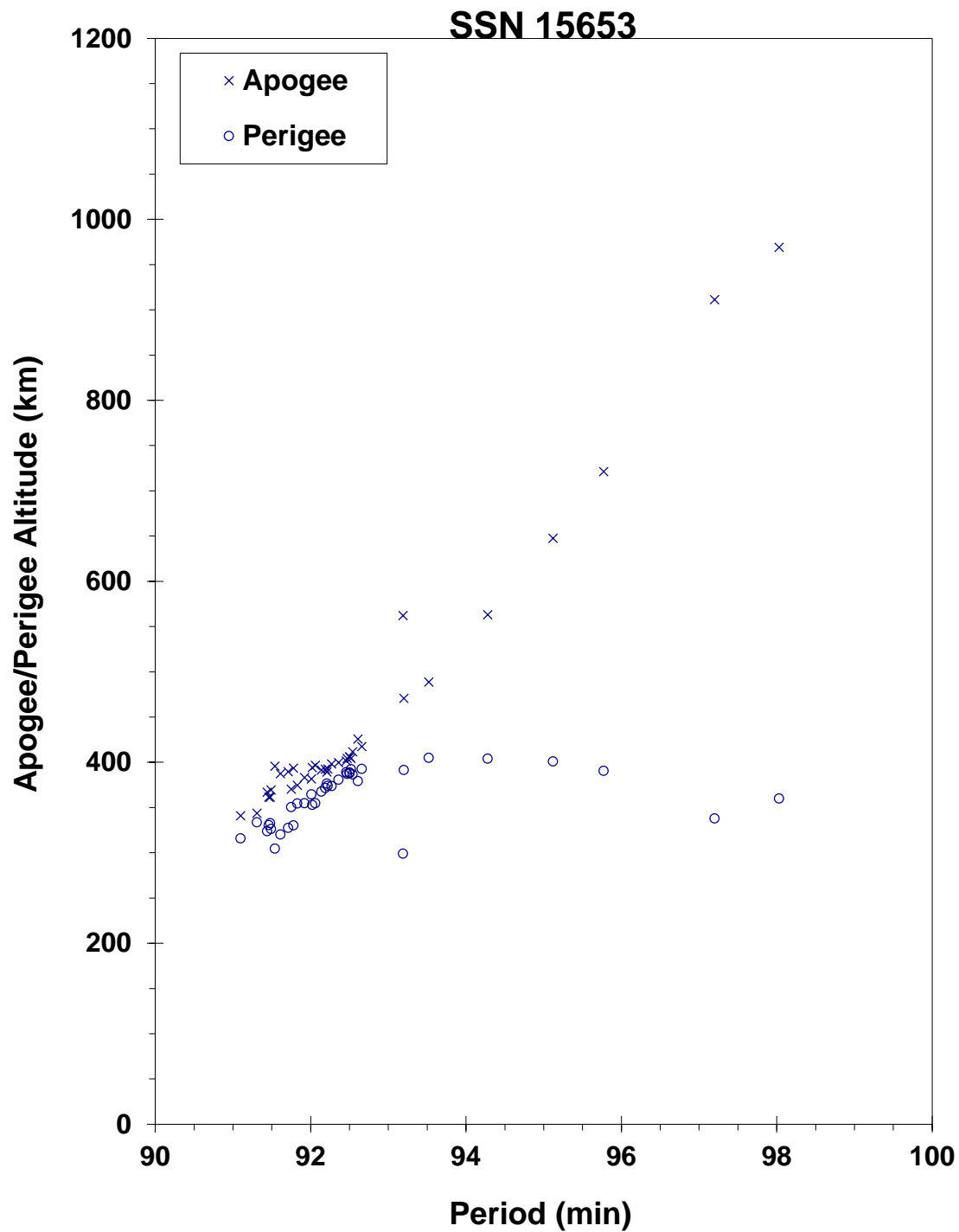
COMMENTS

Cosmos 1646 was the sixteenth spacecraft of the Cosmos 699-type to experience a fragmentation. Spacecraft had been in natural decay for nearly 20 months prior to the event. Many debris reentered before being officially cataloged.

REFERENCE DOCUMENTS

"Artificial Satellite Break-Ups (Part 1): Soviet Ocean Surveillance Satellites", N. L. Johnson, Journal of the British Interplanetary Society, February 1983, pp. 51-58.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1646 debris cloud remnant of 38 fragments about 10 days after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 17 May 1985
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

DATE: 29 Nov 1998 LOCATION: 38.3N, 172.6E
TIME: 0925 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 16420 km

POST-EVENT ELEMENTS

EPOCH: 98332.38151447 MEAN ANOMALY: 98.9018
RIGHT ASCENSION: 344.4719 MEAN MOTION: 4.35077855212150
INCLINATION: 52.0277 MEAN MOTION DOT/2: .00009109
ECCENTRICITY: .5772516 MEAN MOTION DOT DOT/6: 00000-0
ARG. OF PERIGEE: 209.7130 BSTAR: .030939

DEBRIS CLOUD DATA

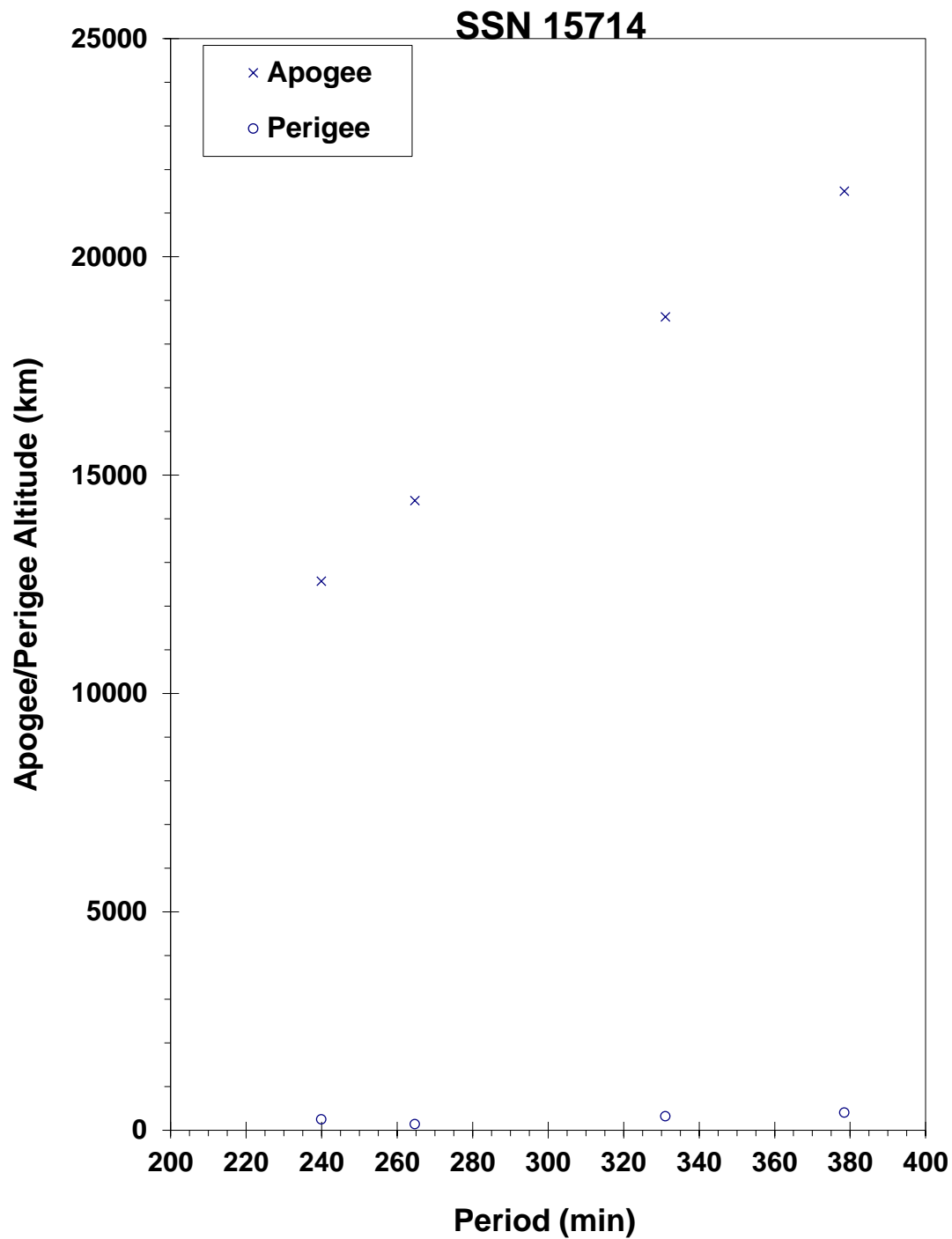
MAXIMUM ΔP : 91.18 min
MAXIMUM ΔI : .76 deg

COMMENTS

This is the 18th event of the Proton Block DM SOZ Ullage Motor class identified to date; it is the seventh associated with a GLONASS mission. This mission was conducted before the engineering defect was identified and passivation measures implemented. In this orbit, debris may be long-lived but hard to track. A total of 60 debris objects were detected.

REFERENCE DOCUMENT

"1998 Ends with Eighth Satellite Breakup", The Orbital Debris Quarterly News, NASA JSC, January 1999. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv4i1.pdf>.



Cosmos 1650-1652 ullage motor debris cloud of 4 fragments within 1 day of the event as reconstructed from the US SSN database.

COSMOS 1654

1985-039A

15734

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 23.53 May 1985
DRY MASS (KG): 5700
MAIN BODY: Cone-cylinder; 2.7 m diameter by 6.3 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

| | | | |
|-----------|-------------|-----------------|----------------|
| DATE: | 21 Jun 1985 | LOCATION: | 8N, 292E (asc) |
| TIME: | 1047 GMT | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | 200 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 85172.01363851 | MEAN ANOMALY: | 313.0734 |
| RIGHT ASCENSION: | 1.2391 | MEAN MOTION: | 16.11890623 |
| INCLINATION: | 64.8566 | MEAN MOTION DOT/2: | .00311214 |
| ECCENTRICITY: | .0086971 | MEAN MOTION DOT DOT/6: | .000034493 |
| ARG. OF PERIGEE: | 47.8764 | BSTAR: | .00015520 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 22.1 min*
MAXIMUM ΔI : 1.5 deg*

*Based on uncataloged debris data

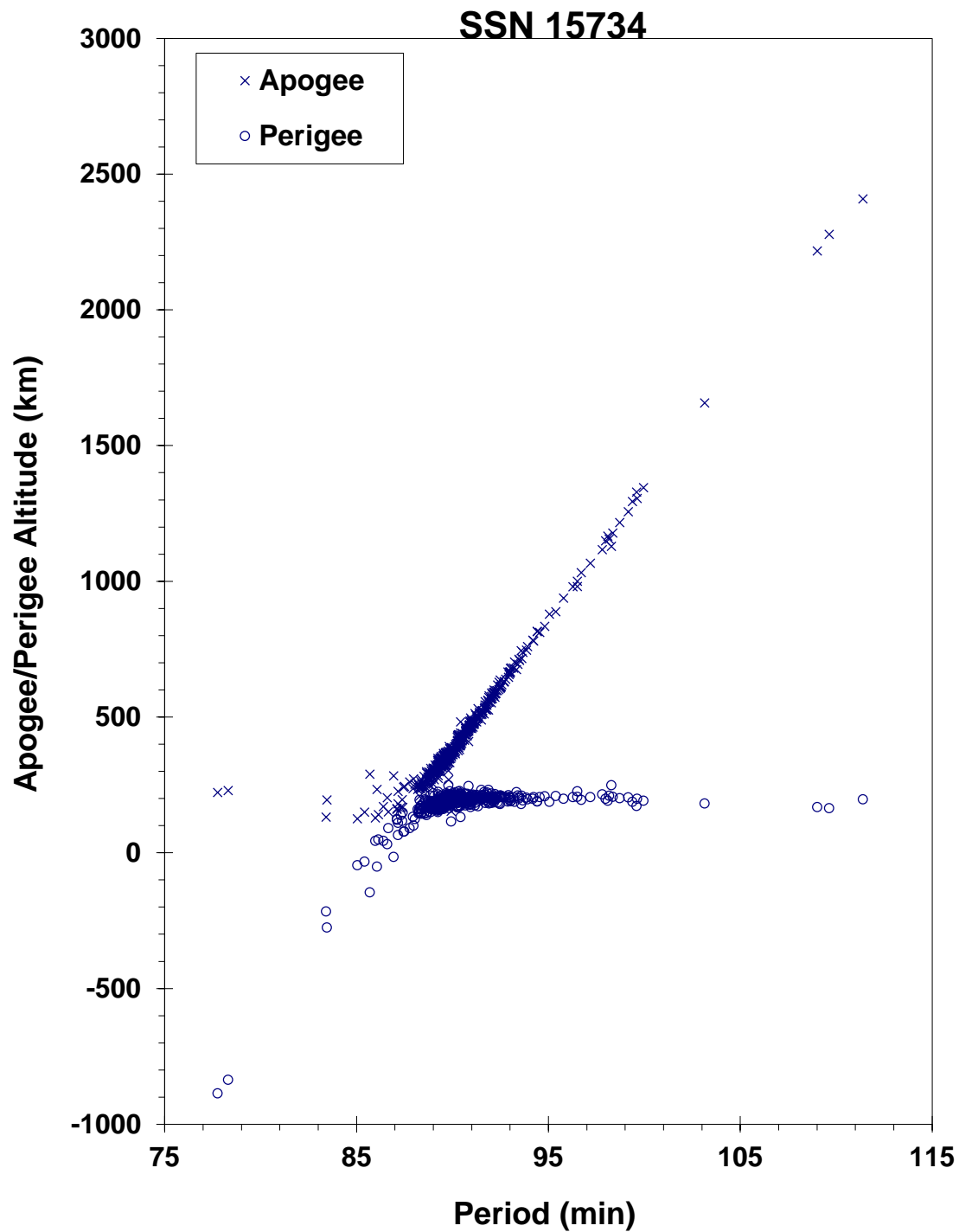
COMMENTS

Spacecraft was destroyed after a malfunction prevented controlled reentry and landing in the Soviet Union. Most debris reentered before being officially cataloged.

REFERENCE DOCUMENTS

Analysis of the Fragmentation of Kosmos 1654, G.T. DeVere, Technical Report CS86-BMDSC-0003, Teledyne Brown Engineering, Colorado Springs, October 1985.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1654 debris cloud remnant of 543 fragments seen 9 hours after the event
by the US SSN PARCS radar.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 30.62 May 1985
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

DATE: 5 Jan 1988 LOCATION: 66N, 151E (asc)
TIME: 0147 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 860 km

PRE-EVENT ELEMENTS

EPOCH: 88002.58690356 MEAN ANOMALY: 91.9605
RIGHT ASCENSION: 205.7335 MEAN MOTION: 14.17143400
INCLINATION: 66.5867 MEAN MOTION DOT/2: .00000144
ECCENTRICITY: .0034143 MEAN MOTION DOT DOT/6: .0
ARG. OF PERIGEE: 267.7562 BSTAR: .000088961

DEBRIS CLOUD DATA

MAXIMUM ΔP : 3.3 min
MAXIMUM ΔI : 0.0 deg

COMMENTS

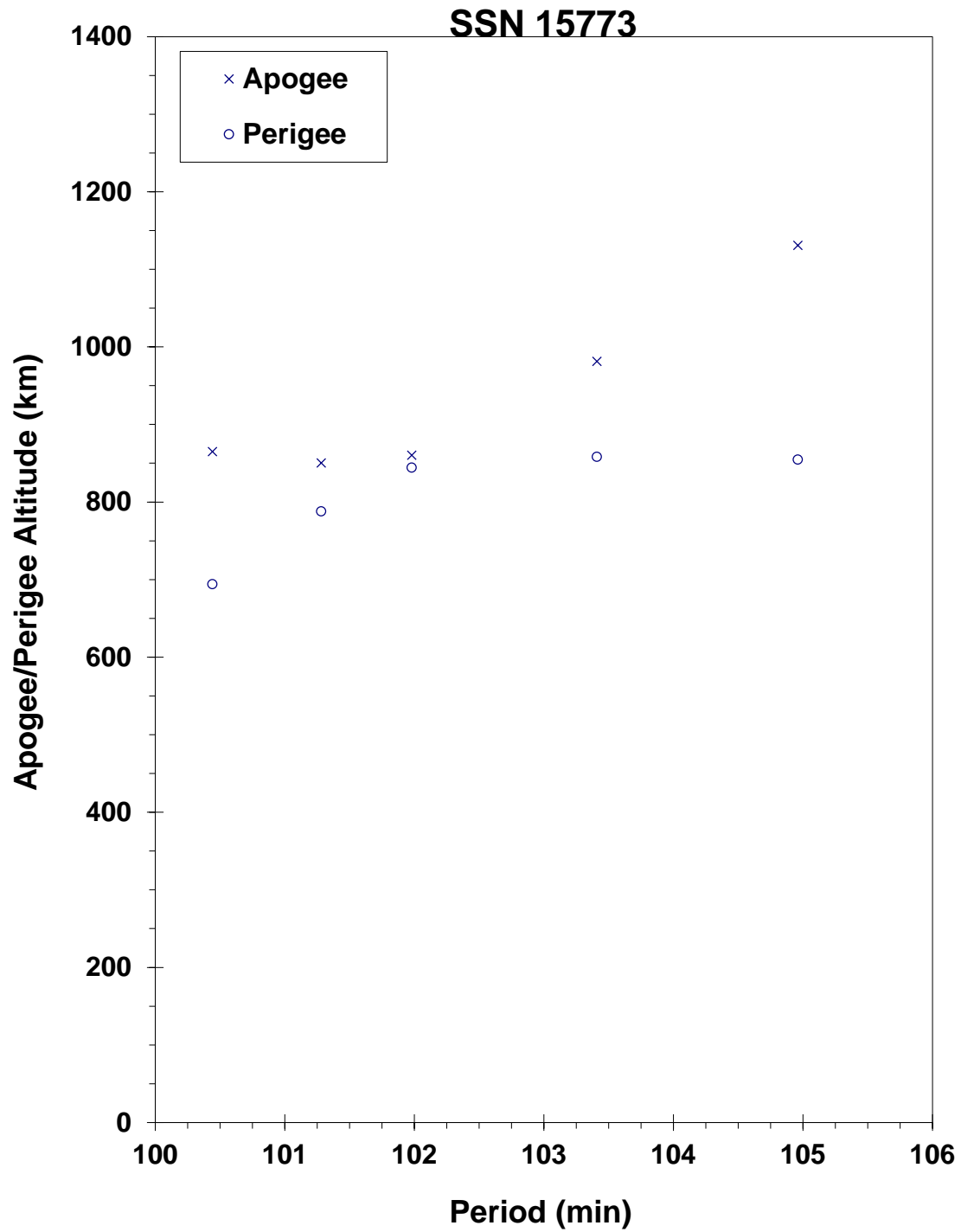
Parent satellite was one of two small engine units that are routinely released after the first burn of the Proton fourth stage. The nature of these objects was identified by Dr. Boris V. Cherniatiev, Deputy Constructor for the Energiya NPO, in October 1992. The cause of this fragmentation is assumed to be related to the residual hypergolic propellants on board and failure of the membrane separating the fuel and oxidizer. NAVSPASUR observed two additional, uncataloged fragments associated with this event. This was the second in a series of fragmentations of this object type.

REFERENCE DOCUMENTS

The Fragmentation of Proton Debris, D. J. Nauer, TBE Technical Report CS93-LKD-004, Teledyne Brown Engineering, Colorado Springs, 31 December 1992.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

"Identification and Resolution of an Orbital Debris Problem with the Proton Launch Vehicle", B. V. Cherniatiev et al, Proceedings of the First European Conference on Space Debris, April 1993.



Fragments from Cosmos 1656 debris as determined 2 weeks after the event. Elements from the US SSN database as published by NASA Goddard Space Flight Center.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 19.07 Sep 1985
DRY MASS (KG): 3000
MAIN BODY: Cylinder; 1.3 m diameter by 17 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge (?)

EVENT DATA

DATE: 18 Dec 1986 LOCATION: 22S, 292 E (asc)
TIME: 2017 GMT ASSESSED CAUSE: Unknown
ALTITUDE: 415 km

PRE-EVENT ELEMENTS

EPOCH: 86351.87879723 MEAN ANOMALY: 315.5258
RIGHT ASCENSION: 337.4852 MEAN MOTION: 15.45249396
INCLINATION: 65.0089 MEAN MOTION DOT/2: .00011076
ECCENTRICITY: .0068048 MEAN MOTION DOT DOT/6: .0
ARG. OF PERIGEE: 45.1423 BSTAR: .00021714

DEBRIS CLOUD DATA

MAXIMUM ΔP : 2.3 min*
MAXIMUM ΔI : 0.7 deg*

*Based on uncataloged debris data

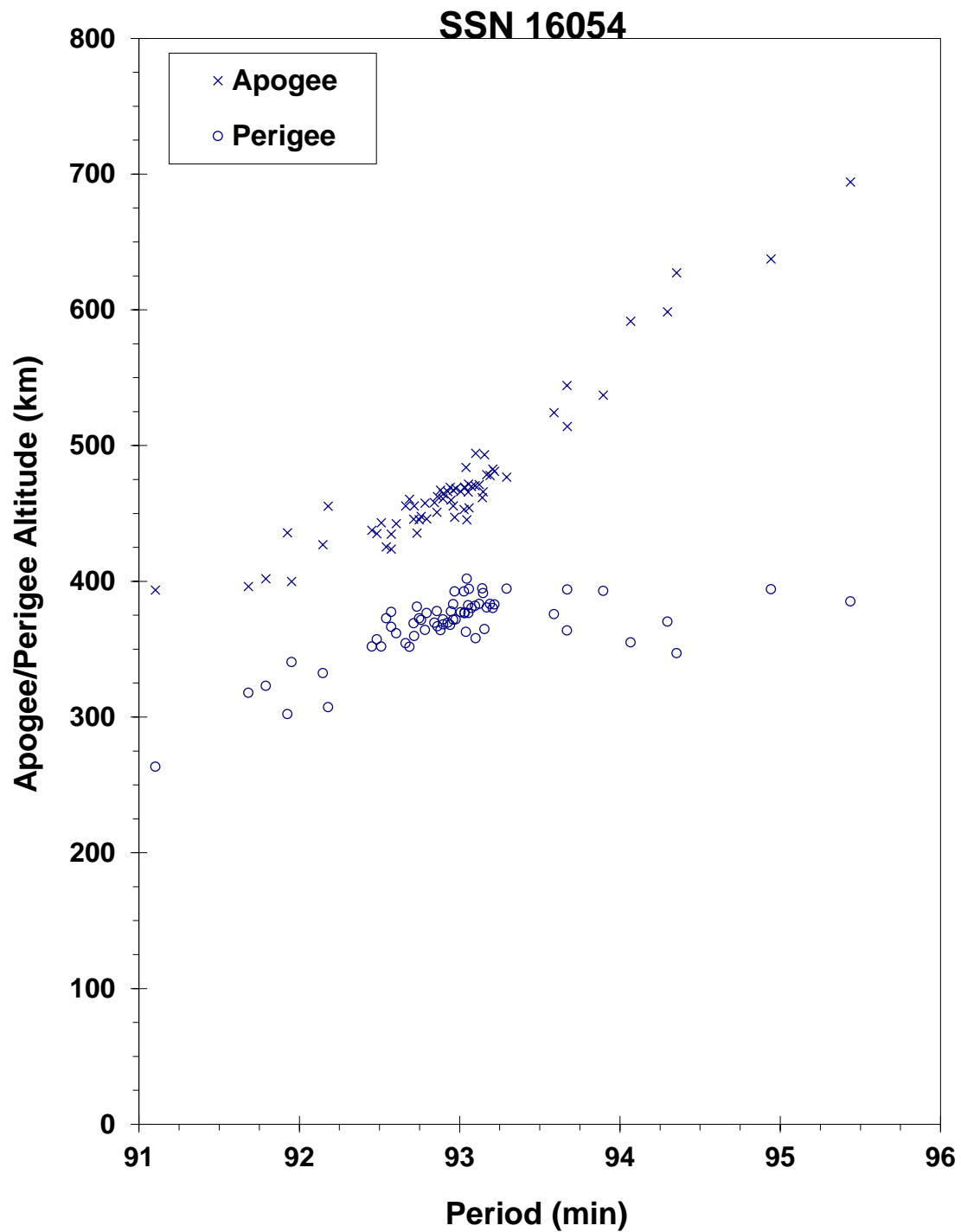
COMMENTS

Cosmos 1682 was the 14th spacecraft of the Cosmos 699-type to experience a fragmentation. Spacecraft had been in natural decay for 2 months prior to the event. Many debris reentered before being officially cataloged.

REFERENCE DOCUMENTS

"Artificial Satellite Break-Ups (Part 1): Soviet Ocean Surveillance Satellites", N. L. Johnson, Journal of the British Interplanetary Society, February 1983, p. 51-58.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1682 debris cloud remnant of 66 fragments about 1 week after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 9.90 Oct 1985
DRY MASS (KG): 220
MAIN BODY: Cylinder; 1.0 m diameter by 1.5 m length
MAJOR APPENDAGES: Gravity gradient boom
ATTITUDE CONTROL: Gravity gradient
ENERGY SOURCES: Battery

EVENT DATA

DATE: 22 Nov 1985 LOCATION: 31N, 326E (dsc)
TIME: 0840 GMT ASSESSED CAUSE: Battery
ALTITUDE: 1415 km

PRE-EVENT ELEMENTS

EPOCH: 85320.62059878 MEAN ANOMALY: 91.0897
RIGHT ASCENSION: 345.1807 MEAN MOTION: 12.62038878
INCLINATION: 82.6124 MEAN MOTION DOT/2: .00000022
ECCENTRICITY: .0002812 MEAN MOTION DOT DOT/6: .0
ARG. OF PERIGEE: 268.9870 BSTAR: .000099999

DEBRIS CLOUD DATA

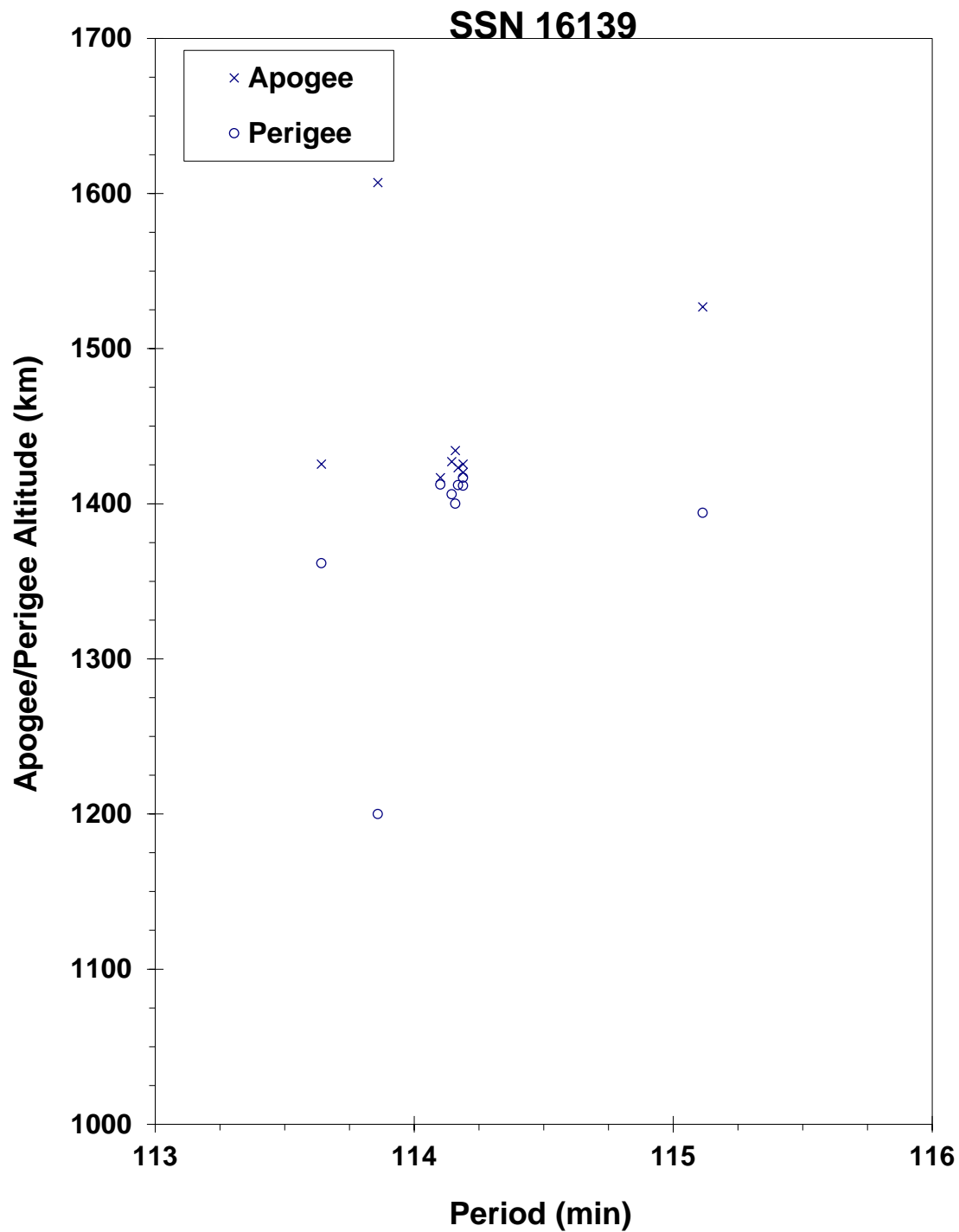
MAXIMUM ΔP : 1.0 min
MAXIMUM ΔI : 0.1 deg

COMMENTS

Cosmos 1691 was one of six independent payloads on this launch, which was only the second in this program. Cosmos 1691 was the last payload deployed and may be referred to as Cosmos 1695 in the former Soviet Union. One fragment was administratively decayed in February 1989. No other payloads in this program have fragmented. This event is assessed to be the second known NiH₂ battery failure as indicated by Dr. K. M. Suitshev during the early 1992 Space Debris Conference in Moscow. See also reference below.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1691 debris cloud of 9 fragments 2 days after the event as reconstructed from Naval Space Surveillance System database.

SATELLITE DATA

TYPE: Rocket Body
OWNER: CIS
LAUNCH DATE: 22.93 Nov 1985
DRY MASS (KG): 1360
MAIN BODY: Cone-cylinder; 2.1 m diameter by 3.3 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: Unknown

EVENT DATA

DATE: 4 May 2006 LOCATION: 67N, 17E (dsc)
TIME: 1604 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 635 km

PRE-EVENT ELEMENTS

EPOCH: 06123.63561455 MEAN ANOMALY: 329.9835
RIGHT ASCENSION: 319.0019 MEAN MOTION: 14.8137473
INCLINATION: 82.5005 MEAN MOTION DOT/2: 0.00000107
ECCENTRICITY: .0021969 MEAN MOTION DOT DOT/6: 0.0
ARG. OF PERIGEE: 30.2640 BSTAR: 0.000010657

DEBRIS CLOUD DATA

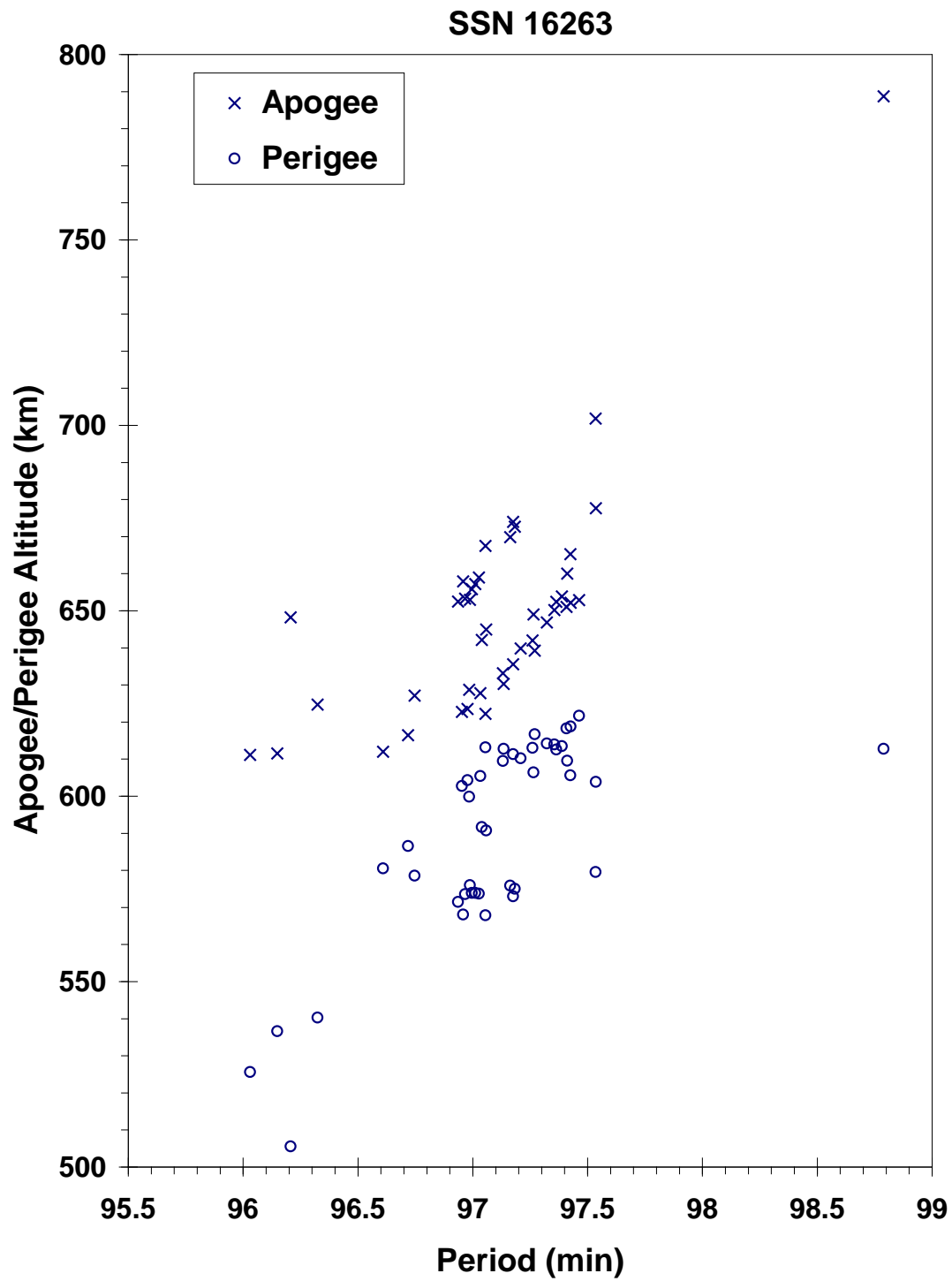
MAXIMUM ΔP : 1.6 min
MAXIMUM ΔI : 0.2 deg

COMMENTS

This is the 5th event of the Tsyklon third stage (SL-14) identified to date.

REFERENCE DOCUMENT

“First Satellite Breakups of 2006”, The Orbital Debris Quarterly News, NASA JSC, July 2006.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv10i3.pdf>.



Tsyklon third stage debris cloud of 44 fragments six days after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Zenit Second Stage
OWNER: CIS
LAUNCH DATE: 28.40 Dec 1985
DRY MASS (KG): 9000
MAIN BODY: Cylinder; 3.9 m diameter by 12 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 28 Dec 1985 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 85363.19328410 | MEAN ANOMALY: | 84.6199 |
| RIGHT ASCENSION: | 281.3886 | MEAN MOTION: | 14.77971051 |
| INCLINATION: | 71.0178 | MEAN MOTION DOT/2: | 0.00065991 |
| ECCENTRICITY: | 0.0306365 | MEAN MOTION DOT DOT/6: | 0 |
| ARG. OF PERIGEE: | 271.9949 | BSTAR: | 0.0041108 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

The Zenit second stage low thrust engine used to perform final orbit insertion exploded. Four pieces of debris cataloged with this mission are probably not associated with the breakup.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 24.91 Dec 1985
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

DATE: 29 Dec 1991 LOCATION: 25.3N, 331.9E
TIME: 0903 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 4730 km

PRE-EVENT ELEMENTS

EPOCH: 91333.40579226 MEAN ANOMALY: 46.8976
RIGHT ASCENSION: 48.0333 MEAN MOTION: 4.23089679
INCLINATION: 65.2547 MEAN MOTION DOT/2: .00000167
ECCENTRICITY: .5645362 MEAN MOTION DOT DOT/6: .0
ARG. OF PERIGEE: 245.7447 BSTAR: .0012603

DEBRIS CLOUD DATA

MAXIMUM ΔP : 5.7 min *
MAXIMUM ΔI : 0.8 deg *

* based upon uncataloged debris data

COMMENTS

Parent satellite was one of two small engine units that are routinely released after the first burn of the Proton fourth stage. The nature of these objects was identified by Dr. Boris V. Cherniatiev, Deputy Constructor for the Energiya NPO, in October 1992. The cause of this fragmentation is assumed to be related to the residual hypergolic propellants on board and failure of the membrane separating the fuel and oxidizer. There were 26 objects associated with this event on 30 December per a telecon with NAVSPASUR (Edna Jenkins). Only 2 analyst satellites were generated and insufficient data was available for a Gabbard diagram. This was the fourth in a series of fragmentations of this object type.

REFERENCE DOCUMENTS

The Fragmentation of Proton Debris, D. J. Nauer, TBE Technical Report CS93-LKD-004, Teledyne Brown Engineering, Colorado Springs, 31 December 1992.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

"Identification and Resolution of an Orbital Debris Problem with the Proton Launch Vehicle", B. V. Cherniatiev et al, Proceedings of the First European Conference on Space Debris, April 1993.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Ariane 1 Third Stage
OWNER: France
LAUNCH DATE: 22.07 Feb 1986
DRY MASS (KG): 1400
MAIN BODY: Cylinder; 2.6 m diameter by 10.3 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: On-board propellants, range safety package

EVENT DATA

| | | | |
|-----------|-------------|-----------------|---------------|
| DATE: | 13 Nov 1986 | LOCATION: | 7N, 42E (asc) |
| TIME: | 1940 GMT | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | 805 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 86305.08337689 | MEAN ANOMALY: | 300.1947 |
| RIGHT ASCENSION: | 18.0087 | MEAN MOTION: | 14.22163662 |
| INCLINATION: | 98.6973 | MEAN MOTION DOT/2: | .00000203 |
| ECCENTRICITY: | .0021203 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 60.1312 | BSTAR: | .000099999 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 6.2 min
MAXIMUM ΔI : 1.2 deg

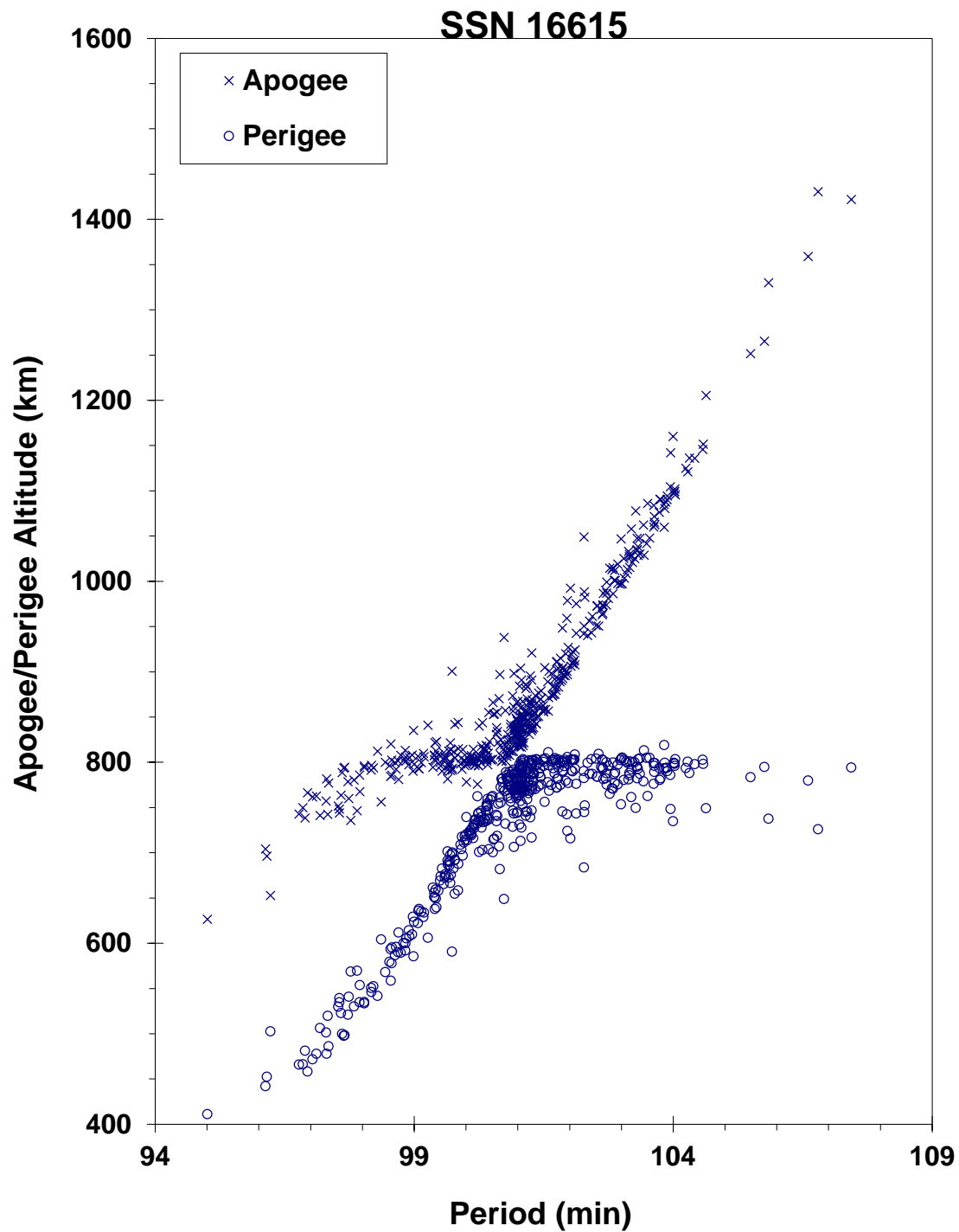
COMMENTS

Event occurred approximately 9 months after the rocket body had successfully deployed the SPOT 1 and Viking payloads. First use of Ariane launch vehicle for low Earth orbit. May be related to other Ariane fragmentations.

REFERENCE DOCUMENTS

A Preliminary Analysis of the Fragmentation of the Spot 1 Ariane Third Stage, N. L. Johnson, Technical Report CS87-LKD-003, Teledyne Brown Engineering, Colorado Springs, March 1987.

Orbital Debris from Upper Stage Breakup, J.P. Loftus, Jr., ed., Vol. 121, Progress in Astronautics and Aeronautics, AIAA, 1989.



Spot 1 R/B debris cloud of 463 fragments 3 months after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 4.21 Aug 1986
DRY MASS (KG): 3000
MAIN BODY: Cylinder; 1.3 m diameter by 17 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge (?)

EVENT DATA

DATE: 21 Sep 1987 LOCATION: 60S, 174E (dsc)
TIME: 1205 GMT ASSESSED CAUSE: Unknown
ALTITUDE: 320 km

PRE-EVENT ELEMENTS

EPOCH: 87263.81808697 MEAN ANOMALY: 70.4851
RIGHT ASCENSION: 122.5376 MEAN MOTION: 15.63167584
INCLINATION: 65.0147 MEAN MOTION DOT/2: .00078200
ECCENTRICITY: .0099296 MEAN MOTION DOT DOT/6: .0
ARG. OF PERIGEE: 288.4915 BSTAR: .00065556

DEBRIS CLOUD DATA

MAXIMUM ΔP : 1.9 min*
MAXIMUM ΔI : 0.0 deg*

*Based on uncataloged debris data

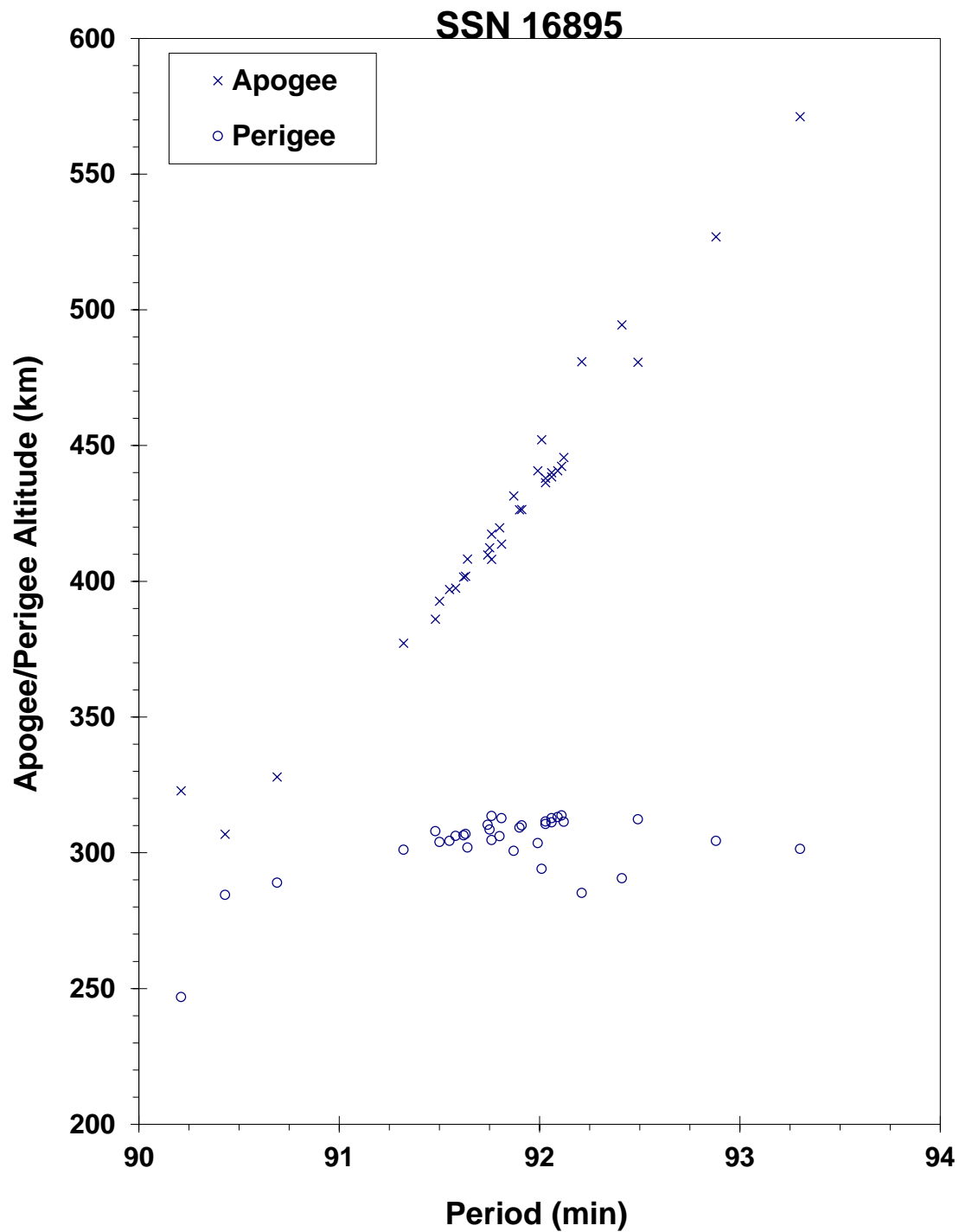
COMMENTS

Cosmos 1769 was the fifteenth spacecraft of the Cosmos 699-type to experience a fragmentation. Spacecraft was regularly maneuvered until 17 Sep 1987 when the vehicle began to decay naturally. Most debris reentered before being officially cataloged.

REFERENCE DOCUMENTS

"Artificial Satellite Break-Ups (Part 1): Soviet Ocean Surveillance Satellites", N. L. Johnson, Journal of the British Interplanetary Society, February 1983, pp. 51-58.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1769 debris cloud remnant of 34 fragments 3 days after the event as reconstructed from Naval Space Surveillance System database.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 5.63 Sep 1986
DRY MASS (KG): 930
MAIN BODY: Cylinder-cone; 1.2 m diameter by 4.6 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge (?)

EVENT DATA

| | | | |
|-----------|------------|-----------------|-----------------|
| DATE: | 5 Sep 1986 | LOCATION: | 15N, 166E (asc) |
| TIME: | 1752 GMT | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | 220 km | | |

POST-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 86250.63774662 | MEAN ANOMALY: | 335.3264 |
| RIGHT ASCENSION: | 28.1524 | MEAN MOTION: | 15.28976390 |
| INCLINATION: | 39.0665 | MEAN MOTION DOT/2: | .01159823 |
| ECCENTRICITY: | .0390567 | MEAN MOTION DOT DOT/6: | .0000050922 |
| ARG. OF PERIGEE: | 26.7075 | BSTAR: | .0028192 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 424.1 min*
MAXIMUM ΔI : 4.4 deg*

*Based on uncataloged debris data

COMMENTS

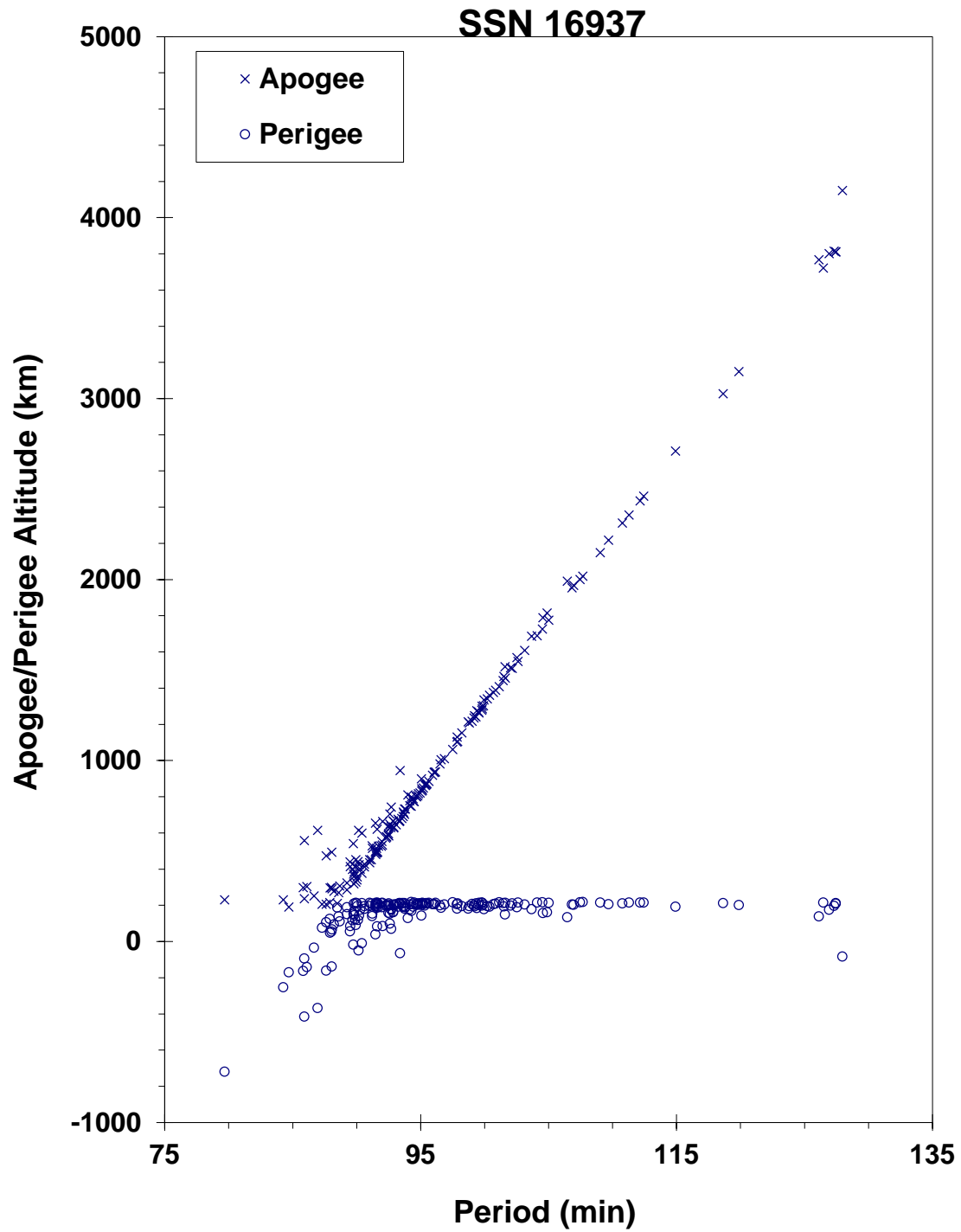
USA 19 deliberately collided with USA 19 R/B at high relative velocity. Both satellites were thrusting at the time of impact. Element set above is post-event and is best estimate of orbit at time of the event. Most debris reentered before being officially cataloged.

REFERENCE DOCUMENTS

The Collision of Satellites 16937 and 16938: A Preliminary Report, N. L. Johnson, Technical Report CS87-LKD-002, Teledyne Brown Engineering, Colorado Springs, 3 December 1986.

The Collision of Satellites 16937 and 16938: Debris Characterization, R. L. Kling, Technical Report CS87-LKD-005, Teledyne Brown Engineering, Colorado Springs, 15 May 1987.

Hazard Analysis of the Breakup of Satellites 16937 and 16938, Technical Report JSC 22471(U), NASA Lyndon B. Johnson Space Center, Houston, 27 February 1987.



USA 19 debris cloud remnant of 187 fragments 1 day after the event as seen by the US SSN radar FPS-85 at Eglin AFB, Florida.

SATELLITE DATA

TYPE: Delta Second Stage (3920) with auxiliary payload
OWNER: US
LAUNCH DATE: 5.63 Sep 1986
DRY MASS (KG): 1455
MAIN BODY: Cylinder-nozzle; 1.4 m diameter by 4.8 m length
MAJOR APPENDAGES: Mini-skirt; 2.4 m by 0.3 m
ATTITUDE CONTROL: Active
ENERGY SOURCES: On-board propellants

EVENT DATA

DATE: 5 Sep 1986 LOCATION: 15N, 166E (asc)
TIME: 1752 GMT ASSESSED CAUSE: Deliberate
ALTITUDE: 220 km

POST-EVENT ELEMENTS

EPOCH: 86249.96053354 MEAN ANOMALY: 307.9381
RIGHT ASCENSION: 10.4654 MEAN MOTION: 15.50608380
INCLINATION: 22.7830 MEAN MOTION DOT/2: .00138611
ECCENTRICITY: .0288474 MEAN MOTION DOT DOT/6: .0
ARG. OF PERIGEE: 54.7772 BSTAR: .00033298

DEBRIS CLOUD DATA

MAXIMUM ΔP : 53.6 min*
MAXIMUM ΔI : 2.5 deg*

*Based on uncataloged debris data

COMMENTS

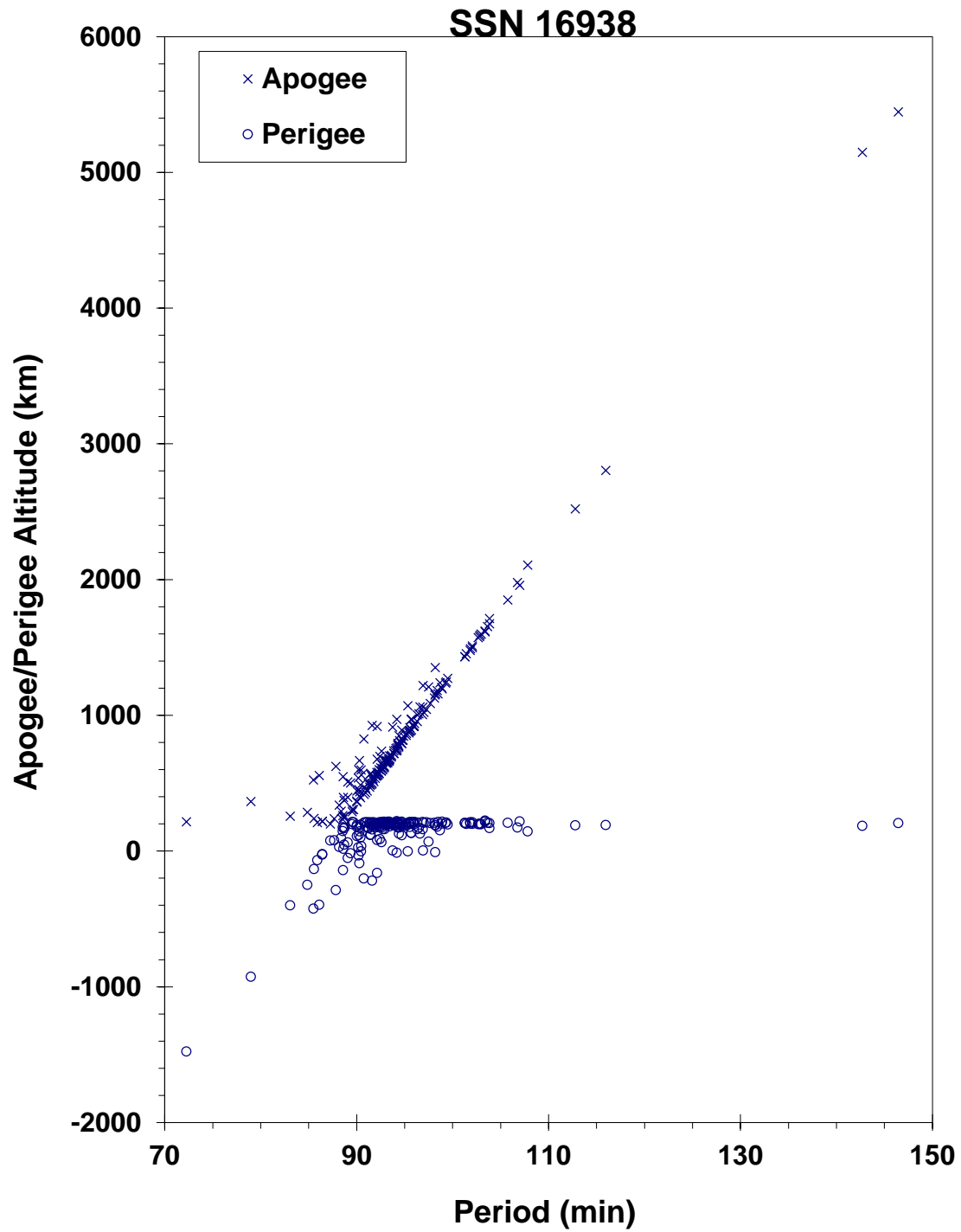
USA 19 R/B was deliberately struck by USA 19 at high relative velocity. Both satellites were thrusting at the time of impact. Element set above is post-event and is best estimate of orbit at time of the event. Most debris reentered before being officially cataloged.

REFERENCE DOCUMENTS

The Collision of Satellites 16937 and 16938: A Preliminary Report, N. L. Johnson, Technical Report CS87-LKD-002, Teledyne Brown Engineering, Colorado Springs, 3 December 1986.

The Collision of Satellites 16937 and 16938: Debris Characterization, R. L. Kling, Technical Report CS87-LKD-005, Teledyne Brown Engineering, Colorado Springs, 15 May 1987.

Hazard Analysis of the Breakup of Satellites 16937 and 16938, Technical Report JSC 22471(U), NASA Lyndon B. Johnson Space Center, Houston, 27 February 1987.



USA 19 R/B debris cloud of 190 fragments 1 day after the event as seen
by the US SSN FPS-85 radar at Eglin AFB, Florida.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 15.47 Jan 1987
DRY MASS (KG): 6300
MAIN BODY: Sphere-cylinder; 2.4 m diameter by 6.5 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

DATE: 29 Jan 1987 LOCATION: 73N, 122E (asc)
TIME: 0555 GMT ASSESSED CAUSE: Deliberate
ALTITUDE: 390 km

PRE-EVENT ELEMENTS

EPOCH: 87028.91020168 MEAN ANOMALY: 178.1696
RIGHT ASCENSION: 256.7724 MEAN MOTION: 15.60427146
INCLINATION: 72.8163 MEAN MOTION DOT/2: .00008569
ECCENTRICITY: .0043147 MEAN MOTION DOT DOT/6: .0
ARG. OF PERIGEE: 182.0100 BSTAR: .000099999

DEBRIS CLOUD DATA

MAXIMUM ΔP : 9.1 min*
MAXIMUM ΔI : 0.1 deg*

*Based on PARCS observations

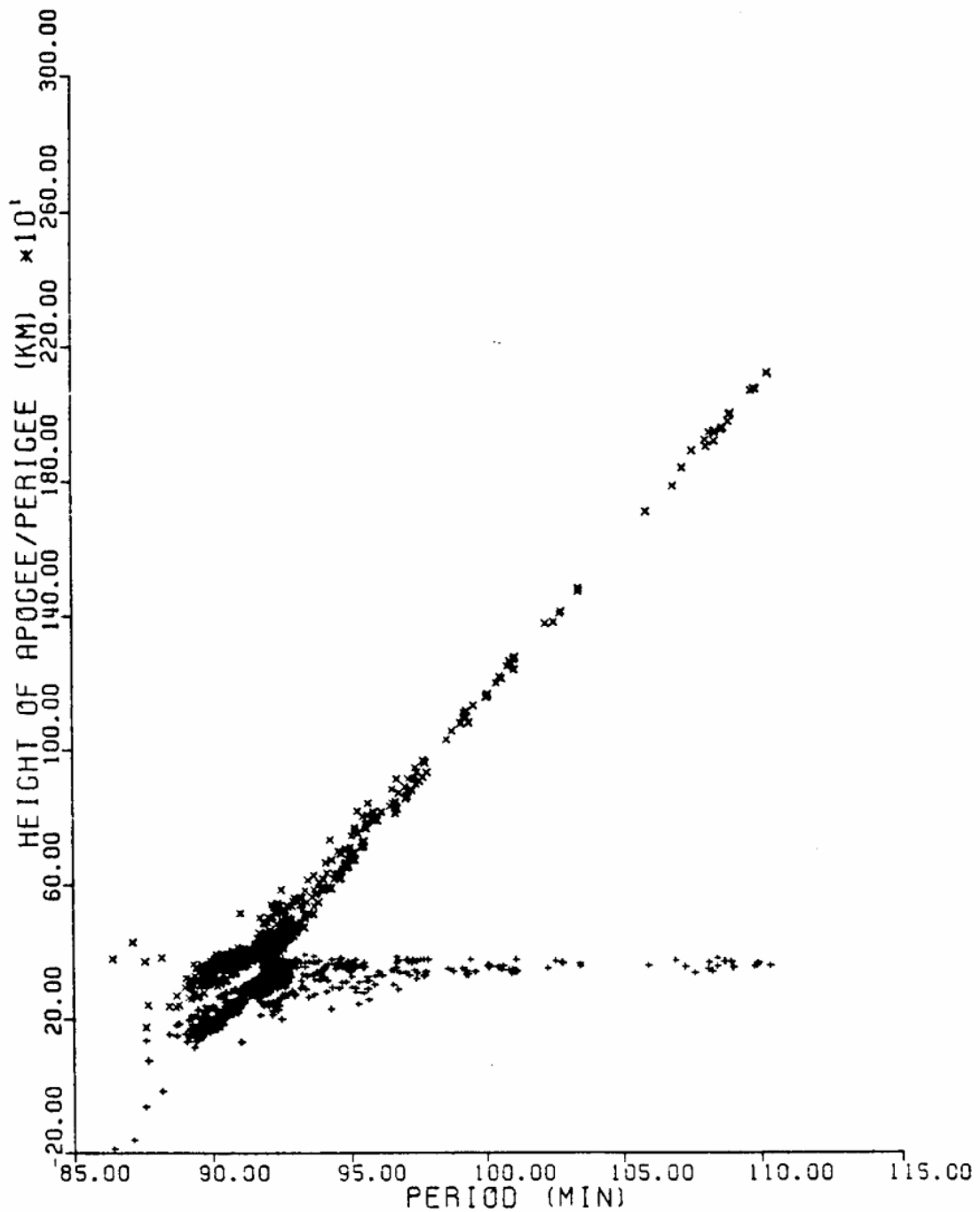
COMMENTS

Spacecraft apparently destroyed after a malfunction prevented controlled reentry and landing in the Soviet Union. A total of 846 separate fragments were observed during one pass over a U.S. Space Surveillance Network radar (PARCS) 2 days after the event.

REFERENCE DOCUMENTS

The Fragmentation of Kosmos 1813, R. L. Kling and J. S. Dowdy, Technical Report CS87-LKD-004, Teledyne Brown Engineering, Colorado Springs, 8 May 1987.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1813 debris cloud as reconstructed from PARCS radar observations taken about 10 hours after the breakup. A total of 846 fragments were identified with Cosmos 1813. This diagram is taken from the cited reference document.

SATELLITE DATA

| | |
|-------------------|--|
| TYPE: | Payload |
| OWNER: | CIS |
| LAUNCH DATE: | 20.20 Feb 1987 |
| DRY MASS (KG): | 1500 |
| MAIN BODY: | Cylinder; 2.4 m diameter by 4 m length |
| MAJOR APPENDAGES: | Gravity-gradient boom; 10 small solar panels |
| ATTITUDE CONTROL: | Gravity gradient |
| ENERGY SOURCES: | Battery |

EVENT DATA

| | | | |
|-----------|-------------|-----------------|----------------|
| DATE: | 17 Dec 1987 | LOCATION: | 15S, 18E (dsc) |
| TIME: | 1739 GMT | ASSESSED CAUSE: | Battery |
| ALTITUDE: | 1485 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 87351.61079422 | MEAN ANOMALY: | 147.6712 |
| RIGHT ASCENSION: | 184.5746 | MEAN MOTION: | 12.40947361 |
| INCLINATION: | 73.6064 | MEAN MOTION DOT/2: | .0 |
| ECCENTRICITY: | .0028819 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 212.2988 | BSTAR: | .0 |

DEBRIS CLOUD DATA

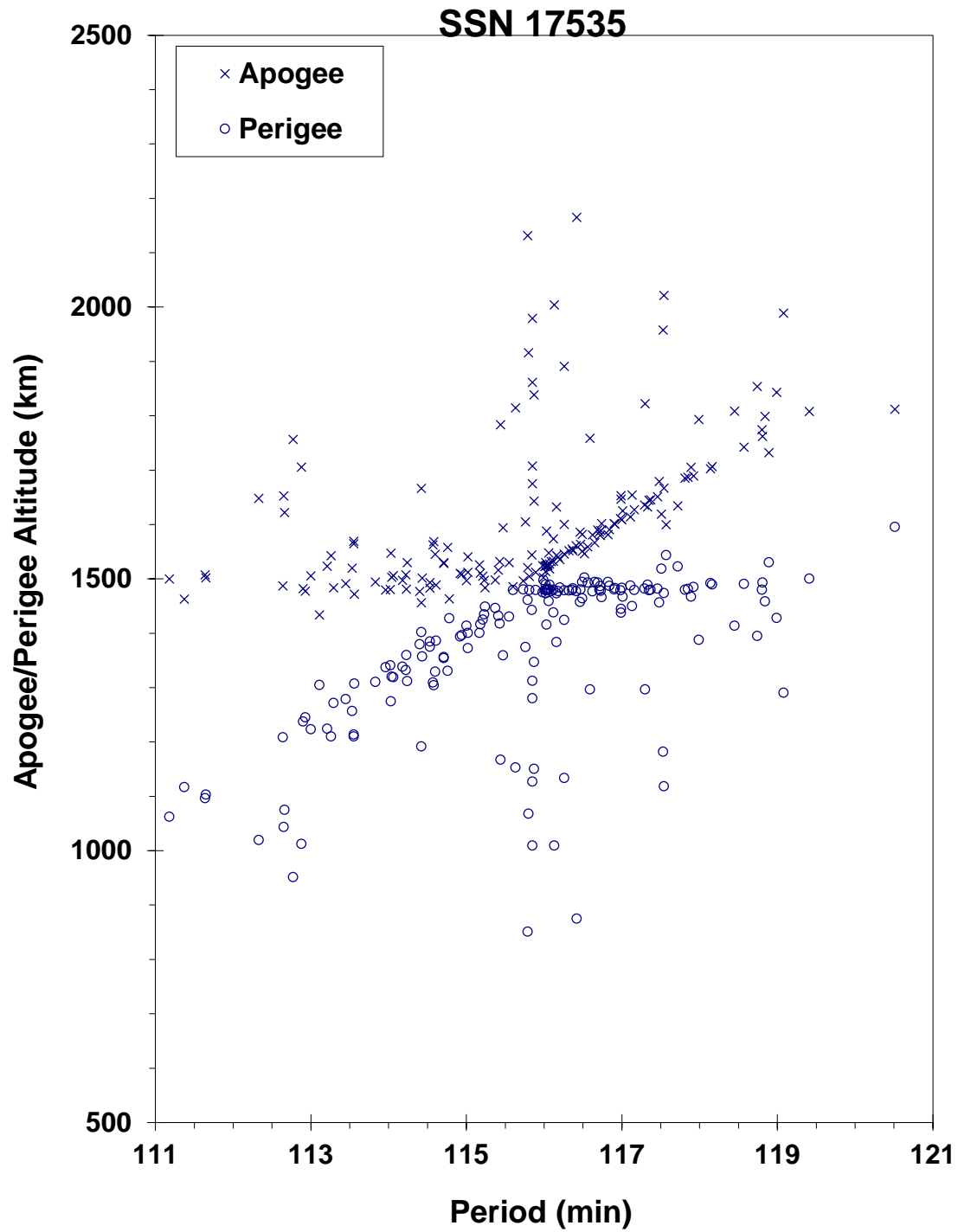
| | |
|----------------------|---------|
| MAXIMUM ΔP : | 4.9 min |
| MAXIMUM ΔI : | 1.4 deg |

COMMENTS

Cosmos 1823 has been acknowledged by the Soviet Union as a geodetic spacecraft, the eighth in a series that debuted in 1981. The spacecraft is known to have been operating 3 months before the event. USSR acknowledged mission termination as of 19 December 1987. Unusually strong radial velocity components are evident in cloud analyses over a period of many months. This event has been confirmed to be the third known failure of the NiH₂ battery as reported by Dr. K. M. Suitashev at the February, 1992 Space Debris Conference held in Moscow.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1823 debris cloud of 165 fragments 2 weeks after the event as reconstructed from Naval Space Surveillance System database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 9.67 Jul 1987
DRY MASS (KG): 5700
MAIN BODY: Cone-cylinder; 2.7 m diameter by 6.3 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

DATE: 26 Jul 1987
TIME: 1539 GMT
ALTITUDE: 245 km
LOCATION: 57S, 239E (asc)
ASSESSED CAUSE: Deliberate

PRE-EVENT ELEMENTS

EPOCH: 87207.60199851
RIGHT ASCENSION: 98.7735
INCLINATION: 67.1494
ECCENTRICITY: .0073576
ARG. OF PERIGEE: 61.7654
MEAN ANOMALY: 300.9577
MEAN MOTION: 16.25421506
MEAN MOTION DOT/2: .01099941
MEAN MOTION DOT DOT/6: .000028662
BSTAR: .00016423

DEBRIS CLOUD DATA

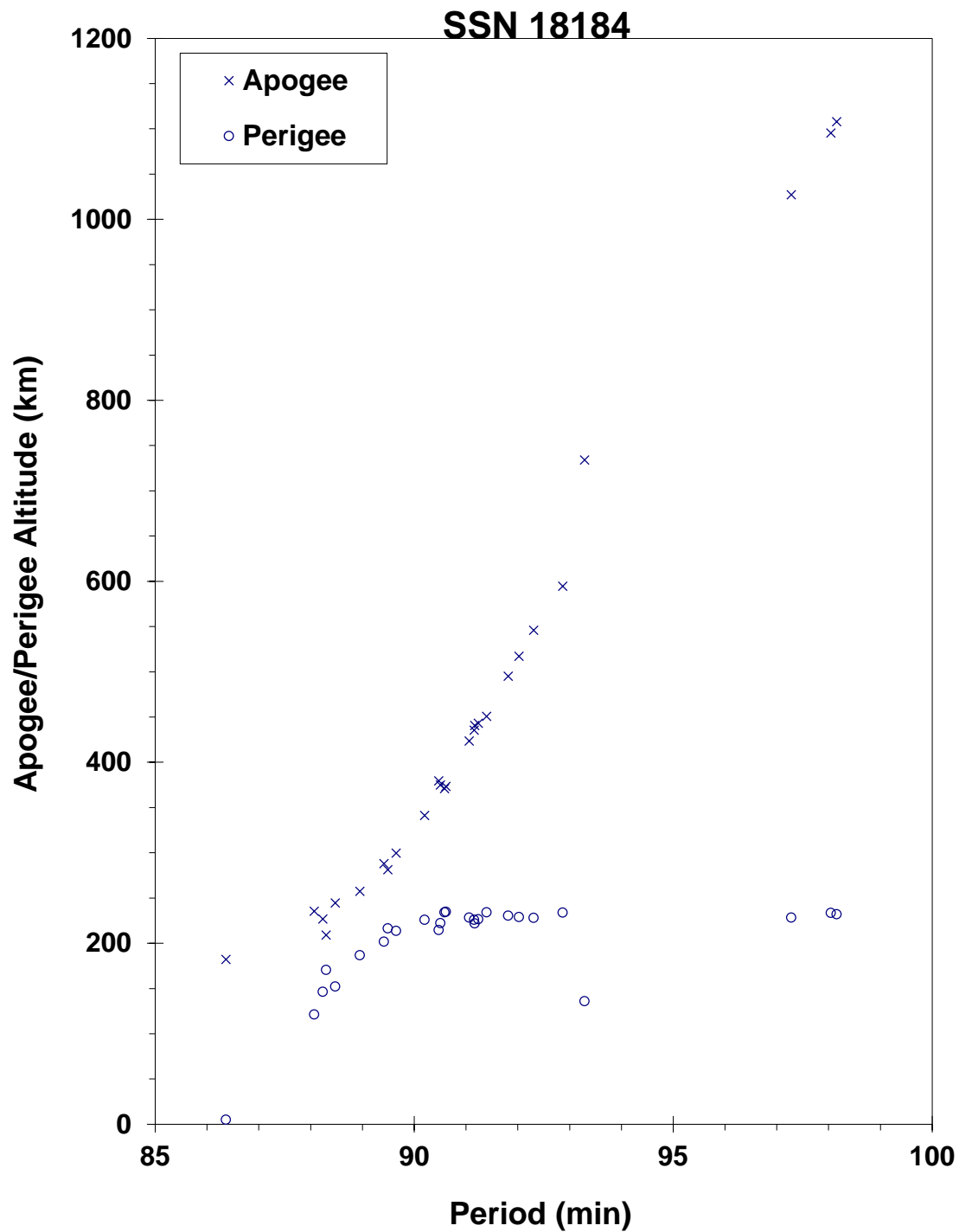
MAXIMUM ΔP : 17.3 min
MAXIMUM ΔI : 0.5 deg

COMMENTS

Spacecraft was destroyed after a malfunction prevented controlled reentry and landing in the Soviet Union. Hundreds of fragments were detected but most reentered before being officially cataloged.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1866 debris cloud of 27 fragments one to 2 days after the event as reconstructed from the US SSN database. Two fragments with orbital periods greater than 103 minutes were cataloged in mid-August 1987.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 16.18 Jul 1987
DRY MASS (KG): 1900
MAIN BODY: Cylinder; 0.8-1.4 m diameter by 3 m length
MAJOR APPENDAGES: Solar arrays, radar and other payload systems
ATTITUDE CONTROL: Gravity gradient
ENERGY SOURCES: Battery, pressurized vessels

EVENT DATA

| | | | |
|-----------|----------------|-----------------|---------|
| DATE: | 27 Nov 1997 | LOCATION: | Unknown |
| TIME: | 0006-0040 GMT? | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | ~630 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 97329.88487815 | MEAN ANOMALY: | 245.1014 |
| RIGHT ASCENSION: | 97.7878 | MEAN MOTION: | 14.83337853 |
| INCLINATION: | 82.5131 | MEAN MOTION DOT/2: | 0.00000439 |
| ECCENTRICITY: | 0.0021357 | MEAN MOTION DOT DOT/6: | 0 |
| ARG. OF PERIGEE: | 115.2417 | BSTAR: | 0.000050420 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

Cosmos 1869 suffered a failure of its radar antenna to deploy immediately after launch. The spacecraft carried other optical sensors, but the vehicle appears to have become non-operational by 1988. At least 20 debris were detected. Virtually all the debris associated with the breakup event exhibited very large area-to-mass ratios, resulting in exceptionally rapid orbital decay.

REFERENCE DOCUMENT

“Recent Satellite Fragmentation Investigations”, N. Johnson, The Orbital Debris Quarterly News, NASA JSC, January 1998, p. 3. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv3i1.pdf>.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Tsyklon Third Stage
OWNER: CIS
LAUNCH DATE: 18.10 Aug 1987
DRY MASS (KG): 1360
MAIN BODY: Cylinder; 2.1 m diameter by 2.4 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: Unknown

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------------------|
| DATE: | 15 Feb 1998 | LOCATION: | 67.8 N, 125.6 E (asc.) |
| TIME: | 2224 GMT | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | 945 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|--------------|
| EPOCH: | 98044.02783074 | MEAN ANOMALY: | 25.0628 |
| RIGHT ASCENSION: | 230.9724 | MEAN MOTION: | 13.84031596 |
| INCLINATION: | 82.5526 | MEAN MOTION DOT/2: | 0.00000025 |
| ECCENTRICITY: | 0.0011144 | MEAN MOTION DOT DOT/6: | 0 |
| ARG. OF PERIGEE: | 334.9992 | BSTAR: | 0.0000096468 |

DEBRIS CLOUD DATA

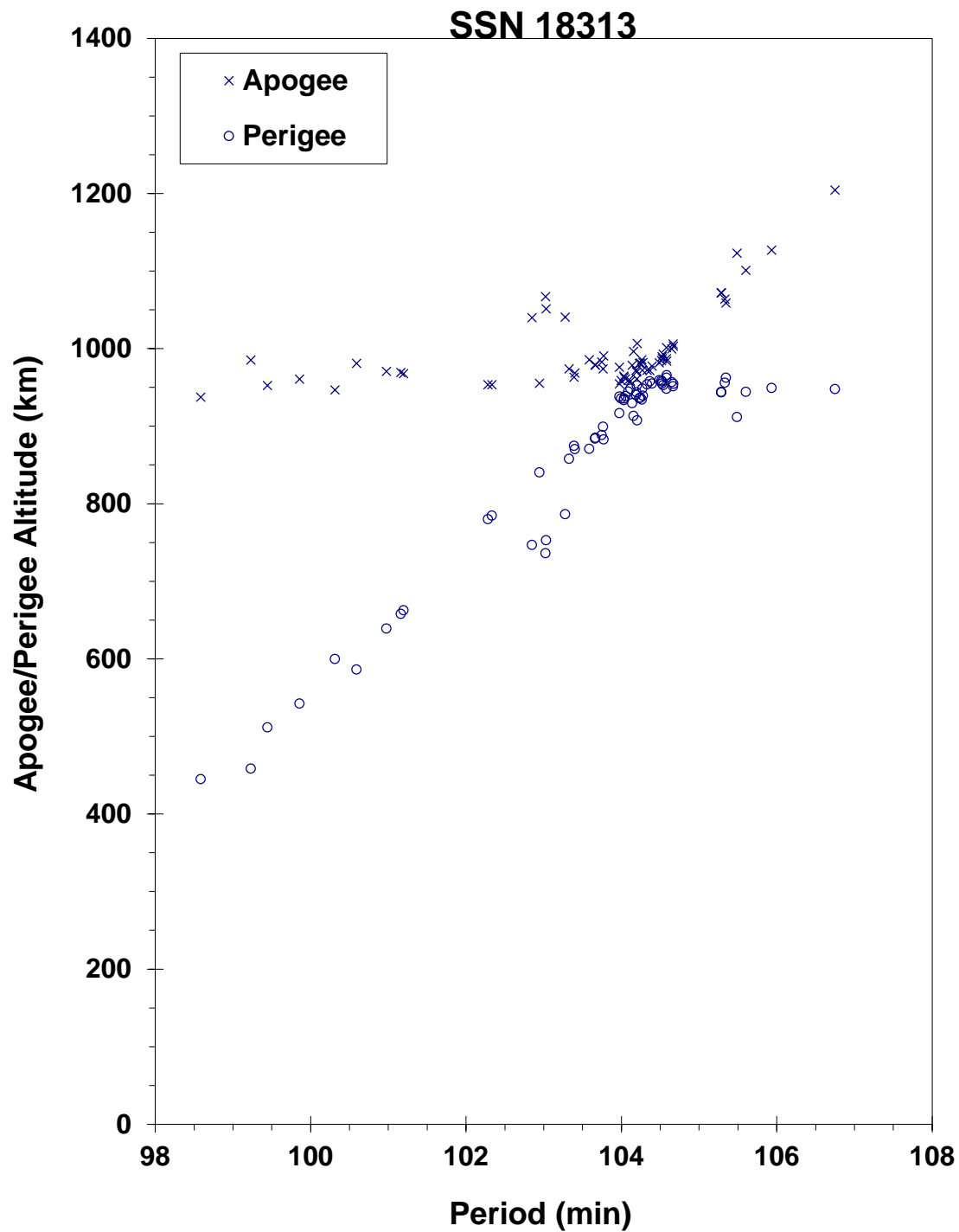
MAXIMUM ΔP : 8.2 min
MAXIMUM ΔI : 0.6 deg

COMMENTS

This is the second time a Ukrainian Tsyklon third stage has experienced a significant breakup. The previous incident in 1988 involved the Cosmos 1045 rocket body at a higher altitude. In both cases, the vehicle was approximately 10 years old. The debris from the current breakup were ejected with a wide range of velocities, from about 15 m/s to more than 250 m/s. Some debris were thrown to altitudes below 500 km, and some exhibited high area-to-mass ratios. Naval Space Command ran COMBO to determine if a tracked object was in vicinity of Meteor 2-16 R/B at the time of the event, and the results were negative.

REFERENCE DOCUMENT

“Three Upper Stage Breakups in One Week Top February Debris Activity”, The Orbital Debris Quarterly News, NASA JSC, April 1998, p. 1. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv3i2.pdf>.



**Meteor 2-16 R/B debris cloud of 67 fragments 1 week after the event
as reconstructed from the US SSN database.**

SATELLITE DATA

TYPE: Ariane 3 Third Stage
OWNER: France
LAUNCH DATE: 16.03 Sep 1987
DRY MASS (KG): 1200
MAIN BODY: Cylinder; 2.6 m diameter by 9.9 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|----------------|-----------------|------------|
| DATE: | 16-19 Sep 1987 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

POST-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 87264.18031994 | MEAN ANOMALY: | 170.9704 |
| RIGHT ASCENSION: | 176.7680 | MEAN MOTION: | 2.22860839 |
| INCLINATION: | 6.8720 | MEAN MOTION DOT/2: | .00014489 |
| ECCENTRICITY: | .7324768 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 182.0665 | BSTAR: | .0038829 |

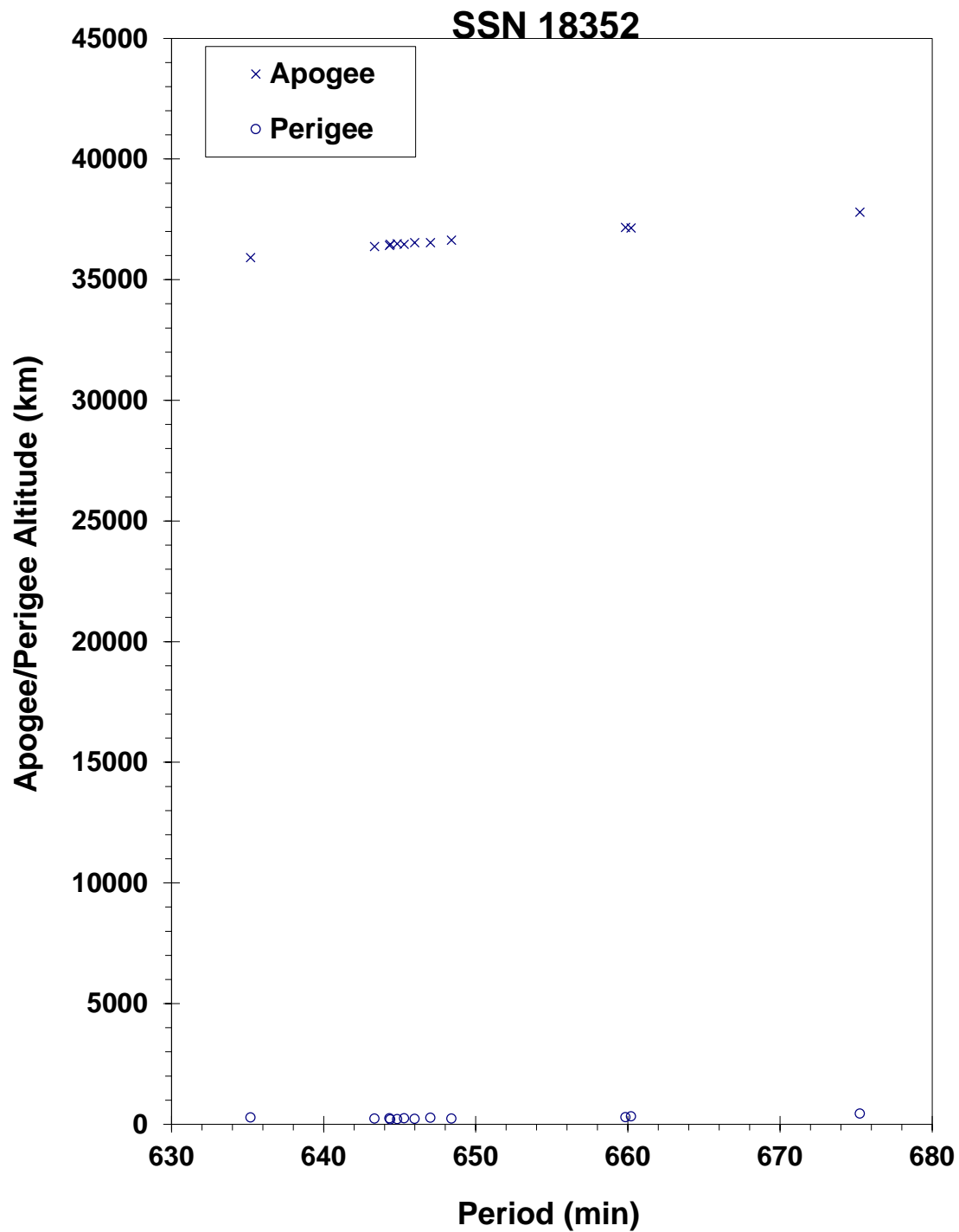
DEBRIS CLOUD DATA

MAXIMUM ΔP : 29.1 min*
MAXIMUM ΔI : 0.9 deg*

*Based on uncataloged debris data

COMMENTS

Above elements are initial published values for the rocket body but are after the event.



AUSSAT K3/ECS 4 R/B debris cloud of 12 fragments about 4 days after launch as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 16.12 Sep 1987
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|--------------------|-----------------|------------|
| DATE: | ~ 01 December 1996 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 96335.26612005 | MEAN ANOMALY: | 175.6198 |
| RIGHT ASCENSION: | 300.4954 | MEAN MOTION: | 4.24439384 |
| INCLINATION: | 64.9068 | MEAN MOTION DOT/2: | 0.00015773 |
| ECCENTRICITY: | 0.5826382 | MEAN MOTION DOT DOT/6: | 0 |
| ARG. OF PERIGEE: | 181.3565 | BSTAR: | 0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 234.1 min
MAXIMUM ΔI : 2.6 deg

COMMENTS

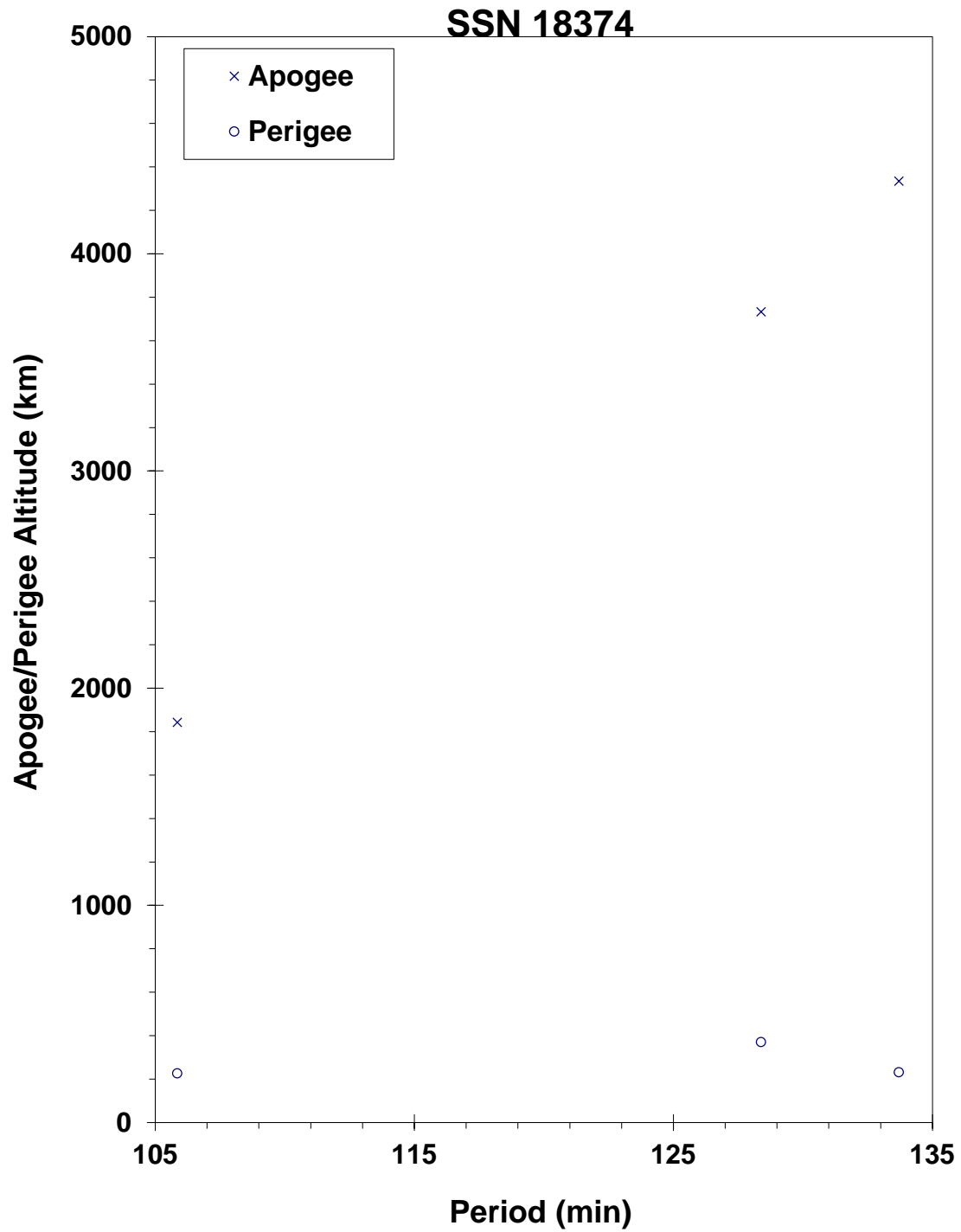
This is the 14th event of this class identified to date.

REFERENCE DOCUMENTS

"Identification and Resolution of an Orbital Debris Problem with the Proton Launch Vehicle", Cherniatiev, Chernyavskiy, Johnson, and McKnight, First European Conference on Space Debris, 5-7 April 1993.

"The Fragmentation of Proton Debris", Nauer, Teledyne Brown Engineering Technical Report CS93LKD-004, 31 December 1992.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1883-1885 auxiliary motor debris cloud of 3 fragments 11 to 14 months after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 16.12 Sep 1987
DRY MASS (KG): ~55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1.0 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 23 Apr 2003 | LOCATION: | Unknown |
| TIME: | ~1800Z | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 03113.46108488 | MEAN ANOMALY: | 332.8061 |
| RIGHT ASCENSION: | 156.9474 | MEAN MOTION: | 4.27871903 |
| INCLINATION: | 65.2438 | MEAN MOTION DOT/2: | .00000068 |
| ECCENTRICITY: | .5548829 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 85.3049 | BSTAR: | .00025672 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 26.0 min*
MAXIMUM ΔI : 1.19 deg*

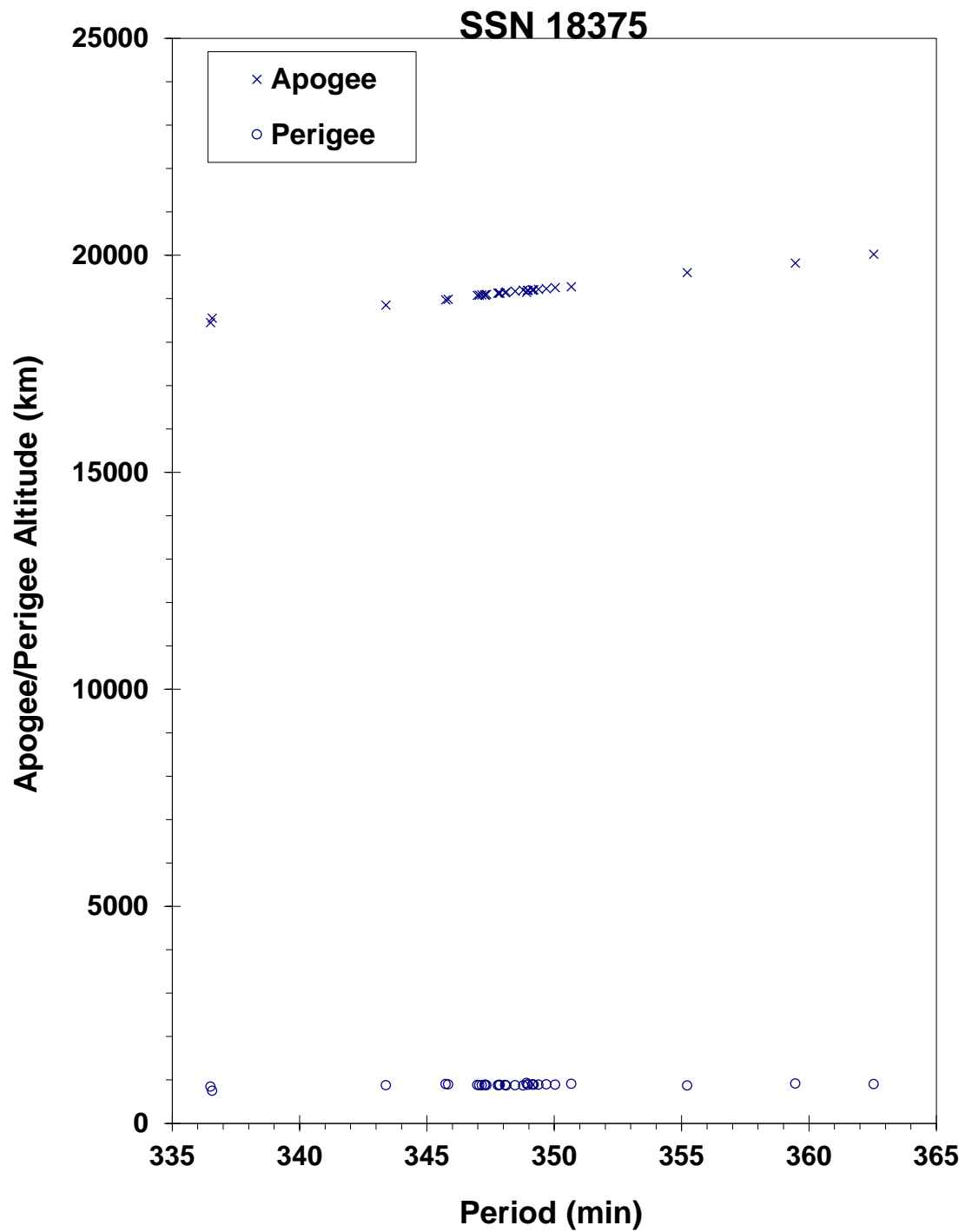
* Based on uncataloged debris data

COMMENTS

This event marks the 27th known breakup of a Proton Blok DM SOZ ullage motor since 1984. This ullage motor was launched before implementation of breakup preventive measures. 31 debris objects were cataloged from this breakup.

REFERENCE DOCUMENT

“Satellite Fragmentations in 2003”, The Orbital Debris Quarterly News, NASA JSC, January 2004. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv8i1.pdf>.



Cosmos 1883-85 auxiliary motor debris cloud of 31 fragments 2 days after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 26.48 Dec 1987
DRY MASS (KG): 6300
MAIN BODY: Sphere-Cylinder; 2.4 m diameter by 6.5 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

| | | | |
|-----------|-------------|-----------------|-----------------|
| DATE: | 31 Jan 1988 | LOCATION: | 11S, 138E (dsc) |
| TIME: | 1109 GMT | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | 250 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 88030.87152193 | MEAN ANOMALY: | 208.0352 |
| RIGHT ASCENSION: | 254.6565 | MEAN MOTION: | 16.07089398 |
| INCLINATION: | 82.5872 | MEAN MOTION DOT/2: | .00174892 |
| ECCENTRICITY: | .0015551 | MEAN MOTION DOT DOT/6: | .000012805 |
| ARG. OF PERIGEE: | 152.1926 | BSTAR: | .00022253 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 5.7 min*
MAXIMUM ΔI : 1.7 deg*

*Based on cataloged and uncataloged debris data

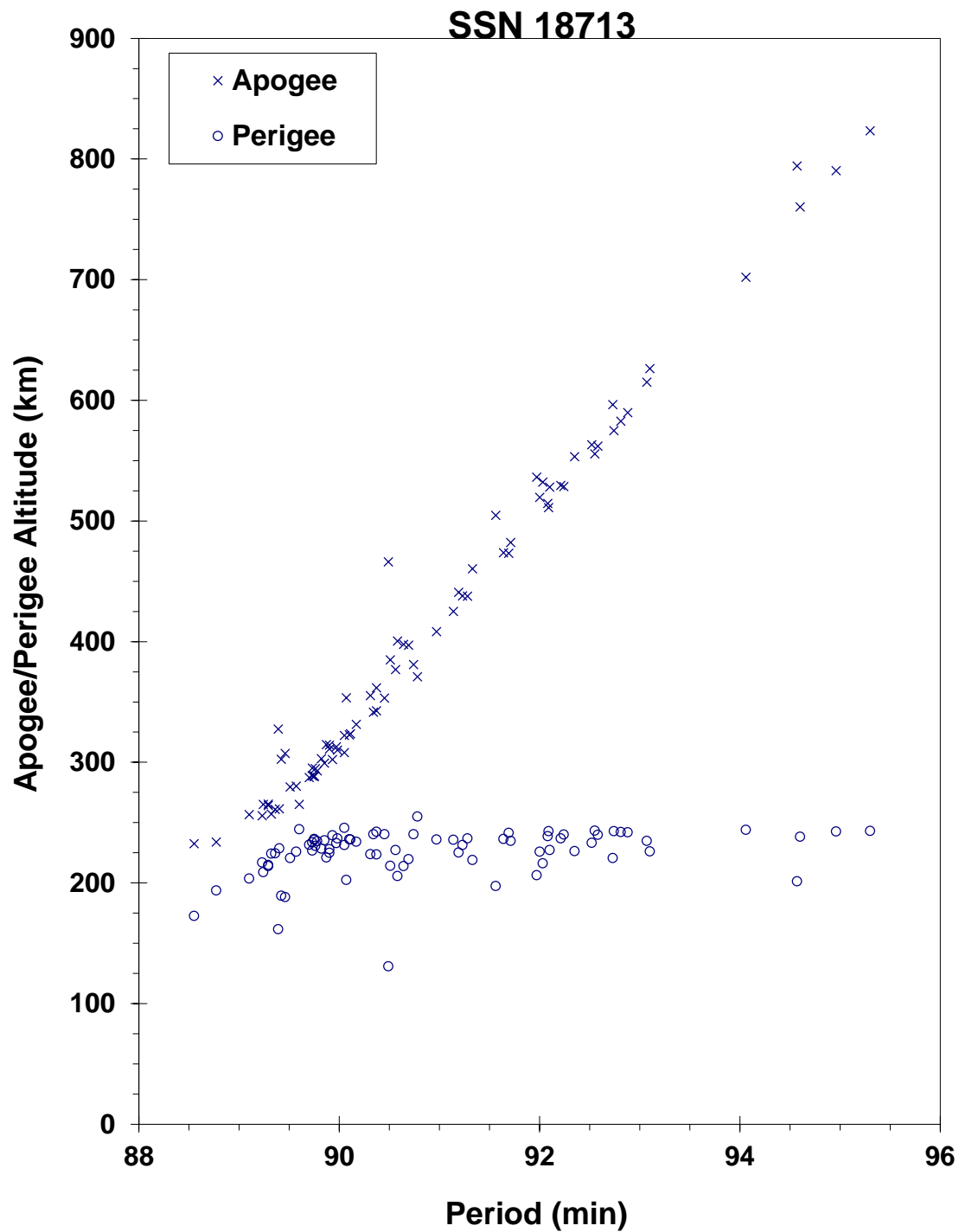
COMMENTS

Spacecraft destroyed after a malfunction prevented controlled reentry and landing in the Soviet Union. Elements for 83 objects remaining in orbit about 10 days after the event were developed. Other debris reentered before being officially cataloged.

REFERENCE DOCUMENTS

The Soviet Year in Space 1988, N. L. Johnson, Teledyne Brown Engineering, 1989, p. 27.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1906 debris cloud remnant of 83 objects 10 days after the event as reconstructed from Naval Space Surveillance System database.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 27.48 Dec 1987
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1.0 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 22 May 1997 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 97141.34020043 | MEAN ANOMALY: | 1.8603 |
| RIGHT ASCENSION: | 253.0389 | MEAN MOTION: | 3.58845480 |
| INCLINATION: | 46.6273 | MEAN MOTION DOT/2: | -0.00000117 |
| ECCENTRICITY: | 0.6287941 | MEAN MOTION DOT DOT/6: | 0 |
| ARG. OF PERIGEE: | 349.7051 | BSTAR: | 0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

This is the 15th event of this class identified to date. At least 72 debris were detected.

REFERENCE DOCUMENTS

"Three Satellite Breakups During May-June," The Orbital Debris Quarterly News, NASA JSC, July 1997, p. 2. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv2i3.pdf>.

"Identification and Resolution of an Orbital Debris Problem with Proton Launch Vehicle", Cherniatiev, Chernyavskiy, Johnson, and McKnight, First European Conference on Space Debris, 5-7 April 1993.

"The Fragmentation of Proton Debris", Nauer, Teledyne Brown Engineering Technical Report CS93-LKD-004, 31 Dec 1992.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

Insufficient data to construct a Gabbard diagram.

METEOR 2-17

1988-005A

18820

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 30.46 January 1988
DRY MASS (KG): 2750
MAIN BODY: Cylinder
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: none at time of event
ENERGY SOURCES: unknown

EVENT DATA

| | | | |
|-----------|----------------|-----------------|---------------------|
| DATE: | 21 June 2005 | LOCATION: | 36.5N, 74.22E (asc) |
| TIME: | 09:14:54.5 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 954 km | | |

PRE-EVENT ELEMENTS

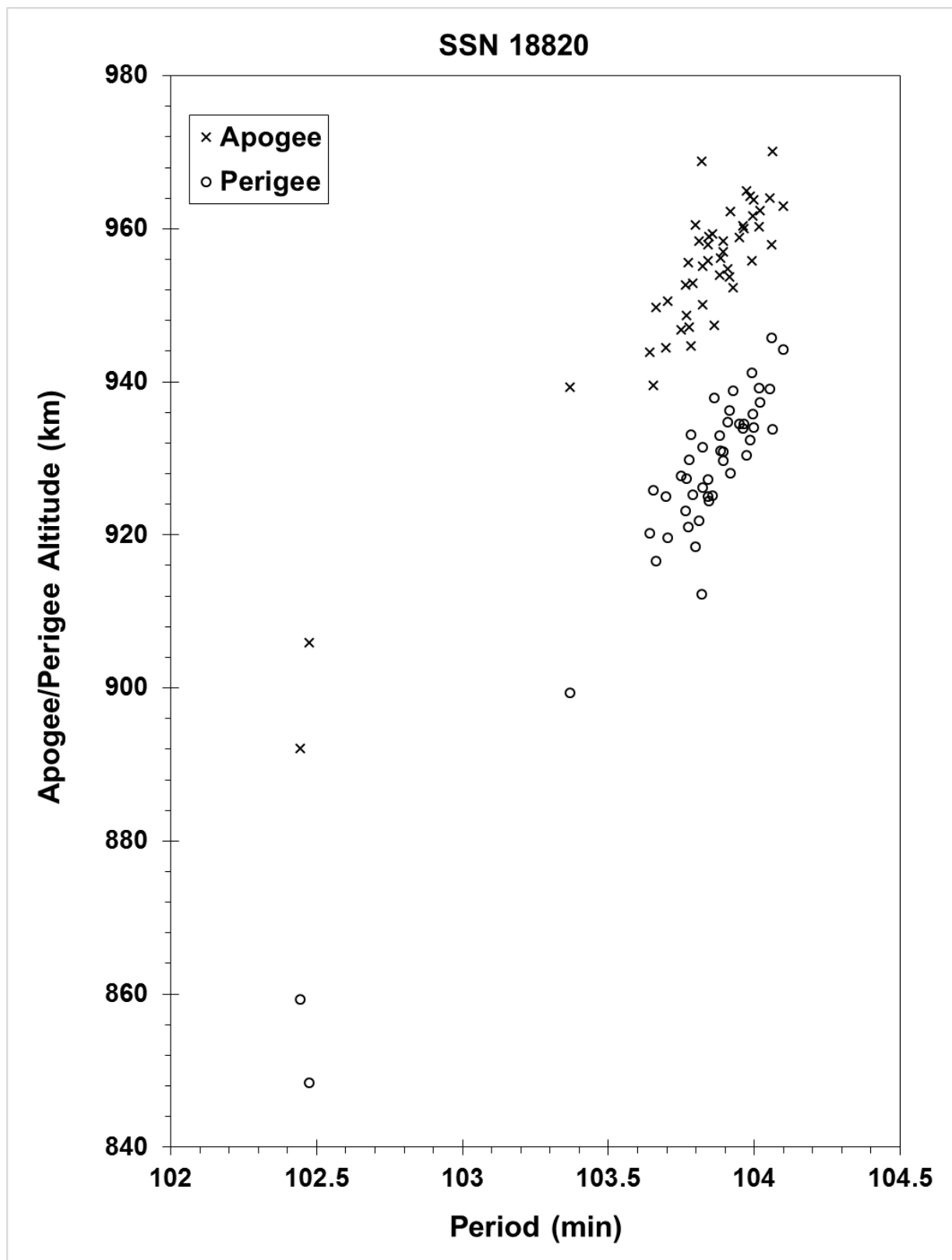
| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 05171.27320873 | MEAN ANOMALY: | 320.4591 |
| RIGHT ASCENSION: | 309.0605 | MEAN MOTION: | 13.85140738 |
| INCLINATION: | 82.5393 | MEAN MOTION DOT/2: | .00000047 |
| ECCENTRICITY: | .0018107 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 39.7894 | BSTAR: | .000027897 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 1.5 min
MAXIMUM ΔI : 0.03 deg

COMMENTS

This meteorological spacecraft suffered both an anomalous event as well as a breakup event.



Meteor 2-17 debris cloud cataloged up to seven years after the event as reconstructed from the US SSN database.

SATELLITE DATA

| | |
|-------------------|--|
| TYPE: | Payload |
| OWNER: | US |
| LAUNCH DATE: | 3.25 February 1988 |
| DRY MASS (KG): | 767 |
| MAIN BODY: | Cylinder; 1.0 m diameter by 3.6 m length |
| MAJOR APPENDAGES: | Solar panel and deployable instrumentation |
| ATTITUDE CONTROL: | none at time of event |
| ENERGY SOURCES: | unknown |

EVENT DATA

| | | | |
|-----------|---------------------|-----------------|---------|
| DATE: | 14-17 December 2012 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 12349.62481750 | MEAN ANOMALY: | 233.9639 |
| RIGHT ASCENSION: | 203.9910 | MEAN MOTION: | 14.26449692 |
| INCLINATION: | 98.8446 | MEAN MOTION DOT/2: | .00000181 |
| ECCENTRICITY: | .0006372 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 126.2134 | BSTAR: | .000094135 |

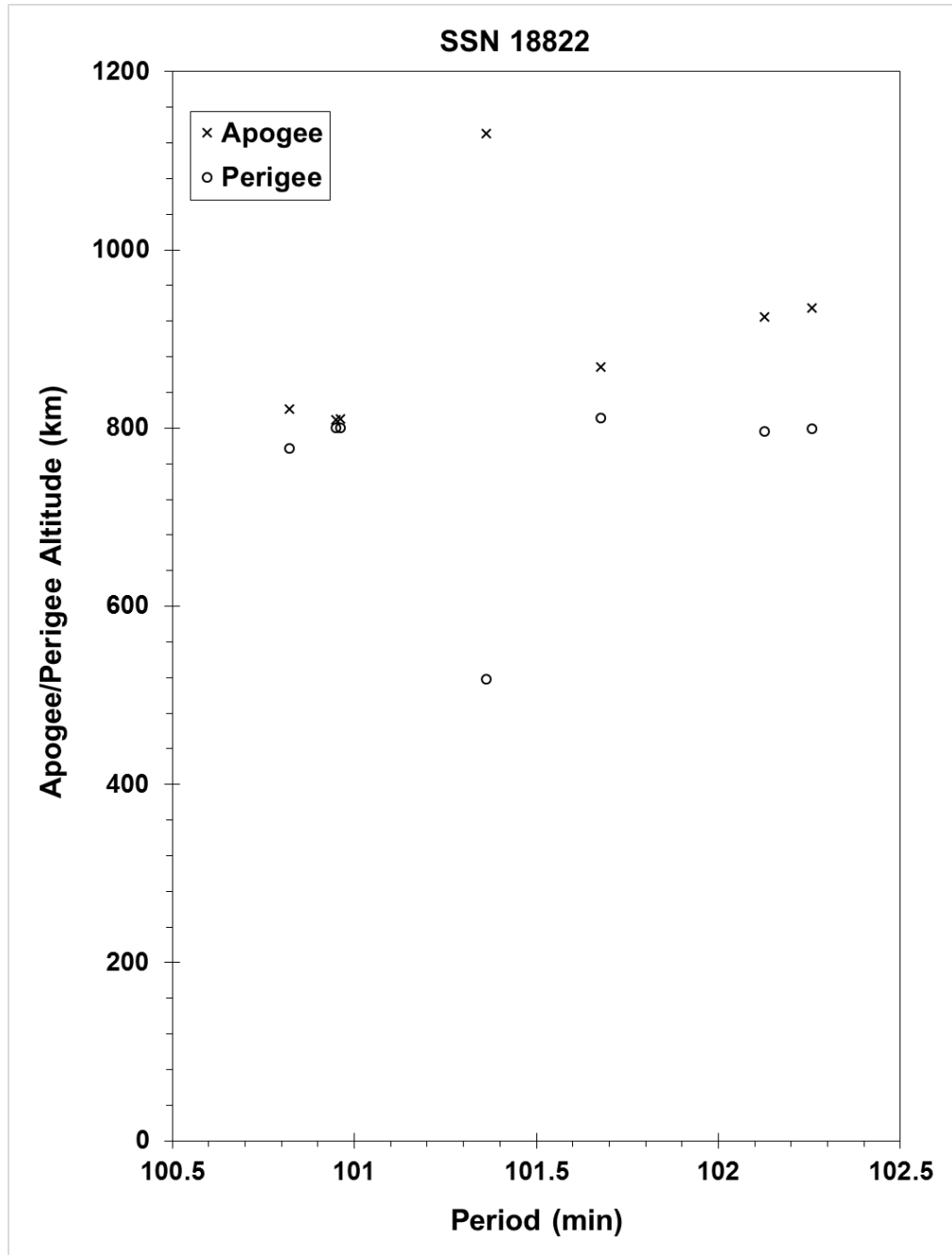
DEBRIS CLOUD DATA

| | |
|----------------------|---------|
| MAXIMUM ΔP : | 1.3 min |
| MAXIMUM ΔI : | 0.2 deg |

COMMENTS

Detected by software.

DMSP 5D-2 F9 debris objects B-F are mission-related debris typical for this spacecraft series. Pieces G-N (SSN 39041-6 inclusive, 43329) appear to be relatively high area-to-mass objects, with only two of the seven objects (39046 and 43329) remaining on orbit as of 4 July 2018. This event may be similar in root cause to the F11 (USA 73) and F13 (USA 109) events.



USA 29 debris cloud cataloged within two weeks of the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 3.15 Feb 1988
DRY MASS (KG): 5700
MAIN BODY: Cone-cylinder; 2.7 m diameter by 6.3 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

| | | | |
|-----------|-------------|-----------------|----------------|
| DATE: | 27 Feb 1988 | LOCATION: | 62N, 98E (asc) |
| TIME: | 0444 GMT | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | 155 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 88058.12322153 | MEAN ANOMALY: | 309.0154 |
| RIGHT ASCENSION: | 264.6529 | MEAN MOTION: | 16.30989909 |
| INCLINATION: | 64.8359 | MEAN MOTION DOT/2: | .03233928 |
| ECCENTRICITY: | .0060041 | MEAN MOTION DOT DOT/6: | .00003669 |
| ARG. OF PERIGEE: | 51.6410 | BSTAR: | .00025587 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 4.2 min*
MAXIMUM ΔI : 1.1 deg*

*Based on uncataloged debris data

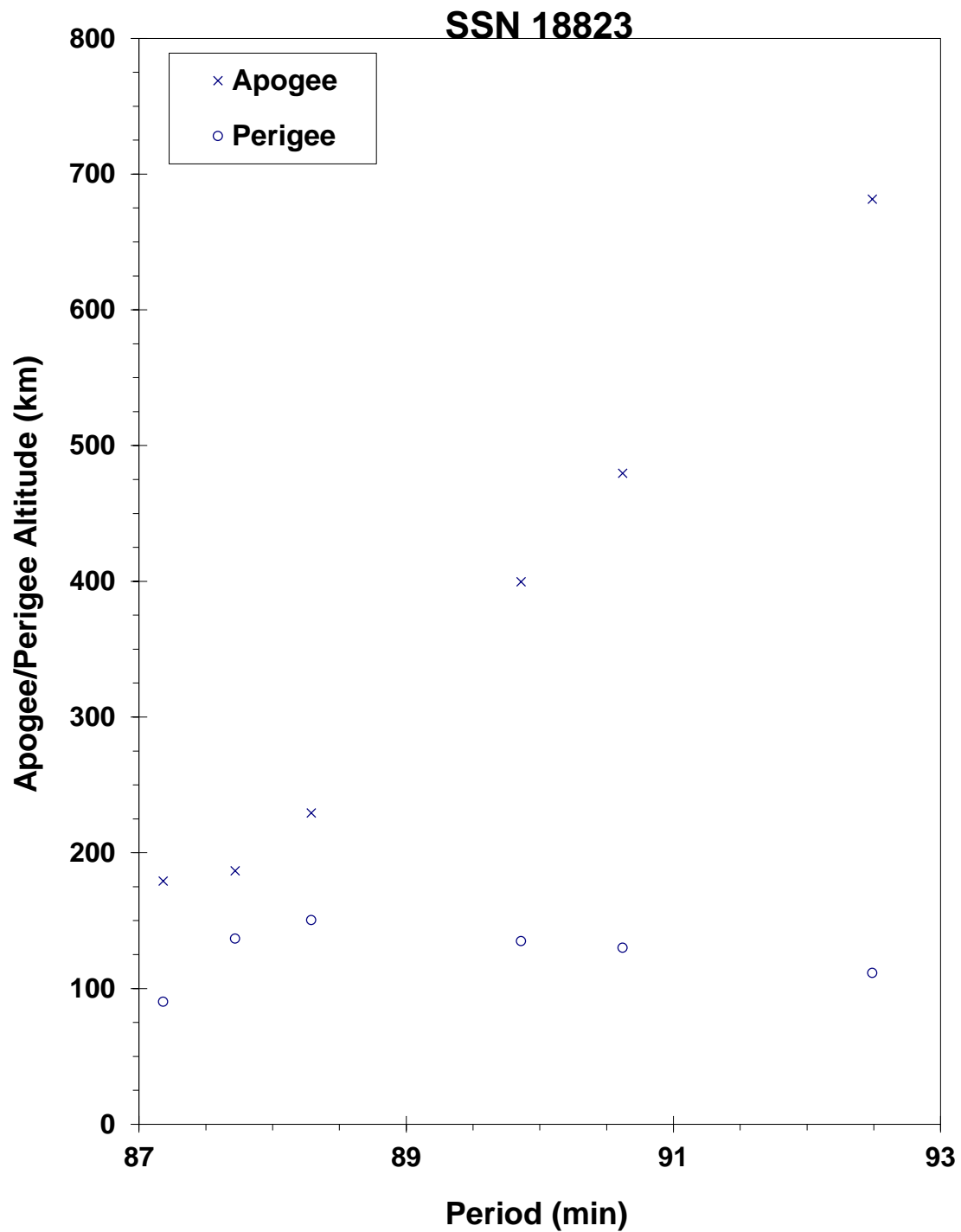
COMMENTS

Spacecraft destroyed after a malfunction prevented controlled reentry and landing in the Soviet Union. Early elements on only 6 objects available. All debris reentered before being officially cataloged.

REFERENCE DOCUMENTS

The Soviet Year in Space 1988, N. L. Johnson, Teledyne Brown Engineering, 1989, p. 31.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 1916 debris cloud remnant of 6 objects within 1 day of the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 22.59 Mar 1988
DRY MASS (KG): 800
MAIN BODY: Cylinder; 2.035 m diameter x 2 m length
MAJOR APPENDAGES: Several short booms
ATTITUDE CONTROL: Gravity-gradient (passive)
ENERGY SOURCES: Batteries

EVENT DATA

| | | | |
|-----------|-------------|-----------------|----------------------|
| DATE: | 23 Dec 1991 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Accidental Collision |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 91356.93360267 | MEAN ANOMALY: | 291.3330 |
| RIGHT ASCENSION: | 126.2142 | MEAN MOTION: | 13.75709229 |
| INCLINATION: | 82.9564 | MEAN MOTION DOT/2: | 0.00000135 |
| ECCENTRICITY: | .0041502 | MEAN MOTION DOT DOT/6: | 0.0 |
| ARG. OF PERIGEE: | 69.2265 | BSTAR: | 0.00012752 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

The collision occurred with a piece of launch debris from Cosmos 926. The debris piece (Sat. No. 13475) was evidently shattered into smaller, non-trackable debris by the impact and the public satellite catalog now notes that it "COLLIDED WITH SATELLITE" in lieu of the standard orbital elements. There were several very close conjunctions during the day in question, the exact time of the collision is unknown. Two pieces of debris were cataloged from the event long before the event was recognized as a collision.

REFERENCE DOCUMENT

"Accidental Collisions of Cataloged Satellites Identified", The Orbital Debris Quarterly News, NASA JSC, April 2005.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv9i2.pdf>.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Ariane 2 Third Stage
OWNER: France
LAUNCH DATE: 17.99 May 1988
DRY MASS (KG): ~1480
MAIN BODY: Cylinder; 2.6 m diameter by 11.7 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|------------|-----------------|--------------|
| DATE: | 9 Jul 2002 | LOCATION: | 4.3 N, 5.7 E |
| TIME: | 1930Z | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | 21,500 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 02190.22071506 | MEAN ANOMALY: | 172.0370 |
| RIGHT ASCENSION: | 187.4675 | MEAN MOTION: | 2.28211164 |
| INCLINATION: | 7.0311 | MEAN MOTION DOT/2: | .00000024 |
| ECCENTRICITY: | .7162572 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 181.6723 | BSTAR: | .0 |

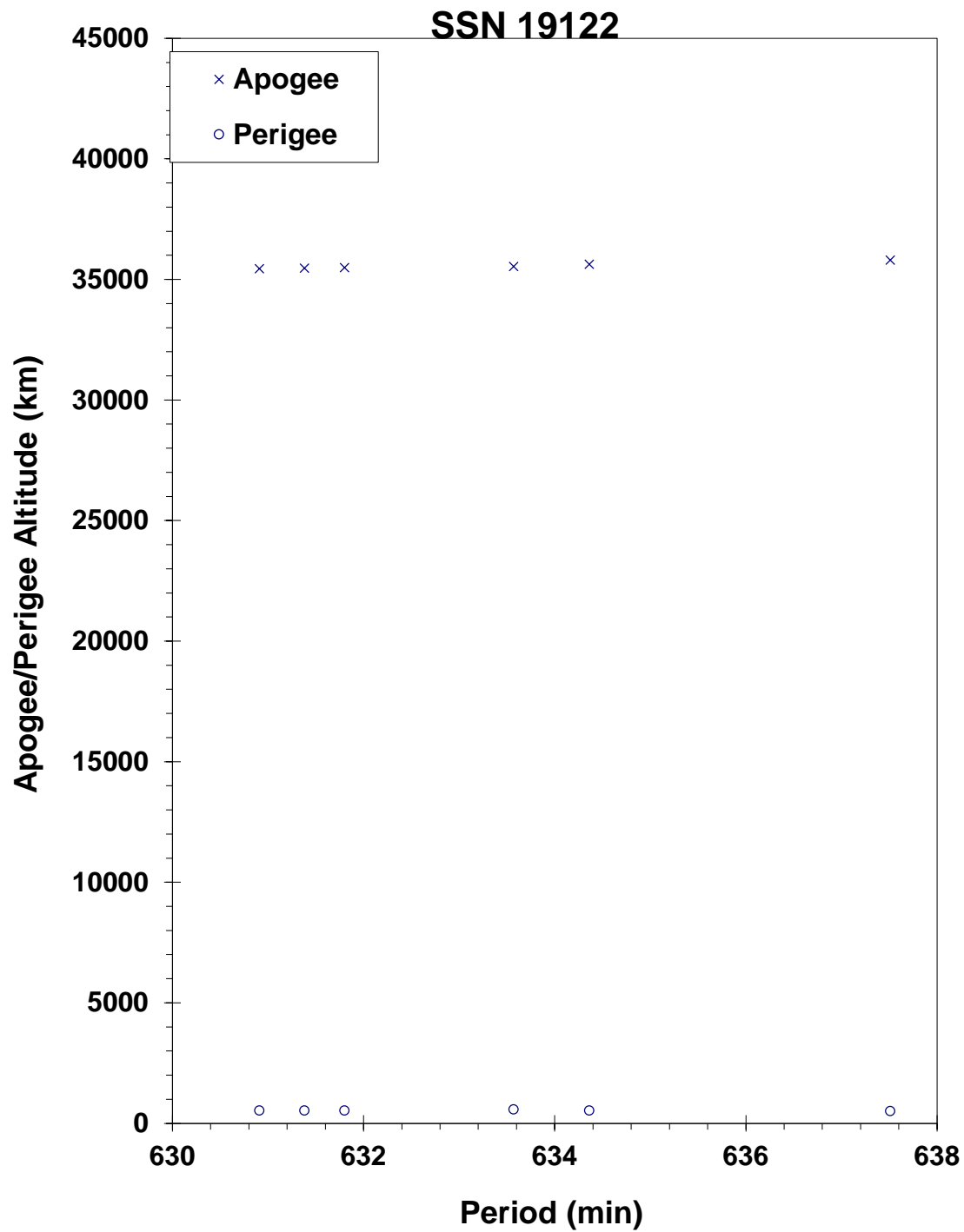
DEBRIS CLOUD DATA

MAXIMUM ΔP : 6.60 min*
MAXIMUM ΔI : 0.33 deg*

* Based on uncataloged debris data

COMMENTS

This is the second breakup of an Ariane 2 third stage officially recognized and the 11th overall breakup of an Ariane upper stage. This stage was launched prior to the implementation of passivation measures. The age of the stage at the time of the breakup was 14 years. Six pieces of debris were initially seen by the SSN, while four were cataloged.



Intelsat 513 R/B debris cloud of 6 fragments 2 weeks after the breakup as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 16.08 Sep 1988
DRY MASS (KG): ~55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1.0 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 04 Aug 2003 | LOCATION: | Unknown |
| TIME: | ~0725Z | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 03214.47923598 | MEAN ANOMALY: | 334.9286 |
| RIGHT ASCENSION: | 239.4643 | MEAN MOTION: | 4.29128214 |
| INCLINATION: | 65.3341 | MEAN MOTION DOT/2: | .00007107 |
| ECCENTRICITY: | .5561230 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 85.1870 | BSTAR: | .071402 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 18.8 min*
MAXIMUM ΔI : 2.79 deg*

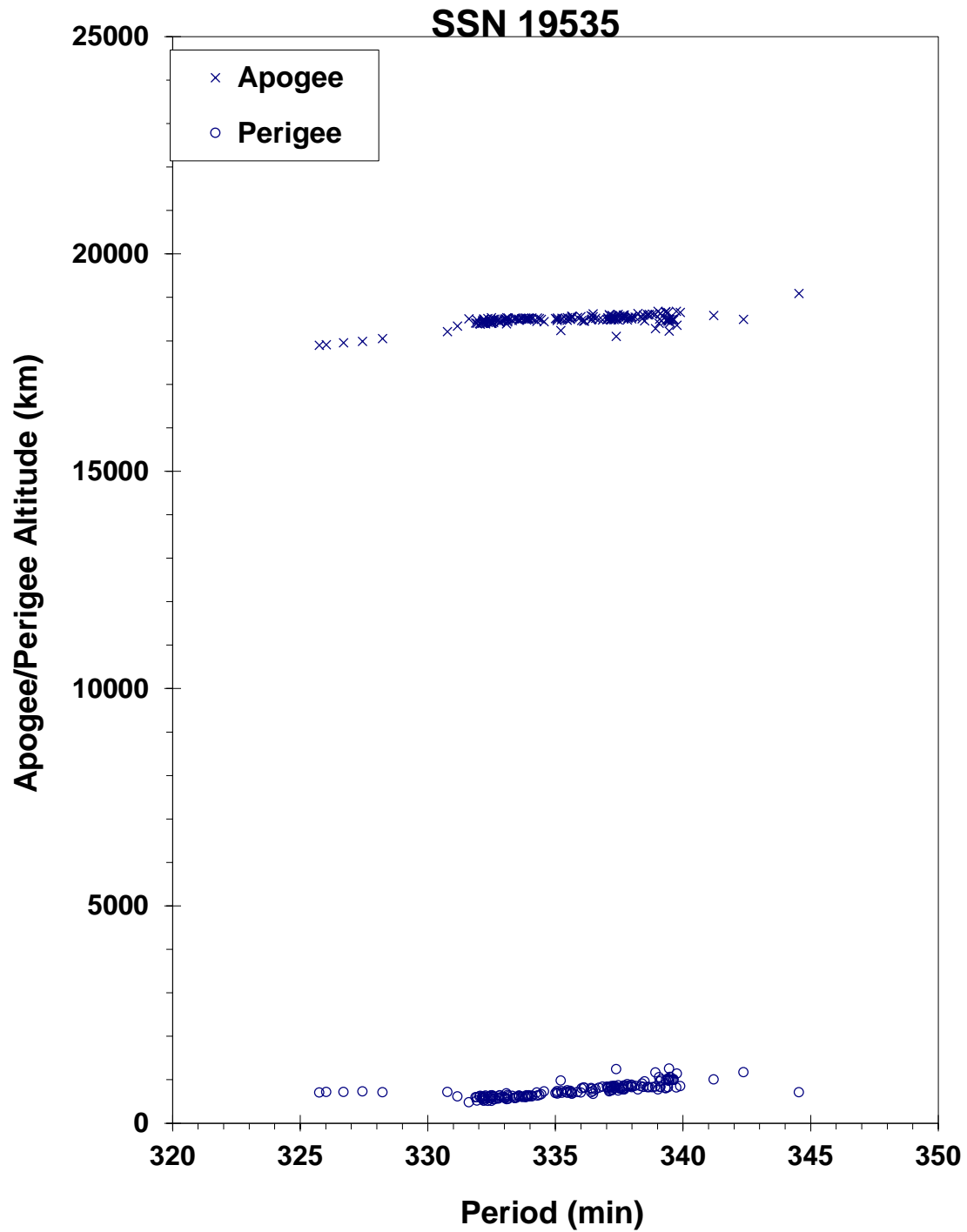
* Based on uncataloged debris data

COMMENTS

This event marks the 28th known breakup of a Proton Block DM SOZ ullage motor since 1984. This ullage motor was launched before implementation of breakup preventive measures. Approximately 175 objects were initially seen by the SSN 1 week after the event. 76 debris objects were cataloged.

REFERENCE DOCUMENT

“Satellite Fragmentations in 2003”, The Orbital Debris Quarterly News, NASA JSC, January 2004. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv8i1.pdf>.



Cosmos 1970-72 auxiliary motor debris cloud of 175 fragments 1 week after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 16 Sep 1988
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1.0 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|------------|-----------------|------------|
| DATE: | 9 Mar 1999 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

POST-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------------|
| EPOCH: | 99067.36656961 | MEAN ANOMALY: | 189.8576 |
| RIGHT ASCENSION: | 108.7309 | MEAN MOTION: | 4.28860956162171 |
| INCLINATION: | 64.6425 | MEAN MOTION DOT/2: | .00000813 |
| ECCENTRICITY: | .5827119 | MEAN MOTION DOT DOT/6: | 00000-0 |
| ARG. OF PERIGEE: | 176.8483 | BSTAR: | .0022335 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

This is the 19th event of the Proton Block DM SOZ Ullage Motor class identified to date; it is the eighth associated with a GLONASS mission. This mission was conducted before the engineering defect was identified and passivation measures implemented. In this highly-eccentric orbit, debris may be long-lived but hard to track. 17 debris objects were detected.

REFERENCE DOCUMENT

“Abandoned Proton Ullage Motors Continue to Create Debris”, The Orbital Debris Quarterly News, NASA JSC, April 1999. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv4i2.pdf>.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Ariane 4 H-10 Third Stage
OWNER: France
LAUNCH DATE: 11.02 Dec 1988
DRY MASS (KG): 1760
MAIN BODY: Cylinder; 2.6 m diameter by 11.4 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|-------------|-----------------|----------------------|
| DATE: | 17 Feb 1998 | LOCATION: | 6.9 N, 157.2 E (dsc) |
| TIME: | 1235 GMT | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | 19630 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 98047.29326560 | MEAN ANOMALY: | 25.3394 |
| RIGHT ASCENSION: | 23.7998 | MEAN MOTION: | 2.25942020 |
| INCLINATION: | 7.3381 | MEAN MOTION DOT/2: | 0.00000046 |
| ECCENTRICITY: | 0.7222736 | MEAN MOTION DOT DOT/6: | 0 |
| ARG. OF PERIGEE: | 248.1711 | BSTAR: | 0.00057969 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

This mission was the second for the Ariane 4 series and occurred prior to implementation of passivation measures. Using observations from the Eglin radar, specialists at Millstone radar found four new pieces from the upper stage. Naval Space Command personnel generated the first two debris element sets and calculated the approximate breakup time noted above.

REFERENCE DOCUMENT

“Three Upper Stage Breakups in One Week Top February Debris Activity”, The Orbital Debris Quarterly News, NASA JSC, April 1998, p. 1. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv3i2.pdf>.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 10 Jan 1989
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|------------|-----------------|------------|
| DATE: | 3 Aug 1998 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

POST-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 98211.80543118 | MEAN ANOMALY: | 172.2753 |
| RIGHT ASCENSION: | 16.7694 | MEAN MOTION: | 4.24137167 |
| INCLINATION: | 64.9243 | MEAN MOTION DOT/2: | .00000287 |
| ECCENTRICITY: | .5776927 | MEAN MOTION DOT DOT/6: | 00000-0 |
| ARG. OF PERIGEE: | 182.6029 | BSTAR: | .0041366 |

DEBRIS CLOUD DATA

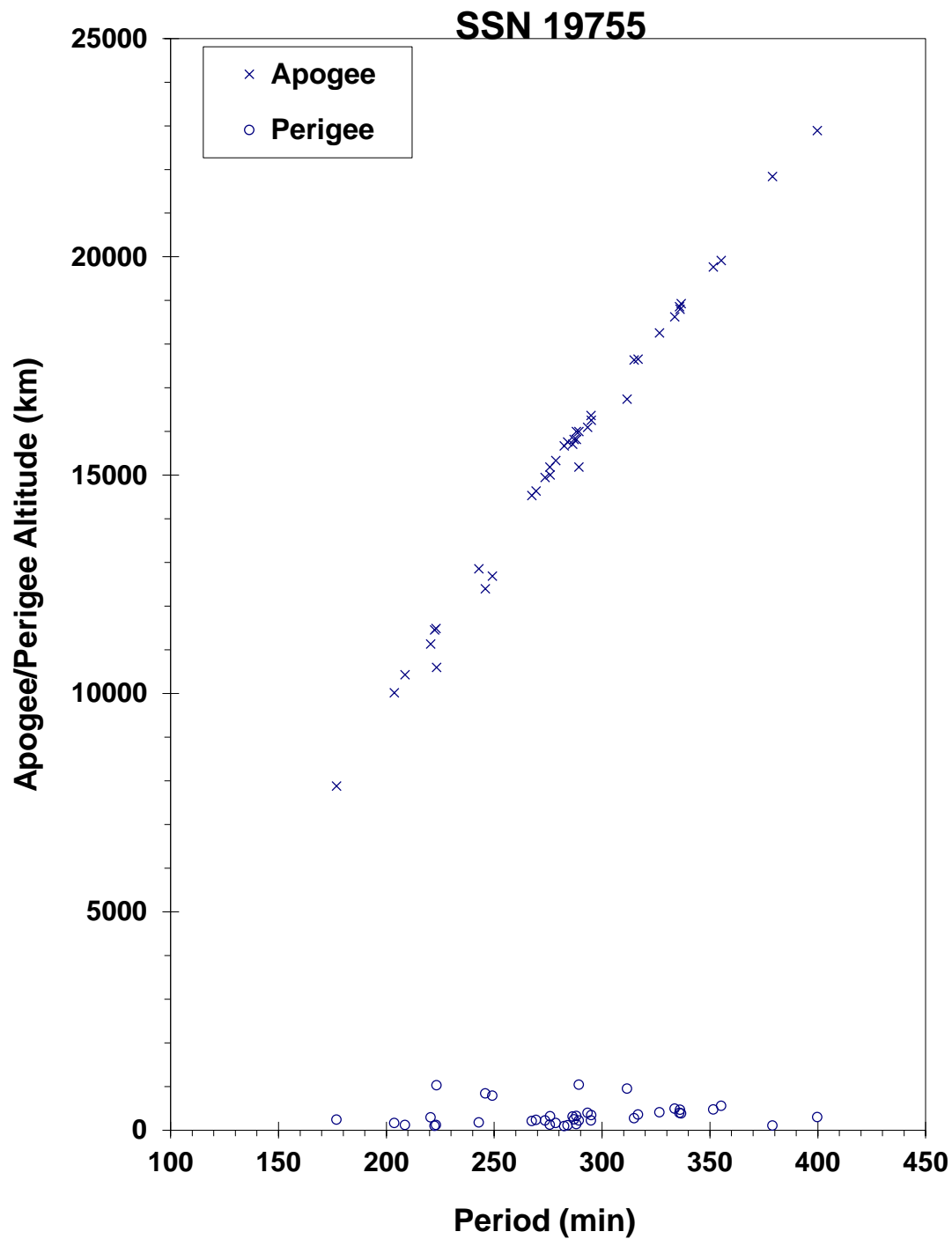
MAXIMUM ΔP : 162.64 min
MAXIMUM ΔI : 3.78 deg

COMMENTS

This is the 17th event of the Proton Block DM SOZ Ullage Motor class identified to date; it is the sixth associated with a GLONASS mission. This mission was conducted before the engineering defect was identified and passivation measures implemented. In this orbit, debris may be long-lived but hard to track. More than 110 debris objects were detected.

REFERENCE DOCUMENT

“Solitary Breakup and Anomalous Events in Third Quarter are Familiar”, The Orbital Debris Quarterly News, NASA JSC, October 1998. Available online at: <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv3i4.pdf>.



Cosmos 1987-1989 ullage motor debris cloud of 39 fragments 7 days after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 10 Jan 1989
DRY MASS (KG): ~55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1.0 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 13 Nov 2003 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 03317.76135862 | MEAN ANOMALY: | 339.1502 |
| RIGHT ASCENSION: | 52.9695 | MEAN MOTION: | 4.24824637 |
| INCLINATION: | 65.4357 | MEAN MOTION DOT/2: | .00000161 |
| ECCENTRICITY: | .5599025 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 72.44443 | BSTAR: | .0017638 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

This event marks the 29th known breakup of a Proton Block DM SOZ ullage motor since 1984, although the event went undetected for over 3 months. This ullage motor was launched before implementation of breakup preventive measures. No debris objects were cataloged from this breakup.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 26.39 Jan 1989
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1 m diameter
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|----------------|-----------------|------------|
| DATE: | 17-18 Dec 1992 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 92351.90838995 | MEAN ANOMALY: | 1.4295 |
| RIGHT ASCENSION: | 266.2338 | MEAN MOTION: | 4.60309514 |
| INCLINATION: | 46.7001 | MEAN MOTION DOT/2: | .00060784 |
| ECCENTRICITY: | .5692927 | MEAN MOTION DOT DOT/6: | .0000093219 |
| ARG. OF PERIGEE: | 353.9854 | BSTAR: | .0015056 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

Parent satellite was one of two small engine units that are routinely released after the first burn of the Proton fourth stage. The nature of these objects was identified by Dr. Boris V. Chernlatiev, Deputy Constructor for the Energiya NPO, in October 1992. The cause of this fragmentation is assumed to be related to the residual hypergolic propellants on board and failure of the membrane separating the fuel and oxidizer. NAVSPASUR observed between 30-40 objects that were associated with this breakup. Only 4 element sets were generated, insufficient for a Gabbard Diagram or BLAST point. This was the seventh in a series of fragmentations of this object type, and was the second located in a geosynchronous transfer orbit.

REFERENCE DOCUMENTS

The Fragmentation of Proton Debris, D. J. Nauer, TBE Technical Report CS93-LKD-004, Teledyne Brown Engineering, Colorado Springs, 31 December 1992.

Analysis of Fragmentations From December 1992 - February 1993, TBE Technical Report CS93-LKD-010, Teledyne Brown Engineering, Colorado Springs, 30 March 1993.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

"Identification and Resolution of an Orbital Debris Problem with the Proton Launch Vehicle", B. V. Cherniatiev et al, Proceedings of the First European Conference on Space Debris, April 1993.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Ariane 2 third stage with VEB
OWNER: France
LAUNCH DATE: 27.06 Jan 1989
DRY MASS (KG): ~1480 kg
MAIN BODY: 2.6 m diameter by 11.7 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants?

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | ~1 Jan 2001 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 00366.06151127 | MEAN ANOMALY: | 45.8970 |
| RIGHT ASCENSION: | 73.3900 | MEAN MOTION: | 2.26500973 |
| INCLINATION: | 8.3781 | MEAN MOTION DOT/2: | .00000580 |
| ECCENTRICITY: | .7188412 | MEAN MOTION DOT DOT/6: | .0000000 |
| ARG. OF PERIGEE: | 225.8250 | BSTAR: | .0040973 |

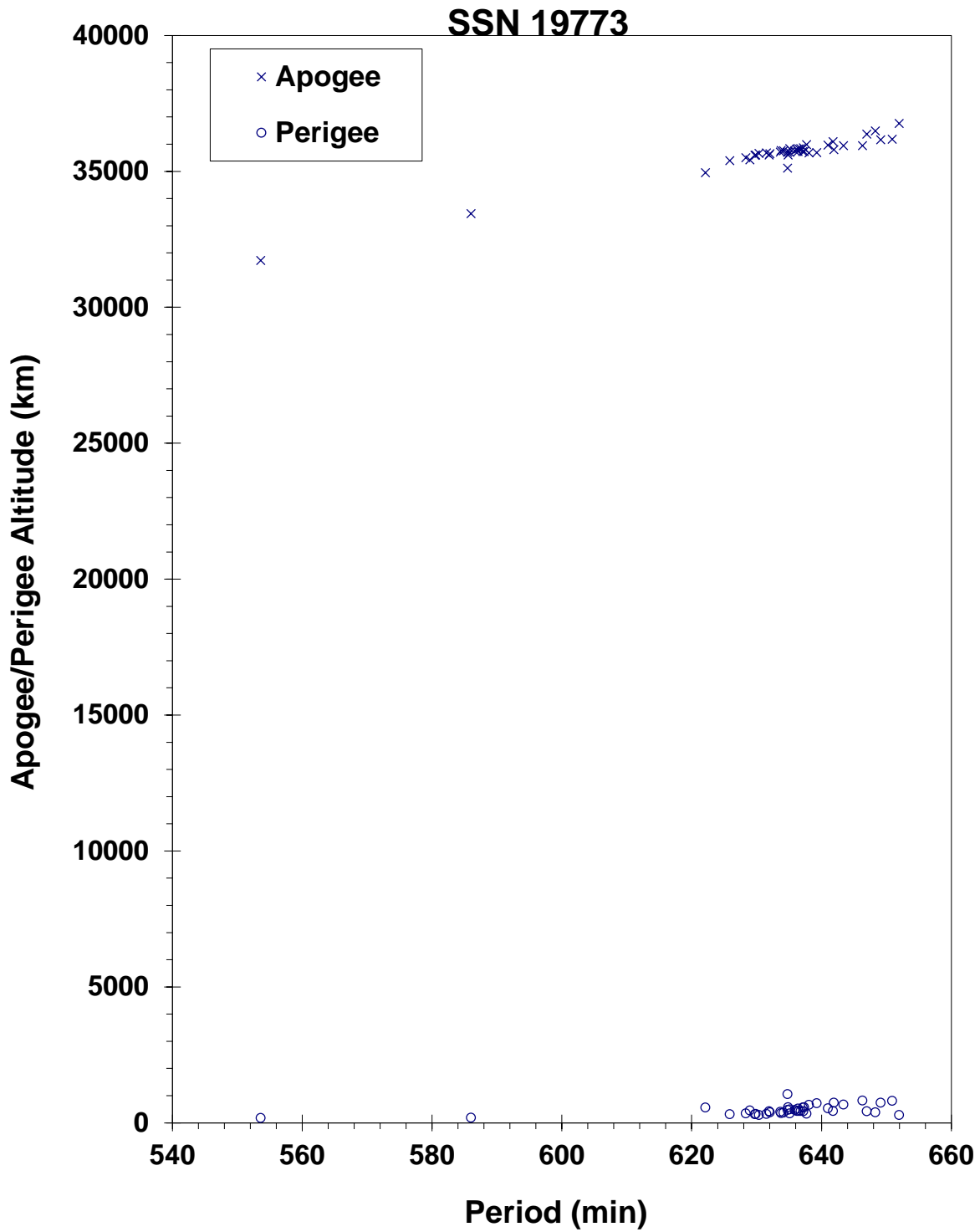
DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown*
MAXIMUM ΔI : Unknown*

* Not calculated due to provisional nature
of orbital data at event time.

COMMENTS

This is the first breakup of an Ariane 2 third stage officially recognized. One Ariane 3 third stage (same as Ariane 2) is known to have broken-up within a few days of launch in 1987. Both vehicles were launched before passivation measures were incorporated with Ariane third stages. Ariane third stage passivation was introduced in January 1990 and has been employed on all Ariane missions since October 1993. The age of the Ariane 2 third stage at the time of the breakup was nearly 12 years.



INTELSAT V F-15 R/B (Ariane 2) debris cloud as constructed using SSN 8XXXX series elements sets (10 January 2001 and before).

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 31.36 May 1989
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellant

EVENT DATA

DATE: 10 Jun 2006 LOCATION: 65S, 100E (dsc)
TIME: 1320 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 17,375 km

PRE-EVENT ELEMENTS

EPOCH: 06161.39815228 MEAN ANOMALY: 338.3349
RIGHT ASCENSION: 289.1150 MEAN MOTION: 4.32576815
INCLINATION: 65.0603 MEAN MOTION DOT/2: -0.00002708
ECCENTRICITY: .5578964 MEAN MOTION DOT DOT/6: 0.0
ARG. OF PERIGEE: 74.2422 BSTAR: -0.018697

DEBRIS CLOUD DATA

MAXIMUM ΔP : 12.9 min
MAXIMUM ΔI : 2.1 deg

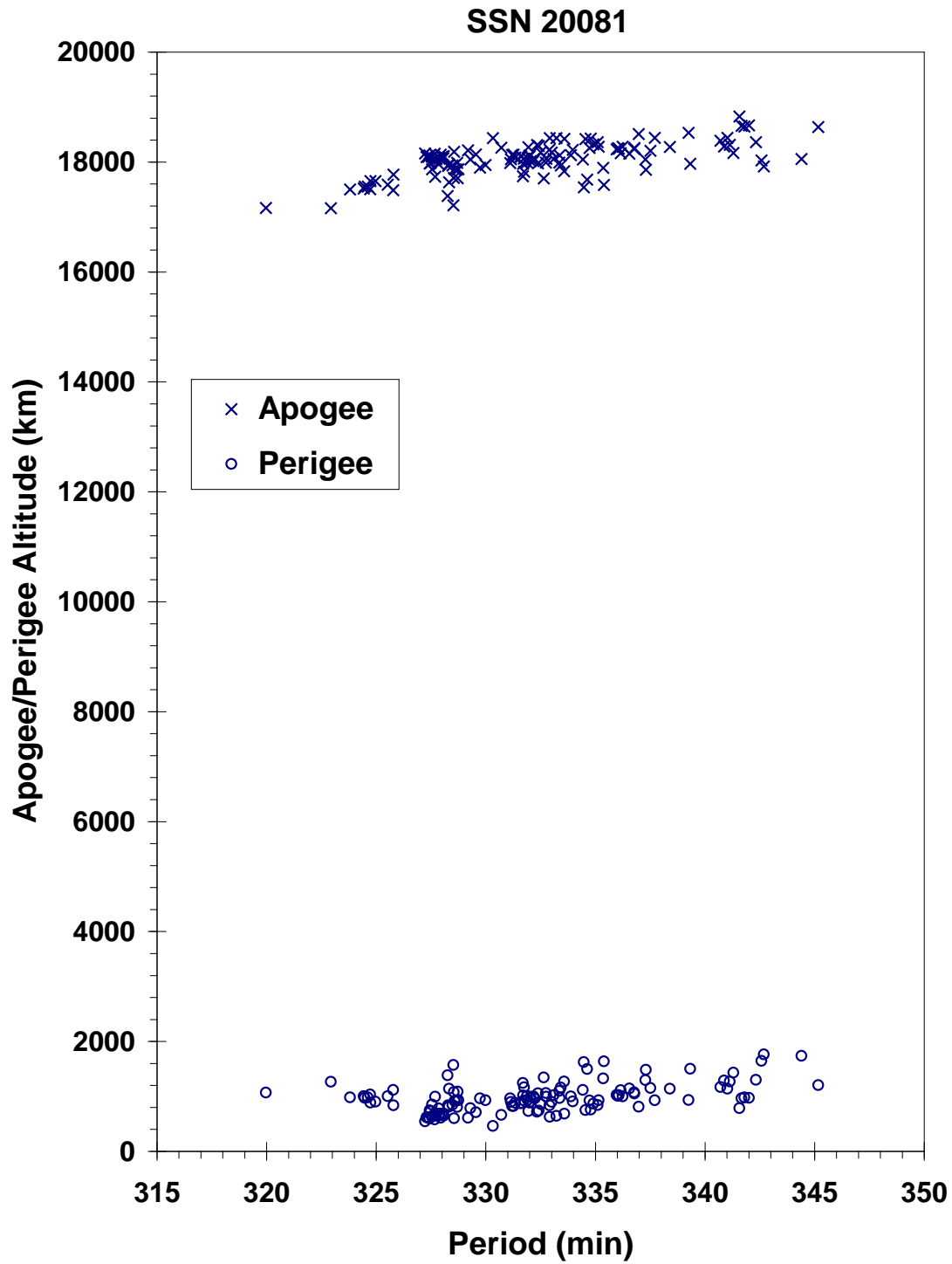
COMMENTS

This event marks the 34th known breakup of a Proton Block DM SOZ ullage motor since 1984. There were two more fragmentation events for this object during July 2006; the second event was on July 3, the third event was July 27. These events resulted in the most debris cataloged of any SOZ ullage motor in history with over 100 pieces cataloged. The majority of debris (>75) were created during the second event on 3 July 2006.

REFERENCE DOCUMENTS

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

“First Satellite Breakups of 2006”, The Orbital Debris Quarterly News, NASA JSC, July 2006.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv10i3.pdf>.



SOZ motor debris cloud around September 2006 with 131 objects, reflecting debris from all three fragmentation events as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 5.95 Jul 1989
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

DATE: 12 Jan 1993 LOCATION: Unknown
TIME: Unknown ASSESSED CAUSE: Propulsion
ALTITUDE: Unknown

PRE-EVENT ELEMENTS

EPOCH: 93004.96424013 MEAN ANOMALY: 353.7659
RIGHT ASCENSION: 215.2912 MEAN MOTION: 2.68234049
INCLINATION: 46.7556 MEAN MOTION DOT/2: .00007021
ECCENTRICITY: .6967354 MEAN MOTION DOT DOT/6: .0
ARG. OF PERIGEE: 45.1358 BSTAR: .0017532

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

Parent satellite was one of two small engine units that are routinely released after the first burn of the Proton fourth stage. The nature of these objects was identified by Dr. Boris V. Chernlatiev, Deputy Constructor for the Energiya NPO, in October 1992. The cause of this fragmentation is assumed to be related to the residual hypergolic propellants on board and failure of the membrane separating the fuel and oxidizer. NAVSPASUR observed 18 objects that were associated with this breakup. Only 5 element sets were generated, and were of insufficient quality for a credible Gabbard Diagram or BLAST point. This was the eighth in a series of fragmentations of this object type, and was the third located in a geosynchronous transfer orbit.

REFERENCE DOCUMENTS

The Fragmentation of Proton Debris, D. J. Nauer, TBE Technical Report CS93-LKD-004, Teledyne Brown Engineering, Colorado Springs, 31 December 1992.

Analysis of Fragmentations From December 1992 - February 1993, TBE Technical Report CS93-LKD-010, Teledyne Brown Engineering, Colorado Springs, 30 March 1993.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

"Identification and Resolution of an Orbital Debris Problem with the Proton Launch Vehicle", B. V. Cherniatiev et al, Proceedings of the First European Conference on Space Debris, April 1993.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 12.63 Jul 1989
DRY MASS (KG): 5700
MAIN BODY: Cone-cylinder; 2.7 m diameter by 6.3 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

DATE: 28 Jul 1989
TIME: 0410-0420 GMT
ALTITUDE: 150 km
LOCATION: 35-65N, 95-140E (asc)
ASSESSED CAUSE: Deliberate

PRE-EVENT ELEMENTS

EPOCH: 89208.98384568
RIGHT ASCENSION: 89.7470
INCLINATION: 67.1441
ECCENTRICITY: .0048139
ARG. OF PERIGEE: 57.9032
MEAN ANOMALY: 302.7810
MEAN MOTION: 16.33519268
MEAN MOTION DOT/2: .03079561
MEAN MOTION DOT DOT/6: .000029506
BSTAR: .00023479

DEBRIS CLOUD DATA

MAXIMUM ΔP : 7.1 min*
MAXIMUM ΔI : 1.3 deg*

*Based on uncataloged debris data

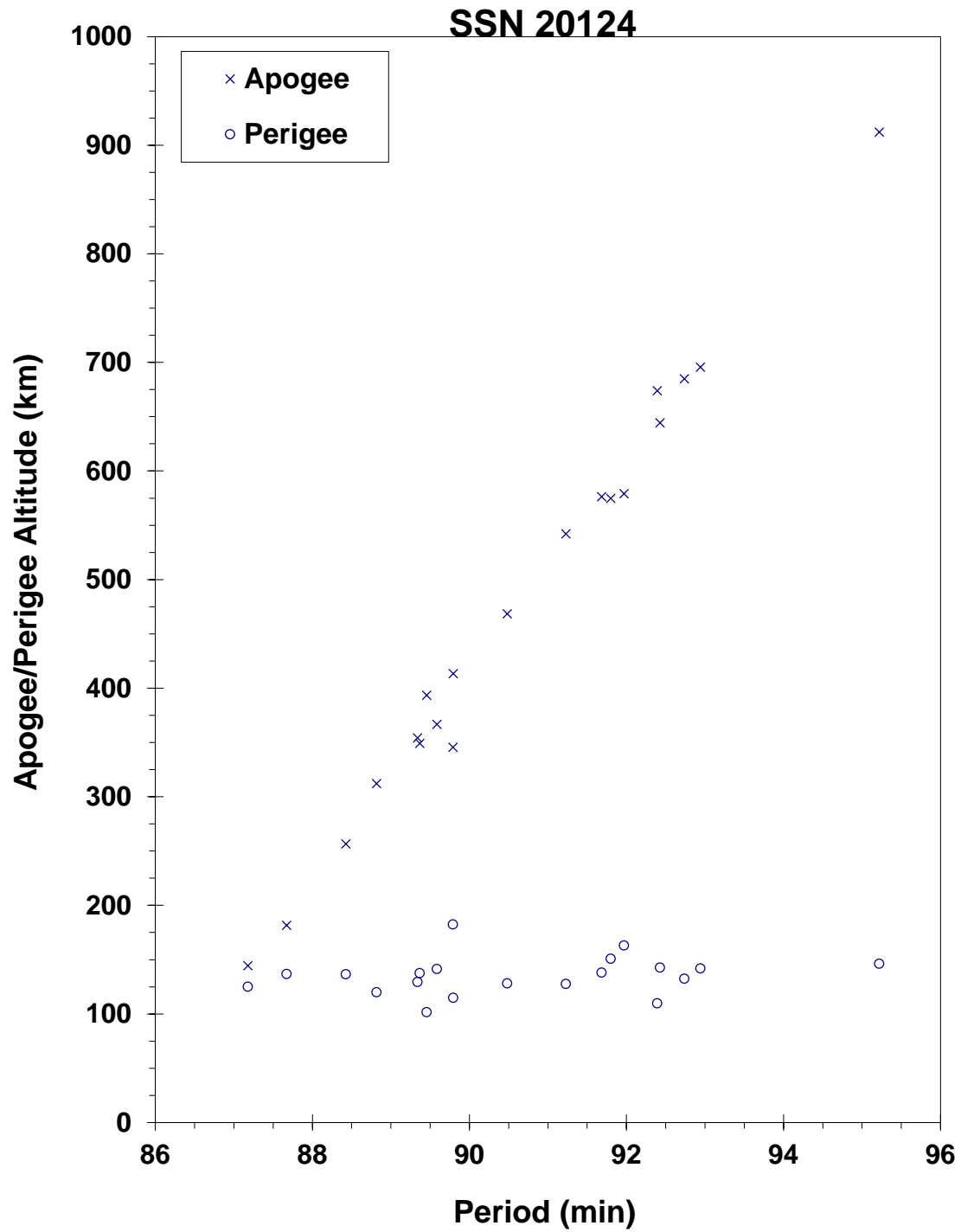
COMMENTS

Spacecraft was destroyed after a malfunction prevented controlled reentry and landing in the Soviet Union. Early element sets on only 20 objects available. Rapid decay of objects made calculation of breakup time and location difficult.

REFERENCE DOCUMENTS

The Fragmentation of Kosmos 2030, N. L. Johnson, Technical Report CS89-TR-JSC-002, Teledyne Brown Engineering, Colorado Springs, Colorado, September 1989.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 2030 debris cloud remnant of 20 objects two to 3 days after the event as reconstructed from the US SSN database. This diagram is taken from the cited reference.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 18.51 Jul 1989
DRY MASS (KG): 6000
MAIN BODY: Cylinder; 2.4 m diameter by 7 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

DATE: 31 Aug 1989 LOCATION: 43N, 111E (dsc)
TIME: 1851 GMT ASSESSED CAUSE: Deliberate
ALTITUDE: 270 km

PRE-EVENT ELEMENTS

EPOCH: 89243.76468690 MEAN ANOMALY: 305.4386
RIGHT ASCENSION: 242.9132 MEAN MOTION: 15.89273241
INCLINATION: 50.5464 MEAN MOTION DOT/2: .00196451
ECCENTRICITY: .0093577 MEAN MOTION DOT DOT/6: .00002154
ARG. OF PERIGEE: 55.5300 BSTAR: .00045172

DEBRIS CLOUD DATA

MAXIMUM ΔP : 7.4 min*
MAXIMUM ΔI : 0.9 deg*

*Based on uncataloged debris data

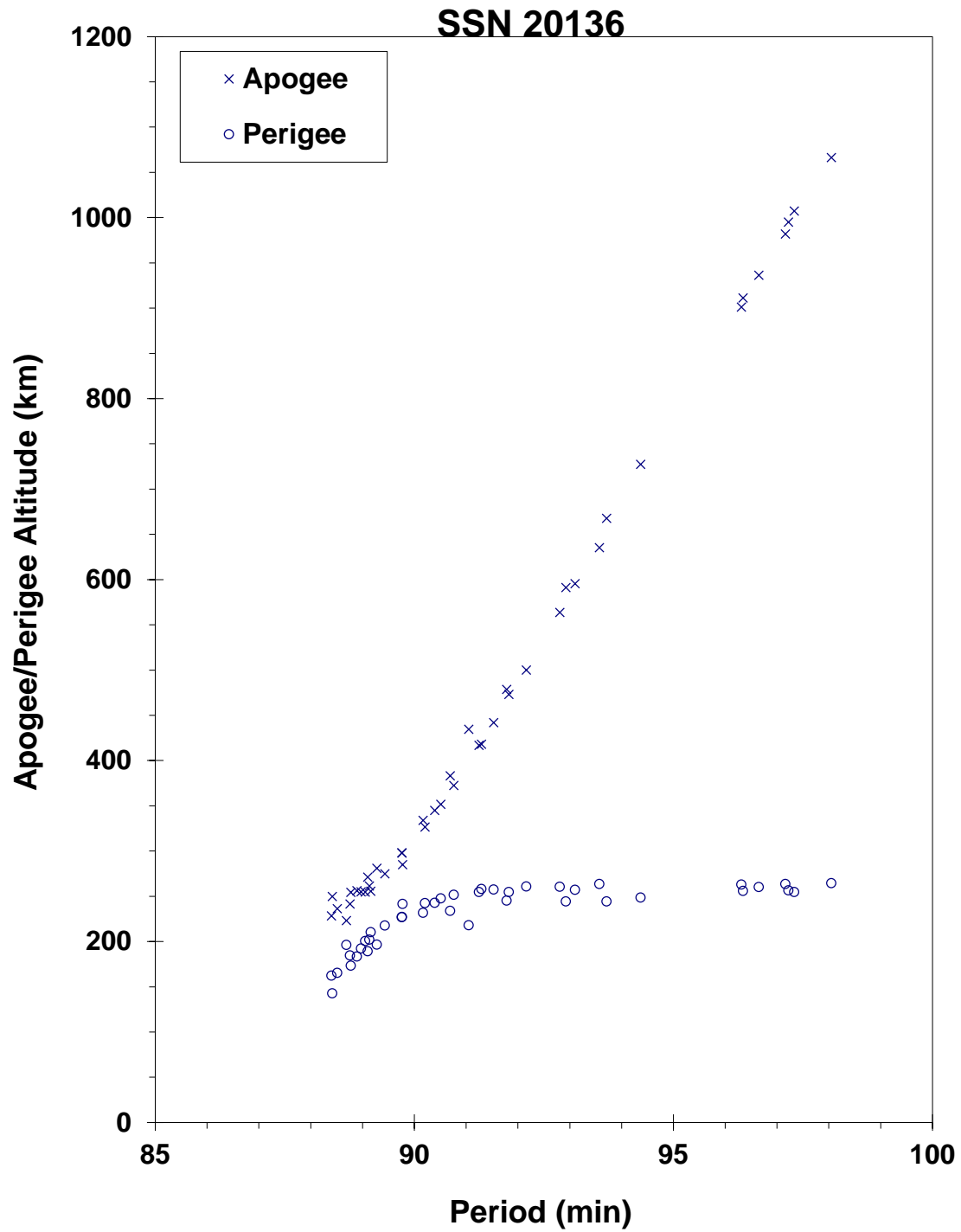
COMMENTS

Spacecraft was destroyed with a planned detonation. Cosmos 2031 was the first of a new series of spacecraft that employs end-of-mission detonation as standard operating procedure. Early elements on 43 objects available. Most debris reentered before being officially cataloged.

REFERENCE DOCUMENTS

The Fragmentation of Kosmos 2031, N. L. Johnson, Technical Report CS89-TR-JSC-003, Teledyne Brown Engineering, Colorado Springs, Colorado, September 1989.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 2031 debris cloud remnant of 43 objects 3 days after the event as reconstructed from Naval Space Surveillance System database. This diagram is taken from the cited reference.

SATELLITE DATA

TYPE: Rocket Body
OWNER: US
LAUNCH DATE: 18.61 Nov 1989
DRY MASS (KG): 920
MAIN BODY: Cylinder; 2.4 m diameter by 6.0 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of event
ENERGY SOURCES: None at time of event

EVENT DATA

DATE: 3 Dec 2006 LOCATION: 52S, 168E (dsc)
TIME: 0337 GMT ASSESSED CAUSE: Unknown
ALTITUDE: 730 km

PRE-EVENT ELEMENTS

EPOCH: 06336.48315357 MEAN ANOMALY: 65.8381
RIGHT ASCENSION: 123.6830 MEAN MOTION: 14.46527792
INCLINATION: 97.0839 MEAN MOTION DOT/2: 0.00000076
ECCENTRICITY: .0073269 MEAN MOTION DOT DOT/6: 0.0
ARG. OF PERIGEE: 293.5127 BSTAR: 0.000029963

DEBRIS CLOUD DATA

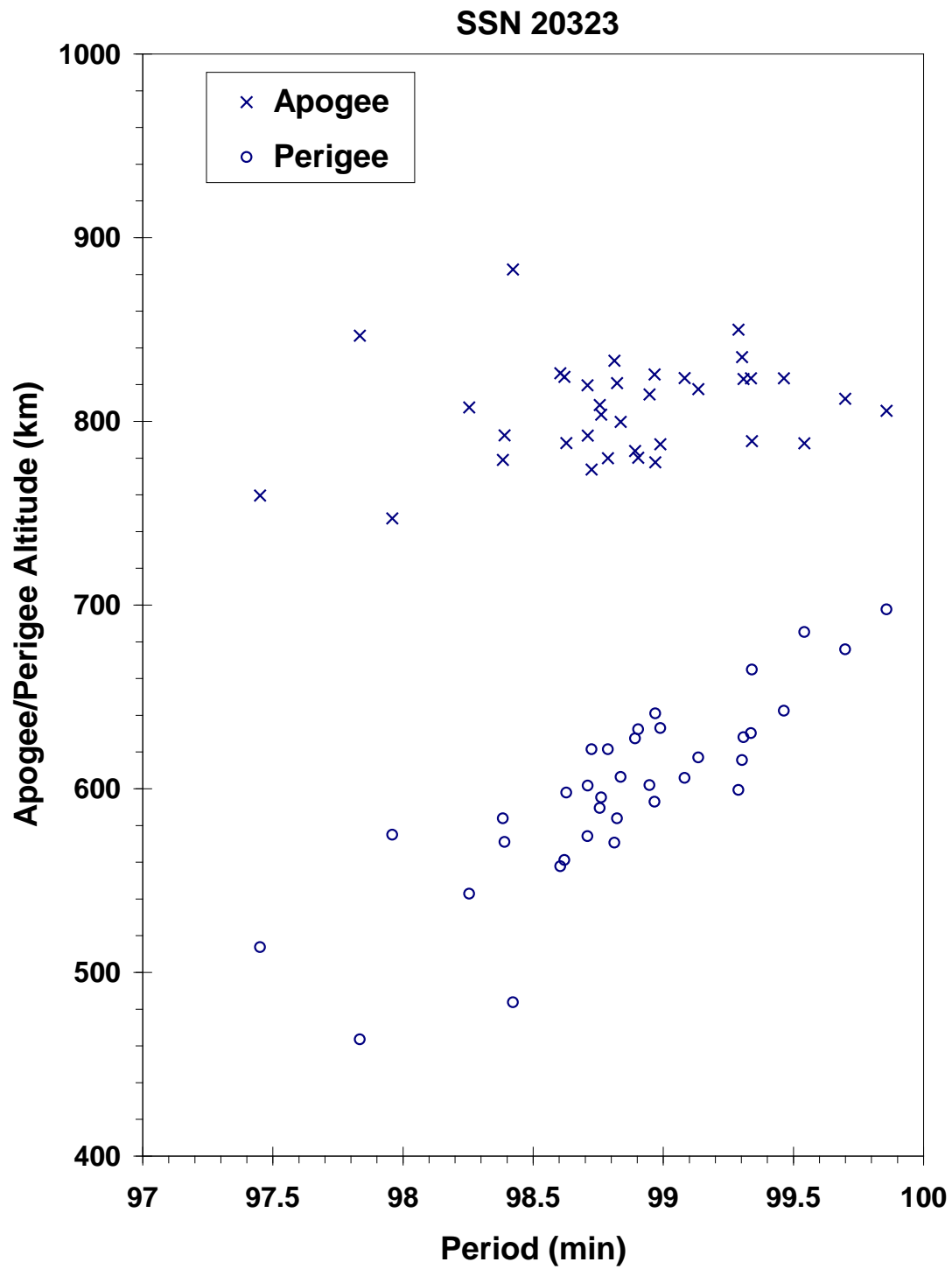
MAXIMUM ΔP : 2.1 min
MAXIMUM ΔI : 0.3 deg

COMMENTS

The Delta 2 rocket body had been passivated and dormant for 17 years. Observation of the object showed a high tumbling rate after the event. A collision with a smaller particle has not been ruled out. For the most part, the cataloged objects from this event were short-lived, i.e., less than 1 month.

REFERENCE DOCUMENT

“Significant Increase in Satellite Breakups During 2006”, The Orbital Debris Quarterly News, NASA JSC, January 2007.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv11i1.pdf>.



COBE R/B debris cloud with 36 fragments, 6 days after the event
as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Tsyklon Third Stage
OWNER: CIS
LAUNCH DATE: 27 Dec 1989
DRY MASS (KG): 1360
MAIN BODY: Cone-cylinder; 2.1 m diameter by 3.3 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: Unknown

EVENT DATA

DATE: 18 Apr 1999 LOCATION: 16.9S, 234.1E
TIME: 0119 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 485 km

POST-EVENT ELEMENTS

EPOCH: 99107.56102679 MEAN ANOMALY: 26.3814
RIGHT ASCENSION: 275.5509 MEAN MOTION: 15.29126555517603
INCLINATION: 73.5159 MEAN MOTION DOT/2: .00003667
ECCENTRICITY: .0010450 MEAN MOTION DOT DOT/6: 00000-0
ARG. OF PERIGEE: 333.6852 BSTAR: .0013164

DEBRIS CLOUD DATA

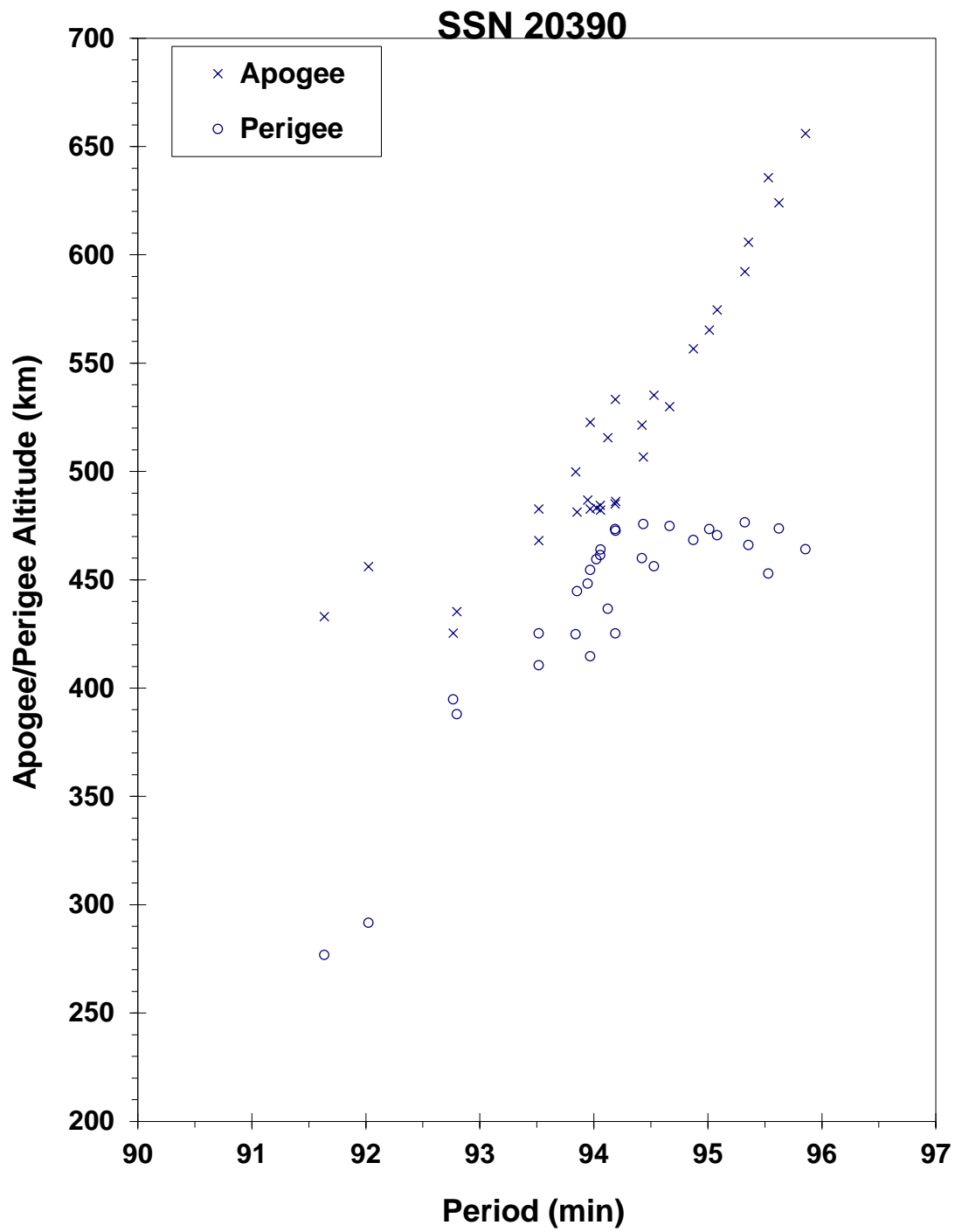
MAXIMUM ΔP : 4.18 min
MAXIMUM ΔI : .66 deg

COMMENTS

This is the 3rd event of the Tsyklon third stage (SL-14) identified to date, and the second within 14 months. All stages have been about 10 years old at the time of breakup. The vehicle is a Ukrainian-produced stage with unknown end-of-mission passivation. Its propellants are UDMH and N2O4. More than 60 debris objects were detected.

REFERENCE DOCUMENT

"Third Tsyklon Upper Stage Breaks Up", The Orbital Debris Quarterly News, NASA JSC, July 1999. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv4i3.pdf>



Cosmos 2053 rocket body debris cloud of 31 fragments 9 days after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 27.47 Dec 1989
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|--------------|-----------------|------------|
| DATE: | Jul 1992 (?) | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|-----------------|------------------------|------------|
| EPOCH: | 92182.661921495 | MEAN ANOMALY: | 6.2737 |
| RIGHT ASCENSION: | 305.7529 | MEAN MOTION: | 2.98492104 |
| INCLINATION: | 47.1115 | MEAN MOTION DOT/2: | .00001757 |
| ECCENTRICITY: | .6700939 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 319.3202 | BSTAR: | .0014976 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

Parent satellite was one of two small engine units that are routinely released after the first burn of the Proton fourth stage. The nature of these objects was identified by Dr. Boris V. Chernlatiev, Deputy Constructor for the Energiya NPO, in October 1992. The cause of this fragmentation is assumed to be related to the residual hypergolic propellants on board and failure of the membrane separating the fuel and oxidizer. NAVSPASUR observed 18 objects that were associated with this breakup. Twelve element sets were generated, but were of insufficient quality for a credible Gabbard Diagram or BLAST point. One object was cataloged on this event in early August 1992. This was the fifth in a series of fragmentations of this object type, and was the first located in a geosynchronous transfer orbit.

REFERENCE DOCUMENTS

The Fragmentation of Proton Debris, D. J. Nauer, TBE Technical Report CS93-LKD-004, Teledyne Brown Engineering, Colorado Springs, 31 December 1992.

Analysis of Fragmentations From December 1992 - February 1993, TBE Technical Report CS93-LKD-010, Teledyne Brown Engineering, Colorado Springs, 30 March 1993.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

"Identification and Resolution of an Orbital Debris Problem with the Proton Launch Vehicle", B. V. Cherniatiev et al, Proceedings of the First European Conference on Space Debris, April 1993.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 20.0 May 1990
DRY MASS (KG): 56
MAIN BODY: ellipsoid; 0.6 m diameter x 1 m length
MAJOR APPENDAGES: none
ATTITUDE CONTROL: none
ENERGY SOURCES: on-board propellants

EVENT DATA

DATE: 18 Nov 2011 LOCATION: 60.2S, 217.8E (asc)
TIME: 0344 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 1292.55 km

PRE-EVENT ELEMENTS

EPOCH: 11321.47402816 MEAN ANOMALY: 6.2919
RIGHT ASCENSION: 24.8398 MEAN MOTION: 4.33127778
INCLINATION: 65.0258 MEAN MOTION DOT/2: 0.00000578
ECCENTRICITY: 0.5722498 MEAN MOTION DOT DOT/6: 0.0
ARG. OF PERIGEE: 332.9281 BSTAR: 0.00072502

DEBRIS CLOUD DATA

MAXIMUM ΔP : 0 min
MAXIMUM ΔI : 0 deg

COMMENTS

This is the 39th known breakup of a Proton-DM SOZ ullage unit since 1984. The breakups are believed to be caused by residual propellants. Russian officials have been aware of the problem since 1992 and have made design changes, although the date of full implementation is unknown. Only one object has been officially cataloged. Due to the difficulty in observing and cataloging debris in deep-space, highly elliptical orbits, a much larger debris ensemble may have been created by this event.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations: A Joint U.S.-Russian Investigation. N.L. Johnson et al. Kaman Sciences Corporation, October 2005.

“Only a Few Minor Satellite Breakups in 2011”, The Orbital Debris Quarterly News, NASA JSC, January 2012.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/ODQNv16i1.pdf>.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 19 May 1990
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|--------------|-----------------|------------|
| DATE: | ~28 Mar 1999 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

POST-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------------|
| EPOCH: | 99087.88291821 | MEAN ANOMALY: | 123.5812 |
| RIGHT ASCENSION: | 319.9610 | MEAN MOTION: | 4.24414150137202 |
| INCLINATION: | 64.8090 | MEAN MOTION DOT/2: | 0.00000311 |
| ECCENTRICITY: | 0.5789417 | MEAN MOTION DOT DOT/6: | 0.0 |
| ARG. OF PERIGEE: | 199.4305 | BSTAR: | 0.0040281 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

This is the 20th event of the Proton Block DM SOZ Ullage Motor class identified to date; it is the ninth associated with a GLONASS mission. This mission was conducted before the engineering defect was identified and passivation measures implemented. In this orbit, debris may be long-lived but hard to track. 76 debris objects were detected.

REFERENCE DOCUMENT

“Abandoned Proton Ullage Motors Continue to Create Debris”, The Orbital Debris Quarterly News, NASA JSC, April 1999. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv4i2.pdf>.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: CZ-4A Final Stage
OWNER: PRC
LAUNCH DATE: 3.04 Sep 1990
DRY MASS (KG): 1000
MAIN BODY: Cylinder-Nozzle; 2.9 m diameter by ~5 m length
MAJOR APPENDAGES: none
ATTITUDE CONTROL: none
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|------------|-----------------|----------------|
| DATE: | 4 Oct 1990 | LOCATION: | 81S, 68E (asc) |
| TIME: | 2014 GMT | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | 895 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|---------------|------------------------|-------------|
| EPOCH: | 90276.6451544 | MEAN ANOMALY: | 162.6773 |
| RIGHT ASCENSION: | 310.6975 | MEAN MOTION: | 14.01192890 |
| INCLINATION: | 98.9340 | MEAN MOTION DOT/2: | .000003118 |
| ECCENTRICITY: | .0010179 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 197.4122 | BSTAR: | .0002183343 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 5.8 min
MAXIMUM ΔI : 0.1 deg

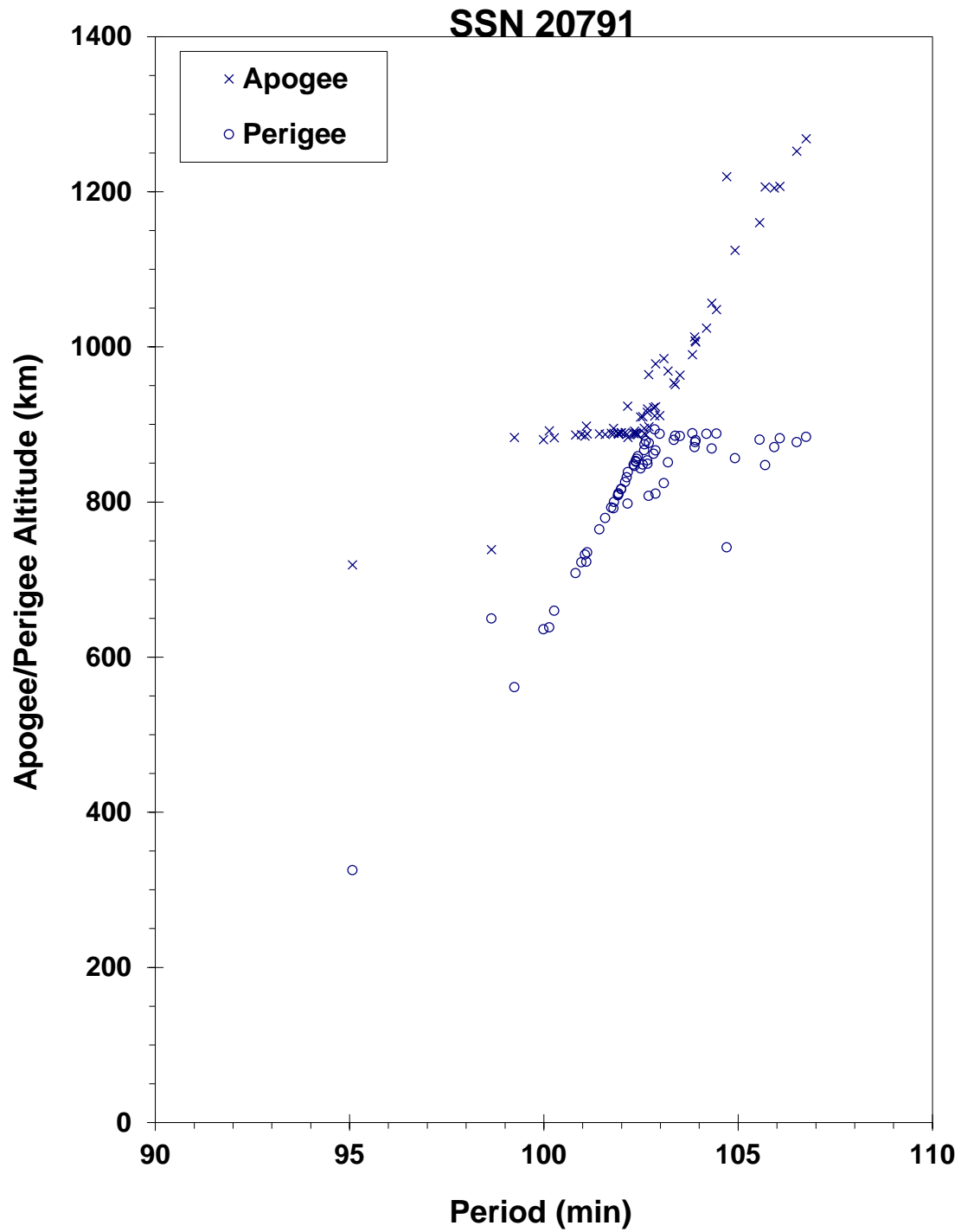
COMMENTS

This second flight of the CZ-4 final stage successfully deployed three payloads (one weather satellite and two inflated balloons) into a sun-synchronous orbit. Propellants used were N₂O₄ and UDMH. An estimated 70-75 fragments were detected soon after the event.

REFERENCE DOCUMENTS

The Fragmentation of Fengyun 1-2 R/B, N. L. Johnson, Technical Report CS90-TR-JSC-013, Teledyne Brown Engineering, Colorado Springs, Colorado, November 1990.

“Analyzing the Cause of LM-4 (A)’s Upper Stage’s Disintegration and the Countermeasures”, W. X. Zhang and S. Y. Liao, 5th International Conference of Pacific Basin Societies, 6-9 Jun 1993, Shanghai.



Fengyun 1-2 R/B debris cloud remnant of 65 objects 5 days after the event as reconstructed from Naval Space Surveillance System database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 1.46 Oct 1990
DRY MASS (KG): 6000
MAIN BODY: Cylinder; 2.4 m diameter by 7 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

DATE: 30 Nov 1990
TIME: 1720 GMT
ALTITUDE: 210 km
LOCATION: 54N, 157E (dsc)
ASSESSED CAUSE: Deliberate

PRE-EVENT ELEMENTS

EPOCH: 90334.45391019
RIGHT ASCENSION: 347.9431
INCLINATION: 64.7547
ECCENTRICITY: .0065418
ARG. OF PERIGEE: 155.2258
MEAN ANOMALY: 205.3252
MEAN MOTION: 16.12811753
MEAN MOTION DOT/2: .00671617
MEAN MOTION DOT DOT/6: .000035339
BSTAR: .00040815

DEBRIS CLOUD DATA

MAXIMUM ΔP : >7.3 min*
MAXIMUM ΔI : 0.3 deg*

*Based on uncataloged debris data

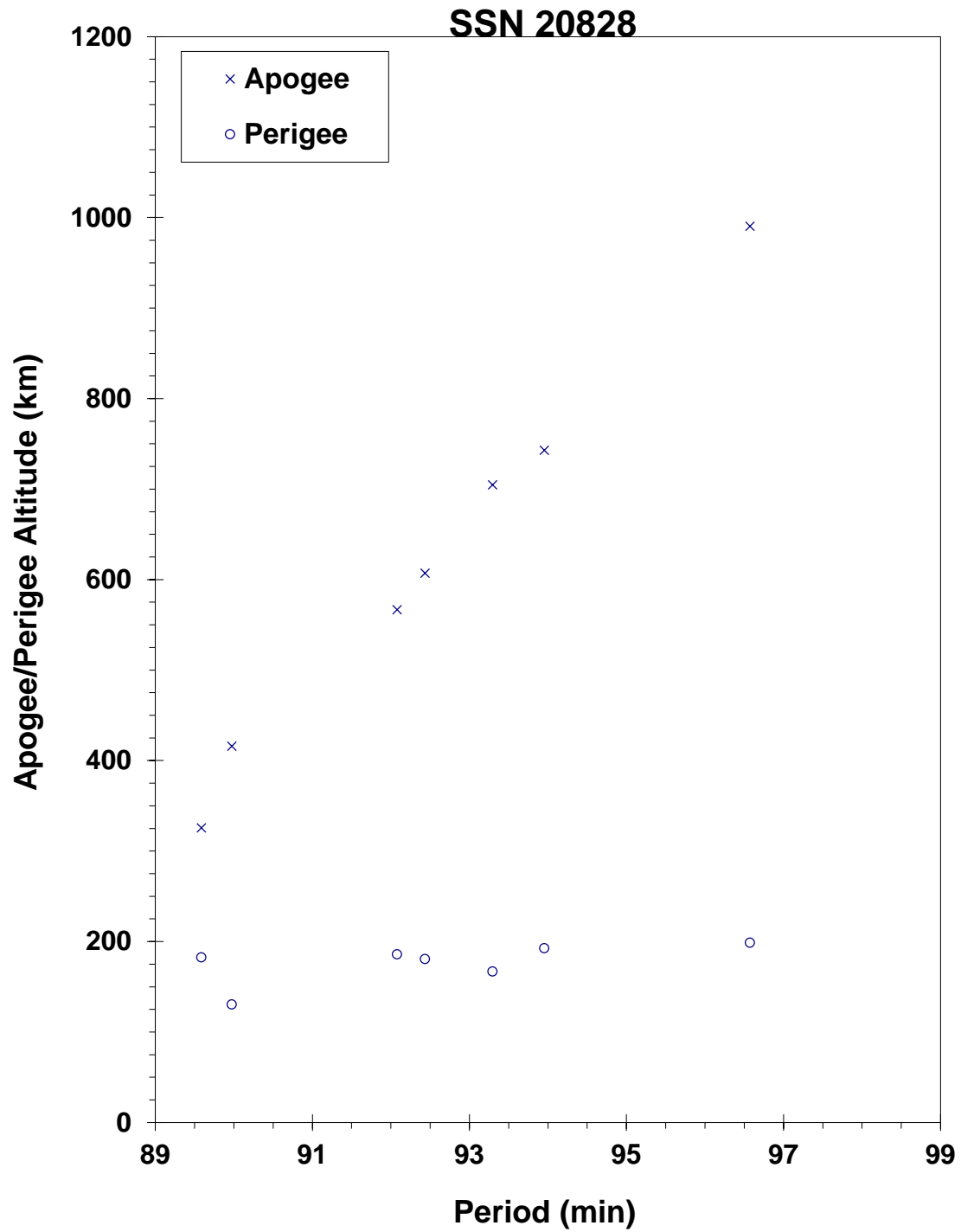
COMMENTS

Spacecraft was destroyed with a planned detonation. Second fragmentation of the Cosmos 2031 subclass. Early elements on only 7 objects available. Most debris reentered before being officially cataloged.

REFERENCE DOCUMENTS

The Fragmentation of Kosmos 2101, N. L. Johnson, Technical Report CS91-TR-JSC-002, Teledyne Brown Engineering, Colorado Springs, Colorado, January 1991.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 2101 debris cloud remnant of 7 objects 3 days after the event was reconstructed from Naval Space Surveillance System database. This diagram is taken from the cited reference.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 23 Nov 1990
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|---------------------|-----------------|------------|
| DATE: | Approx. 14 Dec 1995 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 95348.79476448 | MEAN ANOMALY: | 300.3633 |
| RIGHT ASCENSION: | 140.3319 | MEAN MOTION: | 5.84898259 |
| INCLINATION: | 46.4887 | MEAN MOTION DOT/2: | 0.00111293 |
| ECCENTRICITY: | 0.4967539 | MEAN MOTION DOT DOT/6: | 0.00000006 |
| ARG. OF PERIGEE: | 117.7610 | BSTAR: | 0.00074791 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

Parent satellite was one of two small engine units that are routinely released after the first burn of the Proton fourth stage. The nature of these objects was identified by Dr. Boris V. Chernlatiev, Deputy Constructor for the Energiya NPO, in October 1992. The cause of this fragmentation is assumed to be related to the residual hypergolic propellants on board and failure of the membrane separating the fuel and oxidizer. NAVSPASUR observed 69 objects that were associated with this breakup.

REFERENCE DOCUMENTS

The Fragmentation of Proton Debris, D. J. Nauer, TBE Technical Report CS93-LKD-004, Teledyne Brown Engineering, Colorado Springs, 31 December 1992.

Analysis of Fragmentations From December 1992 - February 1993, TBE Technical Report CS93-LKD-010, Teledyne Brown Engineering, Colorado Springs, 30 March 1993.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

"Identification and Resolution of an Orbital Debris Problem with the Proton Launch Vehicle", B. V. Cherniatiev et al, Proceedings of the First European Conference on Space Debris, April 1993.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 1.66 Dec 1990
DRY MASS (KG): 767
MAIN BODY: Cylinder; 1.0 m diameter by 3.6 m length
MAJOR APPENDAGES: 1 solar panel
ATTITUDE CONTROL: Active, 3 axis
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|------------|-----------------|----------------|
| DATE: | 1 Dec 1990 | LOCATION: | 6N, 232E (dsc) |
| TIME: | 1610 GMT | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | 850 km | | |

POST-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|--------------|
| EPOCH: | 90335.71008487 | MEAN ANOMALY: | 0.9090 |
| RIGHT ASCENSION: | 4.0350 | MEAN MOTION: | 14.29892145 |
| INCLINATION: | 98.8600 | MEAN MOTION DOT/2: | -0.00000049 |
| ECCENTRICITY: | 0.0080986 | MEAN MOTION DOT DOT/6: | 0.0 |
| ARG. OF PERIGEE: | 359.1948 | BSTAR: | -0.000010171 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : >2.0 min*
MAXIMUM ΔI : 1.0 deg*

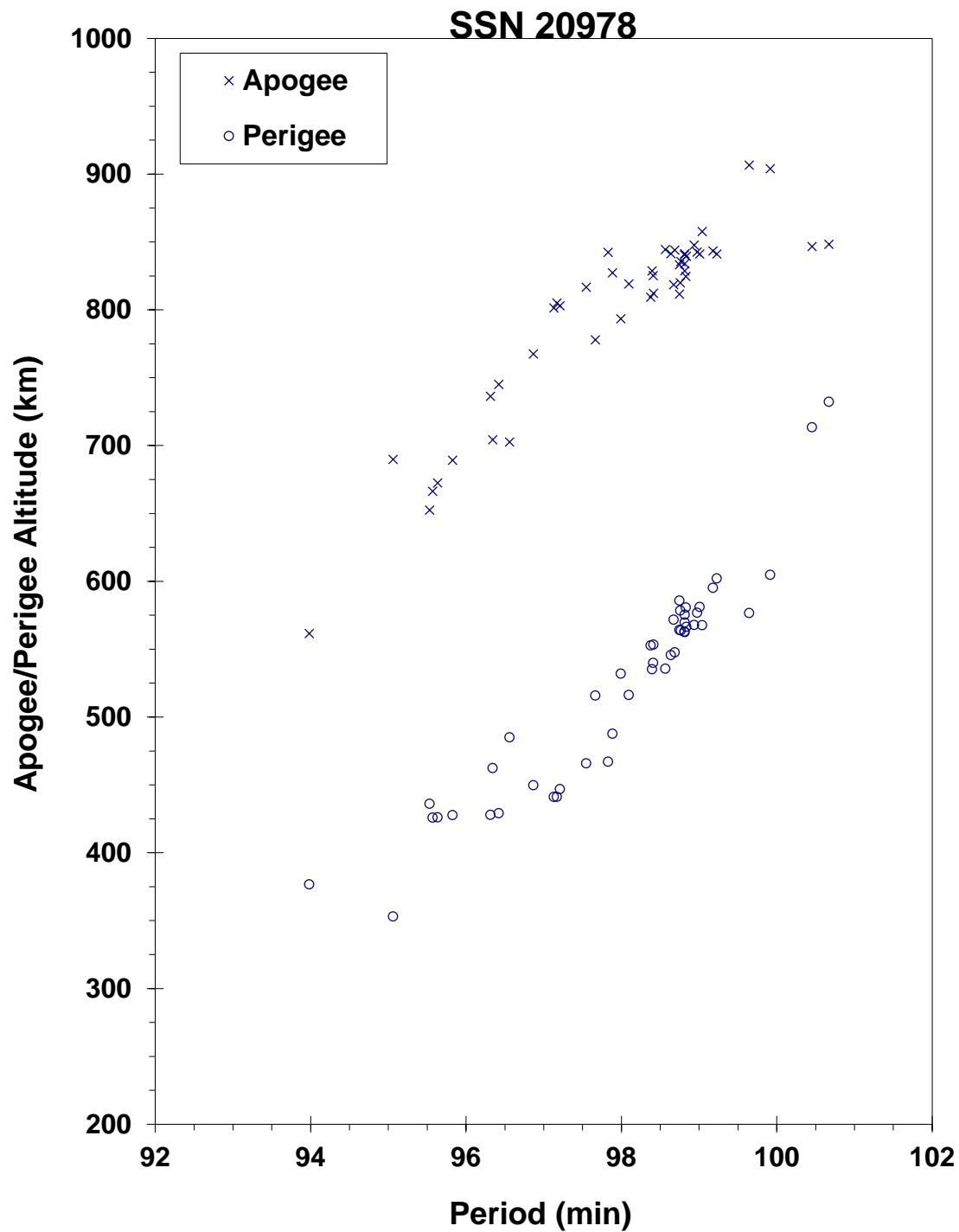
*Based on uncataloged debris data

COMMENTS

During the burn of USA 68's solid-fuel apogee kick motor (STAR-37S, TE-M-364-15), the 20 kg nozzle came apart, terminating thrust. At shutdown USA 68 was in an orbit of 610 km by 850 km. Immediately, a hydrazine orbit make-up system was activated, providing an additional 32.3 m/s DV. More than 40 pieces of non-Mission related debris were observed within a day of the event. The observed debris may include components of the USA 68 sun shield and AKM nozzle shield (total mass 2 kg). Most debris decayed very rapidly. The payload remained operational.

REFERENCE DOCUMENT

The Fragmentation of USA 68, N.L. Johnson, Technical Report CS91-TR-JSC-005, Teledyne Brown Engineering, Colorado Springs, Colorado, March 1991.



USA 68 debris cloud remnant of 48 fragments 12 days after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 8.11 Dec 1990
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1.0 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

DATE: 21 Feb 2003 LOCATION: 34.11 S, 208.61 E
TIME: ~0300Z ASSESSED CAUSE: Propulsion
ALTITUDE: ~17650 km

PRE-EVENT ELEMENTS

EPOCH: 03051.93857279 MEAN ANOMALY: 300.1330
RIGHT ASCENSION: 5.3297 MEAN MOTION: 4.24855437
INCLINATION: 65.3642 MEAN MOTION DOT/2: -0.00000082
ECCENTRICITY: 0.5638383 MEAN MOTION DOT DOT/6: 0.0
ARG. OF PERIGEE: 126.1785 BSTAR: -0.00014197

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

No debris was cataloged for this breakup. This is the 26th event of the Proton Block DM SOZ ullage motor since 1984. This ullage motor was launched prior to Russian recognition of the problem and before implementation of preventive measures.

REFERENCE DOCUMENT

“Satellite Fragmentations in 2003”, The Orbital Debris Quarterly News, NASA JSC, January 2004. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv8i1.pdf>.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 8.11 Dec 1990
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1.0 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 14 Mar 1998 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 98072.07217599 | MEAN ANOMALY: | 85.4178 |
| RIGHT ASCENSION: | 306.4512 | MEAN MOTION: | 4.23530449 |
| INCLINATION: | 65.0803 | MEAN MOTION DOT/2: | 0.00000895 |
| ECCENTRICITY: | 0.5724061 | MEAN MOTION DOT DOT/6: | 0 |
| ARG. OF PERIGEE: | 216.7168 | BSTAR: | 0.0025728 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

This is the 16th event of this class identified to date; it is the fifth associated with a GLONASS mission. More than 110 debris detected, but element sets developed for only a few.

REFERENCE DOCUMENTS

"Identification and Resolution of an Orbital Debris Problem with the Proton Launch Vehicle", Cherniatiev, Chernyavskiy, Johnson, and McKnight, First European Conference on Space Debris, 5-7 April 1993.

"The Fragmentation of Proton Debris", Nauer, Teledyne Brown Engineering Technical Report CS93-LKD-004, 31 December 1992.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

"Three Upper Stage Breakups in One Week Top February Debris Activity", The Orbital Debris Quarterly News, NASA JSC, April 1998, p. 1-2. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv3i2.pdf>.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Ariane 4 H-10 Third Stage
OWNER: France
LAUNCH DATE: 15.97 Jan 1991
DRY MASS (KG): 1760
MAIN BODY: Cylinder; 2.6 m diameter by 11.4 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|-------------------------|-----------------|------------|
| DATE: | Late Apr-early May 1996 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 96106.15481796 | MEAN ANOMALY: | 312.6005 |
| RIGHT ASCENSION: | 104.8696 | MEAN MOTION: | 2.66496263 |
| INCLINATION: | 6.7146 | MEAN MOTION DOT/2: | 0.00007071 |
| ECCENTRICITY: | 0.6989841 | MEAN MOTION DOT DOT/6: | 0 |
| ARG. OF PERIGEE: | 132.7372 | BSTAR: | 0.0012265 |

DEBRIS CLOUD DATA

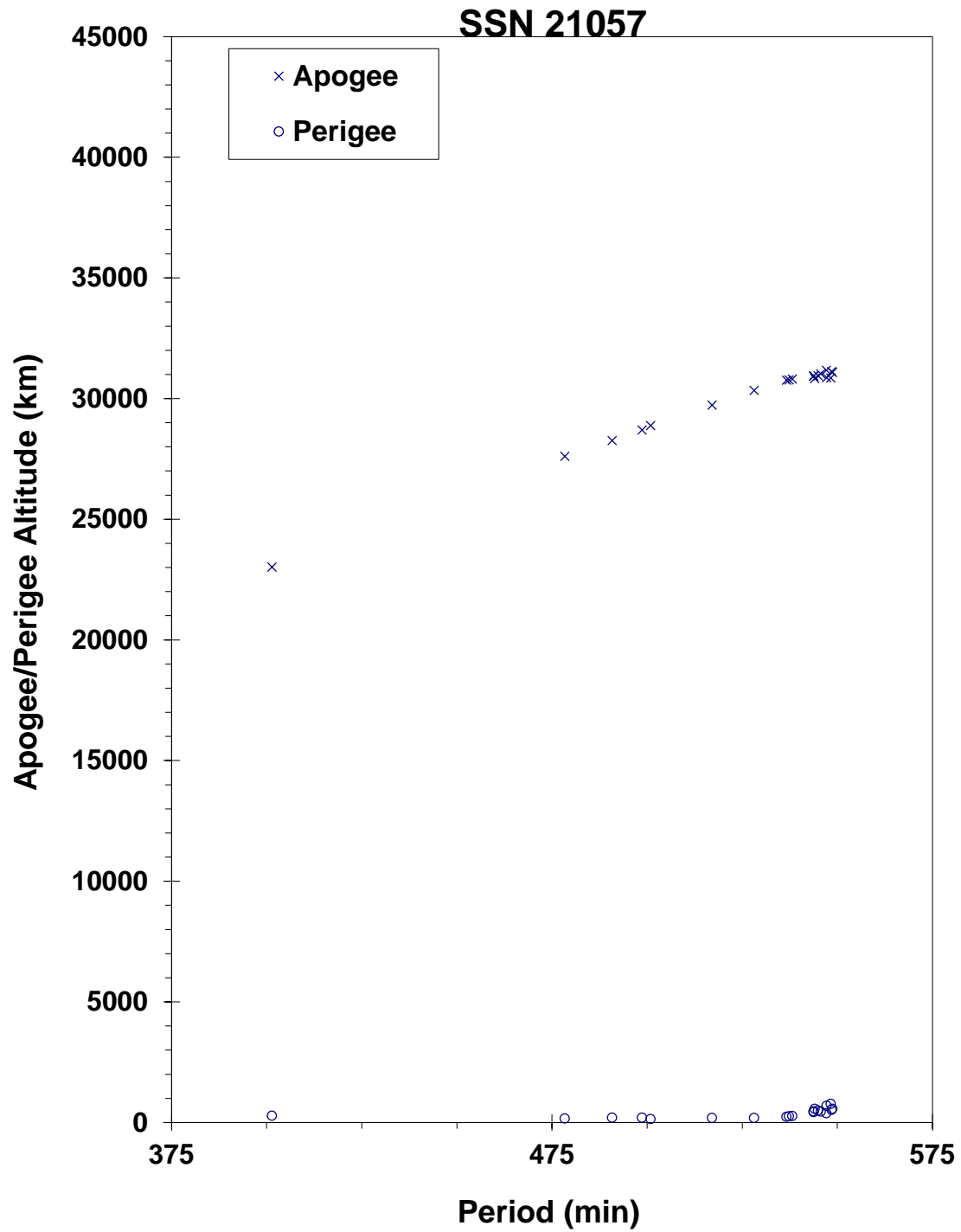
MAXIMUM ΔP : 147.3 min
MAXIMUM ΔI : 1.3 deg

COMMENTS

The event was first recognized by Naval Space Command analysts in early May 1996. Element sets for as many as 20 debris were developed. Since deliberate passivation of Ariane GTO stages was not implemented until 1993, the vehicle was not purged of its residual propellants or pressurants.

REFERENCE DOCUMENT

"Newly Recognized 1996 Breakup", N. L. Johnson, The Orbital Debris Quarterly News, April 1997, p. 2. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNV2i2.pdf>.



Italsat 1/Eutelsat 2 F2 R/B debris cloud of 20 fragments 1 year after
the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Cosmos Second Stage
OWNER: CIS
LAUNCH DATE: 12.12 Feb 1991
DRY MASS (KG): 1435
MAIN BODY: Cylinder; 2.4 m diameter by 6.6 m length
MAJOR APPENDAGES: Payload deployment mechanism
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: Unknown

EVENT DATA

DATE: 5 Mar 1991 LOCATION: 43S, 140E (asc)
TIME: 1345 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 1560 km

PRE-EVENT ELEMENTS

EPOCH: 91062.94236834 MEAN ANOMALY: 112.8991
RIGHT ASCENSION: 166.0317 MEAN MOTION: 12.19552620
INCLINATION: 74.0386 MEAN MOTION DOT/2: 0.00000005
ECCENTRICITY: 0.0166507 MEAN MOTION DOT DOT/6: 0.0
ARG. OF PERIGEE: 245.0348 BSTAR: 0.000099999

DEBRIS CLOUD DATA

MAXIMUM ΔP : 4.3 min*
MAXIMUM ΔI : 0.3 deg*

*Based on uncataloged debris data

COMMENTS

This is the second known fragmentation of the Cosmos second stage and the first in more than 25 years and 370 missions. Like the earlier event (Cosmos 61-63 R/B), this rocket body successfully completed its multiple payload delivery before breakup. NAVSPASUR determined that several minor separations occurred both prior to and after the main breakup cited above (see NAVSPASUR report referenced below).

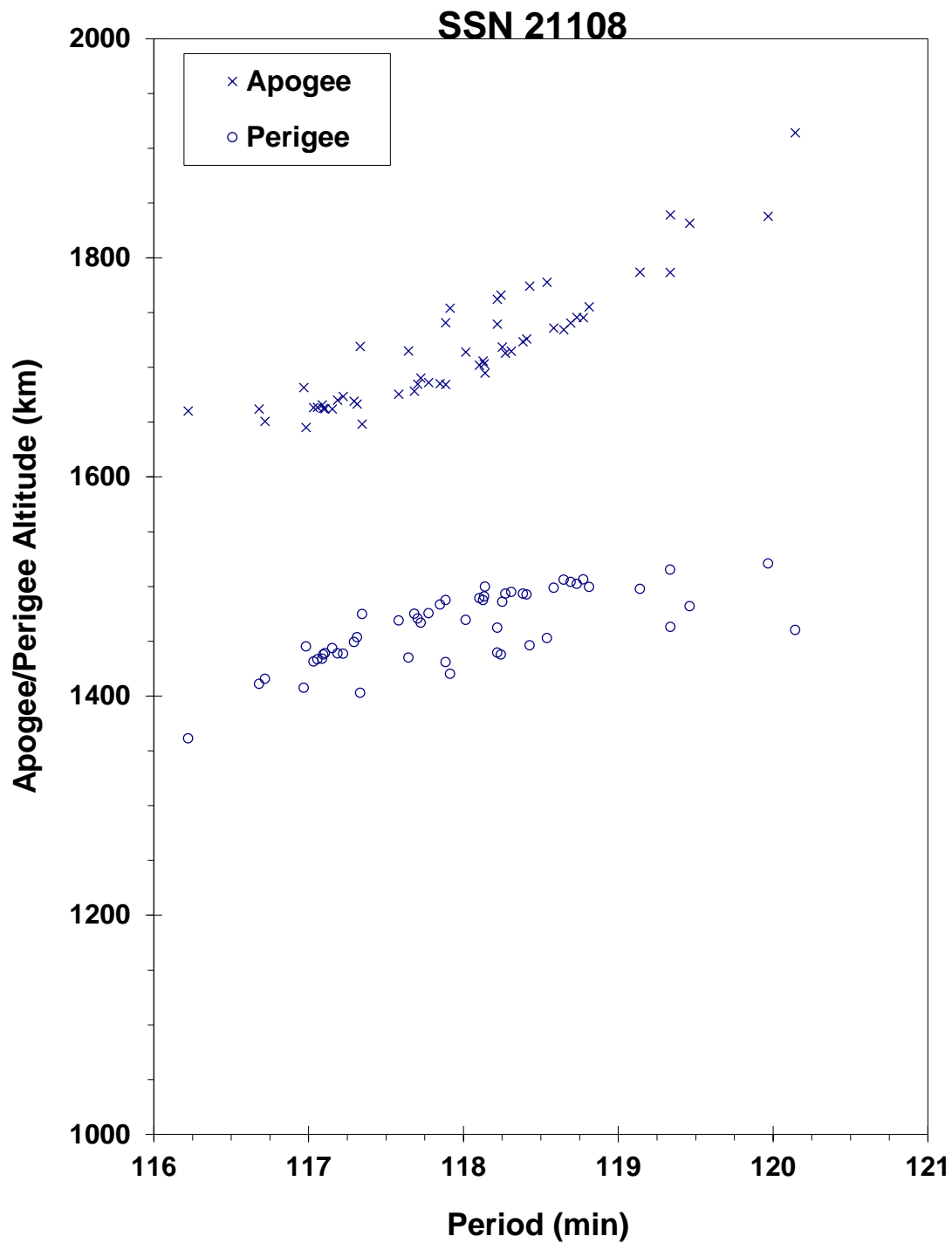
REFERENCE DOCUMENTS

Cosmos 2125-2132 Rocket Body (U), Fragmentation and Breakup Report (U), E.L. Jenkins and R.E. Farmer, Naval Space Surveillance Center, Dahlgren, Virginia, April, 1991.

A Preliminary Analysis of the Fragmentations of the Kosmos 2125-2132 Rocket Body, N.L. Johnson, Technical Report CS91-TR-JSC-007, Teledyne Brown Engineering, Colorado Springs, Colorado, April 1991.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

“The Recent Fragmentations of LEO Upper Stages”, G. Chernyavskiy et al, 45th IAF Congress, 1994.



Cosmos 2125-32 R/B debris cloud of 54 objects 5 days after the major breakup event as reconstructed from Naval Space Surveillance System database. This diagram is taken from the reference cited at the top of this page.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 12 Feb 1991
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Unknown
ENERGY SOURCES: On-board propellants

EVENT DATA

DATE: 7 May 1994
TIME: 0930 GMT
ALTITUDE: 16195 km
LOCATION: 10N, 112E
ASSESSED CAUSE: Propulsion

PRE-EVENT ELEMENTS

EPOCH: 94126.31580012
RIGHT ASCENSION: 110.6447
INCLINATION: 46.6223
ECCENTRICITY: 0.6204369
ARG. OF PERIGEE: 160.8637
MEAN ANOMALY: 240.6661
MEAN MOTION: 3.78477656
MEAN MOTION DOT/2: 0.000127656
MEAN MOTION DOT DOT/6: 0.0
BSTAR: 0.00086951

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

Parent satellite was one of two small engine units that are routinely released after the first burn of the Proton fourth stage. The nature of these objects was identified by Dr. Boris V. Chernlatiev, Deputy Constructor for the Energiya NPO, in October 1992. The cause of this fragmentation appears to be related to the residual hypergolic propellants on board and failure of the membrane separating the fuel and oxidizer. NAVSPASUR observed 38 objects that were associated with this breakup. Only 6 element sets were generated. This was the ninth in a series of fragmentations of this object type, and was the fourth located in a geosynchronous transfer orbit. Two possible fragmentation locations were calculated by the NAVSPOC. The numbers above represent the first possible calculated location.

REFERENCE DOCUMENTS

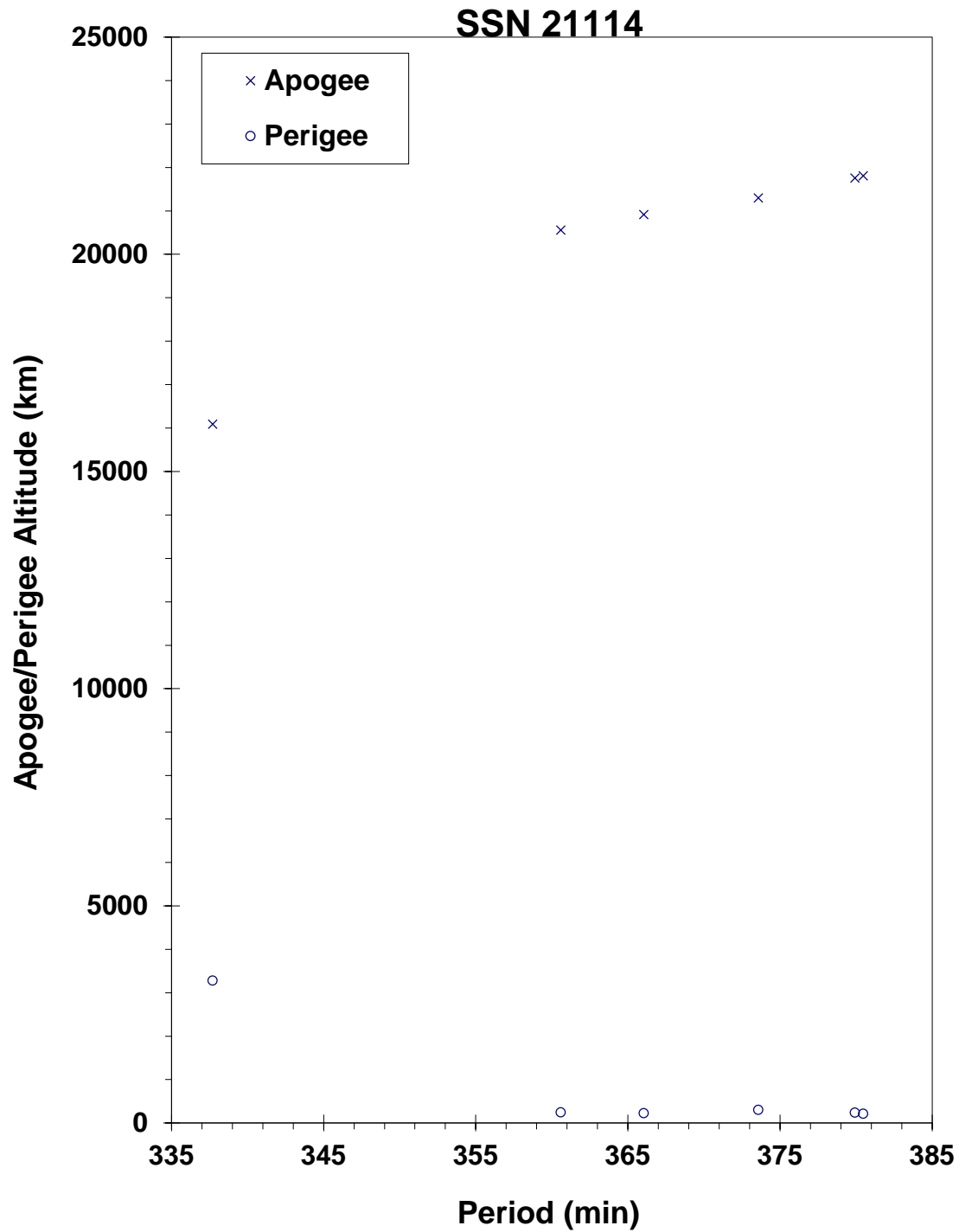
The Fragmentation of Proton Debris, D. J. Nauer, TBE Technical Report CS93-LKD-004, Teledyne Brown Engineering, Colorado Springs, 31 December 1992.

Analysis of Fragmentations From December 1992 - February 1993, TBE Technical Report CS93-LKD-010, Teledyne Brown Engineering, Colorado Springs, 30 March 1993.

The Fragmentation of Cosmos 2133 Debris, I. W. Grissom and D. J. Nauer, TBE Technical Report CS94-LKD-016, Teledyne Brown Engineering, Colorado Springs, 30 June 1994.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

"Identification and Resolution of an Orbital Debris Problem with the Proton Launch Vehicle", B. V. Cherniatiev et al, Proceedings of the First European Conference on Space Debris, April 1993.



Gabbard diagram of six objects from the Cosmos 2133 debris fragmentation.

SATELLITE DATA

TYPE: Ariane 4 H10 Third Stage
OWNER: France
LAUNCH DATE: 2.98 Mar 1991
DRY MASS (KG): 1760
MAIN BODY: Cylinder; 2.6 m diameter by 10 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: On-board propellants

EVENT DATA

DATE: 27 Apr 1994 LOCATION: 0.5S, 79E (dsc)
TIME: 0144 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 270 km

PRE-EVENT ELEMENTS

EPOCH: 94116.17965845 MEAN ANOMALY: 157.2349
RIGHT ASCENSION: 136.1778 MEAN MOTION: 2.86975555
INCLINATION: 6.5808 MEAN MOTION DOT/2: 0.00006058
ECCENTRICITY: 0.6829164 MEAN MOTION DOT DOT/6: 0.0
ARG. OF PERIGEE: 185.9406 BSTAR: 0.001267

DEBRIS CLOUD DATA

MAXIMUM ΔP : 148 min
MAXIMUM ΔI : 1.1 deg

COMMENTS

The fragmentation of this R/B occurred over 37 months after launch. Since deliberate passivation of Ariane GTO stages was not implemented until 1993, the vehicle was not purged of its residual propellants or pressurants. As many as 30 debris fragments were detected.

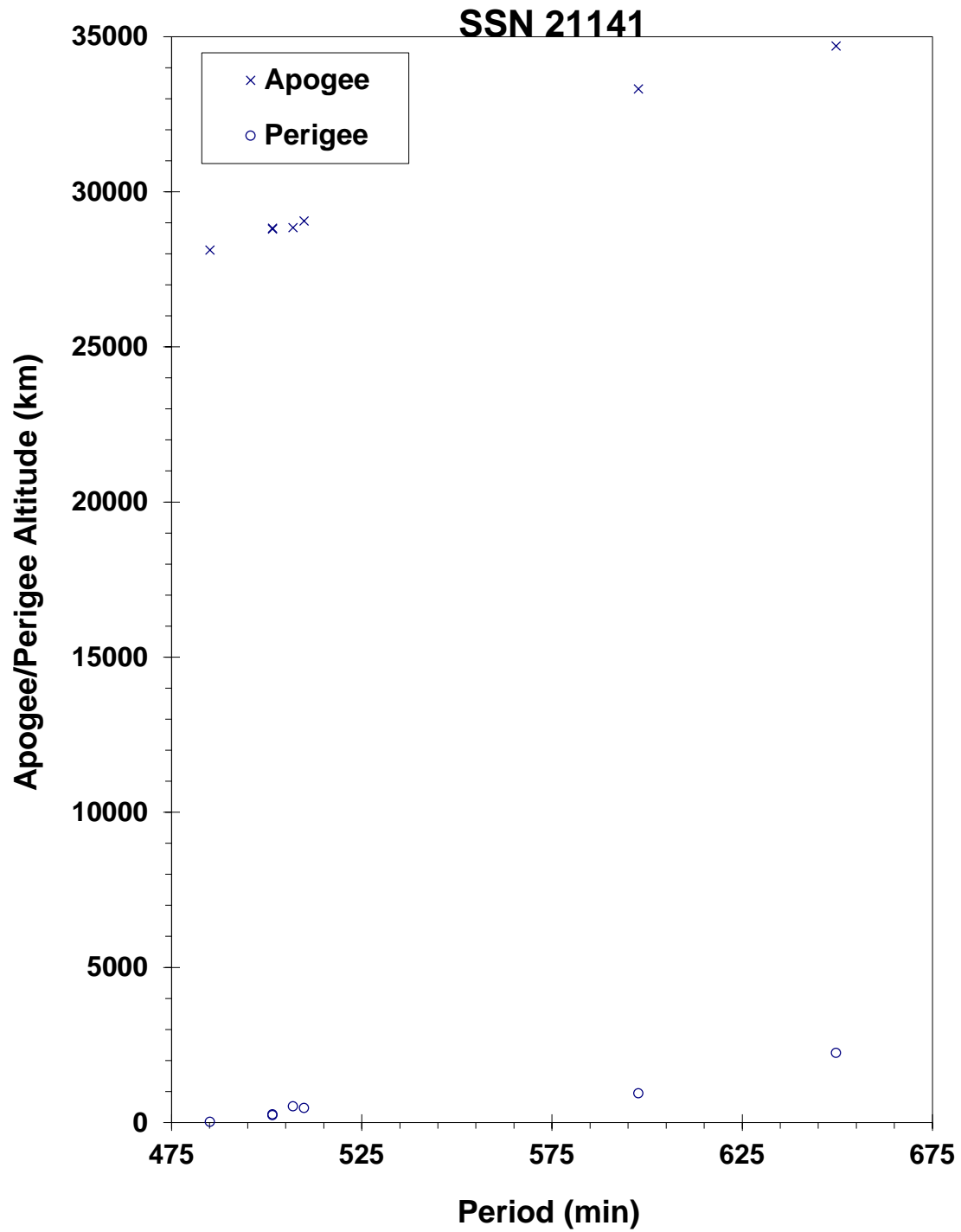
REFERENCE DOCUMENTS

TRW Space Log 1957-1991. TRW Space and Defense Sector, Space and Technology Group. Redondo Beach, CA. 1992.

Space News, "Ariane Rocket Flies After Electrical Glitch Delay, Volume 2, Number 8, 11-17 March 1991.

Rockets of the World. Peter Alway, Ann Arbor, MI, 1993.

The Fragmentation of the Astra 1B/MOP 2 (1) Rocket Body, I. W. Grissom and D. J. Nauer, TBE Technical Report CS94-LKD-014, Teledyne Brown Engineering, Colorado Springs, 15 May 1994.



Astra 1B/MOP 2 R/B debris cloud of 7 fragments as reconstructed
from the US SSN database.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 04.0 Apr 1991
DRY MASS (KG): 55
MAIN BODY: ellipsoid; 0.6 m diameter x 1 m length
MAJOR APPENDAGES: none
ATTITUDE CONTROL: none
ENERGY SOURCES: on-board propellants

EVENT DATA

DATE: 08 Mar 2009 LOCATION: 35.5S, 21.8E (dsc)
TIME: 1000 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 18,544.0 km

PRE-EVENT ELEMENTS

EPOCH: 09066.60605701 MEAN ANOMALY: 350.1859
RIGHT ASCENSION: 138.9363 MEAN MOTION: 4.33919111
INCLINATION: 64.9463 MEAN MOTION DOT/2: 0.00000158
ECCENTRICITY: 0.5689328 MEAN MOTION DOT DOT/6: 0.0
ARG. OF PERIGEE: 41.0118 BSTAR: 0.00042936

DEBRIS CLOUD DATA

MAXIMUM ΔP : 200.0 min
MAXIMUM ΔI : 19.3 deg

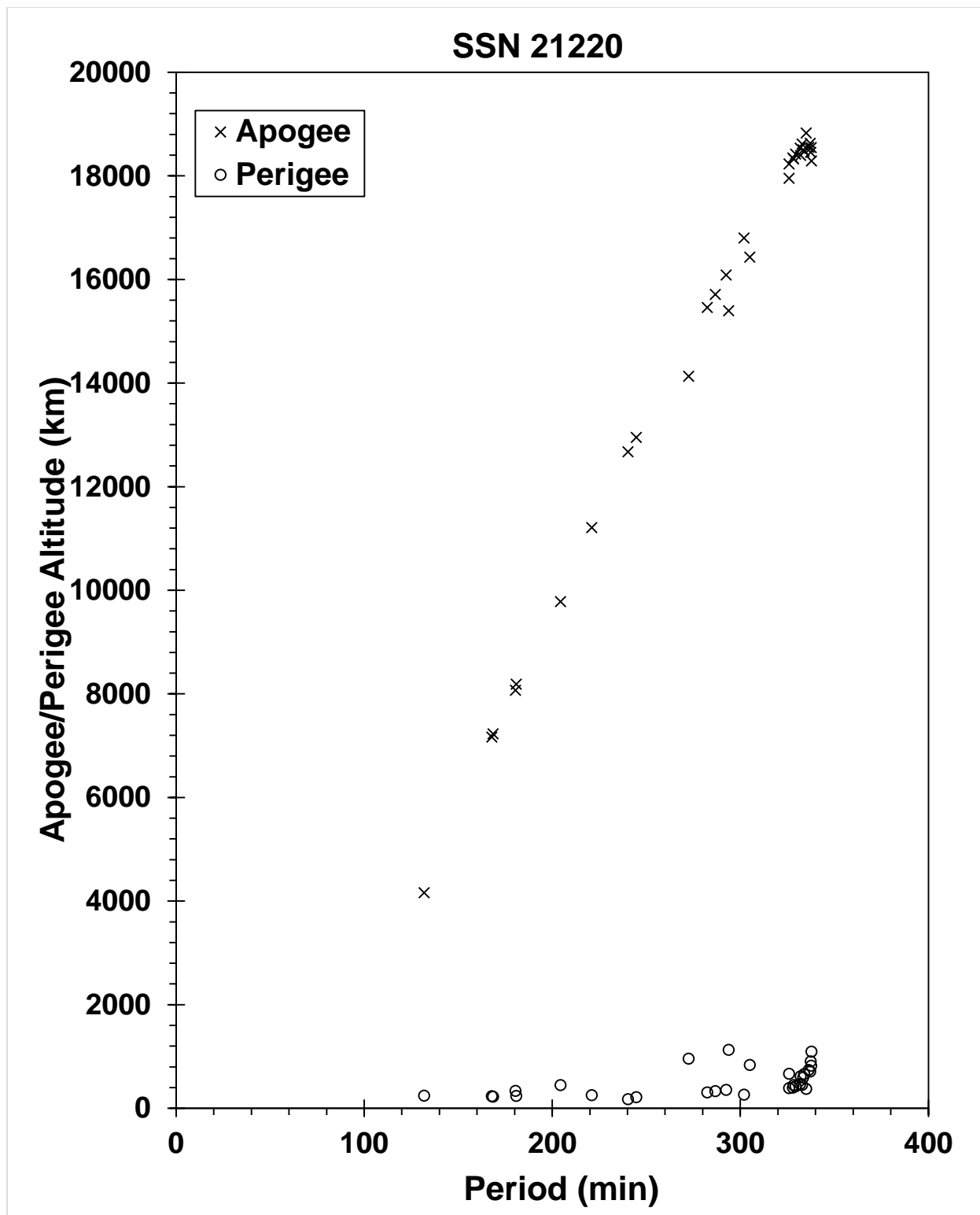
COMMENTS

This is the 37th known breakup of a Proton-DM SOZ ullage unit since 1984. The breakups are believed to be caused by residual propellants. Russian officials have been aware of the problem since 1992 and have made design changes, although the date of full implementation is unknown. Due to the difficulty in observing and cataloging debris in deep-space, highly elliptical orbits, a debris ensemble much larger than that cataloged may have been created by this event.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations: A Joint U.S.-Russian Investigation. N.L. Johnson et al. Kaman Sciences Corporation, October 2005.

“Minor March Satellite Break-Up”, The Orbital Debris Quarterly News, NASA JSC, April 2009.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/ODQNv13i2.pdf>.



The COSMOS ullage motor debris cloud, including 32 fragments cataloged up to 3 months after the event.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 4.45 Apr 1991
DRY MASS (KG): ~55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1.0 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 16 Jun 2001 | LOCATION: | Unknown |
| TIME: | ~0700Z | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 01165.32627059 | MEAN ANOMALY: | 158.6980 |
| RIGHT ASCENSION: | 118.5521 | MEAN MOTION: | 4.28587592 |
| INCLINATION: | 64.5545 | MEAN MOTION DOT/2: | 0.00004370 |
| ECCENTRICITY: | 0.5826262 | MEAN MOTION DOT DOT/6: | 0.0 |
| ARG. OF PERIGEE: | 187.0212 | BSTAR: | 0.0011075 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 236.7 min*
MAXIMUM ΔI : 2.13 deg*

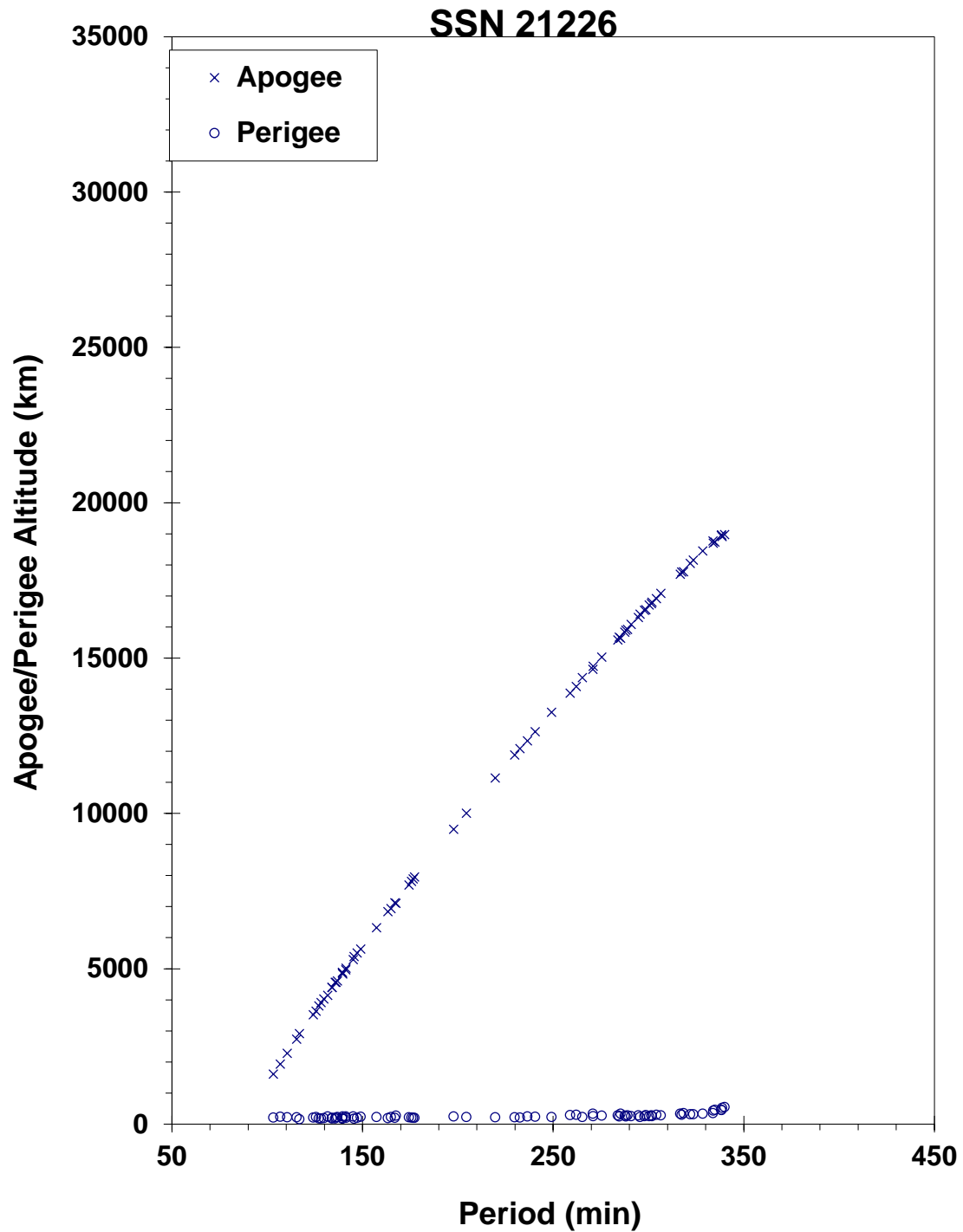
* Based on uncataloged debris data

COMMENTS

The debris from this breakup were difficult for the US Space Surveillance Network to track. Although over 100 debris were initially tracked, none were cataloged. This is the 24th event of the Proton Block DM SOZ ullage motor since 1984. This ullage motor was launched prior to Russian recognition of the problem and before implementation of preventive measures.

REFERENCE DOCUMENT

"Two More Satellite Breakups Detected", The Orbital Debris Quarterly News, NASA JSC, July 2001. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv6i3.pdf>.



Cosmos 2139-41 auxiliary motor debris cloud of 77 fragments within 2 weeks after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Tsyklon Third Stage
OWNER: CIS
LAUNCH DATE: 28.30 Sep 1991
DRY MASS (KG): 1360
MAIN BODY: Cone-cylinder; 2.1 m diameter by 2.4 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: Unknown

EVENT DATA

DATE: 09 Oct 1999
TIME: 1508 GMT
ALTITUDE: 1460 km
LOCATION: 41.5N, 217.9E (dsc)
ASSESSED CAUSE: Propulsion

PRE-EVENT ELEMENTS

EPOCH: 99281.98318497
RIGHT ASCENSION: 96.5043
INCLINATION: 82.5731
ECCENTRICITY: 0.0046780
ARG. OF PERIGEE: 140.1600
MEAN ANOMALY: 220.2415
MEAN MOTION: 12.54216420
MEAN MOTION DOT/2: 0.00000027
MEAN MOTION DOT DOT/6: 0.0
BSTAR: 0.00010000

DEBRIS CLOUD DATA

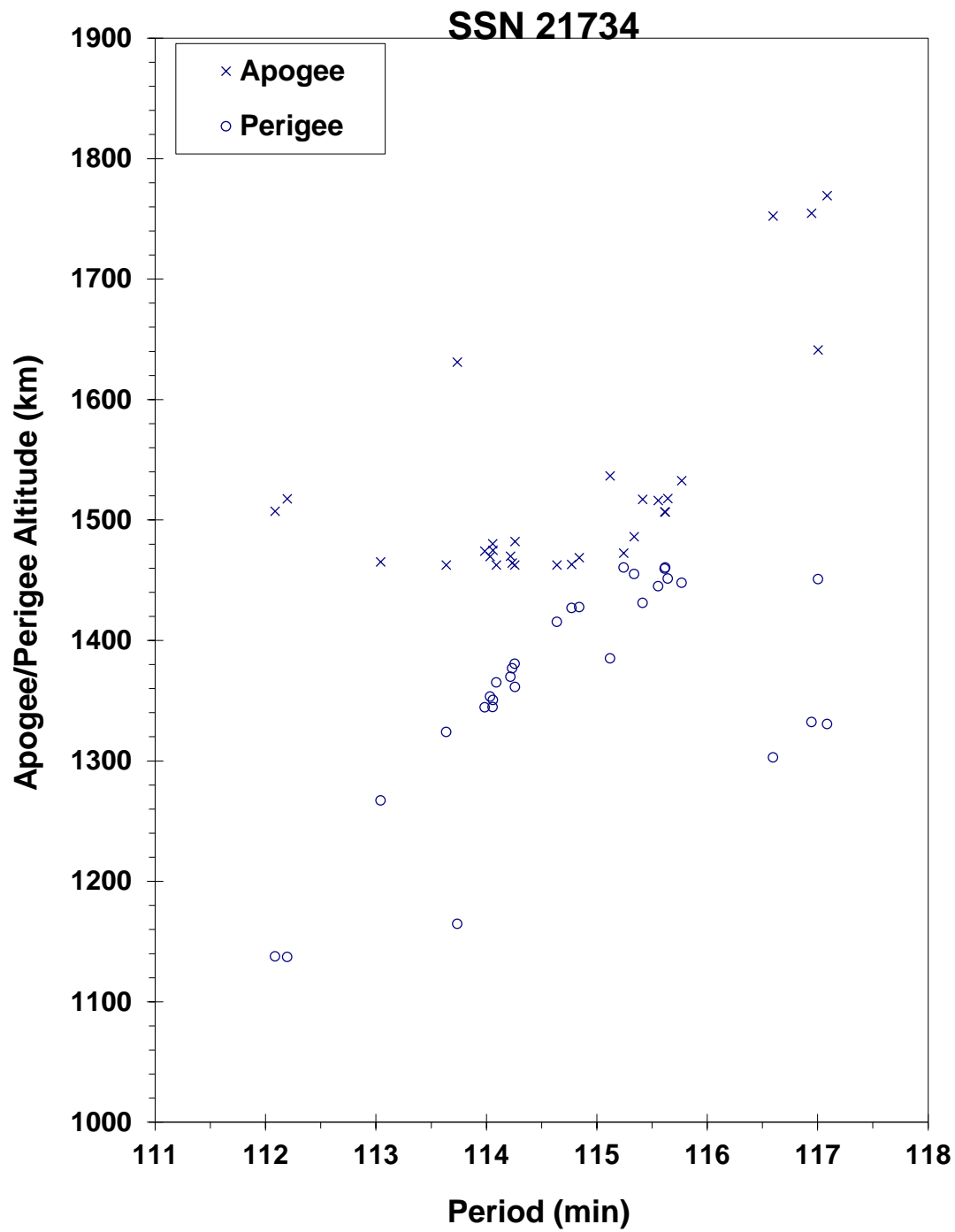
MAXIMUM ΔP : 2.716 min
MAXIMUM ΔI : 0.79 deg

COMMENTS

This is the 4th event of this class identified to date and the second of 1999. All stages have been about 8-10 years old at the time of breakup. The vehicle is a Ukrainian-produced stage, using UDMH and N₂O₄ as propellants. To date these stages have not been passivated at end of mission and may contain up to 300 kg of residual propellants. The issue of Tsyklon orbital stage breakups was discussed with representatives of the National Space Agency of Ukraine during 11-13 October 1999 in Darmstadt, Germany. More than 100 of these stages are currently in Earth orbit. Although the exact cause of these breakups remains unknown, all four events have occurred during periods of high solar flux, i.e., near solar maximum.

REFERENCE DOCUMENT

"Third Tsyklon Upper Stage Breaks Up", The Orbital Debris Quarterly News, NASA JSC, July 1999. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv4i3.pdf>.



Cosmos 2157-2162 rocket body debris cloud of 31 fragments within 1 day of the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 9.55 October 1991
DRY MASS (KG): 6000
MAIN BODY: Cylinder; 2.4 m diameter by 7 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

DATE: 6 December 1991 LOCATION: 55N, 154E (dsc)
TIME: 2021 GMT ASSESSED CAUSE: Deliberate
ALTITUDE: 210 km

PRE-EVENT ELEMENTS

EPOCH: 91340.51933896 MEAN ANOMALY: 213.3470
RIGHT ASCENSION: 37.7884 MEAN MOTION: 16.18797546
INCLINATION: 64.7678 MEAN MOTION DOT/2: 0.00862876
ECCENTRICITY: 0.0054670 MEAN MOTION DOT DOT/6: 0.000035685
ARG. OF PERIGEE: 147.5032 BSTAR: 0.00035926

DEBRIS CLOUD DATA

MAXIMUM ΔP : >9.8 min*
MAXIMUM ΔI : 0.2 deg*

*Based on uncataloged debris data

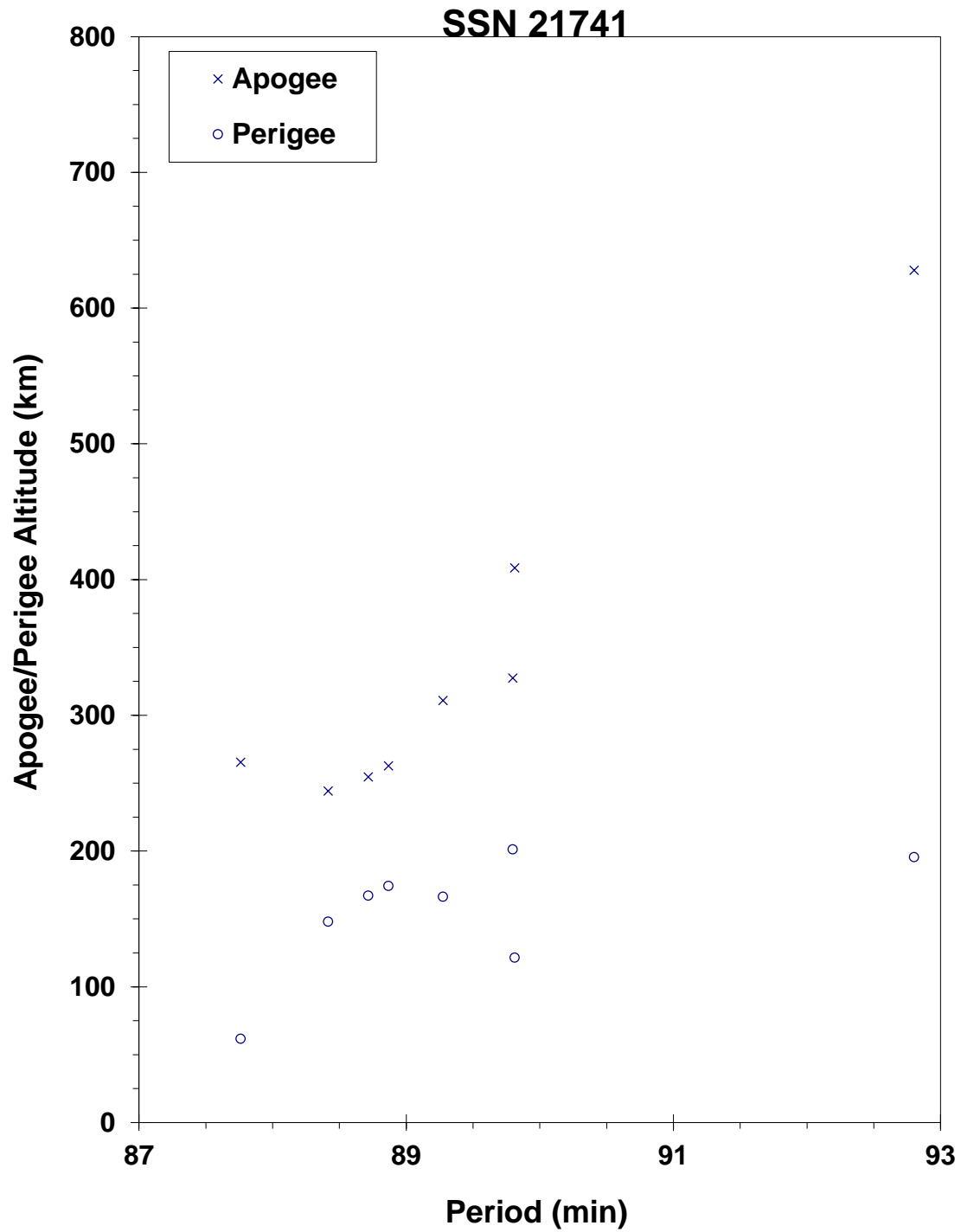
COMMENTS

Spacecraft was destroyed with a planned detonation. Third fragmentation of the Cosmos 2031 subclass. Early elements on only 8 objects available. All debris reentered before being officially cataloged.

REFERENCE DOCUMENTS

The Fragmentation of Kosmos 2163, Technical Report CS92-TR-JSC-002, Teledyne Brown Engineering, Colorado Springs, Colorado, January 1992.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 2163 debris cloud remnant of 8 objects 1 day after the event as reconstructed from the US SSN database. This diagram is taken from the cited reference.

SATELLITE DATA

TYPE: Ariane 4 Third Stage
OWNER: France
LAUNCH DATE: 29.96 Oct 1991
DRY MASS (KG): ~1760
MAIN BODY: Cylinder: 2.6 m diameter by 11.4 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

DATE: 24 Dec 2001 LOCATION: 4.0 N, 344.4 E (asc)
TIME: ~2212Z ASSESSED CAUSE: Propulsion
ALTITUDE: 22,100 km

PRE-EVENT ELEMENTS

EPOCH: 01358.15120659 MEAN ANOMALY: 1.1953
RIGHT ASCENSION: 264.6850 MEAN MOTION: 2.90501578
INCLINATION: 7.1968 MEAN MOTION DOT/2: 0.00042976
ECCENTRICITY: 0.6814056 MEAN MOTION DOT DOT/6: 0.0
ARG. OF PERIGEE: 351.9651 BSTAR: 0.0055981

DEBRIS CLOUD DATA

MAXIMUM ΔP: 4.45 min*
MAXIMUM ΔI: 0.08 deg*

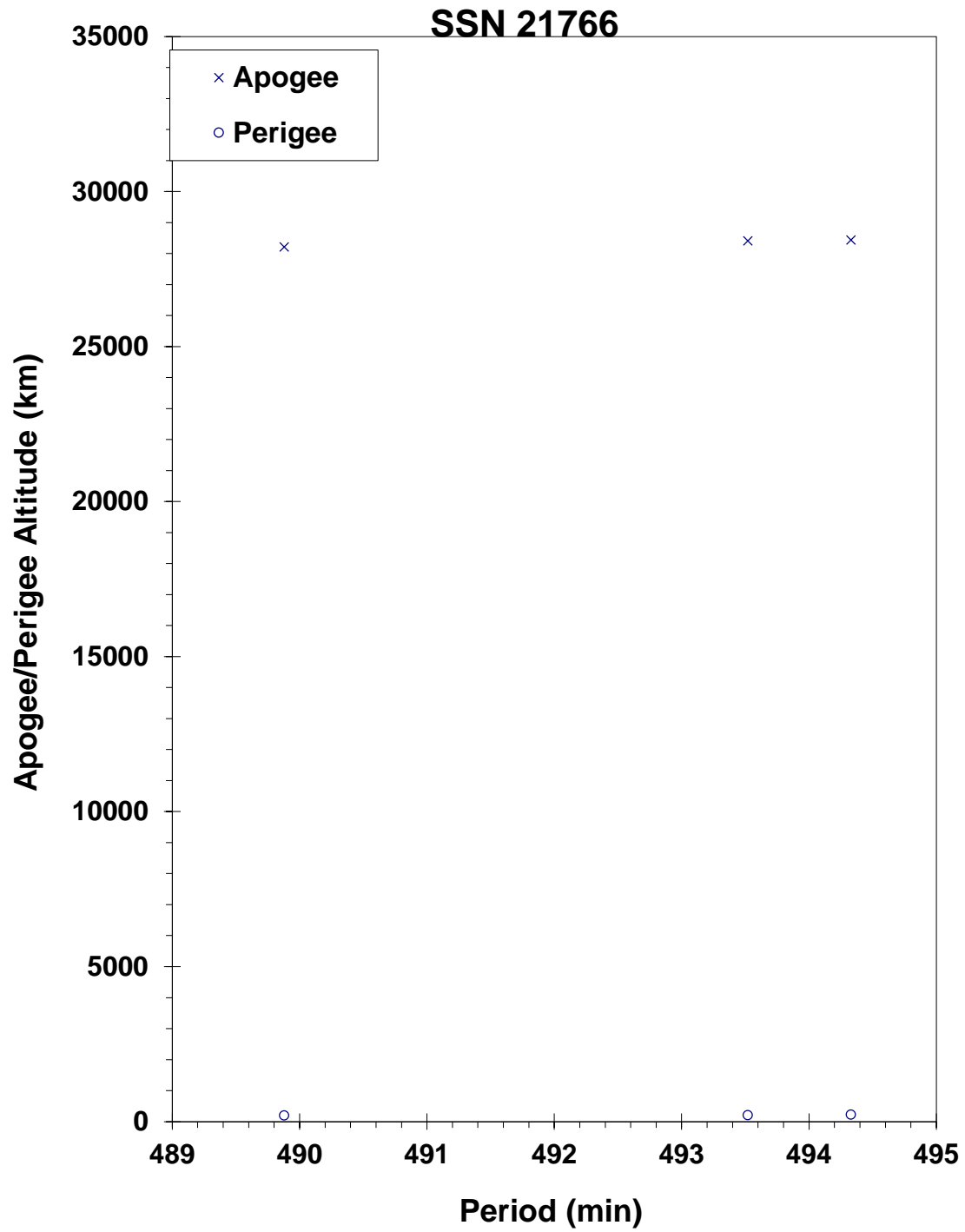
* Based on uncataloged debris data

COMMENTS

This 10-year-old Ariane 4 third stage appears to have suffered a minor fragmentation. The low inclination and high eccentricity of the orbit made debris detection and tracking difficult. Three pieces were initially detected by the SSN and ten objects cataloged a month after the event. The object was seen intact at about 2030 UTC, 24 December. Approximately 25 hours later a debris cloud of eight objects was seen by the same sensor. The perigee of the vehicle prior to breakup was sufficiently high that aerodynamic forces should not have been a factor in the event.

REFERENCE DOCUMENT

"Two Major Satellite Breakups Near End of 2001," The Orbital Debris Quarterly News, NASA JSC, January 2002. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv7i1.pdf>.



Intelsat 601 R/B debris cloud of 3 fragments 3 days after the breakup as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 28.56 Nov 1991
DRY MASS (KG): 767
MAIN BODY: Cylinder; 1.0 m diameter by 3.6 m length
MAJOR APPENDAGES: 1 solar panel
ATTITUDE CONTROL: None at time of event
ENERGY SOURCES: On-board propellant

EVENT DATA

DATE: 15 Apr 2004 LOCATION: 31N, 56E (asc)
TIME: 1454 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 835 km

POST-EVENT ELEMENTS

EPOCH: 04106.47330773 MEAN ANOMALY: 346.2855
RIGHT ASCENSION: 129.0059 MEAN MOTION: 14.15516644
INCLINATION: 98.6744 MEAN MOTION DOT/2: 0.00000244
ECCENTRICITY: .0012890 MEAN MOTION DOT DOT/6: 0.0
ARG. OF PERIGEE: 13.8671 BSTAR: 0.0014668

DEBRIS CLOUD DATA

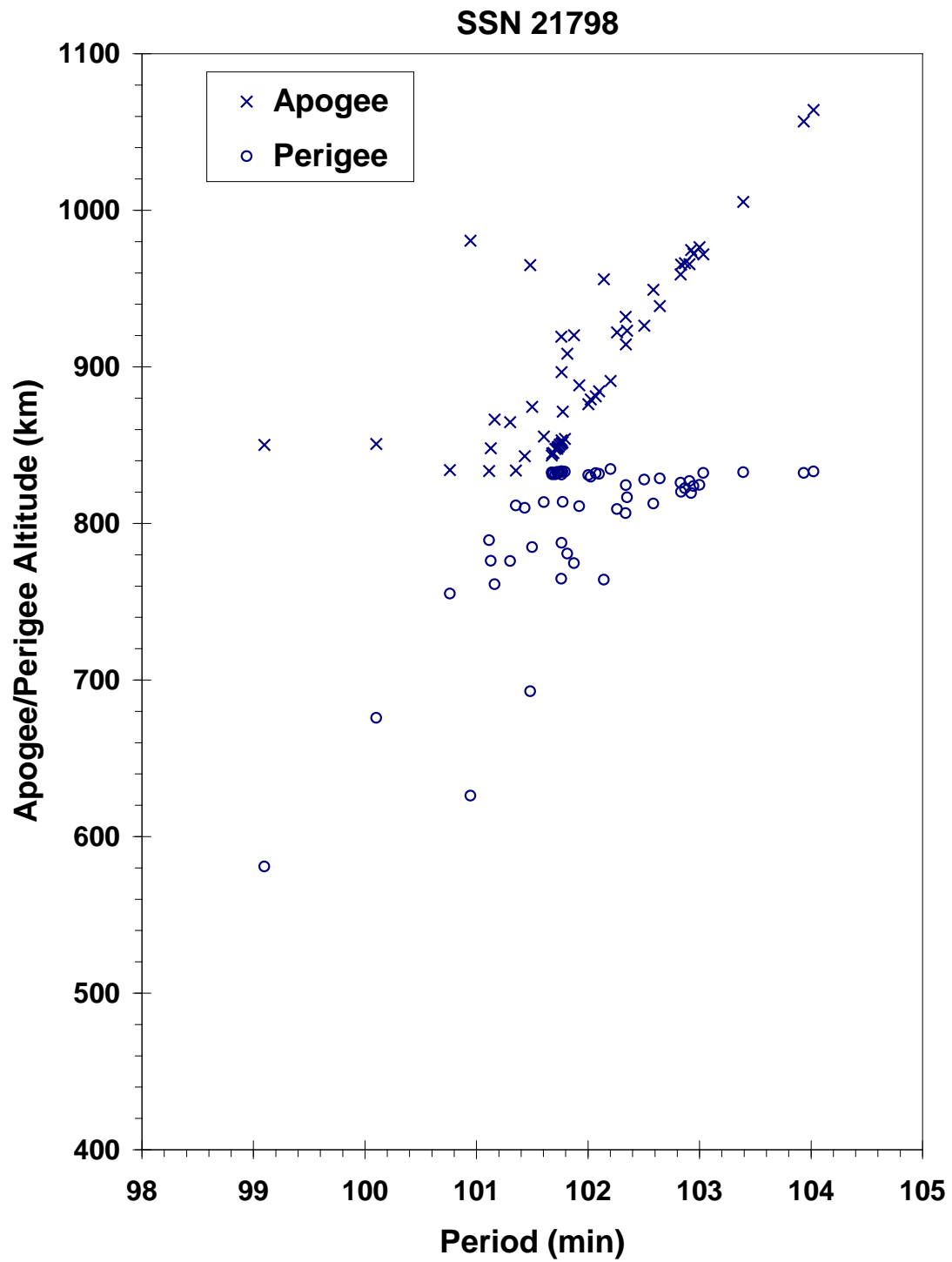
MAXIMUM ΔP : 2.8 min
MAXIMUM ΔI : 0.4 deg

COMMENTS

The spacecraft was non-operational at the time of the event. The electrical power generation system had been passivated by discharging the batteries and disconnecting them from the charging circuit. Virtually no nitrogen remained on board due to a leak detected early in the mission. The only energy source assessed to be on the spacecraft at the time of the event was approximately 6 kg of hydrazine.

REFERENCE DOCUMENT

"Recent Satellite Breakups", The Orbital Debris Quarterly News, NASA JSC, October 2004.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv8i4.pdf>.



USA 73 debris cloud of 56 cataloged fragments 6 weeks after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Ariane 4 H10⁺ Third Stage
OWNER: France
LAUNCH DATE: 15.98 Apr 1992
DRY MASS (KG): 1800
MAIN BODY: Cylinder; 2.6 m diameter by 10 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|-------------------|-----------------|------------|
| DATE: | 21 Apr 1993 (EST) | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 93110.33659871 | MEAN ANOMALY: | 335.5551 |
| RIGHT ASCENSION: | 224.3451 | MEAN MOTION: | 2.28914093 |
| INCLINATION: | 4.03 | MEAN MOTION DOT/2: | 0.000024 |
| ECCENTRICITY: | 0.7248434 | MEAN MOTION DOT DOT/6: | 0.0 |
| ARG. OF PERIGEE: | 110.6851 | BSTAR: | 0.0020699 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

Although analysis indicates that 92-021C fragmented around 21 April 1993, the event was not recognized until eight pieces were cataloged during the period from 30 August 1994 to 2 September 1994. Naval Space Command reported that the eight objects were discovered during the period from July 1993 to July 1994.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Ariane 4 H10 Final Stage
OWNER: France
LAUNCH DATE: 9.95 Jul 1992
DRY MASS (KG): ~1760
MAIN BODY: Cylinder: 2.6 m diameter by 11.4 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|----------|-----------------|------------|
| DATE: | Feb 2002 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 02032.98792301 | MEAN ANOMALY: | 23.4497 |
| RIGHT ASCENSION: | 196.5922 | MEAN MOTION: | 3.10885568 |
| INCLINATION: | 7.0154 | MEAN MOTION DOT/2: | .00036687 |
| ECCENTRICITY: | .6663885 | MEAN MOTION DOT DOT/6: | .0 |
| ARG. OF PERIGEE: | 261.6162 | BSTAR: | .0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 41.05 min*
MAXIMUM ΔI : 0.25 deg*

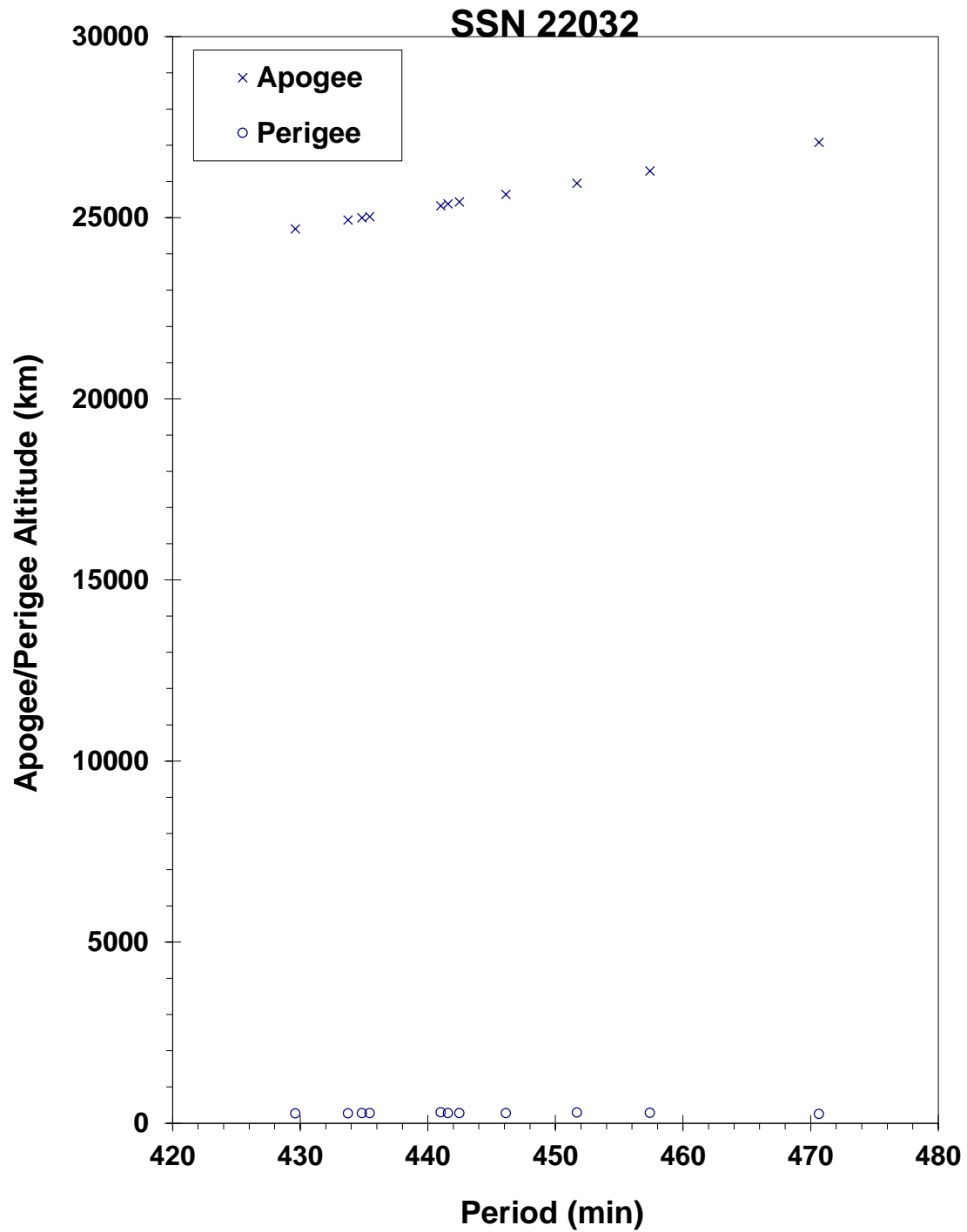
* Based on uncataloged debris data

COMMENTS

This marks the sixth known fragmentation of an Ariane 4 third stage. The last three vehicles involved in such events (1988-109C, 1991-075C, and 1992-041C) had been in orbit 9-10 years at the time of their respective breakups. All flights were conducted prior to the implementation of passivation measures for Ariane GTO missions in September 1993. No Ariane launch vehicle launched since that time is known to have experienced an on-orbit fragmentation.

REFERENCE DOCUMENT

“Second Identified Satellite Breakup of 2002”, [The Orbital Debris Quarterly News](https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv7i3.pdf), NASA JSC, July 2002. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv7i3.pdf>.



Insat 2A/Eutelsat 2F4 R/B debris cloud of 11 fragments 4 months after the breakup as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 30.08 Jul 1992
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellant

EVENT DATA

DATE: 10 Jul 2004 LOCATION: 34S, 240E (asc)
TIME: 0240 GMT (est.) ASSESSED CAUSE: Propulsion
ALTITUDE: 18,525 km

POST-EVENT ELEMENTS

EPOCH: 04190.17957430 MEAN ANOMALY: 263.0579
RIGHT ASCENSION: 223.5224 MEAN MOTION: 4.29145980
INCLINATION: 64.8832 MEAN MOTION DOT/2: 0.00005202
ECCENTRICITY: 0.5757033 MEAN MOTION DOT DOT/6: 0.0
ARG. OF PERIGEE: 149.3553 BSTAR: 0.0055262

DEBRIS CLOUD DATA

MAXIMUM ΔP : 37.6 min
MAXIMUM ΔI : 0.5 deg

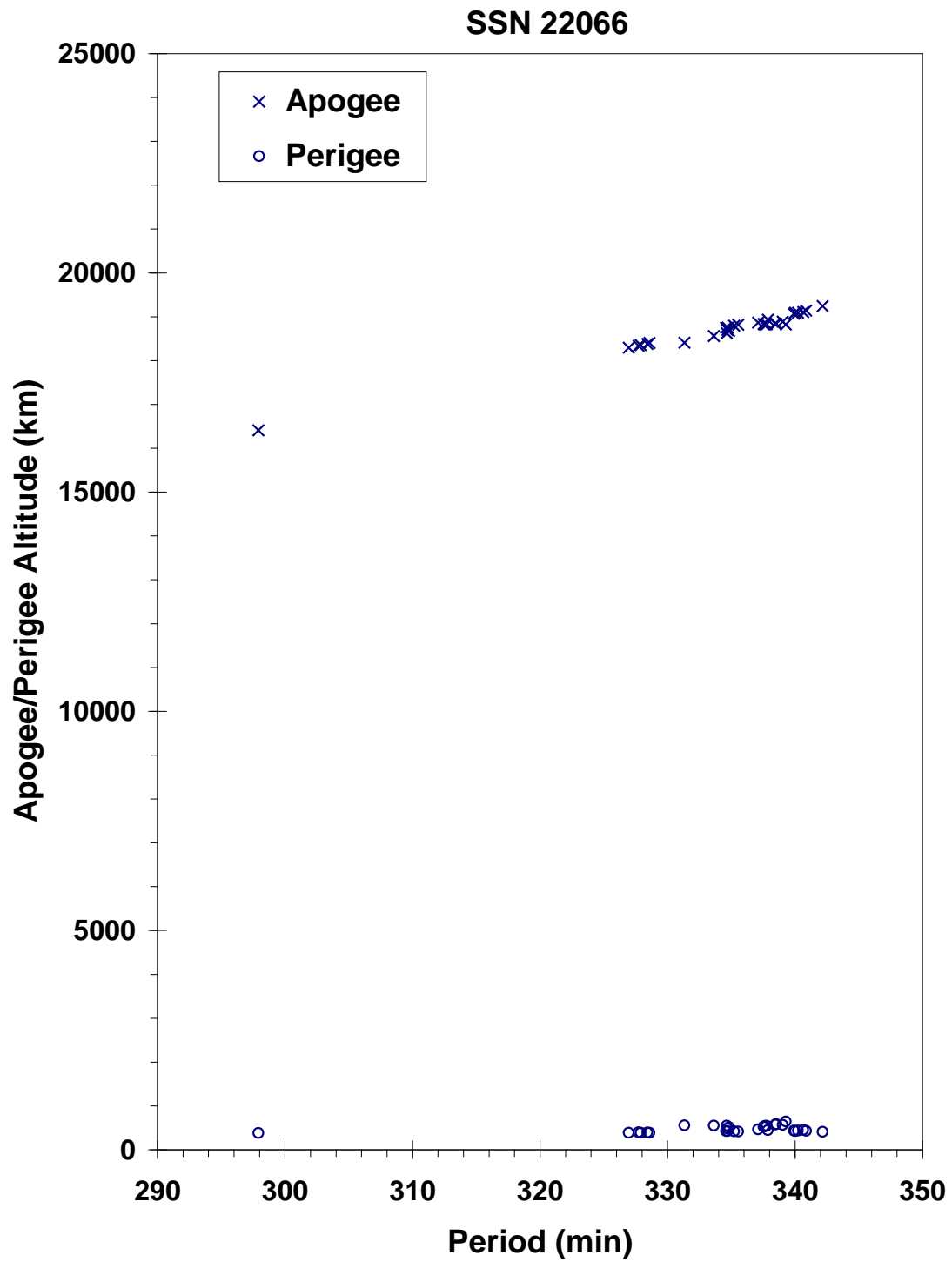
COMMENTS

The object was in a decaying, eccentric orbit; this event marks the 30th known breakup of a Proton Block DM SOZ ullage motor since 1984. More than 100 debris were detected from Sat. No. 22066 by the US SSN. The sister SOZ motor for this launch broke up in an unrelated event ten years earlier (22067).

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

"Recent Satellite Breakups", The Orbital Debris Quarterly News, NASA JSC, October 2004.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv8i4.pdf>.



SOZ motor debris cloud of 31 fragments less than 1 week after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 30.08 Jul 1992
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1.0 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

DATE: Prior to 0824 GMT, 8 Nov 1994 LOCATION: Unknown
TIME: Unknown ASSESSED CAUSE: Propulsion
ALTITUDE: Unknown

PRE-EVENT ELEMENTS

EPOCH: 94310.28602258 MEAN ANOMALY: 9.8460
RIGHT ASCENSION: 65.2049 MEAN MOTION: 4.23571466
INCLINATION: 64.8556 MEAN MOTION DOT/2: 0.00001002
ECCENTRICITY: 0.5708388 MEAN MOTION DOT DOT/6: 0.0
ARG. OF PERIGEE: 316.7786 BSTAR: 0.0033777

DEBRIS CLOUD DATA

MAXIMUM ΔP : 4.2 min
MAXIMUM ΔI : 0.9 deg

COMMENTS

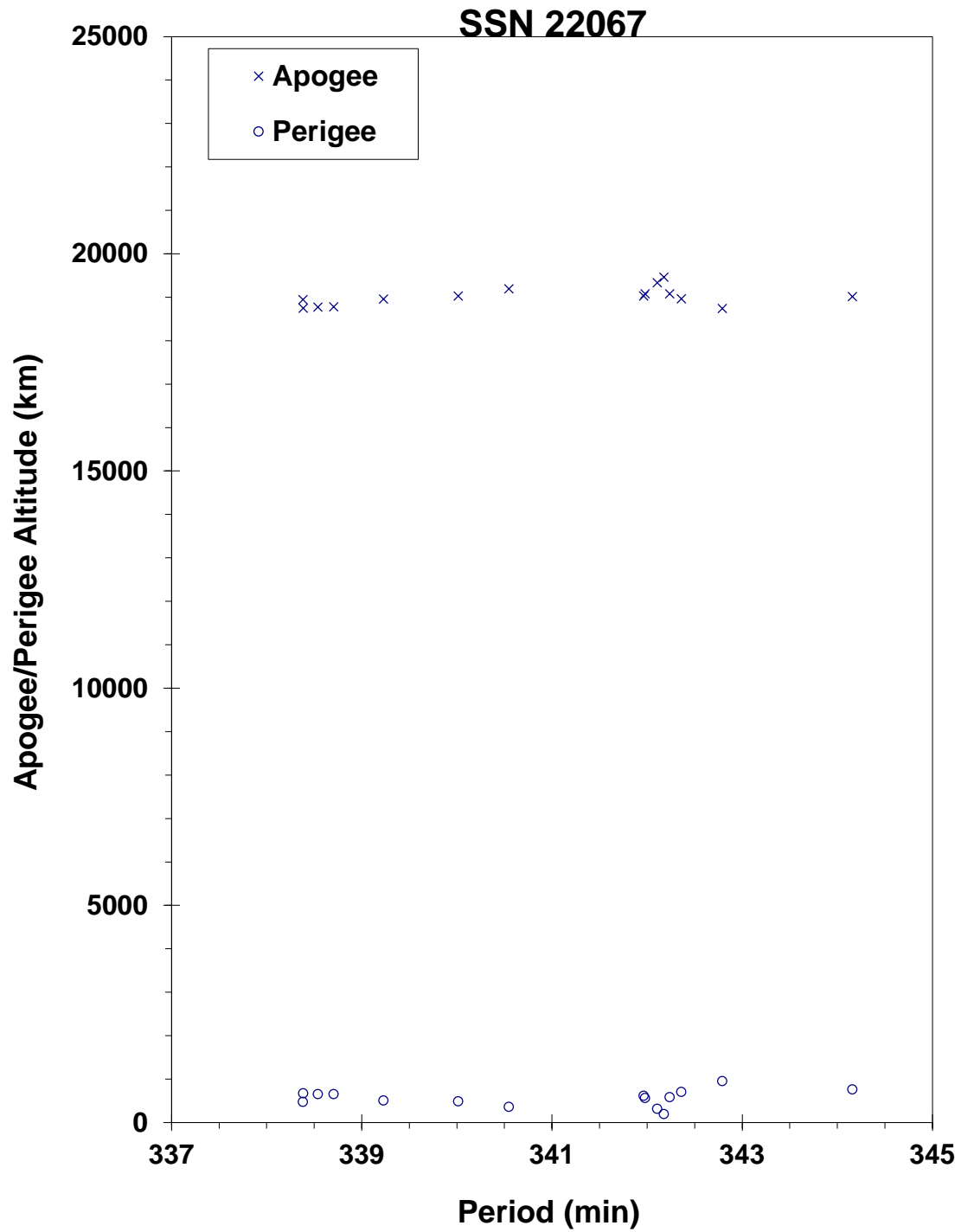
Parent satellite was one of two small engine units that are routinely released after the first burn of the Proton fourth stage. The nature of these objects was identified by Dr. Boris V. Cherniatiev, Deputy Constructor for the Energiya NPO and Mr. Nicholas Johnson of Kaman Sciences, in October, 1992. The cause of this fragmentation appears to be related to the residual hypergolic propellants on board and failure of the membrane separating the fuel and oxidizer. NAVSPASUR observed 32 objects that were associated with this breakup on 8 Nov 94, 36 objects on 9 Nov and 31 objects on 10 Nov. This was the tenth in a series of fragmentations of this object type.

REFERENCE DOCUMENTS

The Fragmentation of Proton Debris, D. J. Nauer, TBE Technical Report CS93-LKD-004, Teledyne Brown Engineering, Colorado Springs, 31 December 1992.

Identification and Resolution of an Orbital Debris Problem with the Proton Launch Vehicle, B. V. Cherniatiev, et al, First European Conference on Space Debris, 5-7 April 1993.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Gabbard Diagram from the Cosmos 2204-2206 debris fragmentation.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 27.55 Nov 1992
DRY MASS (KG): ~55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1.0 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 14 Jul 2001 | LOCATION: | Unknown |
| TIME: | ~1800Z | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|--------------|
| EPOCH: | 01194.12977223 | MEAN ANOMALY: | 232.1640 |
| RIGHT ASCENSION: | 101.3588 | MEAN MOTION: | 9.96766196 |
| INCLINATION: | 46.4697 | MEAN MOTION DOT/2: | 0.01023324 |
| ECCENTRICITY: | 0.2850325 | MEAN MOTION DOT DOT/6: | 0.0000014714 |
| ARG. OF PERIGEE: | 148.6125 | BSTAR: | 0.00050789 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

This event marks the 25th known breakup of a Proton Blok DM SOZ ullage motor since 1984. Due to the moderate eccentricity and altitude of the orbits, the debris were difficult for the US Space Surveillance Network to track. This ullage motor was launched before implementation of breakup preventive measures. No debris were cataloged from this breakup.

REFERENCE DOCUMENT

"New Satellite Breakups Detected", The Orbital Debris Quarterly News, NASA JSC, October 2001. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv6i4.pdf>.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 17.53 Dec 1992
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellant

EVENT DATA

| | | | |
|-----------|--------------|-----------------|------------|
| DATE: | ~22 Apr 2005 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 05112.46798568 | MEAN ANOMALY: | 2.1228 |
| RIGHT ASCENSION: | 268.8209 | MEAN MOTION: | 3.90056983 |
| INCLINATION: | 46.7270 | MEAN MOTION DOT/2: | 0.00018984 |
| ECCENTRICITY: | 0.6142562 | MEAN MOTION DOT DOT/6: | 0.00000003 |
| ARG. OF PERIGEE: | 348.3165 | BSTAR: | 0.0006277 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

This event marks the 32nd known breakup of a Proton Blok DM SOZ ullage motor since 1984. This object had a perigee near 200 km. The event happened sometime between 1100 GMT on April 22, and 1200 GMT on April 23. Only about a dozen pieces were detected by the US SSN.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

"Recent Satellite Breakups", The Orbital Debris Quarterly News, NASA JSC, July 2005.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv9i3.pdf>.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 22.50 Dec 1992
DRY MASS (KG): 6000
MAIN BODY: Cylinder; 2.4 m diameter by 7 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

DATE: 18 Feb 1993 LOCATION: 55N, 157E (dsc)
TIME: 1856 GMT ASSESSED CAUSE: Deliberate
ALTITUDE: 220 km

PRE-EVENT ELEMENTS

EPOCH: 93040.89217375 MEAN ANOMALY: 244.5776
RIGHT ASCENSION: 125.1196 MEAN MOTION: 16.07940666
INCLINATION: 64.8919 MEAN MOTION DOT/2: 0.00301303
ECCENTRICITY: 0.0039285 MEAN MOTION DOT DOT/6: 0.000049705
ARG. OF PERIGEE: 115.8892 BSTAR: 0.00032572

DEBRIS CLOUD DATA

MAXIMUM ΔP : > 2.3 min
MAXIMUM ΔI : > 0.5 deg

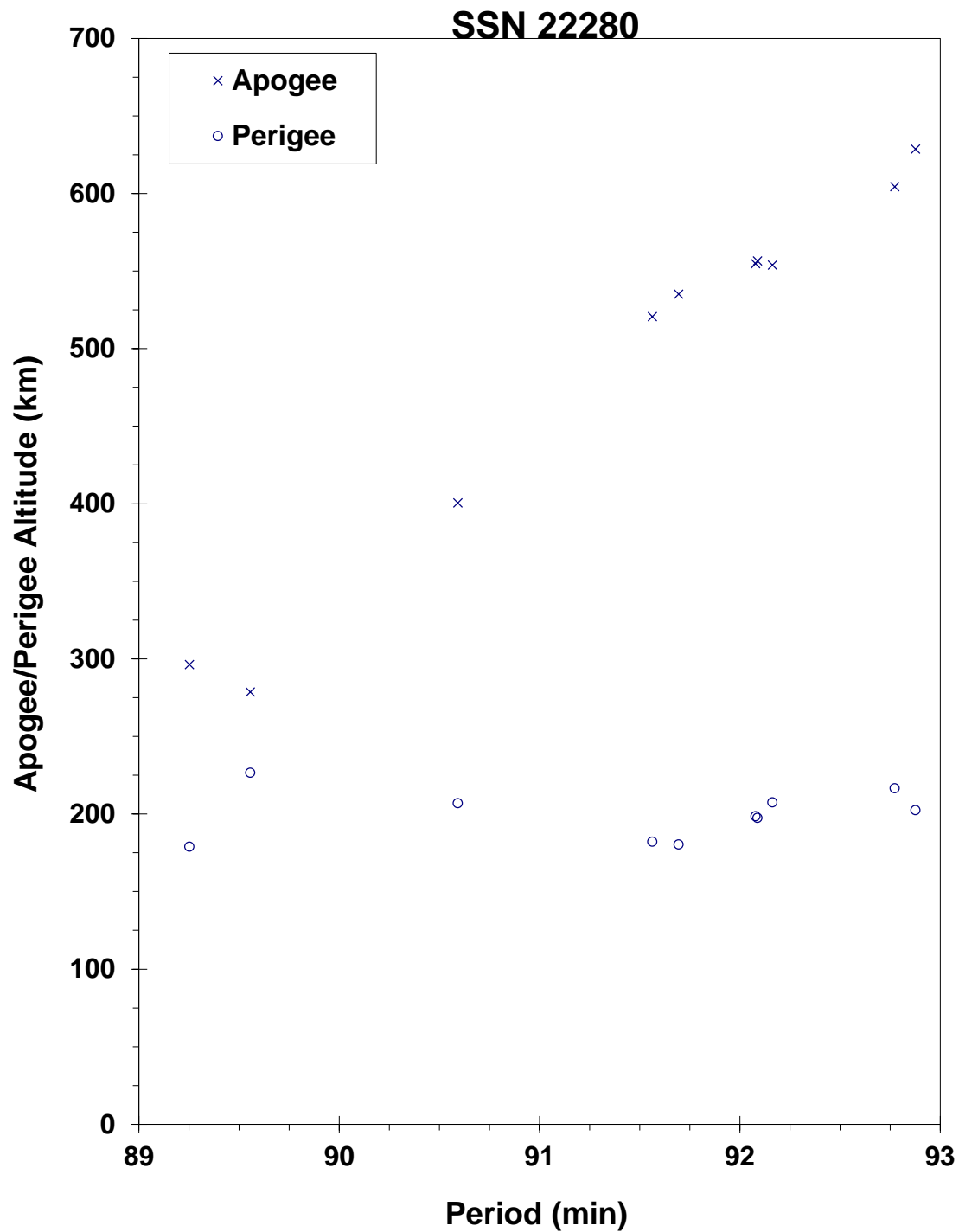
COMMENTS

Spacecraft was destroyed with a planned detonation. Fourth fragmentation of the Cosmos 2031 subclass. Early elements on only 10 objects (including the parent) available; 21 objects were observed by Fylingdales soon after the event.

REFERENCE DOCUMENTS

Analysis of Fragmentations from December 1992 - February 1993, Technical Report CS93-LKD-010, Teledyne Brown Engineering, Colorado Springs, Colorado, 30 March 1993.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 2225 debris cloud remnant of 10 objects 4 days after the event as reconstructed from the data provided by Naval Space Surveillance System in a 22 February 1993 Satellite Support message.

SATELLITE DATA

TYPE: Zenit Second Stage
OWNER: CIS
LAUNCH DATE: 25.25 Dec 1992
DRY MASS (KG): 8300
MAIN BODY: Cylinder; 3.9 m diameter by 12 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: On-board propellants (~900 kg), explosive charge (?)

EVENT DATA (1)

| | | | |
|-----------|-------------|-----------------|------------------|
| DATE: | 26 Dec 1992 | LOCATION: | 63 N, 60 E (asc) |
| TIME: | 0738 GMT | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | 830 km | | |

Note: NAVSPASUR could not correlate with 22285, but 22285 was closest object.

EVENT DATA (2)

| | | | |
|-----------|-------------|-----------------|-------------------|
| DATE: | 26 Dec 1992 | LOCATION: | 44 N, 168 E (asc) |
| TIME: | 2249 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 830 km | | |

EVENT DATA (3)

| | | | |
|-----------|-------------|-----------------|------------------|
| DATE: | 26 Dec 1992 | LOCATION: | 52 N, 63 E (dsc) |
| TIME: | 2310 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 835 km | | |

EVENT DATA (4)

| | | | |
|-----------|-------------|-----------------|-------------------|
| DATE: | 30 Dec 1992 | LOCATION: | 22 S, 172 E (dsc) |
| TIME: | 0903 GMT | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | ~ 830 km | | |

PRE-EVENT ELEMENTS TO EVENT 1

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 92361.30431818 | MEAN ANOMALY: | 289.8749 |
| RIGHT ASCENSION: | 227.4354 | MEAN MOTION: | 14.1258288 |
| INCLINATION: | 71.0274 | MEAN MOTION DOT/2: | -0.00061925 |
| ECCENTRICITY: | 0.0005311 | MEAN MOTION DOT DOT/6: | 0.0 |
| ARG. OF PERIGEE: | 71.7543 | BSTAR: | -0.034134 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 4.3 min *
MAXIMUM ΔI : 1.4 deg *

* Based upon comparison to pre-event elements,
includes all four events, based upon cataloged elements only.

COMMENTS

Four separate events were reported by NAVSPASUR. The first observed event was accompanied by an initial 18 objects, but could not be correlated with the rocket body element set. The rocket body was the closest object to the BLAST point. The second event followed 15 hours later with 96 objects. The third event followed the second by less than 20 minutes and was based upon 51 pieces. The fourth event was accompanied by 3 objects. Element data on 164 objects has been combined into a single Gabbard Diagram.

NAVSPASUR initially generated 164 element sets on the combined debris from these 4 events. On 24 April 1995 object 22366, 1992-093BF, fragmented liberating 1 associated piece.

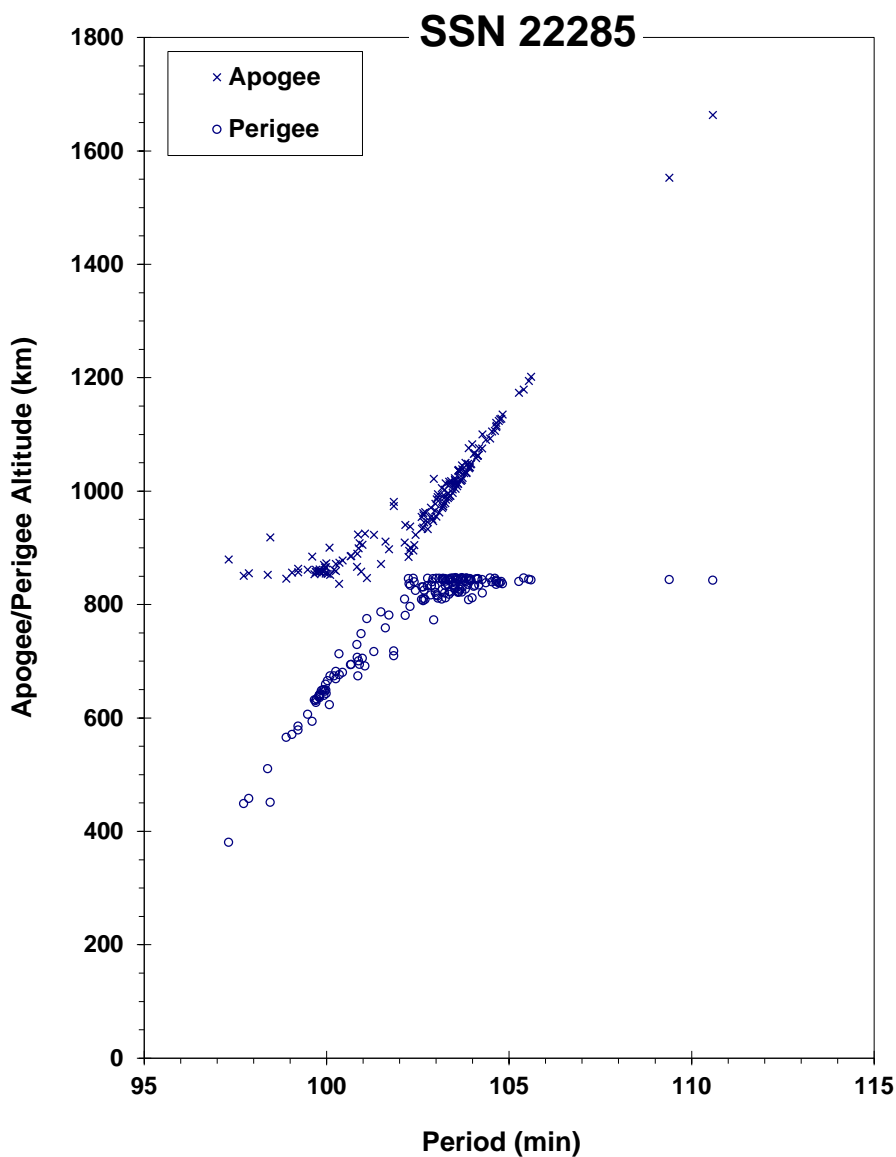
REFERENCE DOCUMENTS

Cosmos 2227 Rocket Body Fragmentation Event, E. L. Jenkins, et. al., NAVSPASUR, Dahlgren, VA.

Analysis of Fragmentations From December 1992 - February 1993, TBE Technical Report CS93-LKD-010, Teledyne Brown Engineering, Colorado Springs, 30 March 1993.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

“The Recent Fragmentations of LEO Upper Stages”, G. Chernyavskiy et al, 45th IAF Congress, October 1994.



Naval Space Surveillance System generated 164 initial element sets on the four events that are plotted on the single Gabbard diagram above.

SATELLITE DATA

TYPE: Zenit Second Stage
OWNER: CIS
LAUNCH DATE: 26.10 Mar 1993
DRY MASS (KG): 8300
MAIN BODY: Cylinder; 3.9 m diameter by 12 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants (~900 kg), explosive charge (?)

EVENT DATA

| | | | |
|-----------|-------------|-----------------|----------------|
| DATE: | 28 Mar 1993 | LOCATION: | 70N, 37E (dsc) |
| TIME: | 0716 GMT | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | 840 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 93088.27687915 | MEAN ANOMALY: | 84.1791 |
| RIGHT ASCENSION: | 258.8192 | MEAN MOTION: | 14.14093359 |
| INCLINATION: | 70.9947 | MEAN MOTION DOT/2: | 0.00255882 |
| ECCENTRICITY: | 0.0006748 | MEAN MOTION DOT DOT/6: | 0.0 |
| ARG. OF PERIGEE: | 275.8565 | BSTAR: | 0.12879 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 3.5 min
MAXIMUM ΔI : 0.1 deg

COMMENTS

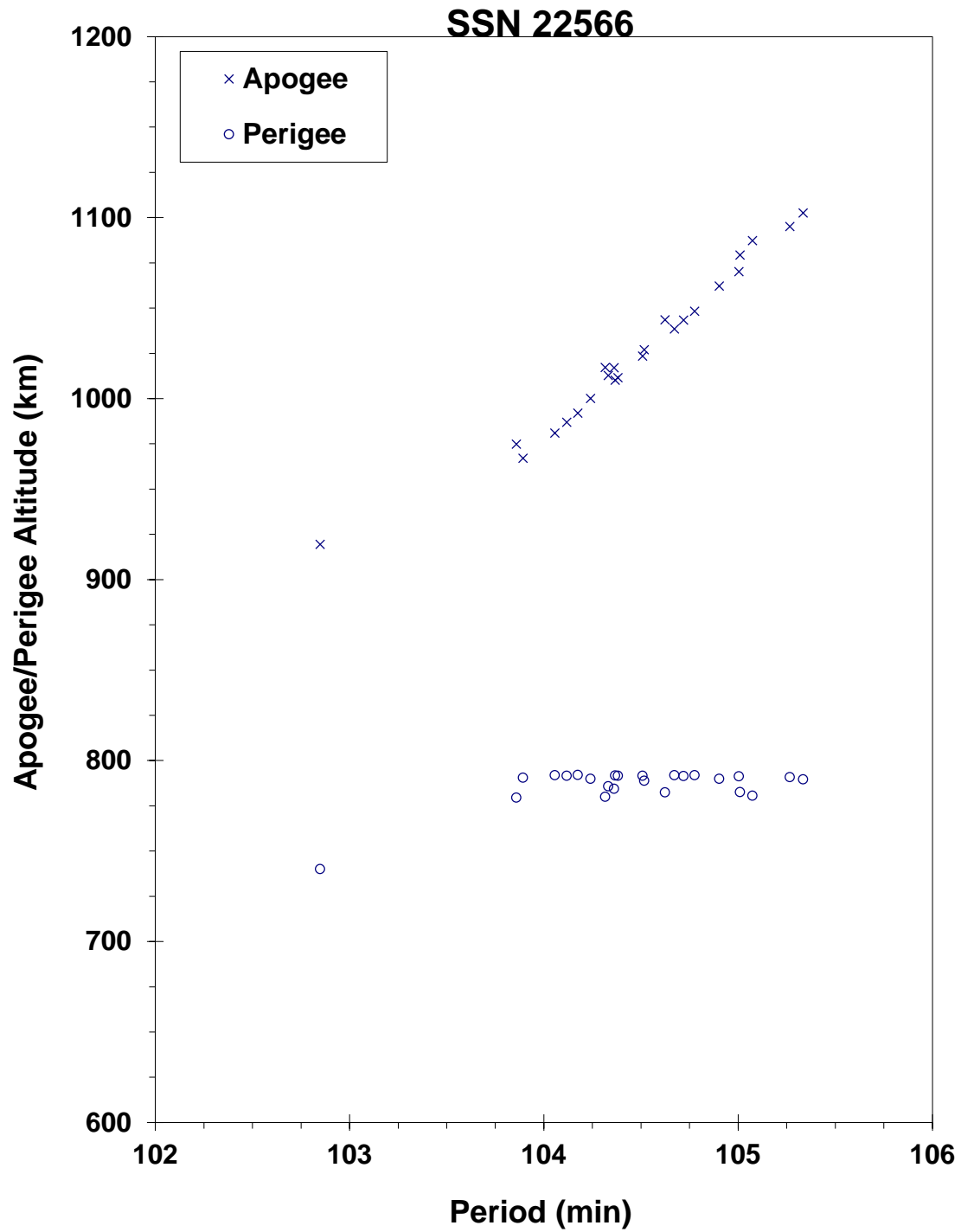
At least 26 initial element sets were generated on this event by NAVSPASUR. The BLAST point was calculated from 12 objects.

REFERENCE DOCUMENTS

The Fragmentation of the Cosmos 2237 & 2243 Rocket Bodies, D. J. Nauer, TBE Technical Report CS93-LKD-016, Teledyne Brown Engineering, Colorado Springs, 15 June 1993.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

“The Recent Fragmentations of LEO Upper Stages”, G. Chernyavskiy et al, 45th IAF Congress, October 1994.



Naval Space Surveillance System generated 24 initial element sets that are plotted on the Gabbard diagram above.

SATELLITE DATA

| | |
|-------------------|--|
| TYPE: | Payload |
| OWNER: | CIS |
| LAUNCH DATE: | 30.50 Mar 1993 |
| DRY MASS (KG): | 3000 |
| MAIN BODY: | Cylinder; 1.3 m diameter by 17 m length |
| MAJOR APPENDAGES: | Solar panels |
| ATTITUDE CONTROL: | Active, 3-axis |
| ENERGY SOURCES: | On-board propellants, explosive charge (?) |

EVENT DATA

| | | | |
|-----------|------------|-----------------|----------------------|
| DATE: | 1 Dec 1994 | LOCATION: | 6.5 S, 243.0 E (dsc) |
| TIME: | 1111 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 275 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|--------------|
| EPOCH: | 94335.21831221 | MEAN ANOMALY: | 119.6648 |
| RIGHT ASCENSION: | 124.7826 | MEAN MOTION: | 16.06466469 |
| INCLINATION: | 65.0063 | MEAN MOTION DOT/2: | 0.00787680 |
| ECCENTRICITY: | 0.0069696 | MEAN MOTION DOT DOT/6: | 0.0000095760 |
| ARG. OF PERIGEE: | 239.7651 | BSTAR: | 0.00073936 |

DEBRIS CLOUD DATA

| | |
|----------------------|---------|
| MAXIMUM ΔP : | Unknown |
| MAXIMUM ΔI : | Unknown |

COMMENTS

Cosmos 2238 was a member of the Cosmos 699-type and the first such spacecraft to breakup for 7 years. As many as 26 debris were detected; however, additional debris probably decayed before they were observed. Cosmos is the 17th fragmentation of a Cosmos 699 class payload.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 27.44 Apr 1993
DRY MASS (KG): 5700
MAIN BODY: Cone-cylinder; 2.7 m diameter by 6.3 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

DATE: 27 Apr 1993 LOCATION: 61N, 81E (asc)
TIME: 1044 GMT ASSESSED CAUSE: Deliberate
ALTITUDE: 200 km

PRE-EVENT ELEMENTS *

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 93119.28633059 | MEAN ANOMALY: | 283.6524 |
| RIGHT ASCENSION: | 51.8515 | MEAN MOTION: | 16.26199828 |
| INCLINATION: | 70.3602 | MEAN MOTION DOT/2: | 0.02823100 |
| ECCENTRICITY: | 0.0032877 | MEAN MOTION DOT DOT/6: | 0.000019668 |
| ARG. OF PERIGEE: | 76.8057 | BSTAR: | 0.00077017 |

* Note: Element Set 1 not generated until ~2 days after the event.

DEBRIS CLOUD DATA

MAXIMUM ΔP : 7.1 min
MAXIMUM ΔI : 0.4 deg

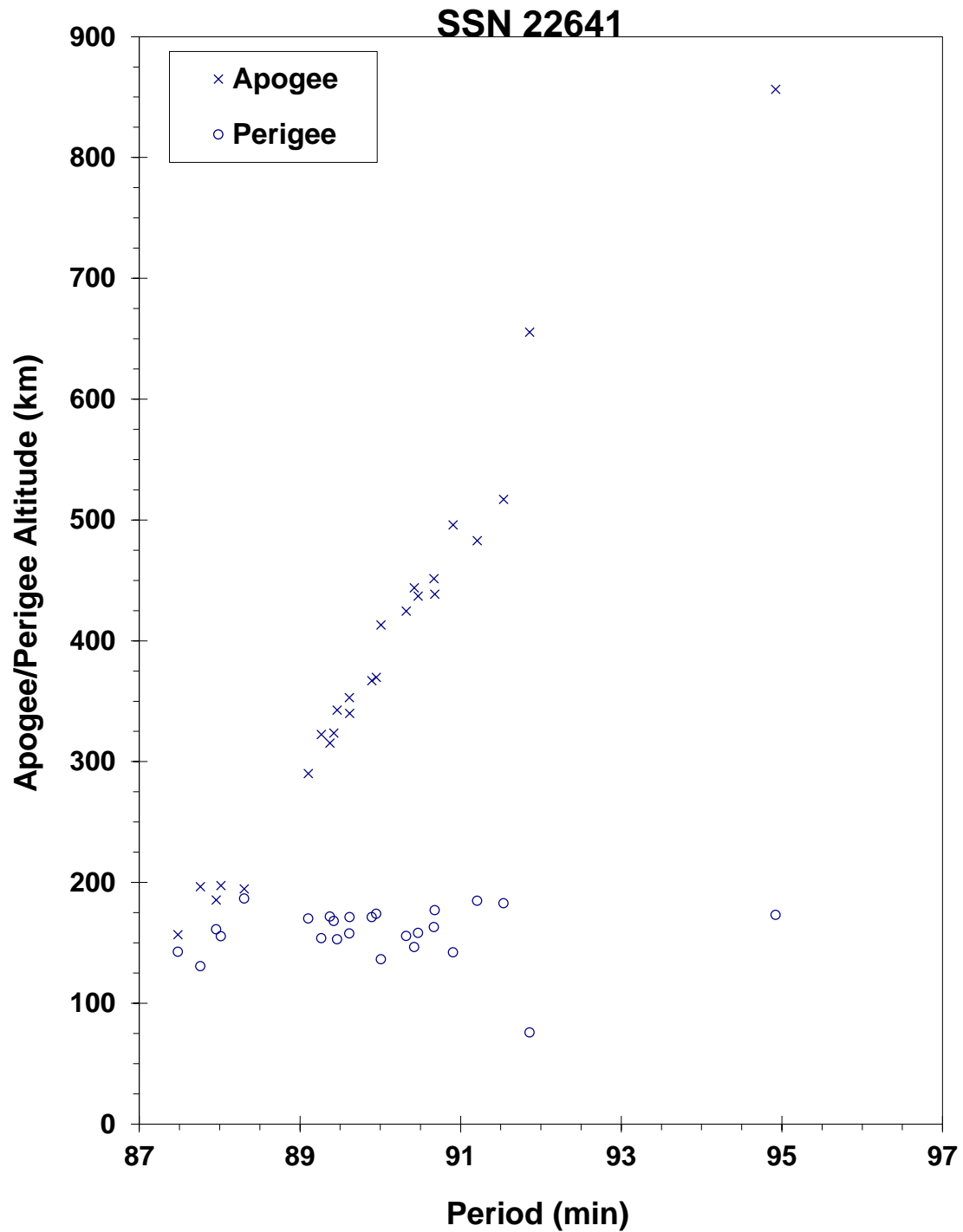
COMMENTS

Although this event was originally thought to be a fragmentation of the Soyuz final stage rocket body, it was actually the payload that fragmented. This event occurred near orbital insertion into the operational orbit. It is unclear whether the payload was attached at the time of the event. The payload malfunctioned and self-destructed. There were 25 initial element sets available after launch. NAVSPASUR reported tracking approximately 27 objects on 30 April 1993, and detected as many as 20 more unknowns. The final official piece count associated with this event was 172 objects. Due to the very low altitude, most objects decayed from this cloud within 2 weeks of launch. No cataloged element sets were released until almost 2 days after the event.

REFERENCE DOCUMENTS

The Fragmentation of the Cosmos 2237 & 2243 Rocket Bodies, D. J. Nauer, TBE Technical Report CS93-LKD-016, Teledyne Brown Engineering, Colorado Springs, 15 June 1993.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Naval Space Surveillance System generated 25 initial element sets on the Cosmos 2243 fragmentation that are plotted on the Gabbard diagram above.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 16 Jun 1993
DRY MASS (KG): 900
MAIN BODY: cylinder; 2.035 m diameter x 2 m length
MAJOR APPENDAGES: gravity gradient boom
ATTITUDE CONTROL: gravity gradient (passive)
ENERGY SOURCES: pressurized vessel; chemical batteries

EVENT DATA

DATE: 10 Feb 2009 LOCATION: 72.5N, 97.9E (asc)
TIME: 1656 GMT ASSESSED CAUSE: Collision
ALTITUDE: 788.597 km

PRE-EVENT ELEMENTS

EPOCH: 09040.49834364 MEAN ANOMALY: 261.5952
RIGHT ASCENSION: 19.4646 MEAN MOTION: 14.31135643
INCLINATION: 74.0355 MEAN MOTION DOT/2: -0.00000001
ECCENTRICITY: 0.0016027 MEAN MOTION DOT DOT/6: 0.0
ARG. OF PERIGEE: 98.7014 BSTAR: 0.0000095251

DEBRIS CLOUD DATA

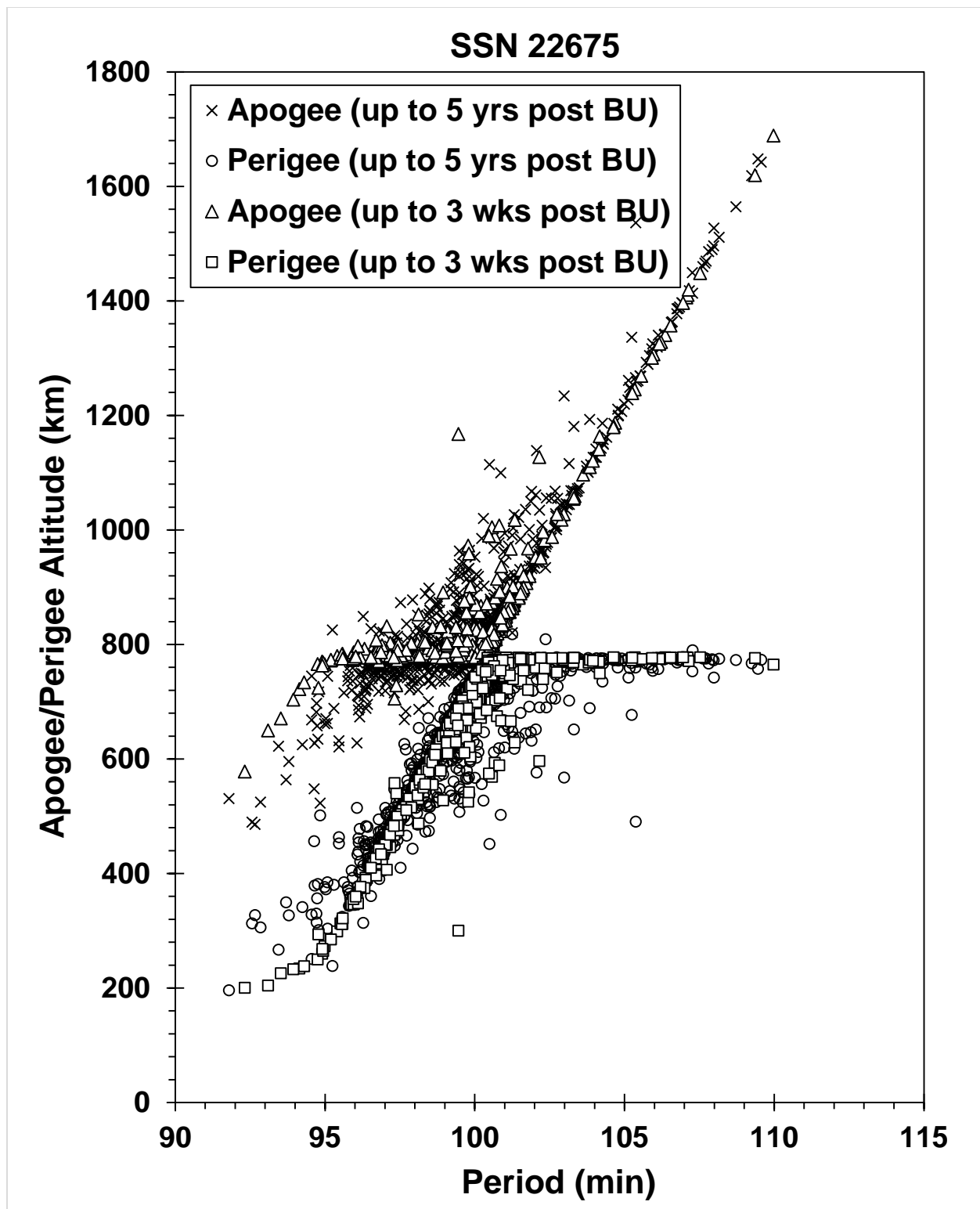
MAXIMUM ΔP : 9.4 min
MAXIMUM ΔI : 24.7 deg

COMMENTS

The spacecraft, a member of the Soviet/Russian Strela-2M low-altitude communications satellite constellation, utilized the versatile KAUR-1 bus (*Kosmicheskii Apparat Universalnogo Ryada-1*, (Космический Аппарат Универсального Ряда), which can be translated as Spacecraft Bus from the Standardized Line-1). This bus consists of a central pressurized cylinder for the payload, an outer cylinder mounting solar cells (the Solar Battery), and a gravity gradient boom for passive stabilization. The spacecraft was derelict at the time of the event. Cosmos 2251 collided with Iridium 33 [q.v.] in the first, accidental collision of large, intact resident space objects.

REFERENCE DOCUMENT

“Satellite Collision Leaves Significant Debris Clouds”, The Orbital Debris Quarterly News, NASA JSC, April 2009.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/ODQNv13i2.pdf>.



The COSMOS 2251 debris cloud, including 1648 fragments cataloged up to 5 years after the event.

SATELLITE DATA

| | |
|-------------------|---|
| TYPE: | Payload |
| OWNER: | CIS |
| LAUNCH DATE: | 14.69 Jul 1993 |
| DRY MASS (KG): | 5700 |
| MAIN BODY: | Cone-cylinder; 2.7 m diameter by 6.3 m length |
| MAJOR APPENDAGES: | Solar panels |
| ATTITUDE CONTROL: | Active, 3-axis |
| ENERGY SOURCES: | On-board propellants, explosive charge |

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 25 Jul 1993 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 93205.96411483 | MEAN ANOMALY: | 292.3177 |
| RIGHT ASCENSION: | 134.4696 | MEAN MOTION: | 16.09525981 |
| INCLINATION: | 67.1310 | MEAN MOTION DOT/2: | 0.00638090 |
| ECCENTRICITY: | 0.0113387 | MEAN MOTION DOT DOT/6: | 0.000023099 |
| ARG. OF PERIGEE: | 68.9805 | BSTAR: | 0.00025239 |

DEBRIS CLOUD DATA

| | |
|----------------------|---------|
| MAXIMUM ΔP : | Unknown |
| MAXIMUM ΔI : | Unknown |

COMMENTS

Spacecraft was destroyed after a malfunction prevented reentry and landing in the CIS. Event identified by Russian officials during investigation cited below.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 7.56 Sep 1993
DRY MASS (KG): 6000
MAIN BODY: Cylinder; 2.4 m diameter by 7 m length
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

DATE: 18 Dec 1993 LOCATION: 65N, 107E (dsc)
TIME: 0711 GMT ASSESSED CAUSE: Deliberate
ALTITUDE: 195 km

PRE-EVENT ELEMENTS

EPOCH: 93352.09835999 MEAN ANOMALY: 294.6647
RIGHT ASCENSION: 209.9170 MEAN MOTION: 16.17608693
INCLINATION: 64.8761 MEAN MOTION DOT/2: 0.00554324
ECCENTRICITY: 0.0065884 MEAN MOTION DOT DOT/6: 0.0
ARG. OF PERIGEE: 66.1310 BSTAR: 0.00022099

DEBRIS CLOUD DATA

MAXIMUM ΔP: 8.7 min *
MAXIMUM ΔI: 0.8 deg *

* Based on uncataloged debris data

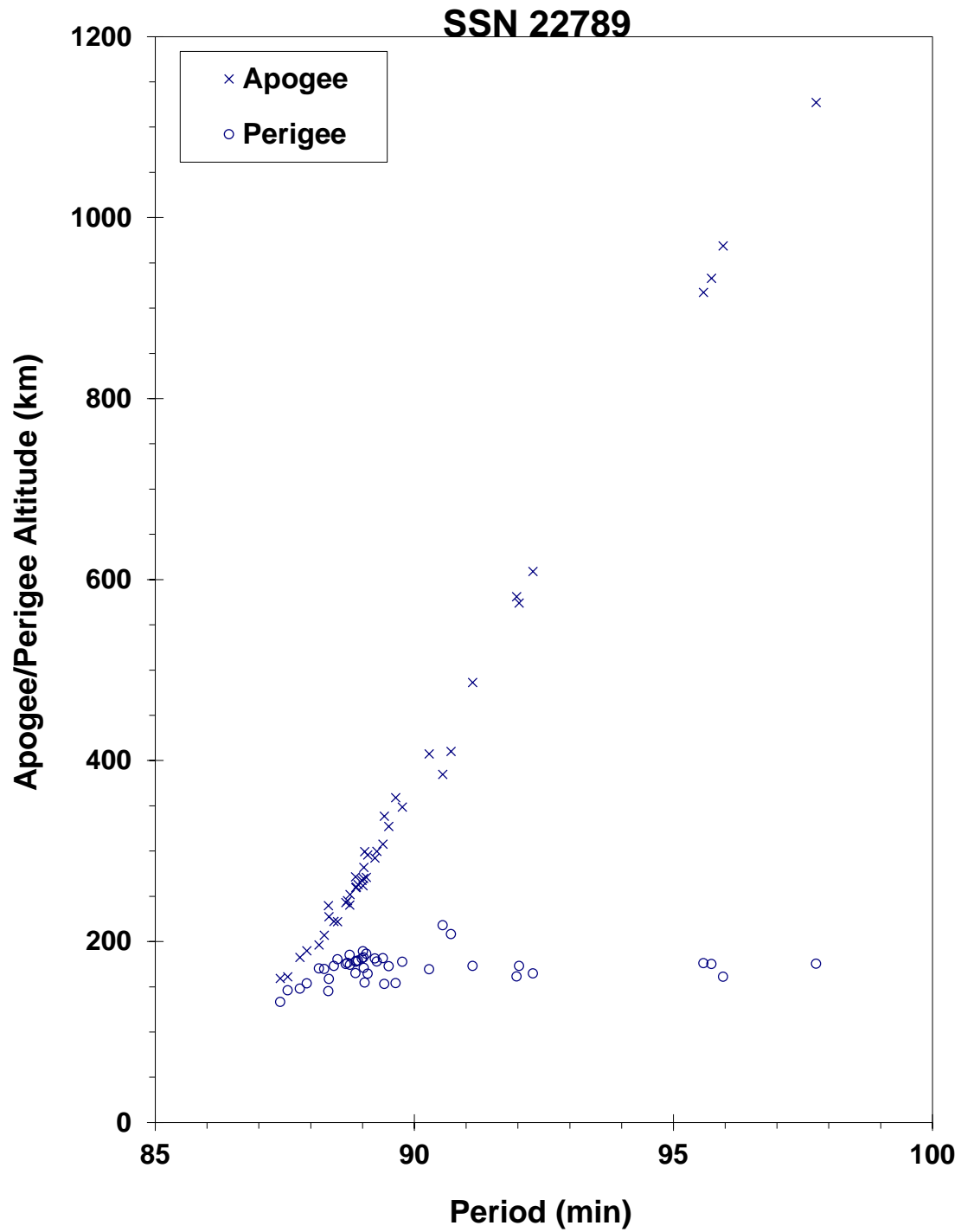
COMMENTS

Spacecraft was destroyed with a planned detonation. Fifth fragmentation of this sub-type (Cosmos 2031 subclass). Early elements on 43 objects (including the parent) were collected; at least 179 objects were reported by the NAVSPOC for early passes through the NAVSPASUR fence.

REFERENCE DOCUMENTS

The Fragmentation of Cosmos 2262, Technical Report CS94-LKD-006, Teledyne Brown Engineering, Colorado Springs, Colorado, 31 December 1993.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.



Cosmos 2262 debris cloud based upon 43 early element sets provided by the NAVSPOC.

SATELLITE DATA

TYPE: Proton Block DM SOZ Ullage Motor
OWNER: CIS
LAUNCH DATE: 18.58 Nov 1993
DRY MASS (KG): ~55 kg
MAIN BODY: ~0.6 m by 0.6 m by 1.0 m
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants?

EVENT DATA

DATE: 6-7 Sep 2000 LOCATION: Unknown
TIME: between 1918-0253 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: Unknown

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|---------------|
| EPOCH: | 00250.18110680 | MEAN ANOMALY: | 305.0033 |
| RIGHT ASCENSION: | 135.7916 | MEAN MOTION: | 6.55809618 |
| INCLINATION: | 46.7439 | MEAN MOTION DOT/2: | 0.00601672 |
| ECCENTRICITY: | 0.4592082 | MEAN MOTION DOT DOT/6: | 0.00000031378 |
| ARG. OF PERIGEE: | 109.1361 | BSTAR: | 0.00059159 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

This is the 22nd breakup event for an object of this class, and the first of the year 2000. The breakups are assessed to be caused by residual propellants. Russian officials have been aware of the problem since 1992 and have made design changes, although the date of full implementation is unknown. The environmental consequence of the breakup will be short-lived; the object is in catastrophic decay from a geosynchronous transfer orbit. Latest estimate of the breakup time is between 1918 GMT, 6 September and 0253 GMT, 7 September.

REFERENCE DOCUMENT

“September Breakup is 22nd in Series”, The Orbital Debris Quarterly News, NASA JSC, October 2000. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNV5i4.pdf>.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Titan II Second Stage
OWNER: US
LAUNCH DATE: 25.69 Jan 1994
DRY MASS (KG): 2860
MAIN BODY: Cylinder; 3.048 m diameter x 7.3 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: On-board propellants

EVENT DATA

DATE: 7 Feb 1994 LOCATION: 59S, 234E (dsc)
TIME: 1719 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 260 km

PRE-EVENT ELEMENTS

EPOCH: 94038.24510489 MEAN ANOMALY: 208.0182
RIGHT ASCENSION: 47.9208 MEAN MOTION: 16.13665058
INCLINATION: 66.9945 MEAN MOTION DOT/2: 0.01050211
ECCENTRICITY: 0.0027030 MEAN MOTION DOT DOT/6: 0.0000059221
ARG. OF PERIGEE: 152.2460 BSTAR: 0.00081413

DEBRIS CLOUD DATA

MAXIMUM ΔP : 5.6 min *
MAXIMUM ΔI : 0.6 deg *

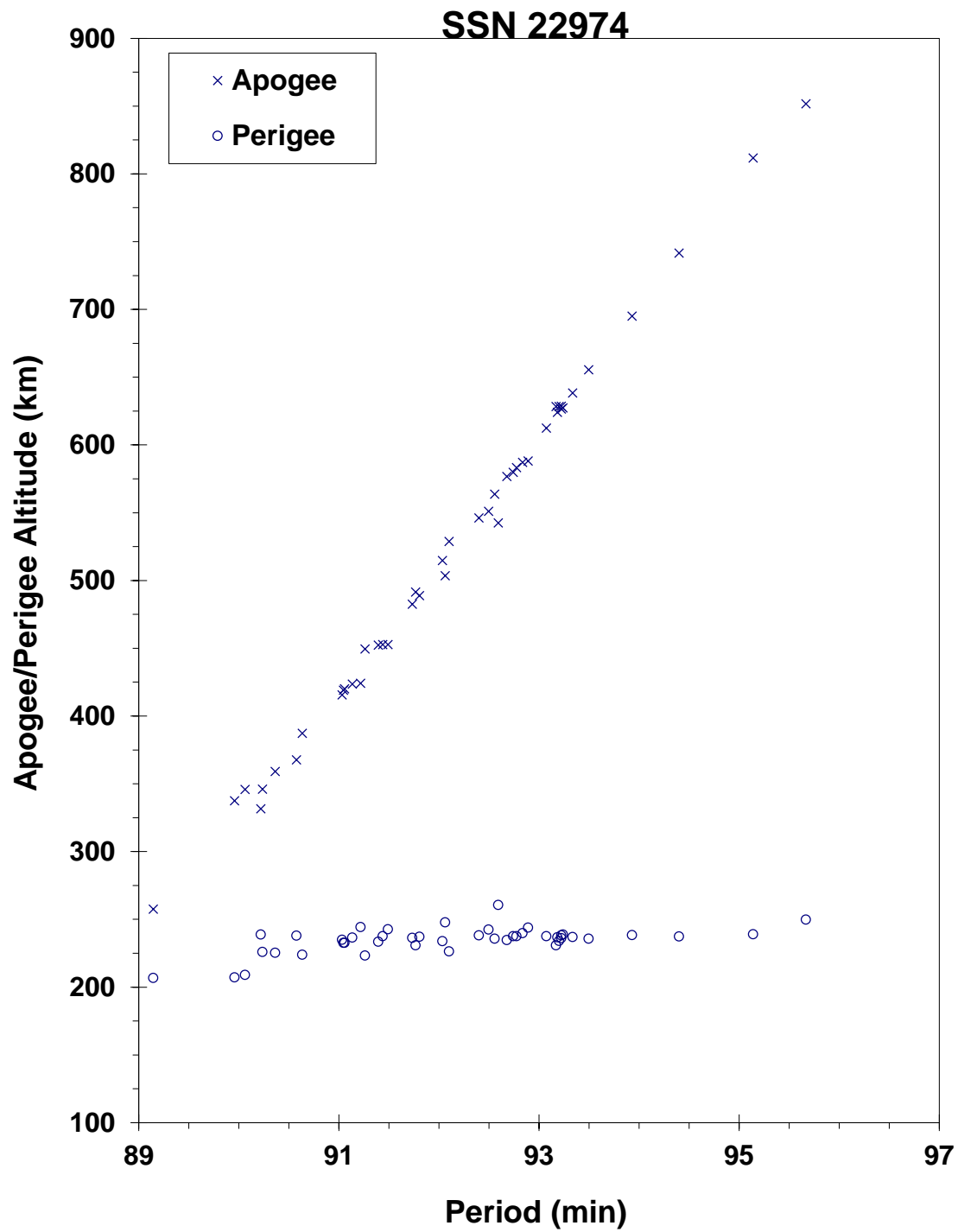
* Based on uncataloged debris data

COMMENTS

First Titan II Second Stage to violently fragment. NAVSPOC reported observing a maximum of 364 objects in the early debris cloud, and the NAVSPOC released 45 element sets. Engineering analysis by the manufacturer (Martin Marietta) indicates no known failure mechanism, although unspent on-board propellants were present.

REFERENCE DOCUMENT

The Fragmentation of the Clementine Rocket Body, TBE Technical Report CS94-LKD-010, Teledyne Brown Engineering, Colorado Springs, Colorado, 31 March 1994.



Gabbard diagram of 45 NAVSPOC element sets.

SATELLITE DATA

TYPE: Pegasus HAPS
OWNER: USA
LAUNCH DATE: 19.71 May 1994
DRY MASS (KG): 97
MAIN BODY: Cylinder; 0.97 m diameter by 0.93 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.
ENERGY SOURCES: On-board propellants and pressurants

EVENT DATA

| | | | |
|-----------|------------|-----------------|------------------|
| DATE: | 3 Jun 1996 | LOCATION: | 67 S, 56 E (asc) |
| TIME: | 1518 GMT | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | 625 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 96155.10100506 | MEAN ANOMALY: | 108.3711 |
| RIGHT ASCENSION: | 197.8565 | MEAN MOTION: | 14.56780581 |
| INCLINATION: | 81.9749 | MEAN MOTION DOT/2: | 0.00000158 |
| ECCENTRICITY: | 0.0165742 | MEAN MOTION DOT DOT/6: | 0 |
| ARG. OF PERIGEE: | 249.9583 | BSTAR: | 0.000025815 |

DEBRIS CLOUD DATA

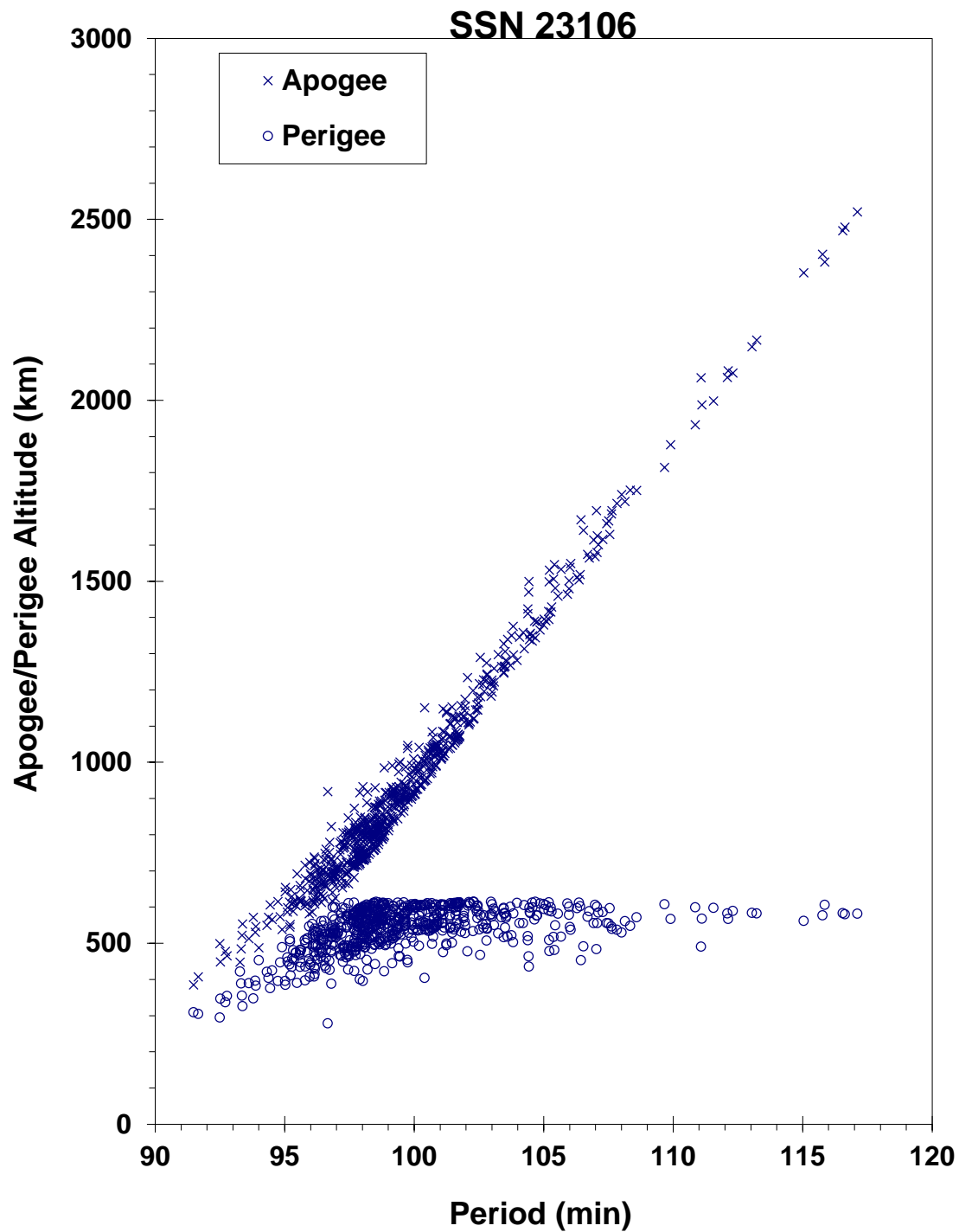
MAXIMUM ΔP : 27.9 min
MAXIMUM ΔI : 2.4 deg

COMMENTS

The Pegasus Hydrazine Auxiliary Propulsion Stage (HAPS) was employed for only the second time. It failed to place its payload into the assigned circular orbit and had an estimated 5-8 kg of residual propellant plus propellant for attitude control on board. The fragmentation of the small, 2-year-old upper stage produced a record number (to that time) of more than 750 tracked debris, and necessitated the addition of a third place in the piece tag format in two line element sets. This is about an order of magnitude more than can be expected for the small dry mass of the R/B of 97 kg. Observations suggest that the debris are physically small with a high radar reflectivity. Investigations suggest that a regulator failure led to overpressurization of the propellant tank that in turn ruptured.

REFERENCE DOCUMENT

"Major Satellite Breakup in June", N. Johnson, Orbital Debris Quarterly News, NASA JSC, September 1996, p. 2 and 11. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNV1i2.pdf>.



Step II R/B debris cloud of 713 fragments as of August 29, 1996 as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 6.99+ Jul 1994
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|---------------------|-----------------|------------|
| DATE: | Approx. 21 Oct 1995 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 95293.99530492 | MEAN ANOMALY: | 2.33725319 |
| RIGHT ASCENSION: | 157.0951 | MEAN MOTION: | 321.8211 |
| INCLINATION: | 47.0485 | MEAN MOTION DOT/2: | 0.00002472 |
| ECCENTRICITY: | 0.7223127 | MEAN MOTION DOT DOT/6: | 0.0000 |
| ARG. OF PERIGEE: | 127.9520 | BSTAR: | 0.0010694 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

Parent satellite was one of two small engine units that are routinely released after the first burn of the Proton fourth stage. The nature of these objects was identified by Dr. Boris V. Chernlatiev, Deputy Constructor for the Energiya NPO, in October 1992. The cause of this fragmentation is assumed to be related to the residual hypergolic propellants on board and failure of the membrane separating the fuel and oxidizer. NAVSPASUR observed 114 objects that were associated with this breakup.

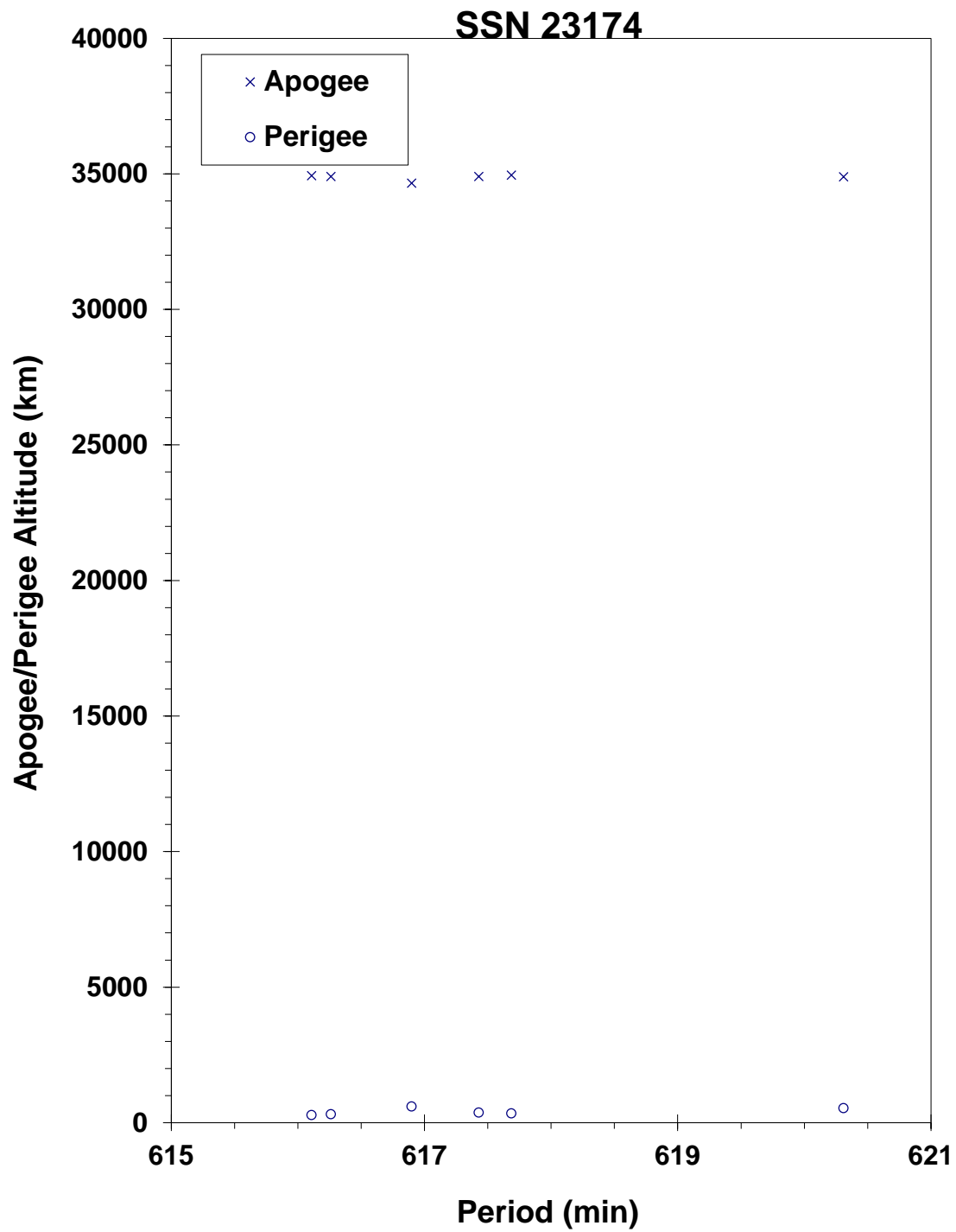
REFERENCE DOCUMENTS

The Fragmentation of Proton Debris, D. J. Nauer, TBE Technical Report CS93-LKD-004, Teledyne Brown Engineering, Colorado Springs, 31 December 1992.

Analysis of Fragmentations From December 1992 - February 1993, TBE Technical Report CS93-LKD-010, Teledyne Brown Engineering, Colorado Springs, 30 March 1993.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

"Identification and Resolution of an Orbital Debris Problem with the Proton Launch Vehicle", B. V. Cherniatiev et al, Proceedings of the First European Conference on Space Debris, April 1993.



Cosmos 2282 ullage motor debris cloud of 6 fragments assembled by NAVSPOC.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 31.60 Oct 1994
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

DATE: Prior to 0547 GMT, 11 May 1995 LOCATION: Unknown
TIME: Unknown ASSESSED CAUSE: Propulsion
ALTITUDE: Unknown

PRE-EVENT ELEMENTS

EPOCH: 95130.00087914 MEAN ANOMALY: 317.6470
RIGHT ASCENSION: 200.4799 MEAN MOTION: 11.93599702
INCLINATION: 46.9113 MEAN MOTION DOT/2: 0.99999999
ECCENTRICITY: 0.2007574 MEAN MOTION DOT DOT/6: 0.000034693
ARG. OF PERIGEE: 63.6124 BSTAR: 0.021116

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

Parent satellite was one of two small engine units that are routinely released after the first burn of the Proton fourth stage. The nature of these objects was identified by Dr. Boris V. Cherniatiev, Deputy Constructor for the Energiya NPO and Mr. Nicholas Johnson of Kaman Sciences, in October, 1992. The cause of this fragmentation appears to be related to the residual hypergolic propellants on board and failure of the membrane separating the fuel and oxidizer. NAVSPASUR observed up to 13 objects that were associated with this breakup on 11 May 95. This was the eleventh in a series of fragmentations of this object type.

REFERENCE DOCUMENTS

The Fragmentation of Proton Debris, D. J. Nauer, TBE Technical Report CS93-LKD-004, Teledyne Brown Engineering, Colorado Springs, 31 December 1992.

Analysis of Fragmentations From December 1992 - February 1993, TBE Technical Report CS93-LKD-010, Teledyne Brown Engineering, Colorado Springs, 30 March 1993.

Identification and Resolution of an Orbital Debris Problem with the Proton Launch Vehicle, B. V. Cherniatiev, et al, First European Conference on Space Debris, 5-7 April 1993.

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 20 Nov 1994
DRY MASS (KG): 56.0
MAIN BODY: ellipsoid; 0.6 m diameter x 1 m length
MAJOR APPENDAGES: none
ATTITUDE CONTROL: none
ENERGY SOURCES: on-board propellants

EVENT DATA

DATE: 7 June 2014
TIME: unknown
ALTITUDE: unknown
LOCATION: unknown
ASSESSED CAUSE: Propulsion

PRE-EVENT ELEMENTS

EPOCH: 14156.48285780
RIGHT ASCENSION: 76.9596
INCLINATION: 64.9844
ECCENTRICITY: 0.5768591
ARG. OF PERIGEE: 344.0988
MEAN ANOMALY: 3.5477
MEAN MOTION: 4.25680746
MEAN MOTION DOT/2: 0.00004224
MEAN MOTION DOT DOT/6: 0.00000
BSTAR: 0.050698

DEBRIS CLOUD DATA

MAXIMUM ΔP : unknown
MAXIMUM ΔI : unknown

COMMENTS

This event was the 41st known fragmentation of Blok-DM ullage motors. The breakups are believed to be caused by residual propellants. Russian officials have been aware of the problem since 1992 and have made design changes, although the date of full implementation is unknown. Although at least 15 small debris were observed by the SSN, none have been officially cataloged.

REFERENCE DOCUMENTS

History of Soviet/Russian Satellite Fragmentations: A Joint U.S.-Russian Investigation. N.L. Johnson et al. Kaman Sciences Corporation, October 2005.

"Flurry of Small Breakups in First Half of 2014", The Orbital Debris Quarterly News, NASA JSC, July 2014.
Available online at <http://orbitaldebris.jsc.nasa.gov/newsletter/pdfs/ODQNV18i3.pdf>.

Insufficient data to construct a Gabbard diagram.

RS-15 R/B

1994-085B

23440

SATELLITE DATA

TYPE: Rokot Third Stage
OWNER: CIS
LAUNCH DATE: 26.13 Dec 1994
DRY MASS (KG): 1000
MAIN BODY: Cylinder; 2.4 m diameter by 2.8 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Unknown
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|-------------|-----------------|-------------------|
| DATE: | 26 Dec 1994 | LOCATION: | 51.6S, 307E (asc) |
| TIME: | 0627 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 1880 km | | |

PRE-EVENT ELEMENTS

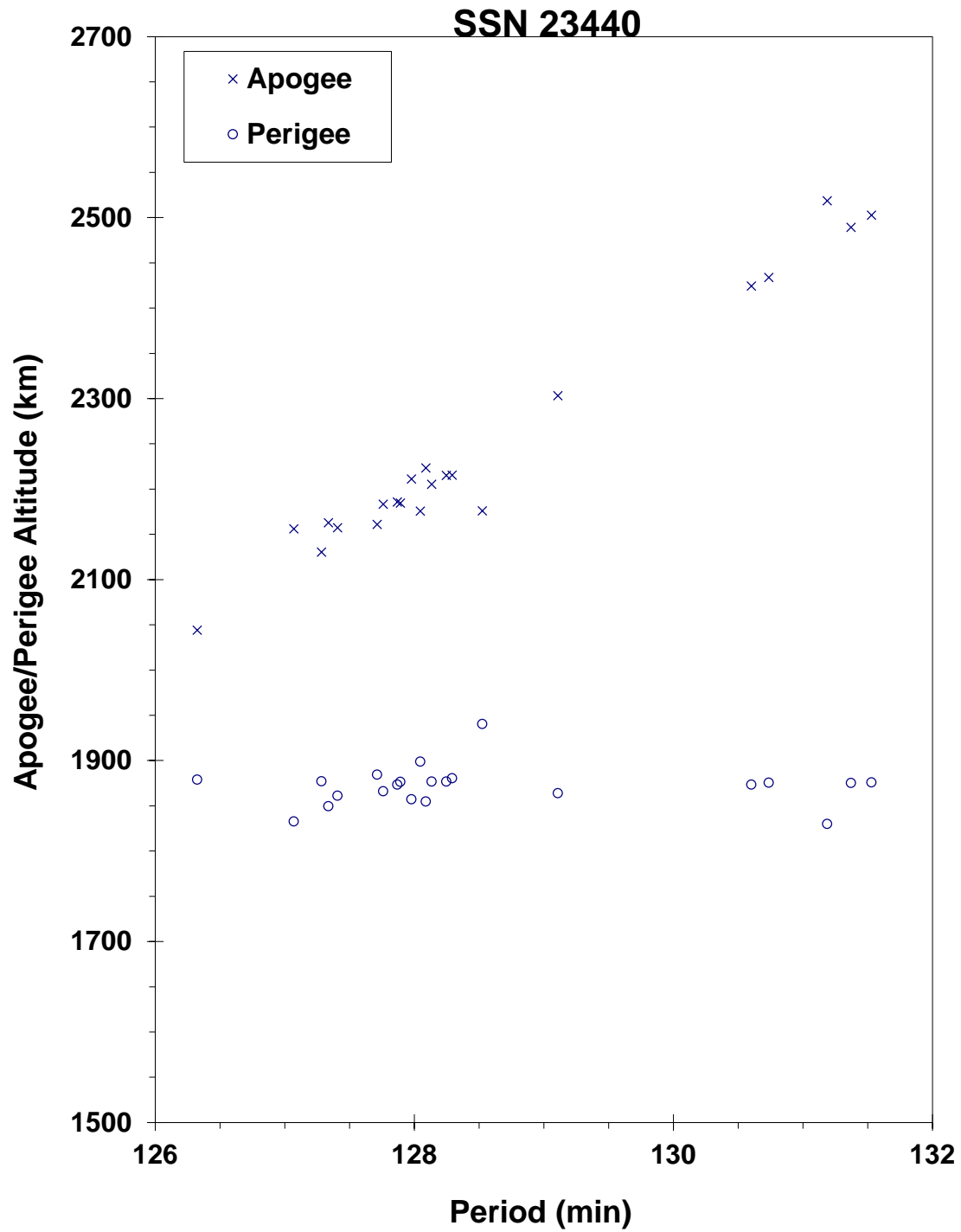
| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 94361.79150546 | MEAN ANOMALY: | 66.1014 |
| RIGHT ASCENSION: | 172.1572 | MEAN MOTION: | 11.27113018 |
| INCLINATION: | 64.8297 | MEAN MOTION DOT/2: | -0.00000043 |
| ECCENTRICITY: | 0.0188748 | MEAN MOTION DOT DOT/6: | 0.00000 |
| ARG. OF PERIGEE: | 292.0126 | BSTAR: | 0.000000 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 4.5 min
MAXIMUM ΔI : 0.2 deg

COMMENTS

Parent satellite was the Rokot third stage. The Rokot is an SS-19 ICBM based vehicle with the third stage being the *Briz-K*, retired after this launch for the improved *Briz-KM* stage. All three stages are fueled with UDMH/N204. NAVSPASUR observed 34 objects that were associated with this breakup, of which 26 entered the public satellite catalog.



Gabbard diagram for RS-15 R/B debris cloud of 23 fragments as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 24 Mar 1995
DRY MASS (KG): 767
MAIN BODY: cylinder; 1.0 m diameter x 3.6 m length
MAJOR APPENDAGES: solar panel
ATTITUDE CONTROL: three-axis
ENERGY SOURCES: on-board propellants and chemical batteries

EVENT DATA

DATE: 3.736 Feb 2015 LOCATION: 40.8S, 7.5E
TIME: 1740 GMT ASSESSED CAUSE: Battery
ALTITUDE: 860.7 km

PRE-EVENT ELEMENTS - unavailable

EPOCH: MEAN ANOMALY:
RIGHT ASCENSION: MEAN MOTION:
INCLINATION: MEAN MOTION DOT/2:
ECCENTRICITY: MEAN MOTION DOT DOT/6:
ARG. OF PERIGEE: BSTAR:

DEBRIS CLOUD DATA

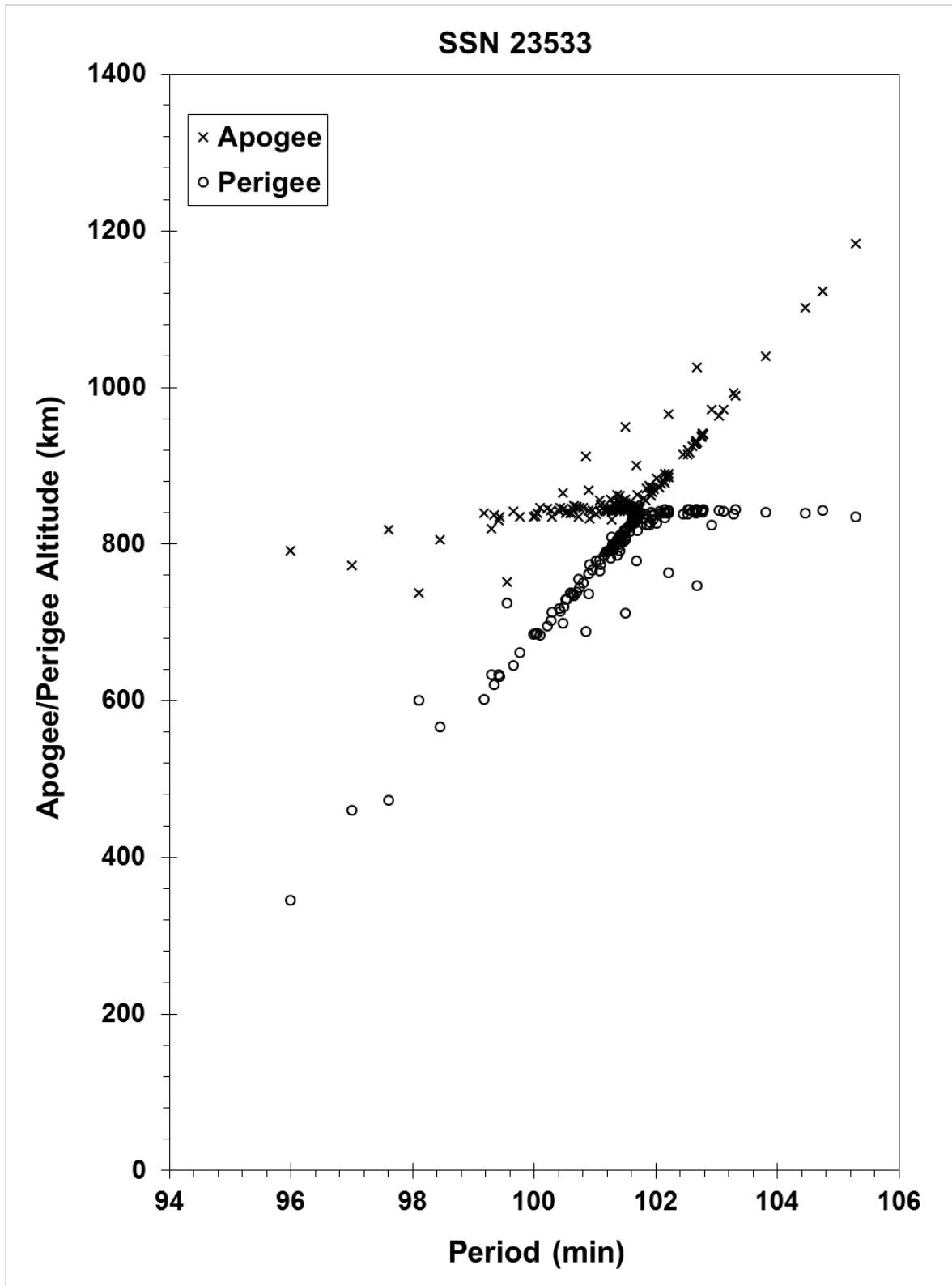
MAXIMUM ΔP : unknown min
MAXIMUM ΔI : unknown deg

COMMENTS

This is the second known breakup of a Defense Meteorological Satellite Program (DMSP) Block 5D-2 spacecraft, the first being DMSP 5D-2 F11 (USA 73). USA 109 was active when the event occurred and supporting telemetry was available for the inquiry. Analysis indicates that the event was due to an explosion of one of the Ni-Cd batteries.

REFERENCE DOCUMENT

“Recent Breakup of a DMSP Satellite”, The Orbital Debris Quarterly News, NASA JSC, April 2015.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv19i2.pdf>.



USA 109 debris cloud. Debris cataloged between 22 and 160 days after event.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 8.20 Jun 1995
DRY MASS (KG): 3000
MAIN BODY: Cylinder; 1.3 m diameter by 17 m length
MAJOR APPENDAGES: Solar arrays
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge?

EVENT DATA

| | | | |
|-----------|--------------|-----------------|--------------------|
| DATE: | 26 June 1997 | LOCATION: | 44 N, 173 E (asc.) |
| TIME: | 0257 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 285 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|--------------|
| EPOCH: | 97176.10173599 | MEAN ANOMALY: | 124.6445 |
| RIGHT ASCENSION: | 342.0749 | MEAN MOTION: | 16.02369895 |
| INCLINATION: | 65.0221 | MEAN MOTION DOT/2: | 0.00306537 |
| ECCENTRICITY: | 0.0084335 | MEAN MOTION DOT DOT/6: | 0.0000069339 |
| ARG. OF PERIGEE: | 234.6794 | BSTAR: | 0.00033322 |

DEBRIS CLOUD DATA

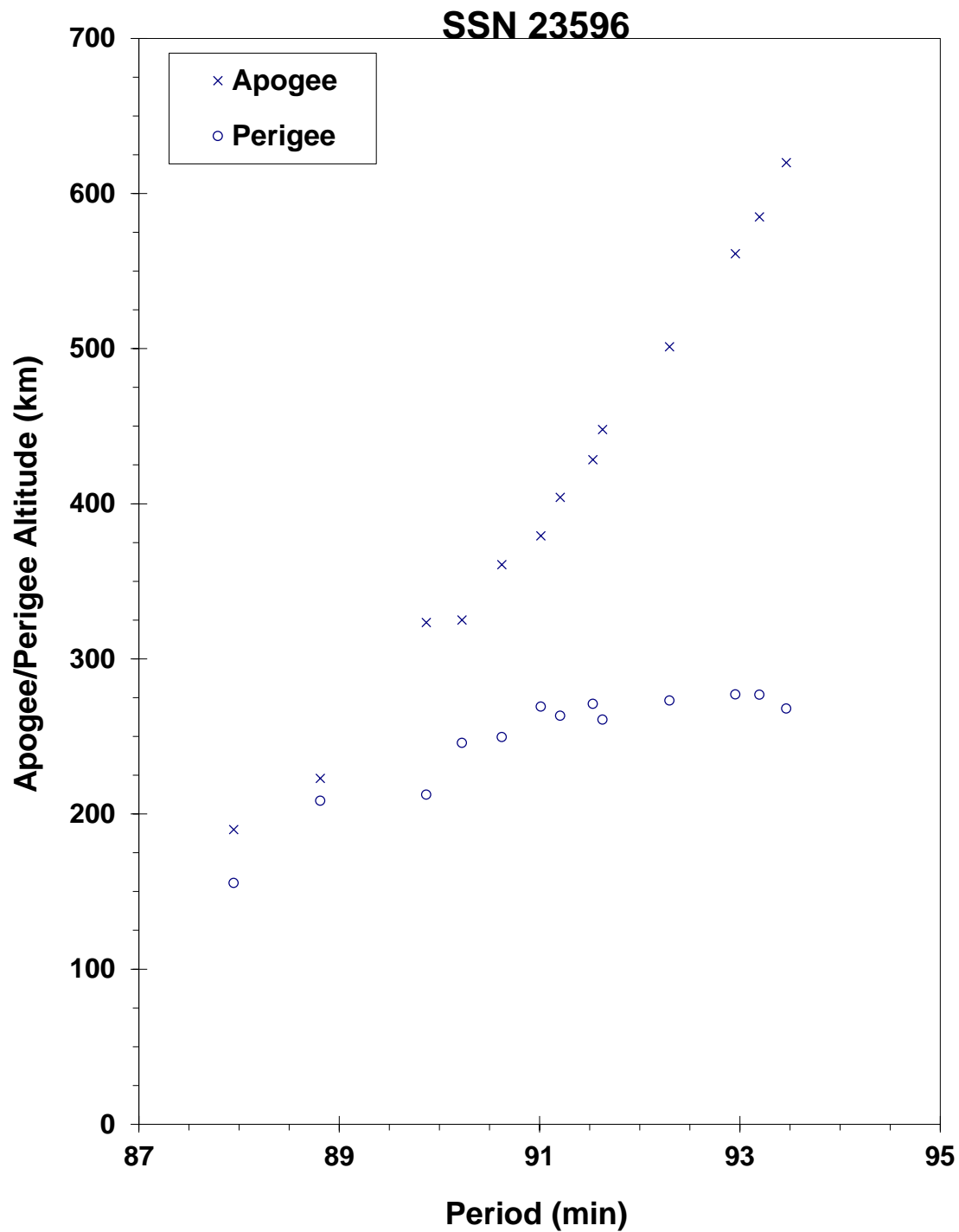
MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

Cosmos 2313 was the second spacecraft of its type to breakup since November 1987. Prior to the current event 17 spacecraft of this class (Cosmos 699) have experienced breakups in low Earth orbit. In the 1980's procedures were introduced to deplete remaining propellants at the end of mission, reducing orbital lifetime at the same time. Cosmos 2313 performed such a maneuver during 22-23 April 1997 and was close to reentry at the time of the event. Earlier spacecraft breakups resulted in up to 150 or more trackable debris. The cause of the event may well not be propellant related, but by reducing the orbital lifetime recent vehicles have decayed before the trigger mechanism could activate. At least 90 debris were detected after this event.

REFERENCE DOCUMENT

"Three Satellite Breakups During May-June", The Orbital Debris Quarterly News, NASA JSC, July 1997, p. 2. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv2i3.pdf>.



Cosmos 2313 debris cloud of 13 fragments 1 day to 2 weeks after the event as reconstructed from the US SSN database.

SATELLITE DATA

| | |
|-------------------|--|
| TYPE: | Payload |
| OWNER: | France |
| LAUNCH DATE: | 7.68 Jul 1995 |
| DRY MASS (KG): | 50 |
| MAIN BODY: | Box; 0.6 m by 0.3 m by 0.3 m |
| MAJOR APPENDAGES: | 6 m long gravity-gradient boom; solar panels |
| ATTITUDE CONTROL: | Gravity-gradient stabilized |
| ENERGY SOURCES: | Battery |

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------------|
| DATE: | 24 Jul 1996 | LOCATION: | 38 S, 60 E (asc) |
| TIME: | 0948 GMT | ASSESSED CAUSE: | Collision |
| ALTITUDE: | 685 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 96205.39273562 | MEAN ANOMALY: | 292.8048 |
| RIGHT ASCENSION: | 141.7519 | MEAN MOTION: | 14.67264268 |
| INCLINATION: | 98.1025 | MEAN MOTION DOT/2: | 0.00000083 |
| ECCENTRICITY: | 0.0008991 | MEAN MOTION DOT DOT/6: | 0 |
| ARG. OF PERIGEE: | 67.4104 | BSTAR: | 0.000023247 |

DEBRIS CLOUD DATA

| | |
|----------------------|-------------|
| MAXIMUM ΔP : | unknown min |
| MAXIMUM ΔI : | unknown deg |

COMMENTS

The incident marked the first time that two objects in the U.S. satellite catalog are known to have accidentally run into one another. The CERISE spacecraft (Satellite Number 23606, International Designator 1995-033B) is a microsatellite of British design. The other participant in the encounter was Satellite Number 18208 (International Designator 1986-019RF), which was generated in November 1986, when ESA's SPOT 1 rocket body broke up into nearly 500 tracked debris. The orbit of this fragmentation debris at the time of the collision was 660 km by 680 km at an inclination of 98.45 degrees. The collision, which occurred with a relative velocity of 14.8 km/s, produced only a single piece of debris large enough to be tracked, i.e., the upper portion of the gravity-gradient boom. Analysis of the manufacturer of the spacecraft bus, Surrey Satellite Technology Ctd. at the University of Surrey, United Kingdom suggested that the 6 m, gravity-gradient boom had been severed at 3.1-3.2 meter from its base.

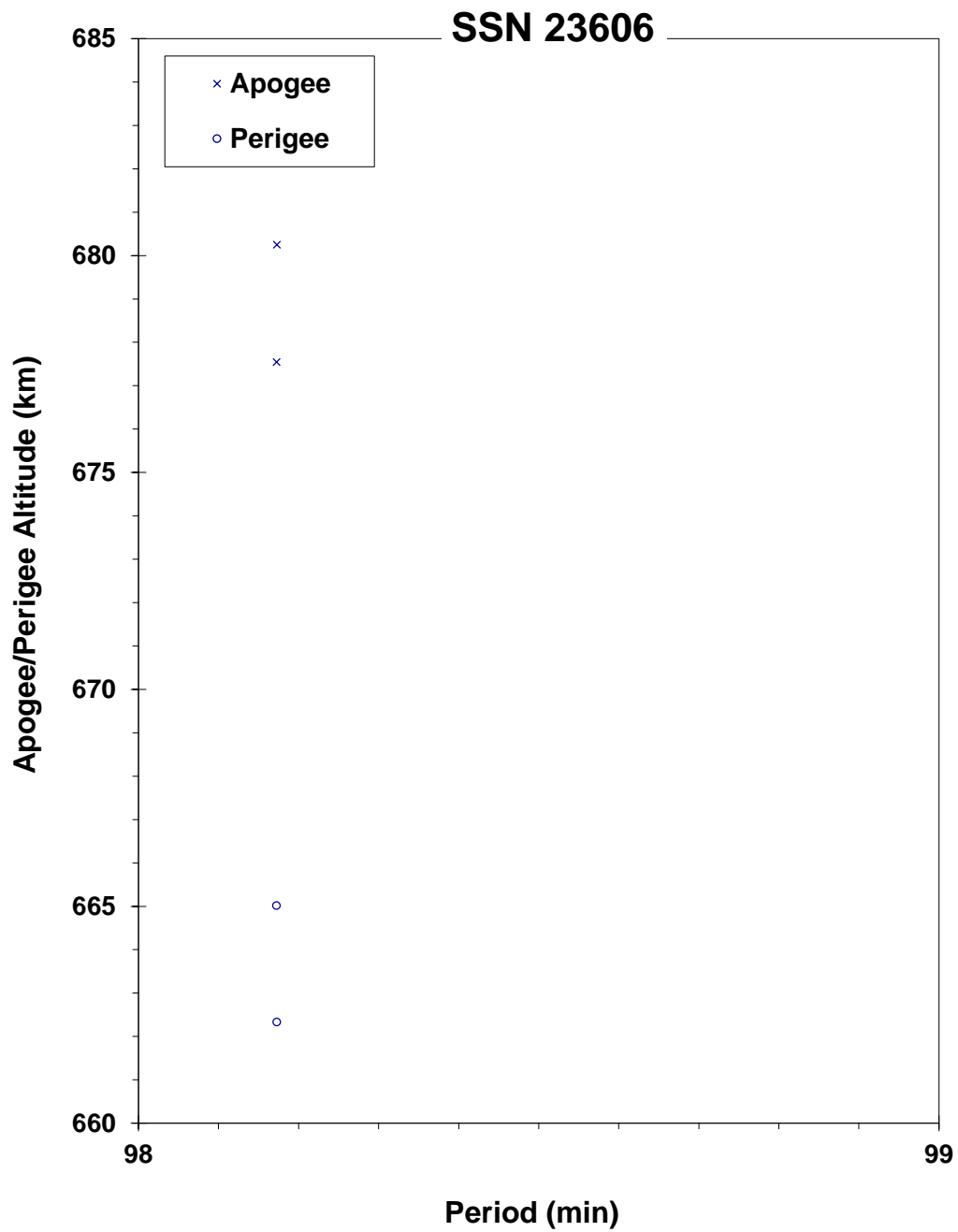
Using USAF Space Command's COMBO (Computation of Miss Between Orbits) program, a close approach of less than 1 km between Satellite 23606 and Satellite 18208 was determined by NASA JSC to have taken place at 0948 GMT on 24 July over the southern Indian Ocean. Naval Space Operations Center (NAVSPOC) at Dahlgren, Virginia, replicated the NASA findings and, using direct observational data and special perturbation theory, was able to refine the miss distance uncertainty to within 137 m. In addition, NAVSPOC identified a minor perturbation in the orbit of Satellite 18208 that occurred about the time of the event.

REFERENCE DOCUMENTS

"First Natural Collision of Cataloged Earth Satellites", N. Johnson, The Orbital Debris Quarterly News, NASA JSC, September 1996, p. 1. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNV1i2.pdf>.

"Collision of CERISE with Space Debris", F. Alby et al, Proceedings of the Second European Conference on Space Debris, SP-393, p. 589-596.

"First 'Confirmed' Natural Collision Between Two Cataloged Satellites", T. Payne, Proceedings of the Second European Conference on Space Debris, SP-393, p. 597-600.



**CERISE debris cloud of 2 fragments 4 days after the event as
reconstructed from the US SSN database.**

SATELLITE DATA

TYPE: Proton Block DM SOZ Ullage Motor
OWNER: CIS
LAUNCH DATE: 24.66 Jul 1995
DRY MASS (KG): ~55 kg
MAIN BODY: ~0.6 m by 0.6 m by 1.0 m
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants?

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 21 Nov 2000 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|---------------|
| EPOCH: | 00324.99357911 | MEAN ANOMALY: | 90.3648 |
| RIGHT ASCENSION: | 200.0539 | MEAN MOTION: | 4.50149006 |
| INCLINATION: | 64.4375 | MEAN MOTION DOT/2: | 0.00164632 |
| ECCENTRICITY: | 0.5787543 | MEAN MOTION DOT DOT/6: | 0.00000030156 |
| ARG. OF PERIGEE: | 213.7574 | BSTAR: | 0.00048999 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

This is the 4th breakup of 2000 and the 23rd breakup of a Proton SOZ motor (see Orbital Debris Quarterly Newsletter V, Issue 4, p. 2.) This object is associated with the 24 July 1995 launch (1995-037) of the Cosmos 2316-2318 satellites. These members of the GLONASS series are equivalent to GPS/Navstar satellites and reside in middle Earth orbit. This object was one of two pieces left in the transfer orbit and is assessed to be one of the SOZ ullage/orientation motor units. As of November 21, 2000, this object had been on orbit 5 years and 121 days.

REFERENCE DOCUMENT

“SOZ Ullage Motor Breakup”, The Orbital Debris Quarterly News, NASA JSC, January 2001. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/odqnv6i1.pdf>.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Proton Blok DM Fourth Stage
OWNER: CIS
LAUNCH DATE: 19.36 Feb 1996
DRY MASS (KG): 3400 (?)
MAIN BODY: Cylinder; 3.7 m diameter by 6.3 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, pressurants, and batteries

EVENT DATA

| | | | |
|-----------|-------------|-----------------|---------------------|
| DATE: | 19 Feb 1996 | LOCATION: | 0.2 N, 88.8 E (dsc) |
| TIME: | 1459 GMT | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | 36511 km | | |

POST-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 96058.46760248 | MEAN ANOMALY: | 359.9314 |
| RIGHT ASCENSION: | 280.4138 | MEAN MOTION: | 2.23172282 |
| INCLINATION: | 48.7 | MEAN MOTION DOT/2: | 0.0002158 |
| ECCENTRICITY: | 0.7321111 | MEAN MOTION DOT DOT/6: | 0.0000 |
| ARG. OF PERIGEE: | 1.7779 | BSTAR: | 0.00068491 |

DEBRIS CLOUD DATA

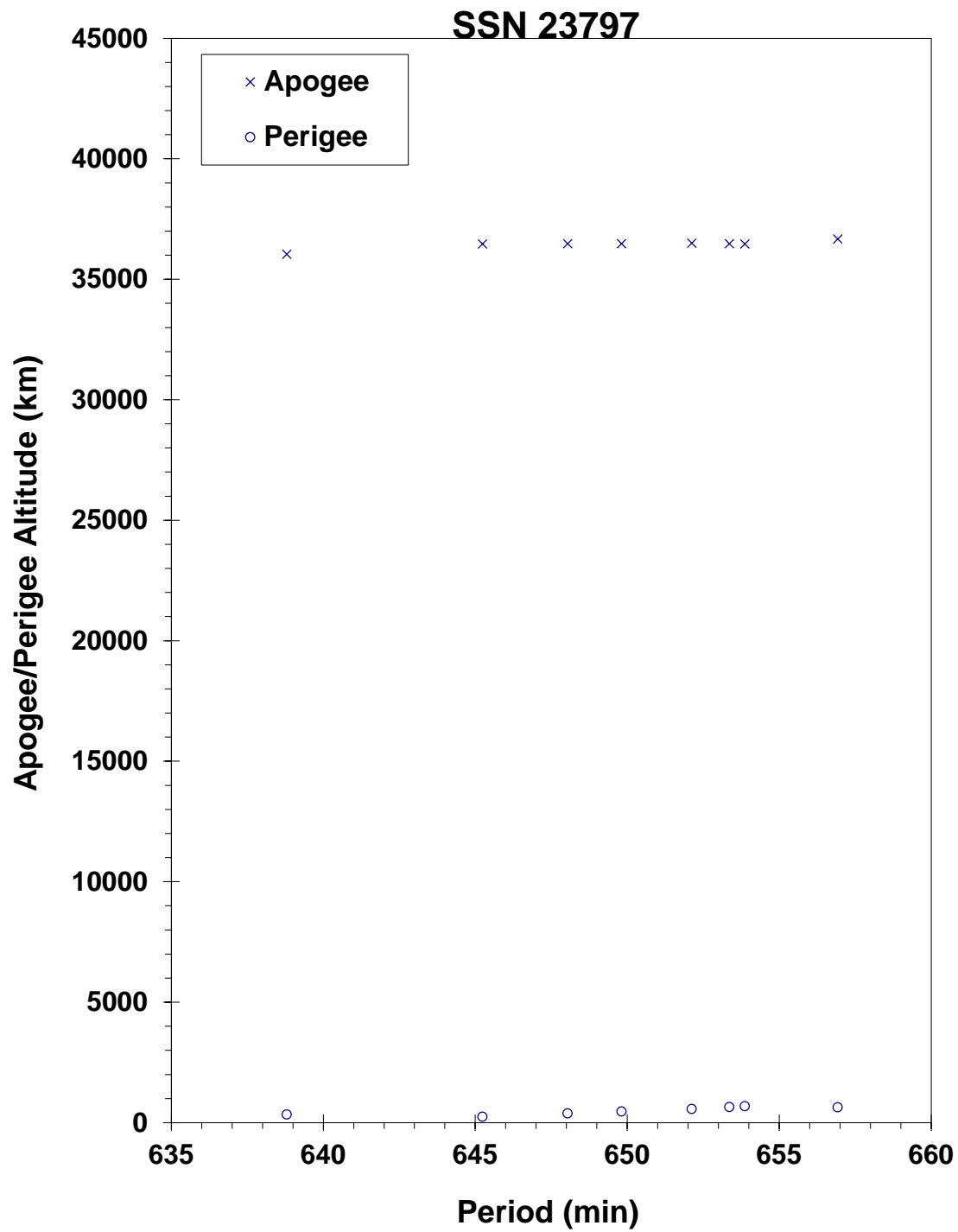
MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

The first burn of this stage was successful and indicates that 23797 was in a transfer orbit. However, prior to the first pass through the NAVSPOC fence, 23797 fragmented. Twenty (20) pieces were observed during this first pass. During a subsequent pass, 196 pieces were observed that were associated with the upperstage. Stage apparently broke up after main engine restart for GEO apogee maneuver.

REFERENCE DOCUMENT

"Satellite Fragmentations in 1996", N. Johnson, The Orbital Debris Quarterly News, NASA JSC, January 1997, p. 1. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNV2i1.pdf>.



Gabbard diagram from Raduga 33 from NAVSPOC elements.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 25.09 May 1996
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Unknown
ENERGY SOURCES: On-board Propellants

EVENT DATA

| | | | |
|-----------|--------------|-----------------|------------|
| DATE: | ~13 Dec 1999 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|--------------|
| EPOCH: | 99347.02294368 | MEAN ANOMALY: | 314.8549 |
| RIGHT ASCENSION: | 194.3249 | MEAN MOTION: | 9.75630550 |
| INCLINATION: | 46.4558 | MEAN MOTION DOT/2: | 0.00969995 |
| ECCENTRICITY: | 0.2950283 | MEAN MOTION DOT DOT/6: | 0.0000015400 |
| ARG. OF PERIGEE: | 75.9037 | BSTAR: | 0.00055450 |

CATALOGED DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

This is 21st breakup event for an object of this class, and the third in 1999. The breakups are believed to be caused by residual propellants. Russian officials have been aware of the problem since 1992 and have made design changes, although the date of full implementation is unknown. This is the most recently launched object to breakup (age = ~3.5 years). The environmental consequence of the breakup was short-lived; the object was in catastrophic decay from a geosynchronous transfer orbit.

REFERENCE DOCUMENTS

The Fragmentation of Proton Debris, D. J. Nauer, TBE Technical Report CS93-LKD-004, Teledyne Brown Engineering, Colorado Springs, 31 December 1992.

Analysis of Fragmentations From December 1992 - February 1993, TBE Technical Report CS93-LKD-010, Teledyne Brown Engineering, Colorado Springs, 30 March 1993.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 15.51 May 1997
DRY MASS (KG): 6000
MAIN BODY: Cylinder; 2.4 m diameter by 7 m length
MAJOR APPENDAGES: Solar arrays
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------------------|
| DATE: | 16 Sep 1997 | LOCATION: | 58.2 N, 157.5 E (asc.) |
| TIME: | 2208 GMT | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | 230 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 97258.16080604 | MEAN ANOMALY: | 247.0345 |
| RIGHT ASCENSION: | 1.1478 | MEAN MOTION: | 16.06645410 |
| INCLINATION: | 64.8485 | MEAN MOTION DOT/2: | 0.00206295 |
| ECCENTRICITY: | 0.0048612 | MEAN MOTION DOT DOT/6: | 0.000026376 |
| ARG. OF PERIGEE: | 113.5945 | BSTAR: | 0.00022999 |

DEBRIS CLOUD DATA

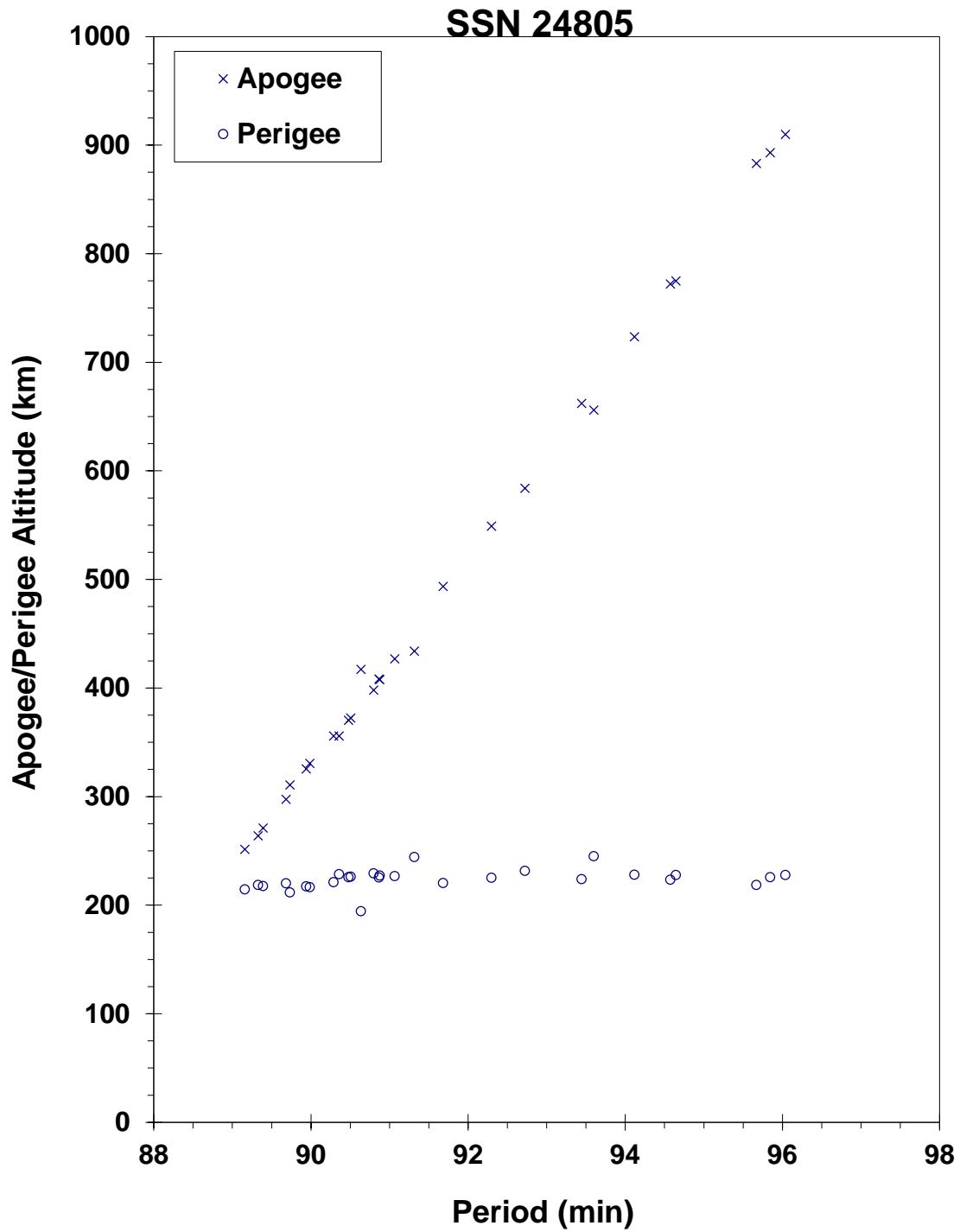
MAXIMUM ΔP : 7.3 min
MAXIMUM ΔI : 0.9 deg

COMMENTS

Cosmos 2343 was the sixth of the Cosmos 2031 class of spacecraft that debuted in 1989 but was not flown since 1993. In all five previous missions (1989-1993), the spacecraft was deliberately exploded at the end of mission. Previous missions of this type include Cosmos 2031, Cosmos 2101, Cosmos 2163, Cosmos 2225, and Cosmos 2262. All such events have occurred over Eastern Russia. This event, as with three of the previous events, occurred over the Kamchatka Peninsula. Highest previous piece count for large debris for this class of vehicle was 180, although more were probably created. Due to the low altitude of the breakup, the debris were short-lived.

REFERENCE DOCUMENT

"International LEO Spacecraft Breakup in September", N. Johnson, [The Orbital Debris Quarterly News](https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv2i4.pdf), NASA JSC, October 1997, p. 2. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv2i4.pdf>.



Cosmos 2343 debris cloud of 28 fragments as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: USA
LAUNCH DATE: 14 Sep 1997
DRY MASS (KG): 556
MAIN BODY: 3.6 m long triangular prism
MAJOR APPENDAGES: single-axis solar panels (2); communication antennas (3)
ATTITUDE CONTROL: three-axis stabilized
ENERGY SOURCES: on-board hydrazine monopropellant; NiH₂ batteries

EVENT DATA

| | | | |
|-----------|-------------|-----------------|--------------------|
| DATE: | 10 Feb 2009 | LOCATION: | 72.5N, 97.9E (asc) |
| TIME: | 1656 GMT | ASSESSED CAUSE: | Collision |
| ALTITUDE: | 788.597 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 09040.36587205 | MEAN ANOMALY: | 274.2112 |
| RIGHT ASCENSION: | 121.8779 | MEAN MOTION: | 14.34219614 |
| INCLINATION: | 86.3996 | MEAN MOTION DOT/2: | 0.00000121 |
| ECCENTRICITY: | 0.0002273 | MEAN MOTION DOT DOT/6: | 0.0 |
| ARG. OF PERIGEE: | 85.9342 | BSTAR: | 0.000036194 |

DEBRIS CLOUD DATA

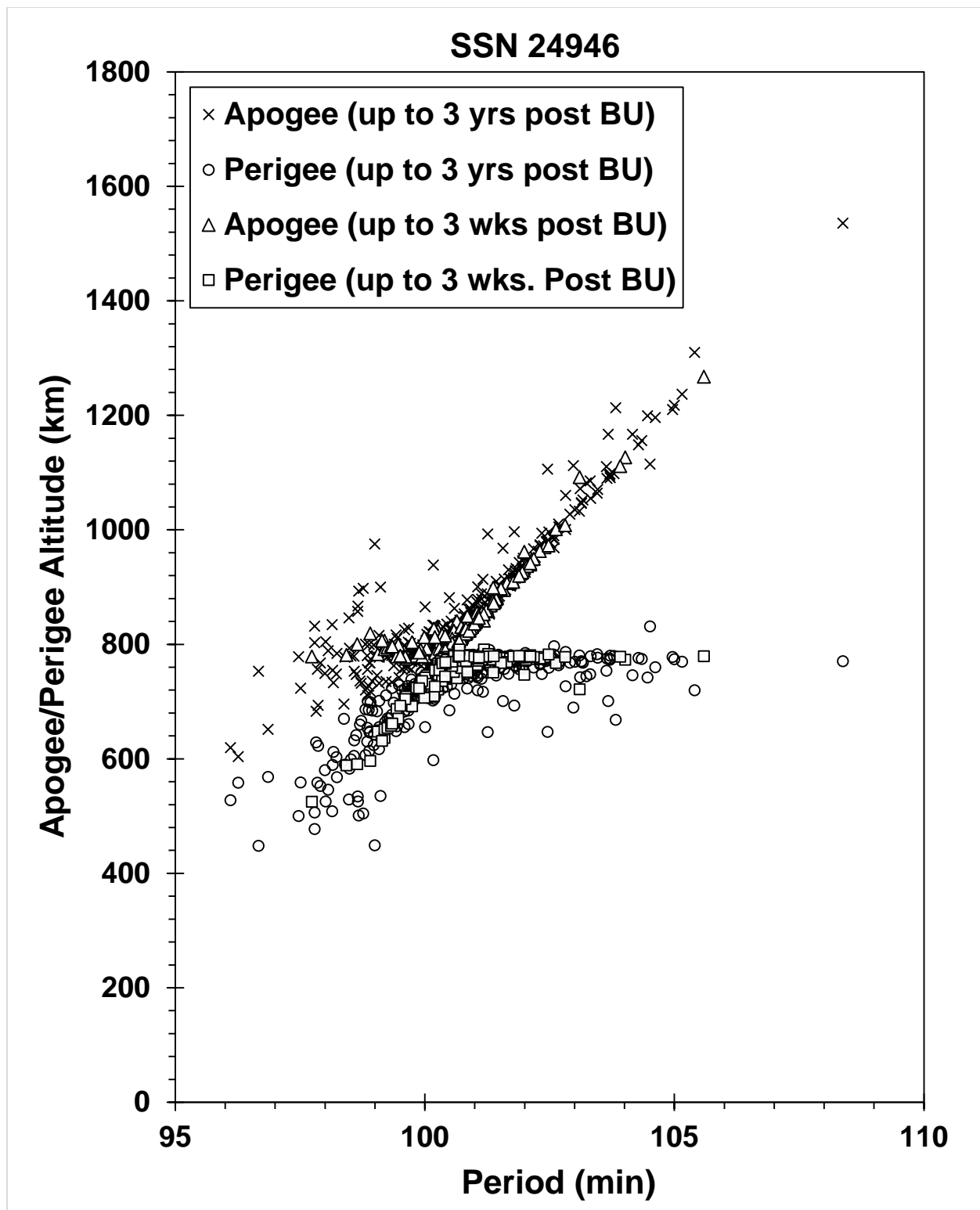
MAXIMUM ΔP : 8.0 min
MAXIMUM ΔI : 0.8 deg

COMMENTS

Iridium 33 was a member of the first generation Iridium low-altitude communication satellite constellation; it utilized a Lockheed Martin LM700A bus and was active at the time of the event. Iridium 33 collided with Cosmos 2251 [q.v.] in the first, accidental collision of large, intact resident space objects.

REFERENCE DOCUMENT

“Satellite Collision Leaves Significant Debris Clouds”, The Orbital Debris Quarterly News, NASA JSC, April 2009.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/ODQNv13i2.pdf>.



The IRIDIUM 33 debris cloud, including 621 fragments cataloged up to 3 years after the event.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 12.71 Nov 1997
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellant

EVENT DATA

DATE: 14 Feb 2007 LOCATION: Unknown
TIME: Unknown ASSESSED CAUSE: Propulsion
ALTITUDE: Unknown

PRE-EVENT ELEMENTS

EPOCH: 07044.95683864 MEAN ANOMALY: 35.6246
RIGHT ASCENSION: 14.2124 MEAN MOTION: 5.48131368
INCLINATION: 46.649 MEAN MOTION DOT/2: 0.00001445
ECCENTRICITY: 0.5113669 MEAN MOTION DOT DOT/6: 0.0
ARG. OF PERIGEE: 267.6957 BSTAR: 0.00013146

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

The object was in a decaying geosynchronous transfer orbit; this event marks the 36th known breakup of a Proton Blok DM SOZ ullage motor since 1984. By accident, an observer in Finland captured about 20 of the debris with two CCD cameras less than 24 hours after the event. The debris would have been too faint to be seen with the naked eye. An estimated 60 debris were detected by the US SSN.

REFERENCE DOCUMENTS

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

“Four Satellite Breakups in February Add to Debris Population”, The Orbital Debris Quarterly News, NASA JSC, April 2007. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv11i2.pdf>.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 9.30 Dec 1997
DRY MASS (KG): 3000
MAIN BODY: 1.3 m diameter by 17 m length, plus solar arrays
MAJOR APPENDAGES: Solar arrays
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge?

EVENT DATA

| | | | |
|-----------|-------------|-----------------|-------------------|
| DATE: | 22 Nov 1999 | LOCATION: | 31.6N, 4.3E (dsc) |
| TIME: | 0440 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 370 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 99325.85267585 | MEAN ANOMALY: | 85.1293 |
| RIGHT ASCENSION: | 332.8746 | MEAN MOTION: | 15.83563975 |
| INCLINATION: | 65.0115 | MEAN MOTION DOT/2: | 0.00295116 |
| ECCENTRICITY: | 0.0134056 | MEAN MOTION DOT DOT/6: | 0.000036131 |
| ARG. OF PERIGEE: | 273.4567 | BSTAR: | 0.00065869 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 2.834 min
MAXIMUM ΔI : 0.22 deg

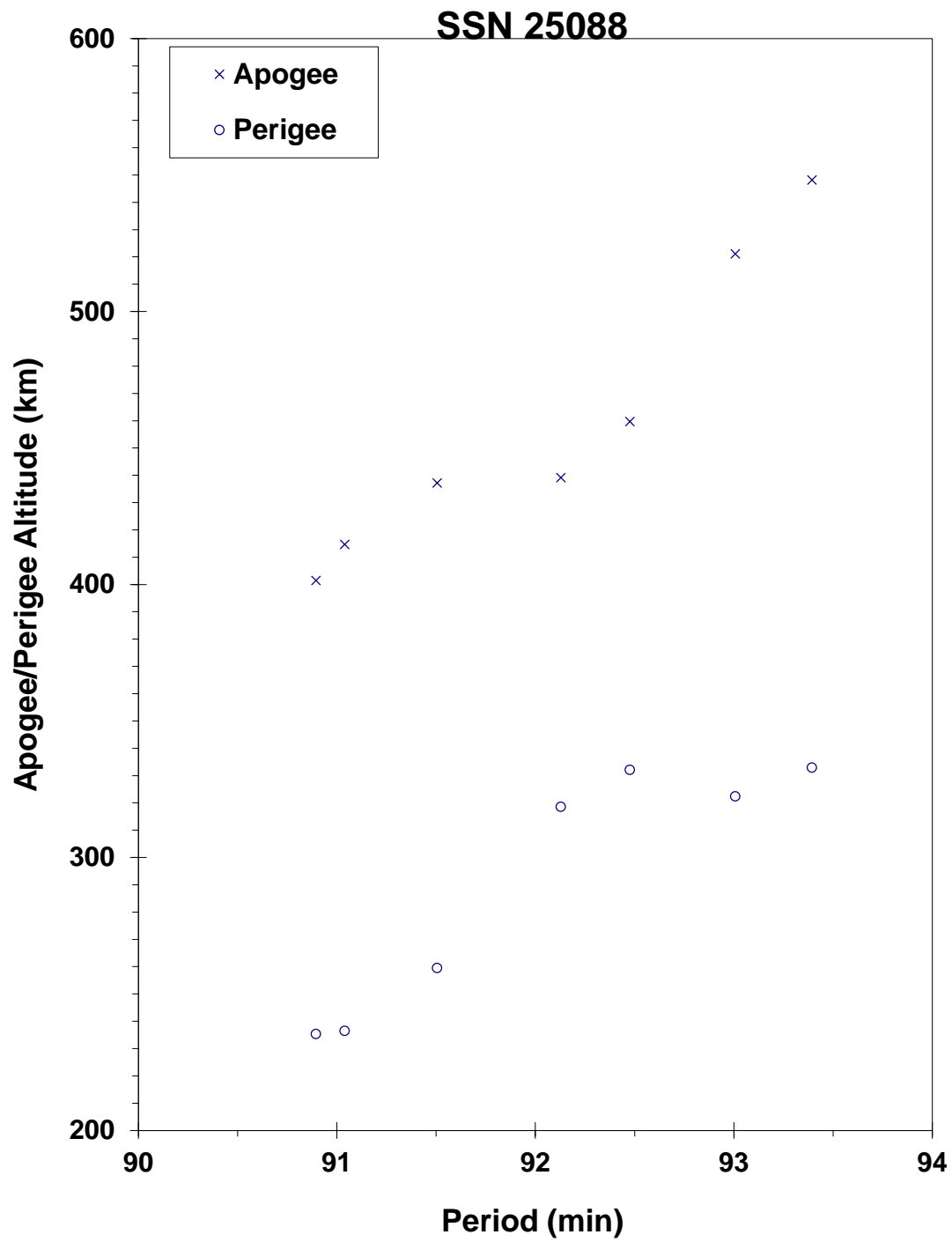
COMMENTS

Cosmos 2347 was the 19th spacecraft of this type known to have experienced a major fragmentation. Such events were common prior to 1988, but only three breakups have occurred during the past 12 years: Cosmos 2347, Cosmos 2238 (1 Dec 1994), and Cosmos 2313 (26 June 1997). In this case Cosmos 2347 had performed a standard end-of-mission maneuver on 19 November 1999, a little more than 2 days before the breakup. Extensive analyses of these events have been conducted, although the cause is still unknown in the open literature.

A second breakup of Cosmos 2347 was discovered on 10 December when the spacecraft's orbit had decayed to 175 km by 250 km. Three dozen new debris were detected after the second event, but the very low altitude made it difficult to assess accurately the number of large debris. Prior spacecraft (especially Cosmos 1220, 1260, and 1306) also experienced multiple fragmentations.

REFERENCE DOCUMENT

"Satellite Breakups Increase in Last Quarter of 1999", The Orbital Debris Quarterly News, NASA JSC, January 2000. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv5i1.pdf>.



Cosmos 2347 debris cloud of 7 fragments within 1 day of the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Proton Blok DM Fourth Stage
OWNER: CIS
LAUNCH DATE: 24.97 Dec 1997
DRY MASS (KG): 3400 (?)
MAIN BODY: Cylinder; 3.7 m diameter by 6.3 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|-------------|-----------------|---------------------|
| DATE: | 25 Dec 1997 | LOCATION: | 0.3 S, 91.2 E (dsc) |
| TIME: | 0550 GMT | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | 35995 km | | |

POST-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 97359.90803672 | MEAN ANOMALY: | 359.8589 |
| RIGHT ASCENSION: | 92.0594 | MEAN MOTION: | 2.26580509 |
| INCLINATION: | 51.4499 | MEAN MOTION DOT/2: | -0.00000507 |
| ECCENTRICITY: | 0.7304004 | MEAN MOTION DOT DOT/6: | 0 |
| ARG. OF PERIGEE: | 1.0181 | BSTAR: | 0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

The commercial Asiasat 3 spacecraft was launched by a Proton launch vehicle at 2319 GMT, 24 December 1997. The fourth stage completed its first burn successfully about 0035 GMT, 25 December, placing the R/B-S/C combination into a GTO. When the vehicle reached its first apogee, the main engine was restarted but shutdown within 1 second, apparently due to a catastrophic failure. The US Space Surveillance Network detected less than 10 objects, and by 9 January only 1-2 were still being observed. The fragmentation is similar to the breakup of the Raduga 33 upper stage on 19 Feb 1996. In that case, nearly 200 debris were detected by the SSN.

REFERENCE DOCUMENT

“Recent Satellite Fragmentation Investigations”, N. Johnson, The Orbital Debris Quarterly News, January 1998, p. 3. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv3i1.pdf>.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: H-II Second Stage
OWNER: Japan
LAUNCH DATE: 21.33 Feb 1998
DRY MASS (KG): 3000
MAIN BODY: Spheroid + cylinder + cone; 4 m diameter by 10.1 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 21 Feb 1998 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

POST-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|--------------|
| EPOCH: | 98054.59975400 | MEAN ANOMALY: | 162.0601 |
| RIGHT ASCENSION: | 294.3031 | MEAN MOTION: | 13.51967368 |
| INCLINATION: | 30.0458 | MEAN MOTION DOT/2: | 0.0002873 |
| ECCENTRICITY: | 0.1097485 | MEAN MOTION DOT DOT/6: | -0.000003104 |
| ARG. OF PERIGEE: | 194.5714 | BSTAR: | 0.00029603 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

According to a NASDA report, a welding failure caused the LE-5A engine shutdown that stranded the Japanese COMETS satellite (1998-011A) in an elliptical orbit. The failure occurred 47 seconds into an apogee-raising maneuver. The report determined from telemetry data that hot combustion gases managed to penetrate special welding, called brazing, between nickel alloy cooling tubes in the lowest part of the combustion chamber near the top of the engine's nozzle skirt. Burning through the tubes, combustion gases quickly caused a fire, which triggered the engine shutdown. The report concludes the accident was caused by a manufacturing flaw and not a fundamental design problem. At least three dozen debris were detected by optical sensors in Hawaii.

REFERENCE DOCUMENTS

Kallender, P., "LE-5A Shutdown Blamed on Welding." Space News, 23-29 March 1998, p. 6.

"The Upper Stage Breakups in One Week Top February Debris Activity", The Orbital Debris Quarterly News, NASA JSC, April 1998, p. 1. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv3i2.pdf>.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Delta II Second Stage Rocket Body
OWNER: US
LAUNCH DATE: 23 Feb 1999
DRY MASS (KG): 921
MAIN BODY: 2.4 m diameter x 5 m long right circular cylinder
MAJOR APPENDAGES: none
ATTITUDE CONTROL: inactive 3-axis
ENERGY SOURCES: none at time of event

EVENT DATA

DATE: 28 April 2014
TIME: unknown
ALTITUDE: unknown
LOCATION: unknown
ASSESSED CAUSE: Unknown

PRE-EVENT ELEMENTS

EPOCH: 14117.05656630
RIGHT ASCENSION: 271.0573
INCLINATION: 96.4621
ECCENTRICITY: 0.0143136
ARG. OF PERIGEE: 329.6124
MEAN ANOMALY: 143.8191
MEAN MOTION: 14.45733299
MEAN MOTION DOT/2: 0.00000245
MEAN MOTION DOT DOT/6: 0.00000
BSTAR: 0.00069149

DEBRIS CLOUD DATA

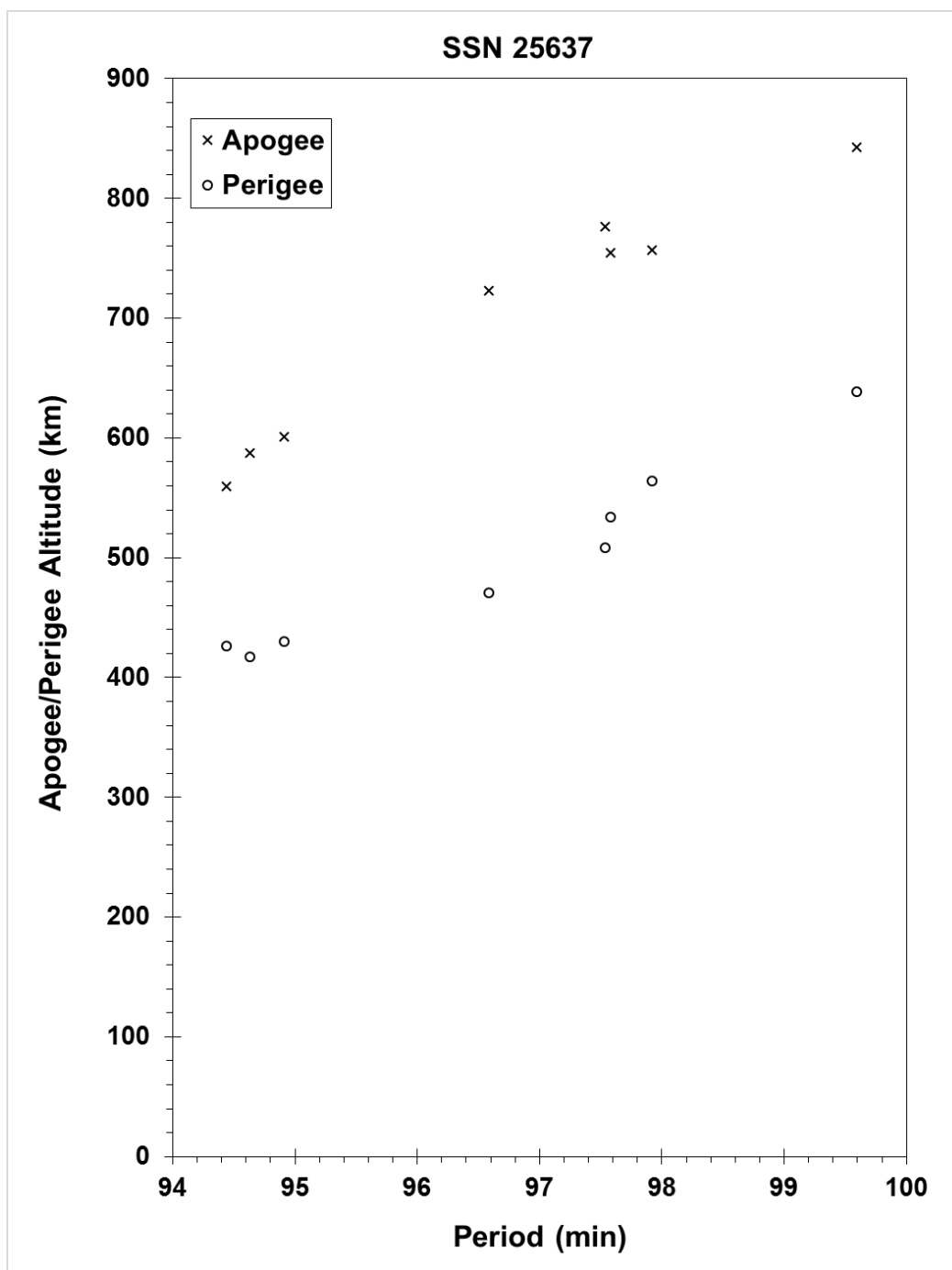
MAXIMUM ΔP : 5.2 minutes
MAXIMUM ΔI : 0.2°

COMMENTS

The cause of this event, occurring over 15 years after launch, is unknown. Seven fragments entered the catalog but all decayed from orbit by 16 May 2014. Since 1982, Delta II second stage rocket bodies have been fully passivated, implying that a small MMOD impact may have caused this event.

REFERENCE DOCUMENT

“Flurry of Small Breakups in First Half of 2014”, The Orbital Debris Quarterly News, NASA JSC, July 2014.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv18i3.pdf>.



ARGOS/ORSTED/SUNSAT R/B debris cloud.

SATELLITE DATA

TYPE: Payload
OWNER: PRC
LAUNCH DATE: 10.06 May 1999
DRY MASS (KG): 950
MAIN BODY: Box; 1.5 m by 1.5 m by 1.5 m
MAJOR APPENDAGES: Solar Panels, 1.5 m by 4 m
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants

EVENT DATA

DATE: 11 Jan 2007
TIME: 2226 GMT
ALTITUDE: 860 km
LOCATION: 35N, 100E (asc)
ASSESSED CAUSE: Deliberate (HVI)

PRE-EVENT ELEMENTS

EPOCH: 07011.90621003
RIGHT ASCENSION: 1.7411
INCLINATION: 98.6464
ECCENTRICITY: 0.0013513
ARG. OF PERIGEE: 266.0357
MEAN ANOMALY: 94.0215
MEAN MOTION: 14.11820274
MEAN MOTION DOT/2: 0.00000180
MEAN MOTION DOT DOT/6: 0.0
BSTAR: 0.00012153

DEBRIS CLOUD DATA

MAXIMUM ΔP : 33.4 min
MAXIMUM ΔI : 5.0 deg

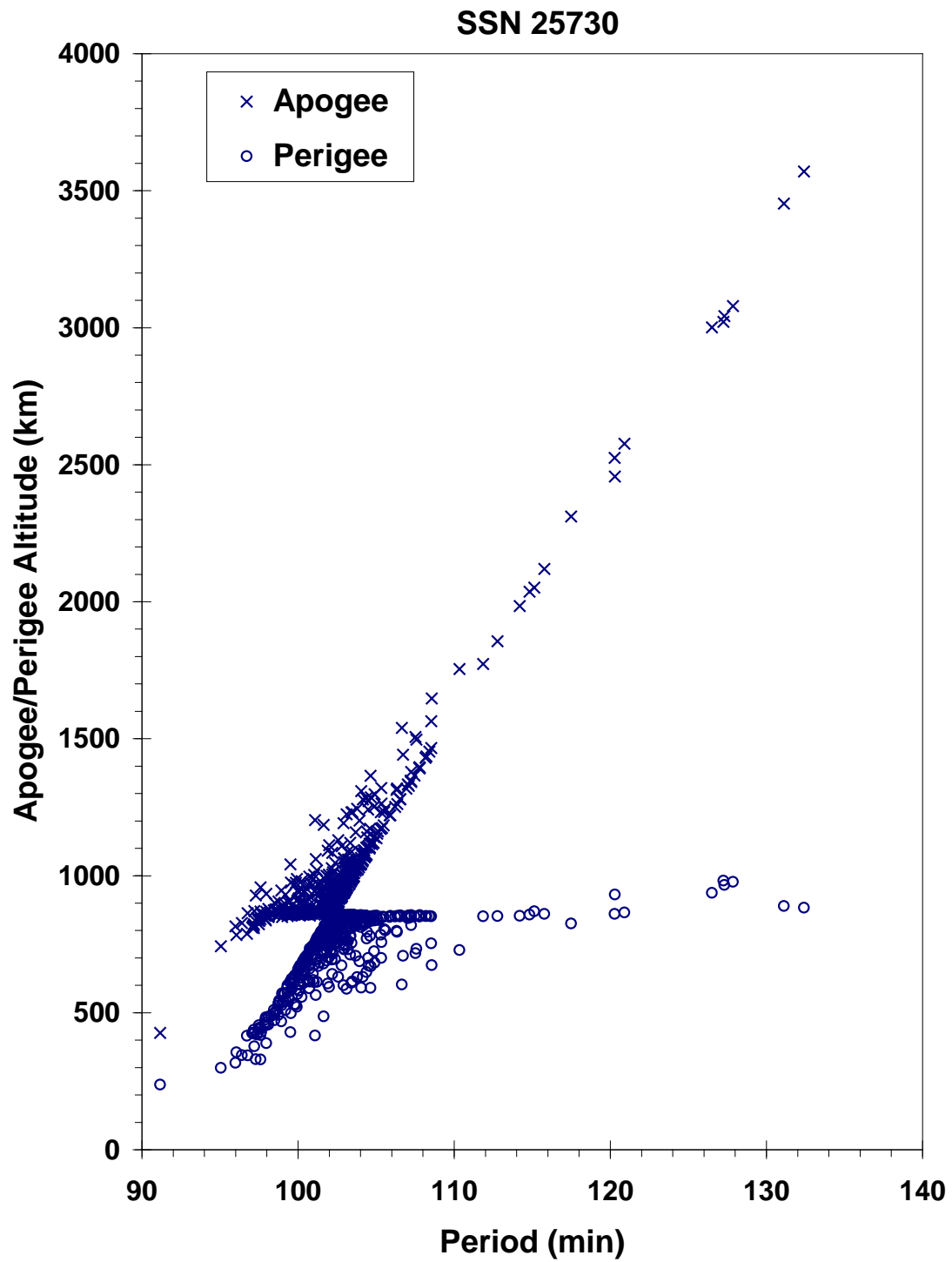
COMMENTS

The debris cloud created by this anti-satellite test represents the worst contamination of low Earth orbit in history. More than half the identified debris were thrown into orbits exceeding a mean altitude of 850 km, meaning that much of the 10 cm and larger debris will be in orbit for decades or centuries.

REFERENCE DOCUMENT

“Chinese Anti-satellite Test Creates Most Severe Orbital Debris Cloud in History”, The Orbital Debris Quarterly News, NASA JSC, April 2007.

Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv11i2.pdf>.



Fengyun 1C debris cloud remnant of 2000+ cataloged fragments 6 months after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: PRC/Brazil
LAUNCH DATE: 14.14 Oct 1999
DRY MASS (KG): 1450
MAIN BODY: Box: 1.8 m by 2.0 m by 2.2 m
MAJOR APPENDAGES: 6.3 m x 2.6 m Solar Panels
ATTITUDE CONTROL: None at time of event
ENERGY SOURCES: On-board propellant?

EVENT DATA

| | | | |
|-----------|-------------|-----------------|-----------------|
| DATE: | 18 Feb 2007 | LOCATION: | 35N, 128E (asc) |
| TIME: | 1107 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 780 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|--------------|
| EPOCH: | 07049.17726620 | MEAN ANOMALY: | 250.9413 |
| RIGHT ASCENSION: | 88.9135 | MEAN MOTION: | 14.34483847 |
| INCLINATION: | 98.2175 | MEAN MOTION DOT/2: | -0.00000185 |
| ECCENTRICITY: | 0.0007520 | MEAN MOTION DOT DOT/6: | 0.0 |
| ARG. OF PERIGEE: | 109.2997 | BSTAR: | -0.000051172 |

DEBRIS CLOUD DATA

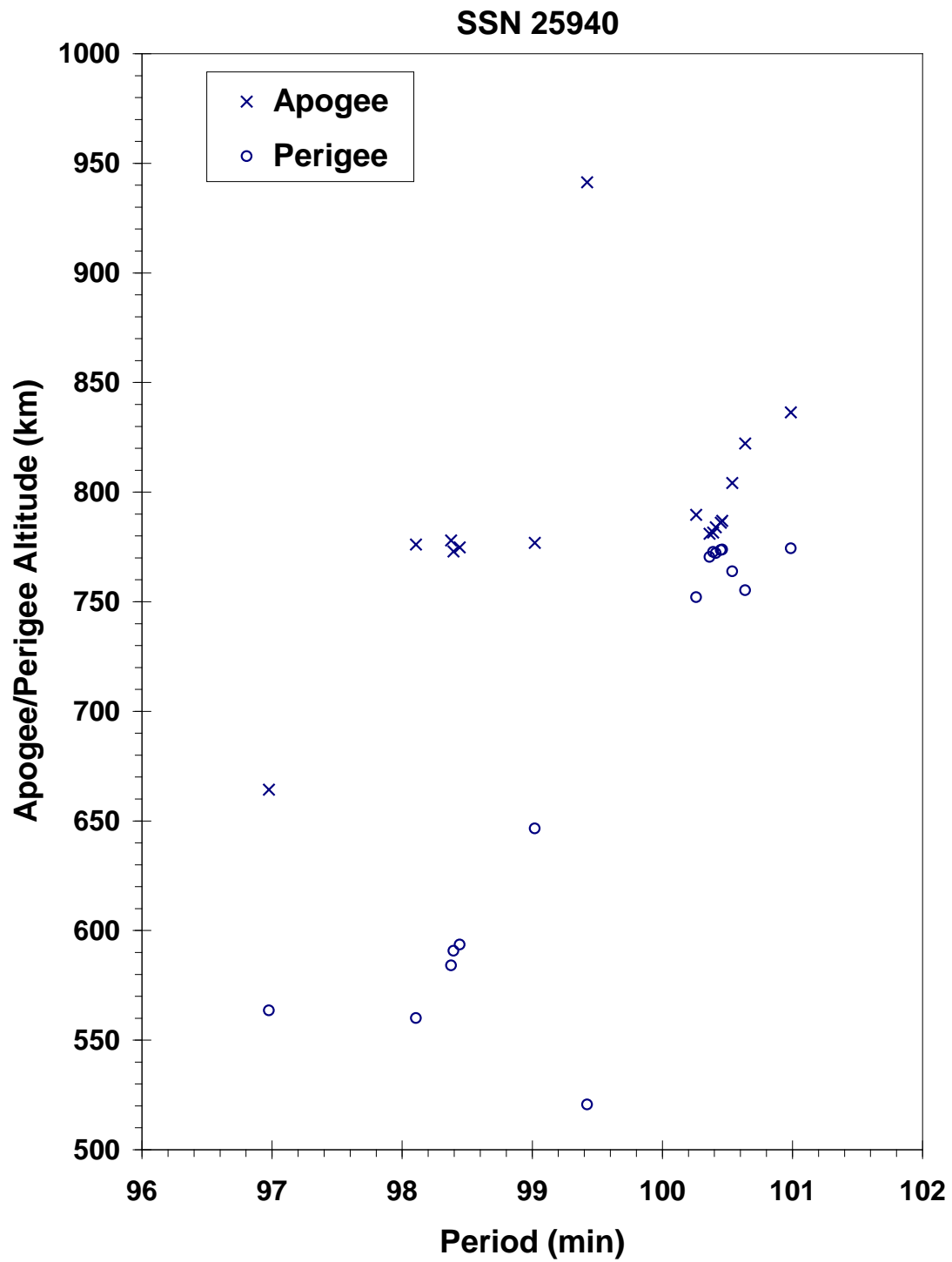
MAXIMUM ΔP : 3.4 min
MAXIMUM ΔI : 0.3 deg

COMMENTS

The spacecraft exceeded its expected lifetime and had been moved to a retirement orbit in August 2003. It is unclear whether or not it was entirely passivated. The rocket body associated with this payload (1999-057C, 25942) experienced an unrelated, but significant fragmentation six months after launch.

REFERENCE DOCUMENT

“Four Satellite Breakups in February Add to Debris Population”, The Orbital Debris Quarterly News, NASA JSC, April 2007. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv11i2.pdf>.



CBERS 1 debris cloud of 16 cataloged fragments 2 weeks after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Long March 4 third stage
OWNER: PRC
LAUNCH DATE: 14.14 Oct 1999
DRY MASS (KG): 1000
MAIN BODY: Cylinder nozzle; 2.9 m diameter by ~5 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

DATE: 11 Mar 2000 LOCATION: 51.2S, 311.5W
TIME: 1304 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 741 km

PRE-EVENT ELEMENTS

EPOCH: 00070.18630445 MEAN ANOMALY: 46.4837
RIGHT ASCENSION: 146.5592 MEAN MOTION: 14.46869143
INCLINATION: 98.5376 MEAN MOTION DOT/2: 0.00001547
ECCENTRICITY: 0.0012358 MEAN MOTION DOT DOT/6: 0.0
ARG. OF PERIGEE: 313.5308 BSTAR: 0.00043864

DEBRIS CLOUD DATA

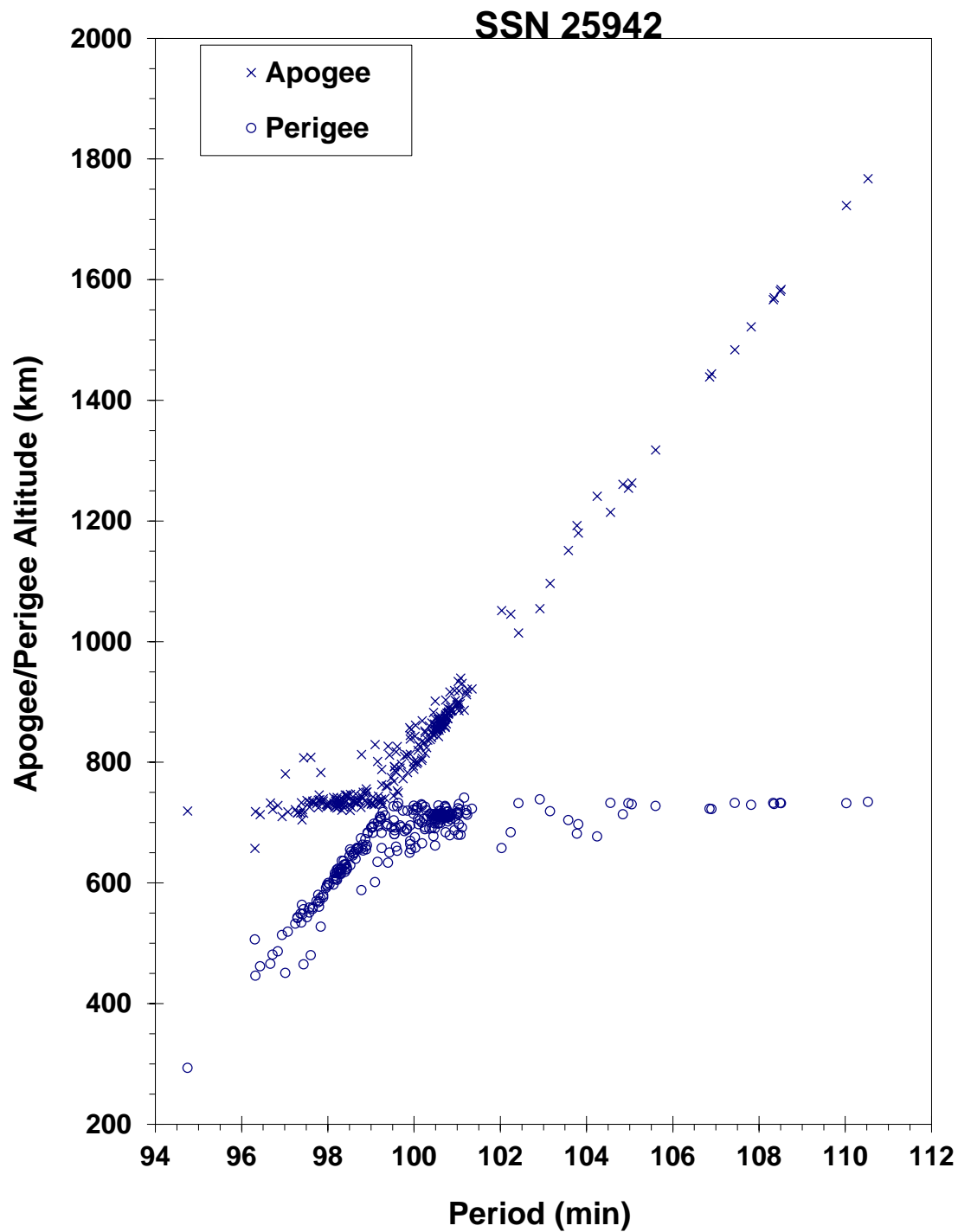
MAXIMUM ΔP : 10.985 min
MAXIMUM ΔI : 0.99 deg

COMMENTS

This is the second Long March 4 to breakup in only four missions. The first breakup (flight 2) occurred on 4 Oct 1990, 1 month after launch. Long March 4 missions did not resume until 1999, when two more were flown. This breakup involved the second 1999 mission (flight 4) and occurred 5 months after launch. This event has created more trackable debris than the 1990 breakup, with more than 300 pieces tracked by the SSN. Chinese officials were aware of the international concern following the 1990 breakup and had pledged to adopt countermeasures before the 1999 missions. Passivation of this vehicle was attempted.

REFERENCE DOCUMENT

"Analyzing the Cause of LM-4 (A)'s Upper Stage's Disintegration and the Countermeasures", W. X. Zang and S. Y. Liao, 5th International Conference of Pacific Basin Societies, 6-9 Jun 1993, Shanghai.



CBERS-1 / SACI-1 long March 4 third stage debris cloud of 280 fragments within 1 day of the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 26.33 Dec 1999
DRY MASS (KG): ~3000
MAIN BODY: Cylinder; 1.3 m diameter by 17 m length, plus solar arrays
MAJOR APPENDAGES: Solar Arrays
ATTITUDE CONTROL: Active 3-axis
ENERGY SOURCES: On-board propellant, explosive charge

EVENT DATA

| | | | |
|-----------|-------------|-----------------|-----------------------|
| DATE: | 21 Nov 2001 | LOCATION: | 38.3 S, 342.3 E (dsc) |
| TIME: | 1414Z | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 410 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 03325.57054648 | MEAN ANOMALY: | 199.8631 |
| RIGHT ASCENSION: | 55.0233 | MEAN MOTION: | 15.51939724 |
| INCLINATION: | 65.0021 | MEAN MOTION DOT/2: | 0.00131711 |
| ECCENTRICITY: | 0.0008788 | MEAN MOTION DOT DOT/6: | 0.0 |
| ARG. OF PERIGEE: | 257.3641 | BSTAR: | 0.0021441 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 10.62 min*
MAXIMUM ΔI : 1.28 deg*

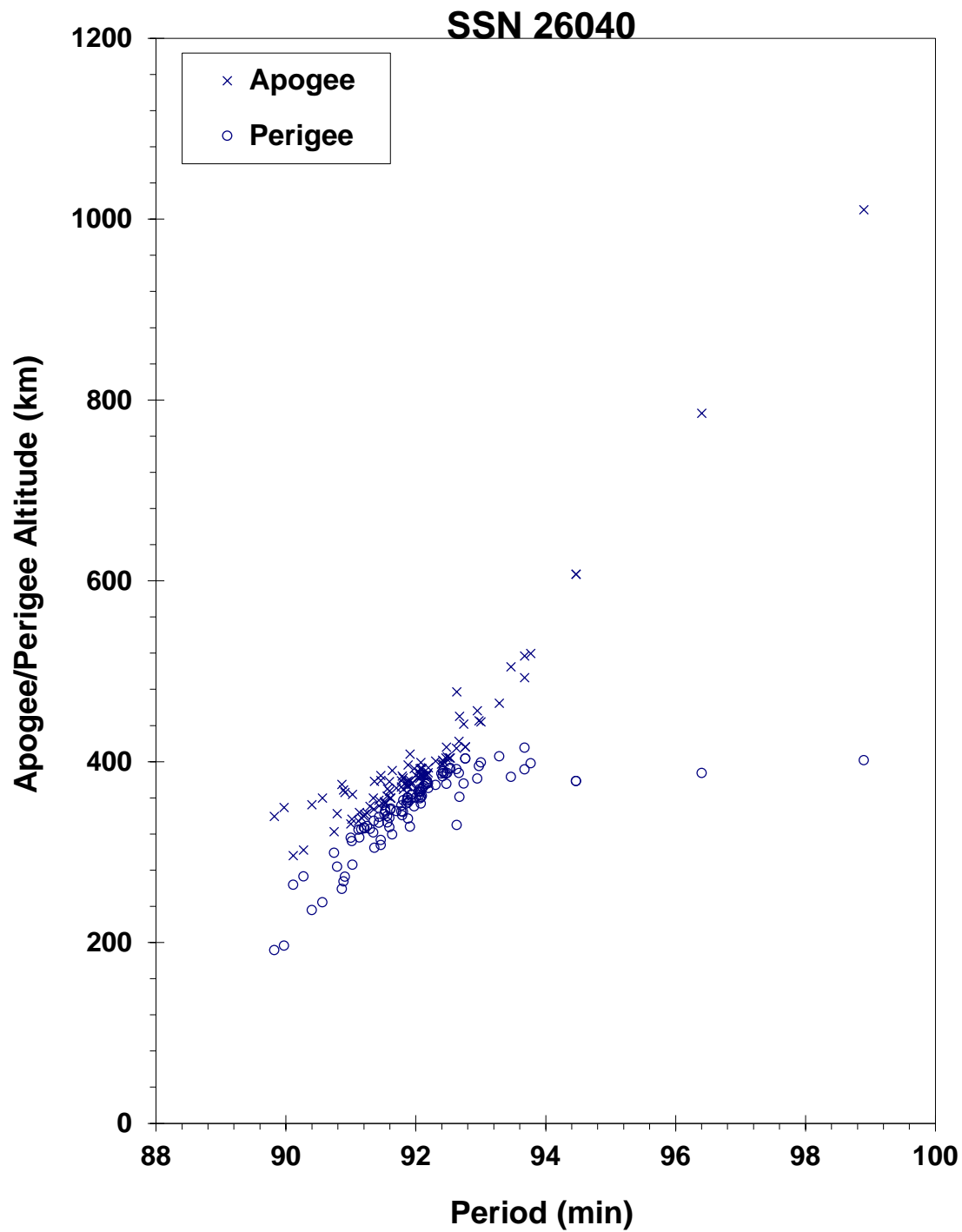
* Based on uncataloged debris data

COMMENTS

Cosmos 2367 was the 20th spacecraft of this type (Cosmos 699 class) known to have experienced a major fragmentation. The previous spacecraft in this series was Cosmos 2347, which experienced two fragmentations, one each in Nov and Dec of 1999. Cosmos 2367 was still in its operational orbit at the time of the event. Over 100 pieces were detected by the SSN 1 week after the breakup. Based upon other observations, the actual number of pieces probably exceeded 300. Although some debris were thrown into orbits with apogees above 1000 km, in general the debris were short-lived.

REFERENCE DOCUMENT

“Two Major Satellite Breakups Near End of 2001”, The Orbital Debris Quarterly News, NASA JSC, January 2002. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv7i1.pdf>.



Cosmos 2367 debris cloud of 103 fragments 1 week after the breakup as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 4.99 Jul 2000
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellant

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | ~1 Sep 2006 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 06244.59746638 | MEAN ANOMALY: | 38.1280 |
| RIGHT ASCENSION: | 18.3906 | MEAN MOTION: | 3.86574836 |
| INCLINATION: | 46.8834 | MEAN MOTION DOT/2: | 0.00005467 |
| ECCENTRICITY: | 0.6151900 | MEAN MOTION DOT DOT/6: | 0.0 |
| ARG. OF PERIGEE: | 248.4110 | BSTAR: | 0.00034737 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 98.8 min
MAXIMUM ΔI : 0.2 deg

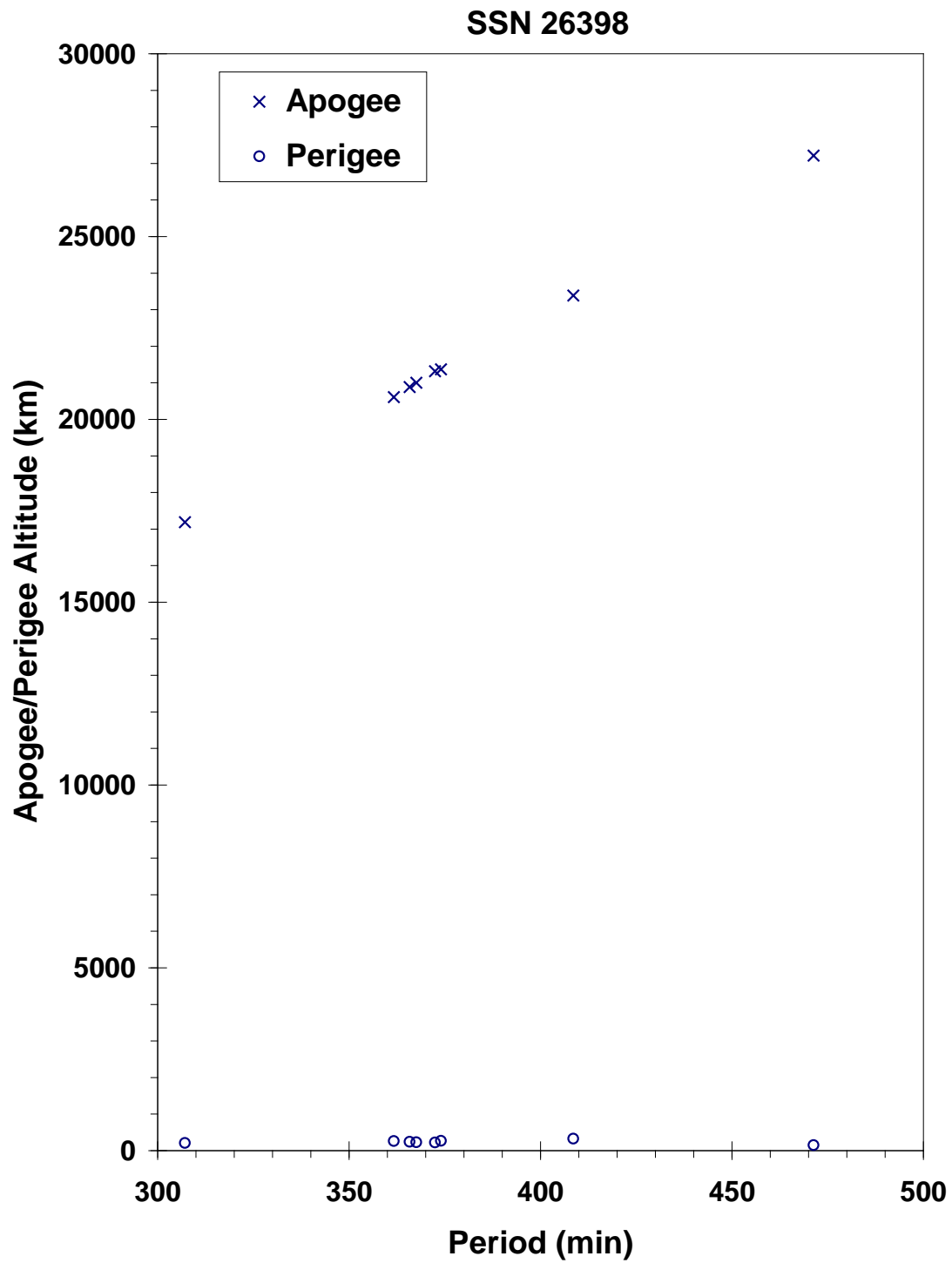
COMMENTS

The object was in a decaying geosynchronous transfer orbit; this event marks the 35th known breakup of a Proton Blok DM SOZ ullage motor since 1984. Only a handful of debris was detected from this event.

REFERENCE DOCUMENTS

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

“Three More Satellites Involved in Fragmentations”, The Orbital Debris Quarterly News, NASA JSC, October 2006. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv10i4.pdf>.



Cosmos 2371 SOZ motor debris cloud of 7 cataloged fragments plus the parent a day after the event. The debris pieces were never cataloged.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 21 Sep 2000
DRY MASS (KG): 1475
MAIN BODY: box; 1.88 m x 1.88 m x 4.2 m long
MAJOR APPENDAGES: solar panel
ATTITUDE CONTROL: none at time of event
ENERGY SOURCES: on-board propellants and batteries

EVENT DATA

| | | | |
|-----------|-------------|-----------------|----------------------|
| DATE: | 25 Nov 2015 | LOCATION: | 77.5 N, 48.8 E (dsc) |
| TIME: | 0950 GMT | ASSESSED CAUSE: | Battery |
| ALTITUDE: | 855 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 15329.31881564 | MEAN ANOMALY: | 226.6499 |
| RIGHT ASCENSION: | 34.9954 | MEAN MOTION: | 14.13117297 |
| INCLINATION: | 98.9249 | MEAN MOTION DOT/2: | 0.00000058 |
| ECCENTRICITY: | 0.0011308 | MEAN MOTION DOT DOT/6: | 0.00000 |
| ARG. OF PERIGEE: | 133.5614 | BSTAR: | 0.00054879 |

DEBRIS CLOUD DATA

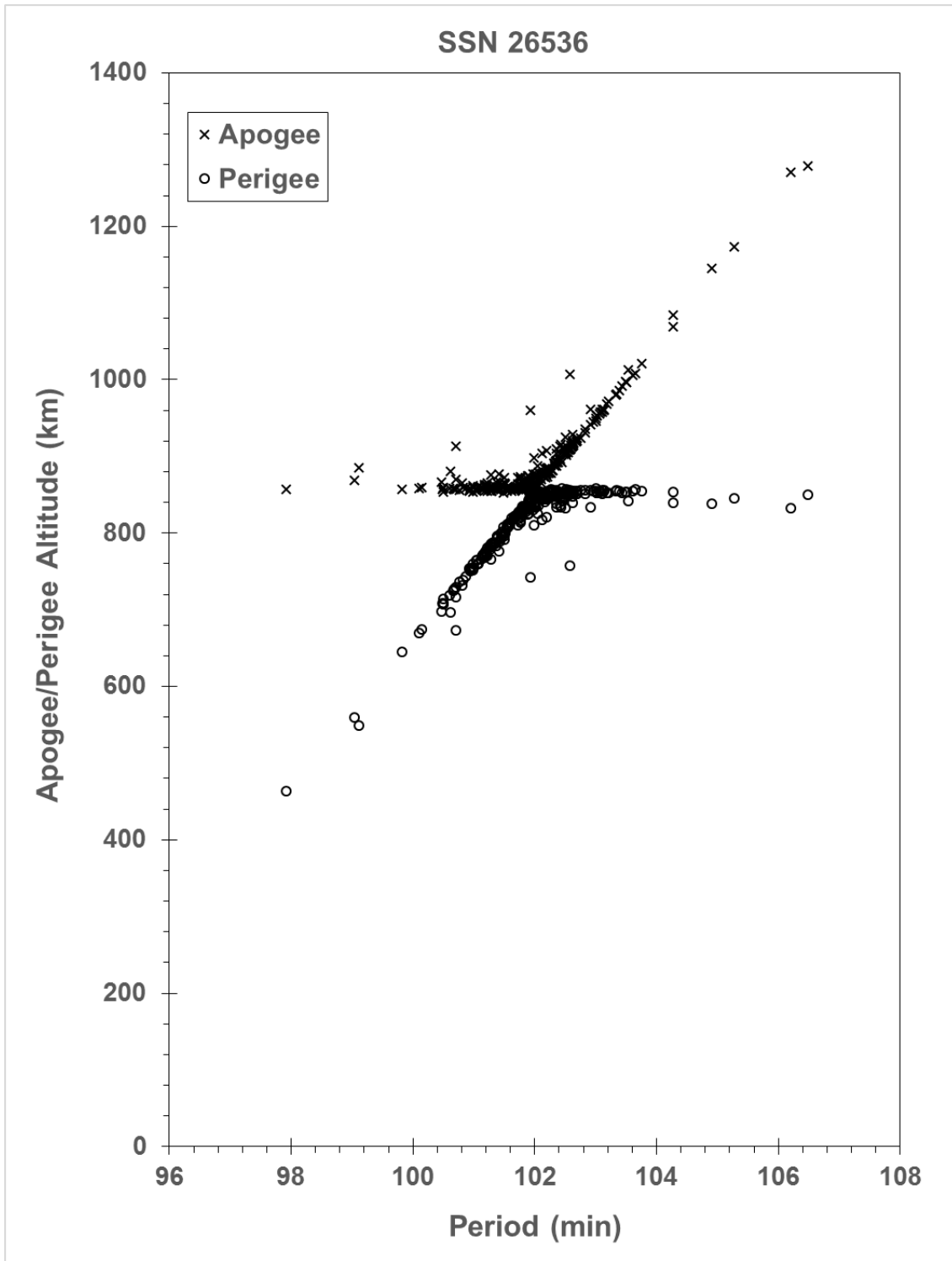
MAXIMUM ΔP : TBD min
MAXIMUM ΔI : TBD deg

COMMENTS

This event was the second known breakup of a NOAA-series spacecraft, the other being NOAA-8. NOAA-16 performed its nominal mission until placed in backup status in 2005. An on-board anomaly ended communication on 6 June 2014 and the spacecraft was decommissioned on 9 June 2014. The event is assessed as likely being a battery explosion.

REFERENCE DOCUMENTS

“Recent NOAA_16 Satellite Breakup”, The Orbital Debris Quarterly News, NASA JSC, April 2016.
Available online at <http://orbitaldebris.jsc.nasa.gov/newsletter/pdfs/ODQNv20i1-2.pdf>.



NOAA 16 debris cloud; debris cataloged between approximately one week and 123 days after event.

SATELLITE DATA

TYPE: PSLV Final Stage
OWNER: India
LAUNCH DATE: 22.20 Oct 2001
DRY MASS (KG): ~900
MAIN BODY: Cylinder; 2.8 m diameter by 2.6 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board hypergolic propellants

EVENT DATA

| | | | |
|-----------|-------------|-----------------|-------------------|
| DATE: | 19 Dec 2001 | LOCATION: | 25 S, 340 E (asc) |
| TIME: | ~1140Z | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | 670 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|--------------|
| EPOCH: | 01352.90695581 | MEAN ANOMALY: | 316.4909 |
| RIGHT ASCENSION: | 65.6004 | MEAN MOTION: | 14.85657962 |
| INCLINATION: | 97.9010 | MEAN MOTION DOT/2: | -0.00000443 |
| ECCENTRICITY: | 0.0088752 | MEAN MOTION DOT DOT/6: | 0.0 |
| ARG. OF PERIGEE: | 44.3375 | BSTAR: | -0.000041058 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 9.86 min*
MAXIMUM ΔI : 3.06 deg*

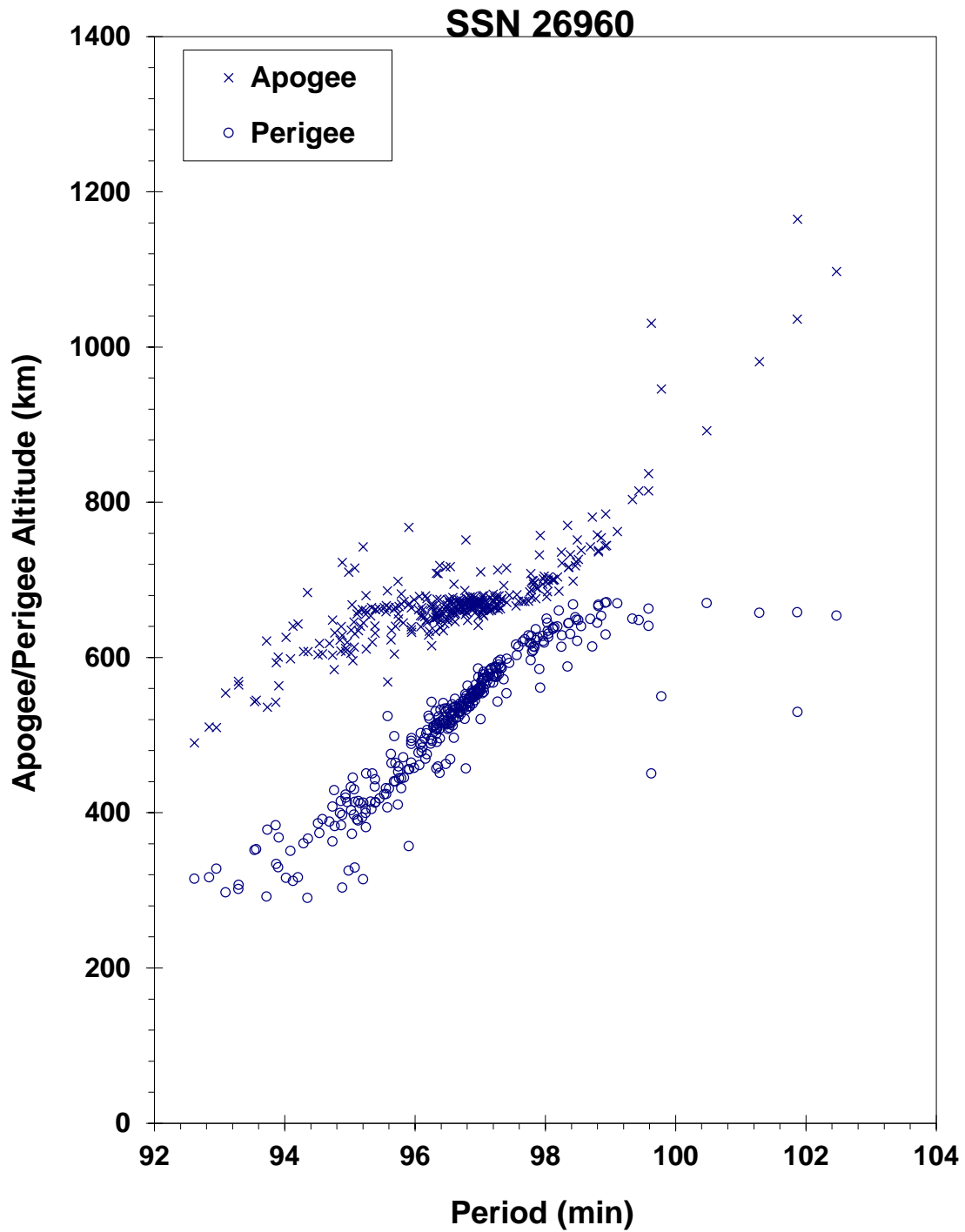
* Based on uncataloged debris data

COMMENTS

This is the first known breakup associated with the Polar Satellite Launch Vehicle (PSLV) fourth stage. While 332 fragments were initially detected by the SSN, 326 debris were cataloged. The vehicle employed hypergolic propellants that were not passivated after payload delivery. Some of the debris could remain in orbit for several years or longer.

REFERENCE DOCUMENT

“Two Major Satellite Breakups Near End of 2001”, The Orbital Debris Quarterly News, NASA JSC, January 2002. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv7i1.pdf>.



TES R/B debris cloud of 332 fragments 1 week after the breakup as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 21.17 Dec 2001
DRY MASS (KG): 3000
MAIN BODY: Cylinder; 1.3 m diameter by 17 m length, plus solar arrays
MAJOR APPENDAGES: Solar Arrays
ATTITUDE CONTROL: Active 3-axis
ENERGY SOURCES: On-board propellant; explosive charge

EVENT DATA

| | | | |
|-----------|-------------|-----------------|-----------------|
| DATE: | 28 Feb 2004 | LOCATION: | 26S, 100E (dsc) |
| TIME: | ~1930 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 265 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 04059.20843210 | MEAN ANOMALY: | 95.7196 |
| RIGHT ASCENSION: | 358.8049 | MEAN MOTION: | 15.87352021 |
| INCLINATION: | 64.9763 | MEAN MOTION DOT/2: | 0.00416036 |
| ECCENTRICITY: | 0.0131275 | MEAN MOTION DOT DOT/6: | 0.000069430 |
| ARG. OF PERIGEE: | 262.9022 | BSTAR: | 0.00074756 |

DEBRIS CLOUD DATA

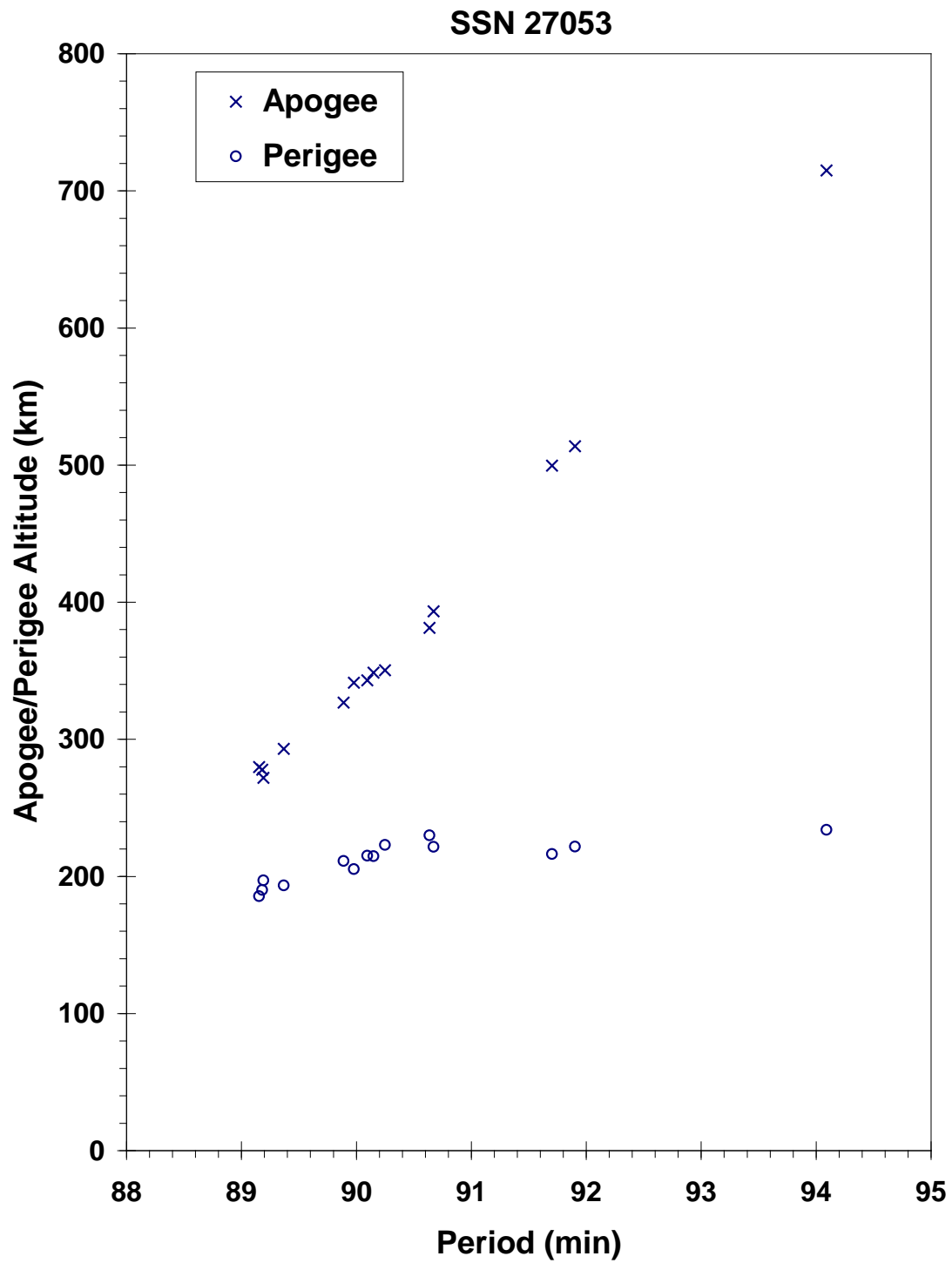
MAXIMUM ΔP : 3.4 min
MAXIMUM ΔI : 0.3 deg

COMMENTS

Cosmos 2383 was the 21st spacecraft of this type (Cosmos 699 class) known to have experienced a major fragmentation. The previous spacecraft in this series was Cosmos 2367, which experienced a fragmentation at an altitude just above the International Space Station in 2001. This event also produced debris crossing the ISS orbit altitude. Approximately 50 debris were detected by the US SSN. Fortunately, all the cataloged debris from this fragmentation were short-lived. Of the 48 spacecraft of this type, 21 have experienced fragmentation events.

REFERENCE DOCUMENT

"Fragmentation of Cosmos 2383", The Orbital Debris Quarterly News, NASA JSC, April 2004.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv8i2.pdf>.



Cosmos 2383 debris cloud of 14 cataloged fragments a few days after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 25.63 Jul 2002
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellant

EVENT DATA

| | | | |
|-----------|------------|-----------------|------------|
| DATE: | 1 Jun 2005 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 05151.71140009 | MEAN ANOMALY: | 222.2107 |
| RIGHT ASCENSION: | 143.5118 | MEAN MOTION: | 15.06786995 |
| INCLINATION: | 63.6569 | MEAN MOTION DOT/2: | 0.00075759 |
| ECCENTRICITY: | 0.0418207 | MEAN MOTION DOT DOT/6: | 0.000007275 |
| ARG. OF PERIGEE: | 140.9987 | BSTAR: | 0.00057187 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 5.2 min
MAXIMUM ΔI : 1.0 deg

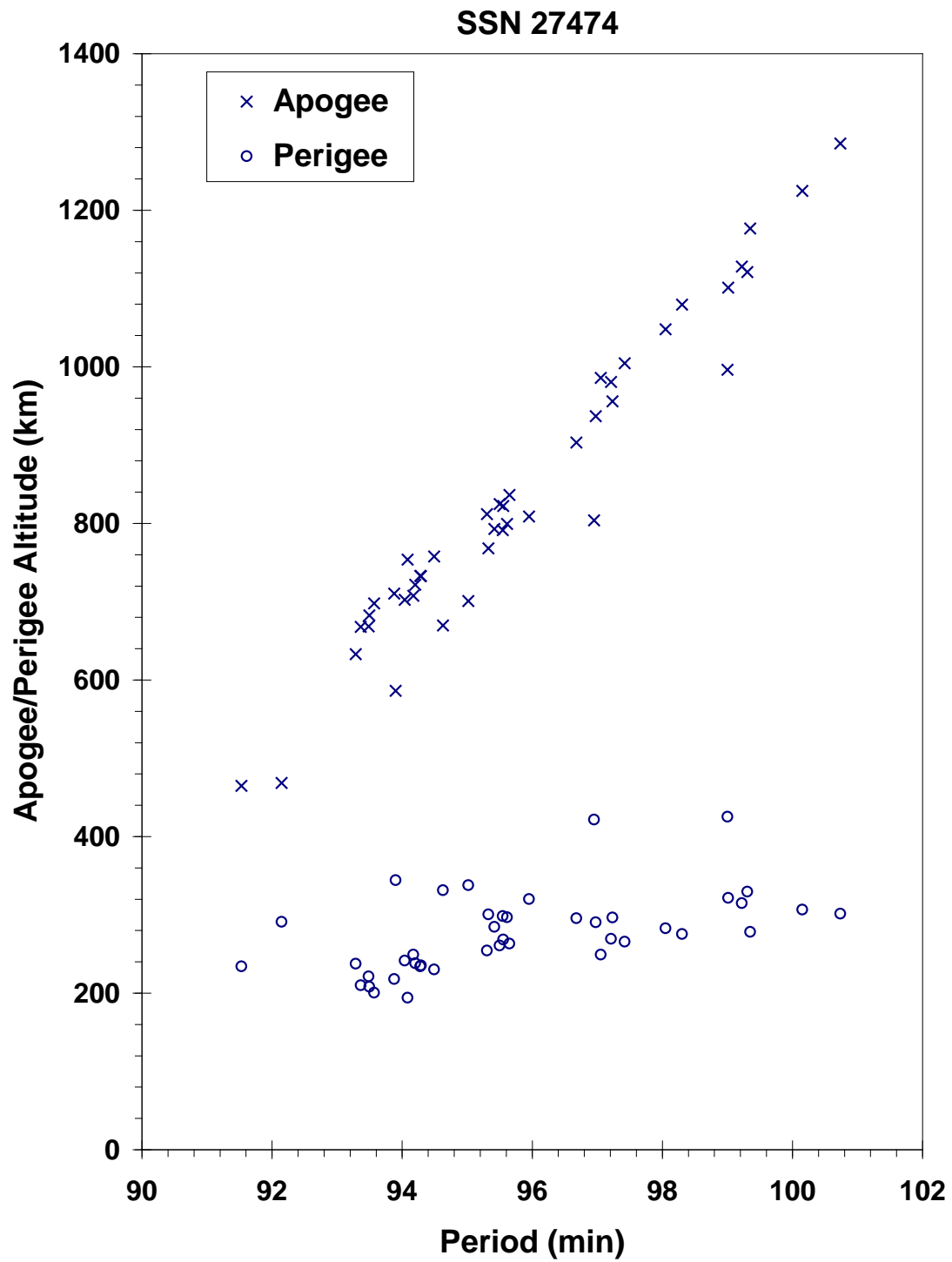
COMMENTS

This event marks the 33rd known breakup of a Proton Block DM SOZ ullage motor since 1984. The motor experienced a second event a month later about 29 June. About 40 new debris were seen after each event.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

“Recent Satellite Breakups”, The Orbital Debris Quarterly News, NASA JSC, July 2005.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNV9i3.pdf>.



SOZ motor debris cloud in July 2005 after the second breakup event.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 25.63 Jul 2002
DRY MASS (KG): 55
MAIN BODY: Ellipsoid; 0.6 m diameter by 1 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellant

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 29 Oct 2004 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Propulsion |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 04302.83463691 | MEAN ANOMALY: | 199.8896 |
| RIGHT ASCENSION: | 56.1075 | MEAN MOTION: | 15.09294101 |
| INCLINATION: | 63.6401 | MEAN MOTION DOT/2: | 0.00146672 |
| ECCENTRICITY: | 0.0436823 | MEAN MOTION DOT DOT/6: | 0.000010335 |
| ARG. OF PERIGEE: | 161.8395 | BSTAR: | 0.00073391 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : Unknown
MAXIMUM ΔI : Unknown

COMMENTS

This event marks the 31st known breakup of a Proton Blok DM SOZ ullage motor since 1984. More than 60 debris were detected by the Naval electronic fence. No debris were in orbit long enough to be cataloged.

REFERENCE DOCUMENTS

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

“Recent Satellite Breakups”, The Orbital Debris Quarterly News, NASA JSC, January 2005.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv9i1.pdf>.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 12.60 Aug 2003
DRY MASS (KG): ~6000
MAIN BODY: Cylinder; 2.4 m diameter by 7 m length
MAJOR APPENDAGES: Solar Arrays
ATTITUDE CONTROL: Active, 3-axis
ENERGY SOURCES: On-board propellants, explosive charge

EVENT DATA

| | | | |
|-----------|------------|-----------------|-----------------------|
| DATE: | 9 Dec 2003 | LOCATION: | 64.8 N, 135.4 E (asc) |
| TIME: | 0129Z | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | 189.33 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|--------------|
| EPOCH: | 03342.92270571 | MEAN ANOMALY: | 296.9639 |
| RIGHT ASCENSION: | 136.8172 | MEAN MOTION: | 16.22926227 |
| INCLINATION: | 64.9062 | MEAN MOTION DOT/2: | 0.01025110 |
| ECCENTRICITY: | 0.0055948 | MEAN MOTION DOT DOT/6: | 0.0000073532 |
| ARG. OF PERIGEE: | 63.7269 | BSTAR: | 0.00028689 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 7.34 min*
MAXIMUM ΔI : 0.08 deg*

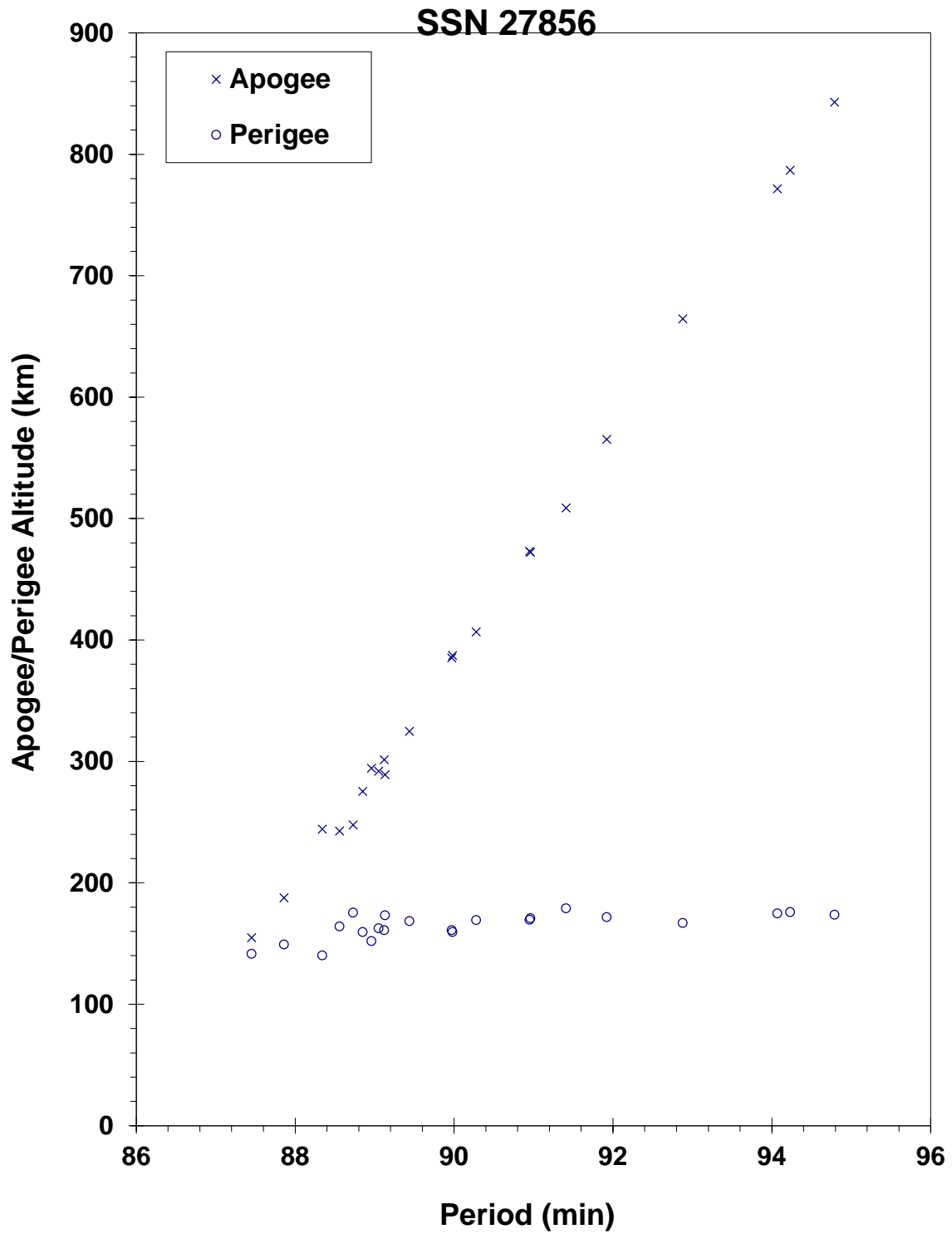
* Based on uncataloged debris data

COMMENTS

Cosmos 2399 was the seventh of the Cosmos 2031 class of spacecraft that debuted in 1989 but was not flown since 1997. In all six previous missions (1989-1997), the spacecraft was deliberately exploded at the end of mission. Previous missions of this type include Cosmos 2031, Cosmos 2101, Cosmos 2163, Cosmos 2225, Cosmos 2262, and Cosmos 2343. All such events have occurred over Eastern Russia. Highest previous piece count for large debris for this class of vehicle was 180, although more were probably created. Approximately 22 debris were detected by the SSN. Due to the low altitude of the breakup, the debris were short-lived.

REFERENCE DOCUMENT

"Satellite Fragmentations in 2003", The Orbital Debris Quarterly News, NASA JSC, January 2004. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNV8i1.pdf>.



Cosmos 2399 debris cloud of 22 fragments 1 day after the breakup as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Rocket Body
OWNER: Japan
LAUNCH DATE: 24.06 Jan 2006
DRY MASS (KG): ~3000
MAIN BODY: Cylinder; 4.0 m diameter by 10.6 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of event
ENERGY SOURCES: On-board propellant

FIRST EVENT DATA

| | | | |
|-----------|------------|-----------------|----------------|
| DATE: | 8 Aug 2006 | LOCATION: | 5N, 131E (asc) |
| TIME: | 1407 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 685 km | | |

SECOND EVENT DATA

| | | | |
|-----------|-------------|-----------------|-----------------|
| DATE: | 27 Aug 2006 | LOCATION: | 76S, 135E (asc) |
| TIME: | 1618 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 695 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 06220.18259253 | MEAN ANOMALY: | 147.5075 |
| RIGHT ASCENSION: | 300.3416 | MEAN MOTION: | 14.8204486 |
| INCLINATION: | 98.1944 | MEAN MOTION DOT/2: | 0.00000143 |
| ECCENTRICITY: | 0.0106899 | MEAN MOTION DOT DOT/6: | 0.0 |
| ARG. OF PERIGEE: | 211.9623 | BSTAR: | 0.000023737 |

DEBRIS CLOUD DATA

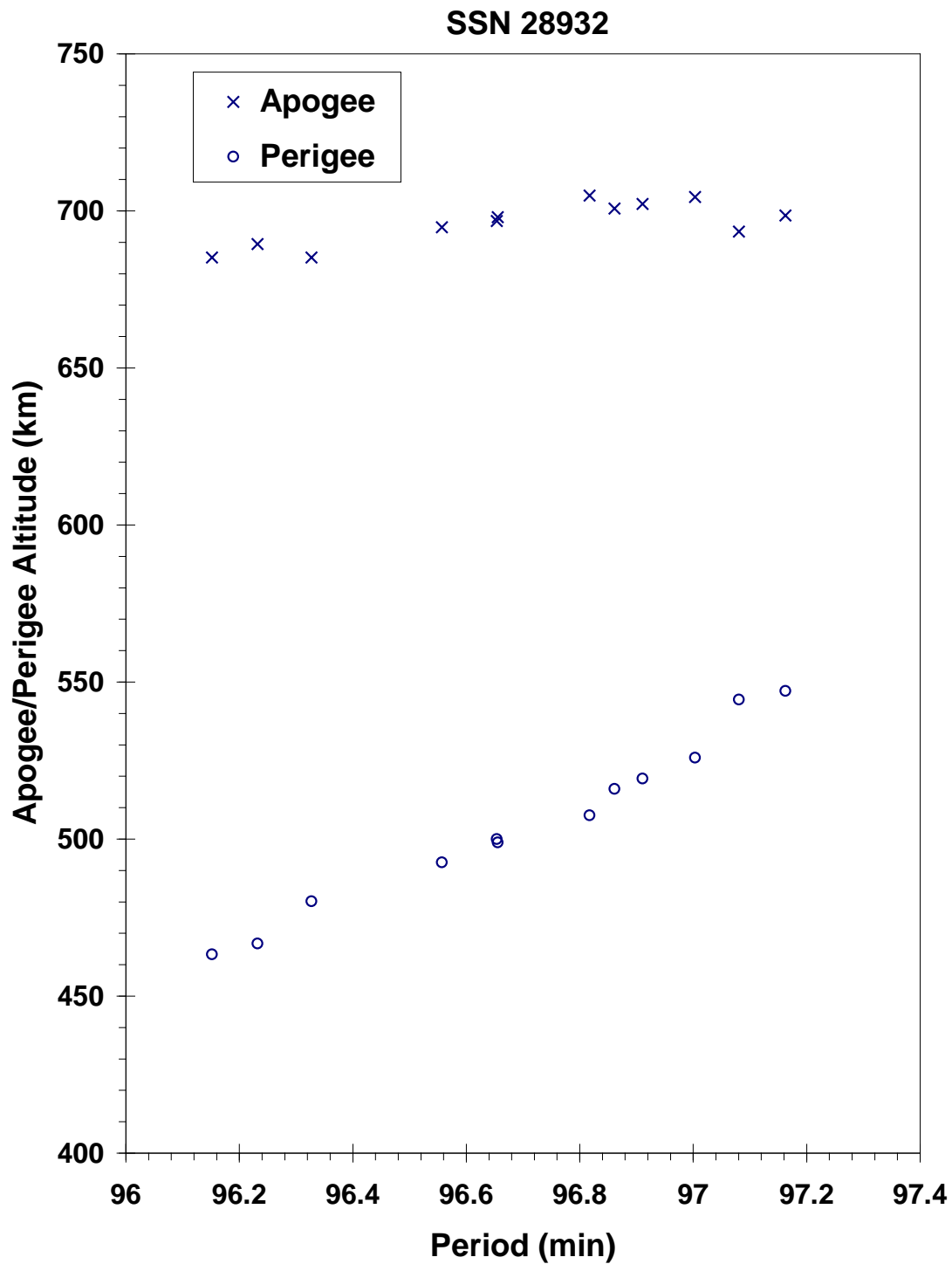
MAXIMUM ΔP : 1.0 min
MAXIMUM ΔI : 0.0 deg

COMMENTS

The first event shed four pieces of debris, the second event shed more than 15 pieces. The parent experienced little if any change in orbit and the debris from both events were ejected with relatively low delta velocity from the parent, making a propulsion explosion unlikely for the cause of breakup. All the debris appeared to have high area-to-mass ratios, causing the ejected debris to decay within six months of the events. The parent body remains in orbit as of August 2007. Another H-IIA second stage (2006-037B) experienced two minor fragmentation events a few months later. Possible source of the debris is light-weight insulation material.

REFERENCE DOCUMENTS

“Three More Satellites Involved in Fragmentations”, The Orbital Debris Quarterly News, NASA JSC, October 2006.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv10i4.pdf>.



The ALOS R/B debris cloud from the second event, a few days after the event. The parent body is the piece with the highest perigee, at the right of the graph.

SATELLITE DATA

TYPE: Rocket Body
OWNER: CIS
LAUNCH DATE: 28.84 Feb 2006
DRY MASS (KG): 2600
MAIN BODY: Cylinder; 2.5 m diameter by 2.6 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None
ENERGY SOURCES: On-board propellants

EVENT DATA

DATE: 19 Feb 2007 LOCATION: 31S, 135E (asc)
TIME: 1721 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 7640 km

PRE-EVENT ELEMENTS

EPOCH: 07050.57138199 MEAN ANOMALY: 134.5177
RIGHT ASCENSION: 213.0506 MEAN MOTION: 5.25304422
INCLINATION: 51.4995 MEAN MOTION DOT/2: 0.00000185
ECCENTRICITY: 0.5083307 MEAN MOTION DOT DOT/6: 0.0
ARG. OF PERIGEE: 197.8403 BSTAR: 0.00029198

DEBRIS CLOUD DATA

MAXIMUM ΔP : 48.6 min
MAXIMUM ΔI : 2.6 deg

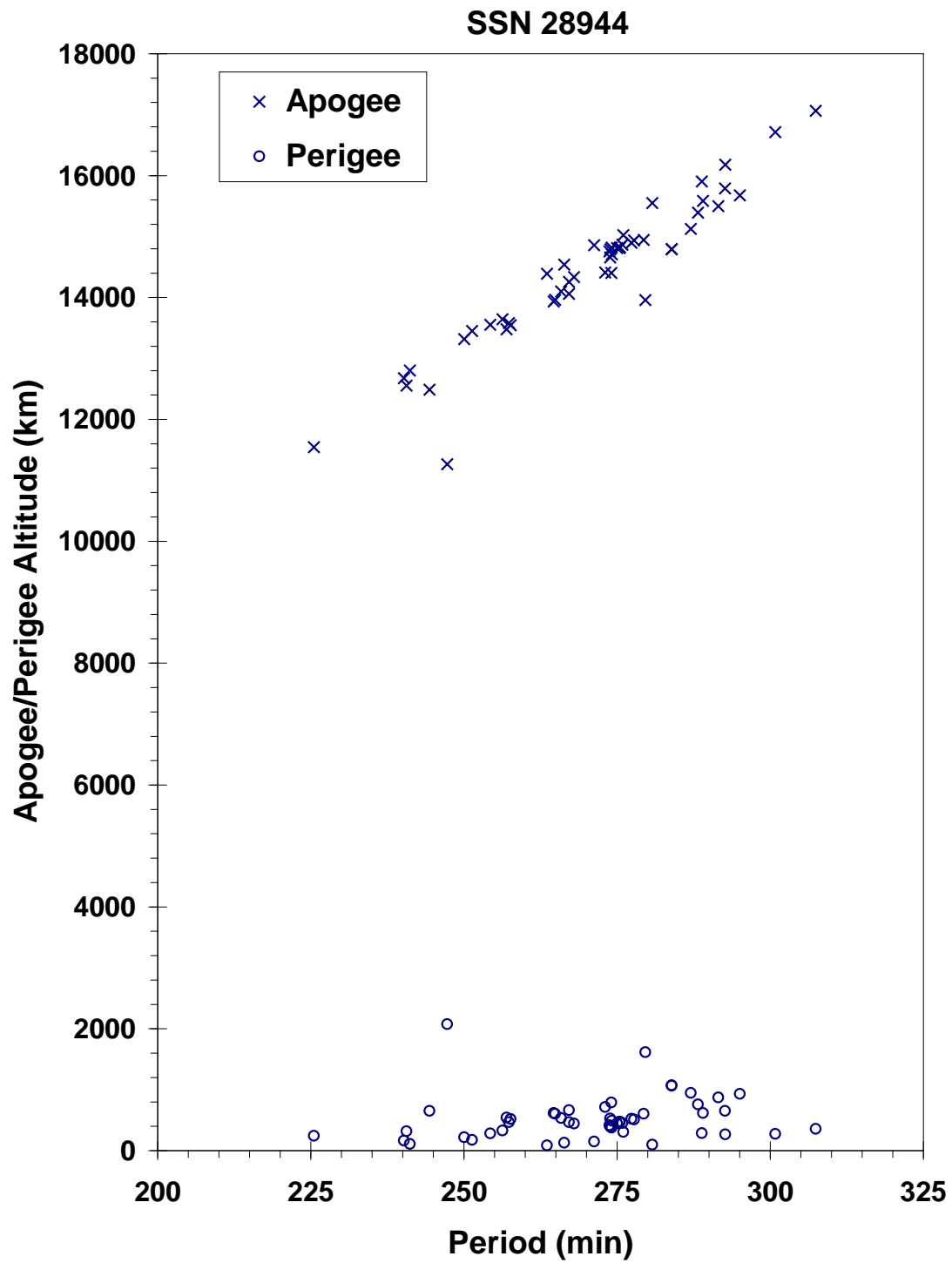
COMMENTS

This upper stage malfunctioned, stranding itself and its payload in an eccentric orbit. The cause of the breakup is assessed to be related to the ~8 metric tons of unused propellants. The breakup occurred over Southern Australia, and was captured by three amateur astronomers, which showed expansion of a faint cloud around the exploded fourth stage. Over a thousand objects 10 cm or larger were estimated by the US SSN.

REFERENCE DOCUMENT

“Two More Incidents Add to Growing Space Debris”, [Space News](#), February 26, 2007.

“Four Satellite Breakups in February Add to Debris Population”, [The Orbital Debris Quarterly News](#), NASA JSC, April 2007.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv11i2.pdf>.



The *Briz-M* R/B debris cloud of uncataloged fragments 10 days after the event.

SATELLITE DATA

TYPE: Payload
OWNER: PRC
LAUNCH DATE: 26 Apr 2006
DRY MASS (KG): 2721.0
MAIN BODY: unknown
MAJOR APPENDAGES: unknown
ATTITUDE CONTROL: unknown
ENERGY SOURCES: unknown

EVENT DATA

DATE: 4 Feb 2010
TIME: 0649 GMT
ALTITUDE: 636.516 km

LOCATION: 56.3N, 140.7E (asc)
ASSESSED CAUSE: Unknown

PRE-EVENT ELEMENTS

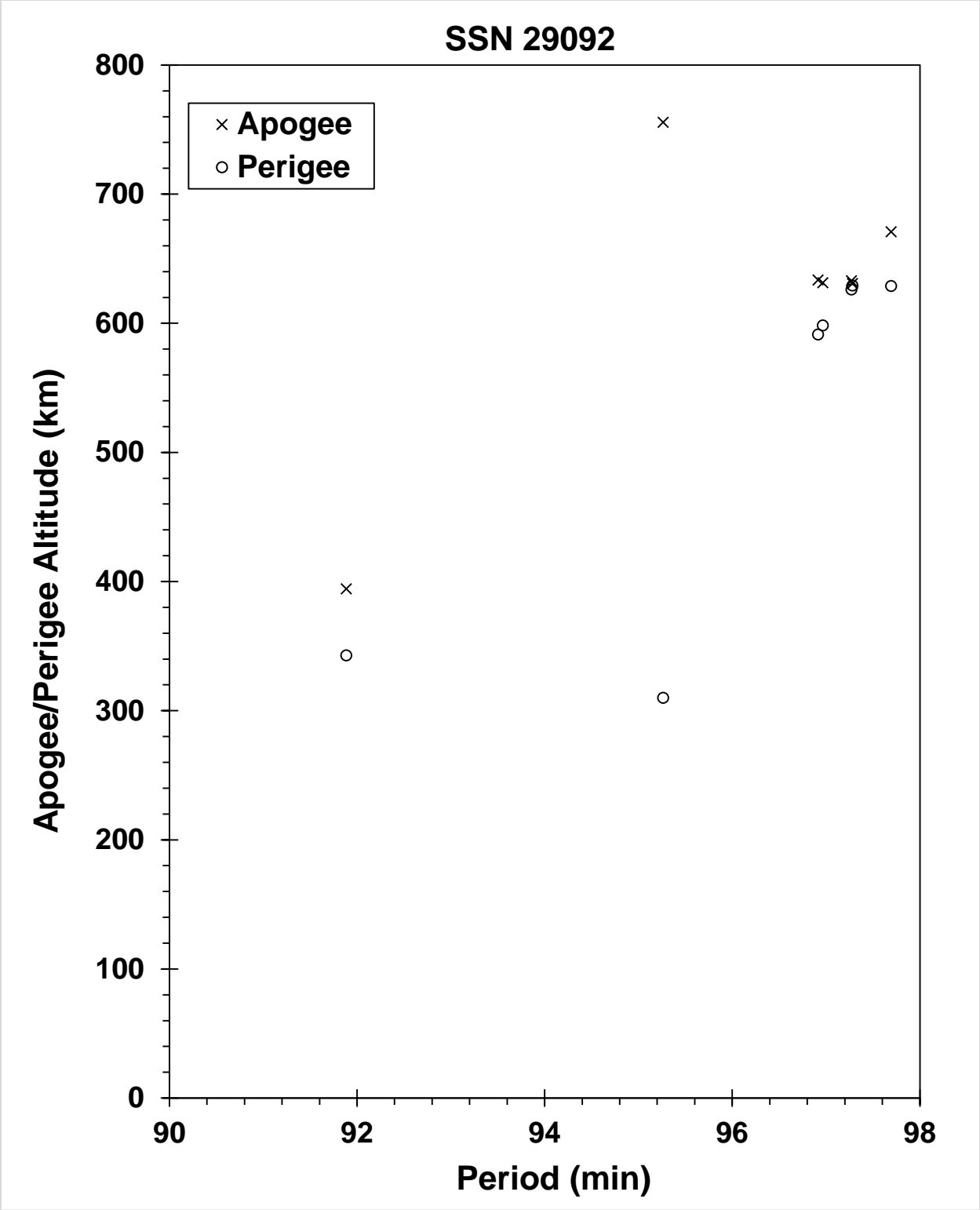
| | | | |
|------------------|----------------|------------------------|--------------|
| EPOCH: | 10034.53017668 | MEAN ANOMALY: | 264.9880 |
| RIGHT ASCENSION: | 28.5835 | MEAN MOTION: | 14.80351885 |
| INCLINATION: | 97.8566 | MEAN MOTION DOT/2: | -0.00000264 |
| ECCENTRICITY: | 0.0001956 | MEAN MOTION DOT DOT/6: | 0.0 |
| ARG. OF PERIGEE: | 95.1578 | BSTAR: | -0.000027865 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 5.4 min
MAXIMUM ΔI : 0.4 deg

COMMENTS**REFERENCE DOCUMENT**

“Old and New Satellite Breakups Identified”, The Orbital Debris Quarterly News, NASA JSC, April 2010.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/ODQNv14i2.pdf>.



Seven cataloged fragments 7 weeks after the event.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 25 Jun 2006
DRY MASS (KG): 3200
MAIN BODY: 1.7 m diameter x 17 m long cylinder
MAJOR APPENDAGES: solar panels
ATTITUDE CONTROL: active, three-axis
ENERGY SOURCES: on-board propellants, explosive charge (?)

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------------------|
| DATE: | 14 Mar 2008 | LOCATION: | 52.247S, 24.130E (dsc) |
| TIME: | 0429 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 420.242 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 08073.77012377 | MEAN ANOMALY: | 145.3615 |
| RIGHT ASCENSION: | 48.2539 | MEAN MOTION: | 15.52978992 |
| INCLINATION: | 65.0437 | MEAN MOTION DOT/2: | 0.00011420 |
| ECCENTRICITY: | 0.0014765 | MEAN MOTION DOT DOT/6: | 0.0 |
| ARG. OF PERIGEE: | 285.7968 | BSTAR: | 0.00018556 |

DEBRIS CLOUD DATA

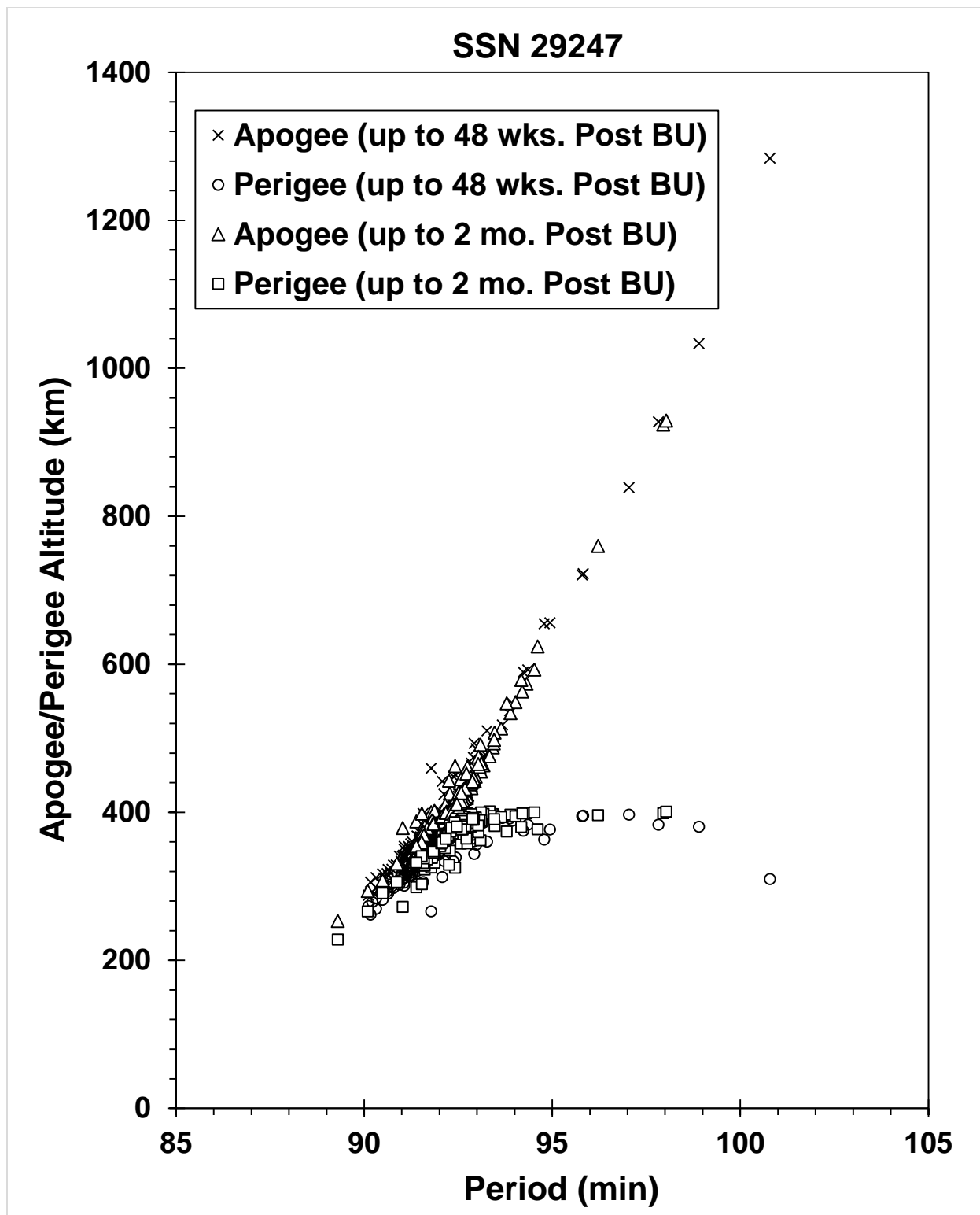
MAXIMUM ΔP : 8.1 min
MAXIMUM ΔI : 1.4 deg

COMMENTS

Cosmos 2421 was the last of the Soviet/Russian series of Electronic Ocean Reconnaissance Satellite (EORSAT) spacecraft. As with other spacecraft of the Cosmos 699 class, the high area-to-mass ratio of the majority of debris produced in this event resulted in a short residence on-orbit and no long-term environmental hazard.

REFERENCE DOCUMENT

“Satellite Breakups During First Quarter of 2008”, The Orbital Debris Quarterly News, NASA JSC, April 2008.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/ODQNv12i2.pdf>.



The COSMOS 2421 debris cloud, including 509 fragments cataloged up to 48 weeks after the event.

SATELLITE DATA

TYPE: Rocket Body
OWNER: Japan
LAUNCH DATE: 11.19 Sep 2006
DRY MASS (KG): ~3000
MAIN BODY: Cylinder; 4.0 m diameter by 10.6 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of event
ENERGY SOURCES: On-board propellant

EVENT DATA

DATE: 28 Dec 2006
TIME: 1729 GMT
ALTITUDE: 470 km
LOCATION: 58S, 131E (asc)
ASSESSED CAUSE: Unknown

PRE-EVENT ELEMENTS

EPOCH: 06361.50058695
RIGHT ASCENSION: 117.7926
INCLINATION: 97.2357
ECCENTRICITY: 0.0043549
ARG. OF PERIGEE: 12.7250
MEAN ANOMALY: 347.5081
MEAN MOTION: 15.35084918
MEAN MOTION DOT/2: 0.00002838
MEAN MOTION DOT DOT/6: 0.0
BSTAR: 0.000083212

DEBRIS CLOUD DATA

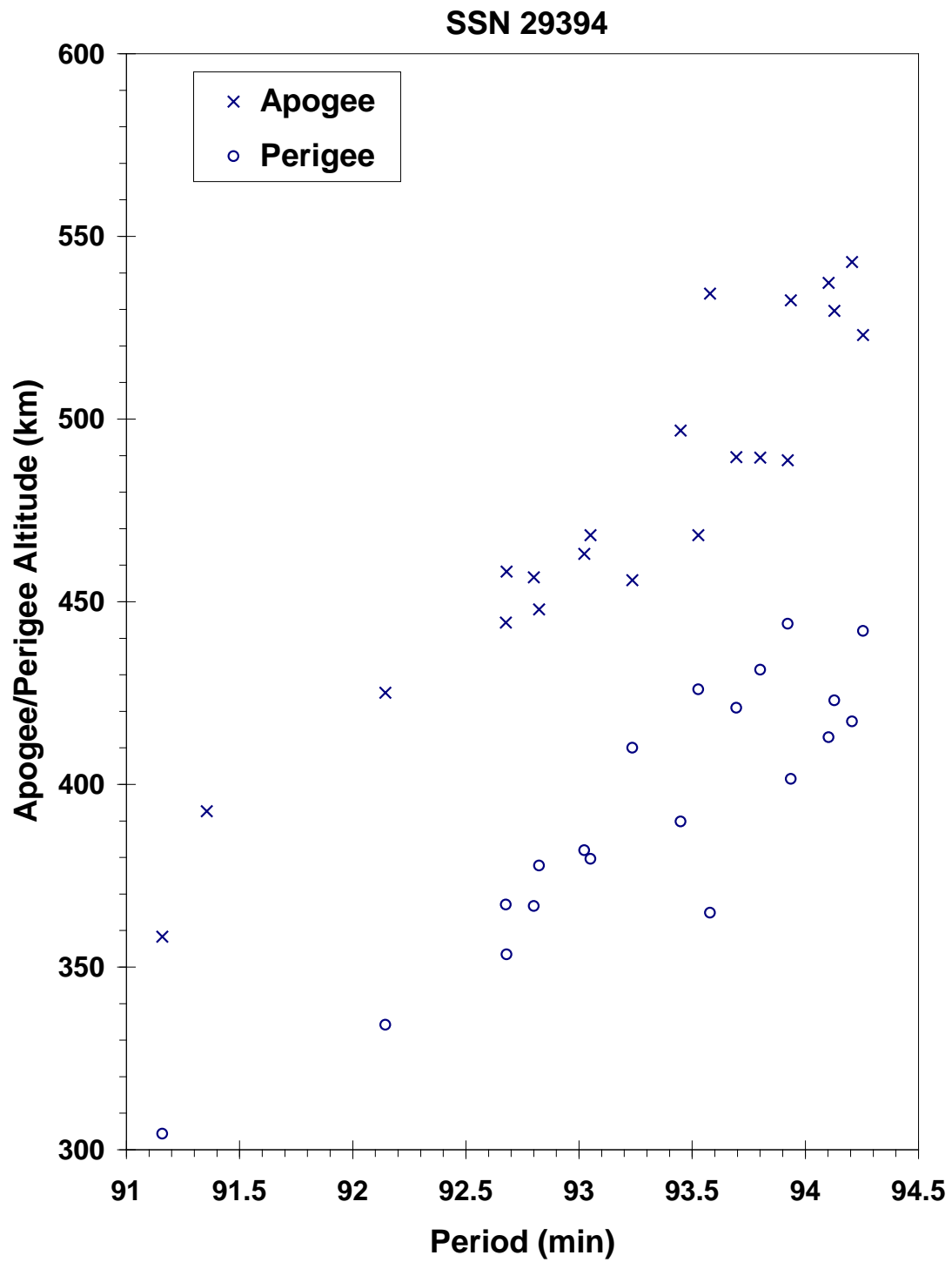
MAXIMUM ΔP : 2.6 min
MAXIMUM ΔI : 0.2 deg

COMMENTS

This was the second fragmentation of an H-IIA second stage in 2006 (see 2006-002B). No debris were cataloged from the first event, but at least 20 new objects were detected. However, the rocket body experienced a second fragmentation event on 25 July 2007, releasing at least 15 new debris.

REFERENCE DOCUMENT

“Significant Increase in Satellite Breakups During 2006”, The Orbital Debris Quarterly News, NASA JSC, January 2007.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv11i1.pdf>.



H-IIA debris cloud 2 weeks after the first event. The debris were never cataloged.

SATELLITE DATA

| | |
|-------------------|--|
| TYPE: | Payload |
| OWNER: | CIS |
| LAUNCH DATE: | 14.57 Sep 2006 |
| DRY MASS (KG): | ~6000 |
| MAIN BODY: | Cylinder; 2.4 m diameter by 7 m length |
| MAJOR APPENDAGES: | Solar Arrays |
| ATTITUDE CONTROL: | Active, 3-axis |
| ENERGY SOURCES: | On-board propellants, explosive charge |

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------|
| DATE: | 17 Nov 2006 | LOCATION: | Unknown |
| TIME: | ~1800 GMT | ASSESSED CAUSE: | Deliberate |
| ALTITUDE: | 210 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|--------------|
| EPOCH: | 06321.75318443 | MEAN ANOMALY: | 8.8408 |
| RIGHT ASCENSION: | 358.1498 | MEAN MOTION: | 16.11335386 |
| INCLINATION: | 64.8900 | MEAN MOTION DOT/2: | 0.00257180 |
| ECCENTRICITY: | 0.0061777 | MEAN MOTION DOT DOT/6: | 0.0000071463 |
| ARG. OF PERIGEE: | 106.6782 | BSTAR: | 0.00018281 |

DEBRIS CLOUD DATA

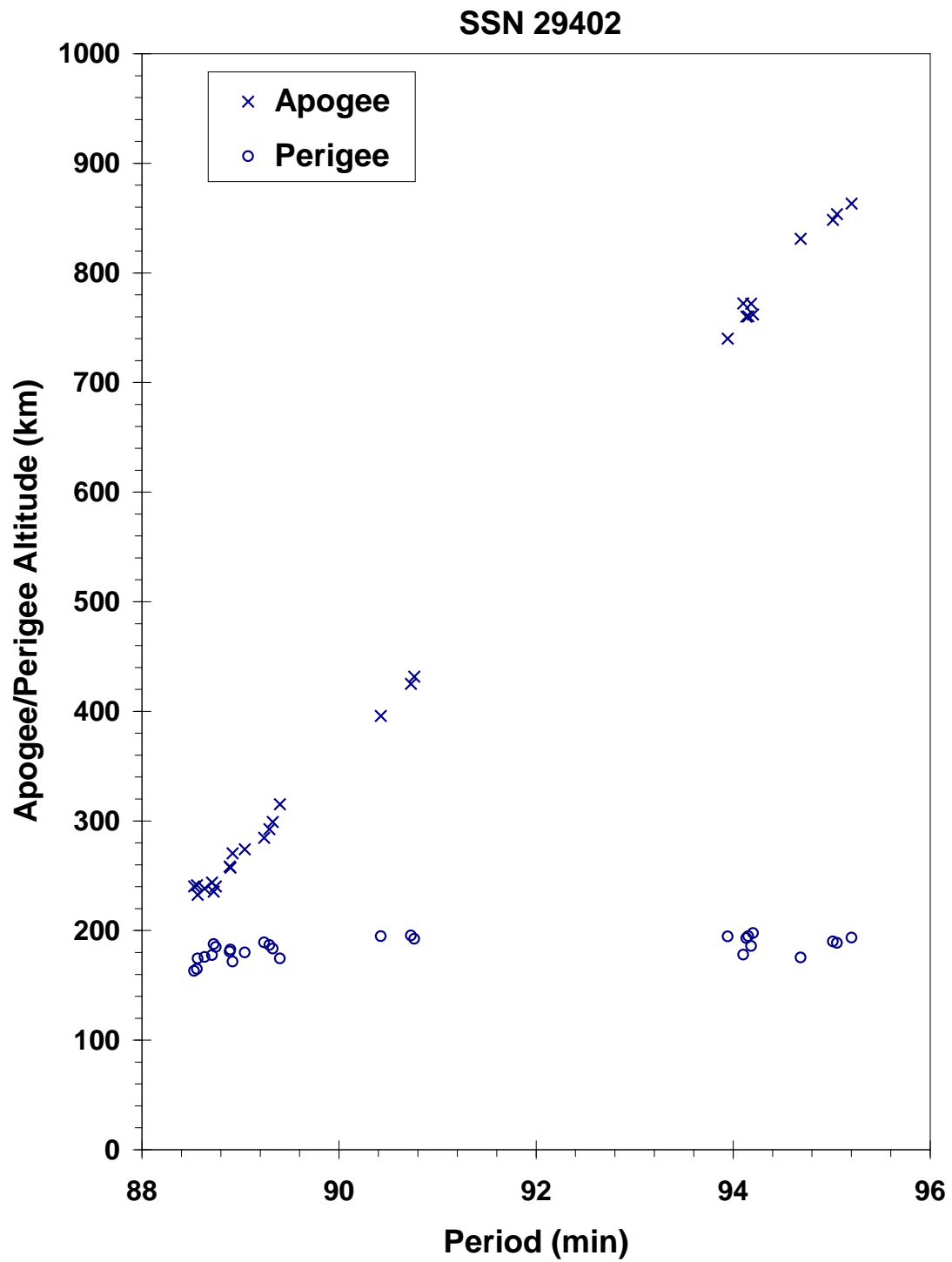
| | |
|----------------------|---------|
| MAXIMUM ΔP : | 5.8 min |
| MAXIMUM ΔI : | 0.5 deg |

COMMENTS

Cosmos 2423 was the eighth of the Cosmos 2031 class of spacecraft, which debuted in 1989. In all seven previous missions, the spacecraft was deliberately exploded at the end of mission. Cosmos 2423 had the shortest lifetime of any of this class of spacecraft, two months instead of a usual four months. Because these spacecraft are deliberately exploded at a low altitude, the lifetime of the trackable debris cloud is usually measured in days.

REFERENCE DOCUMENTS

“Significant Increase in Satellite Breakups During 2006”, The Orbital Debris Quarterly News, NASA JSC, January 2007. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv11i1.pdf>.



SATELLITE DATA

TYPE: Rocket Body
OWNER: US
LAUNCH DATE: 4.58 Nov 2006
DRY MASS (KG): 2850
MAIN BODY: Cylinder; 4.0 m diameter by 12 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of event
ENERGY SOURCES: On-board propellant

EVENT DATA

| | | | |
|-----------|------------|-----------------|---------|
| DATE: | 4 Nov 2006 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 06308.62553010 | MEAN ANOMALY: | 119.1776 |
| RIGHT ASCENSION: | 307.4245 | MEAN MOTION: | 14.13676442 |
| INCLINATION: | 98.7885 | MEAN MOTION DOT/2: | -0.00000044 |
| ECCENTRICITY: | 0.0022197 | MEAN MOTION DOT DOT/6: | 0.0 |
| ARG. OF PERIGEE: | 240.7178 | BSTAR: | 0.0 |

DEBRIS CLOUD DATA

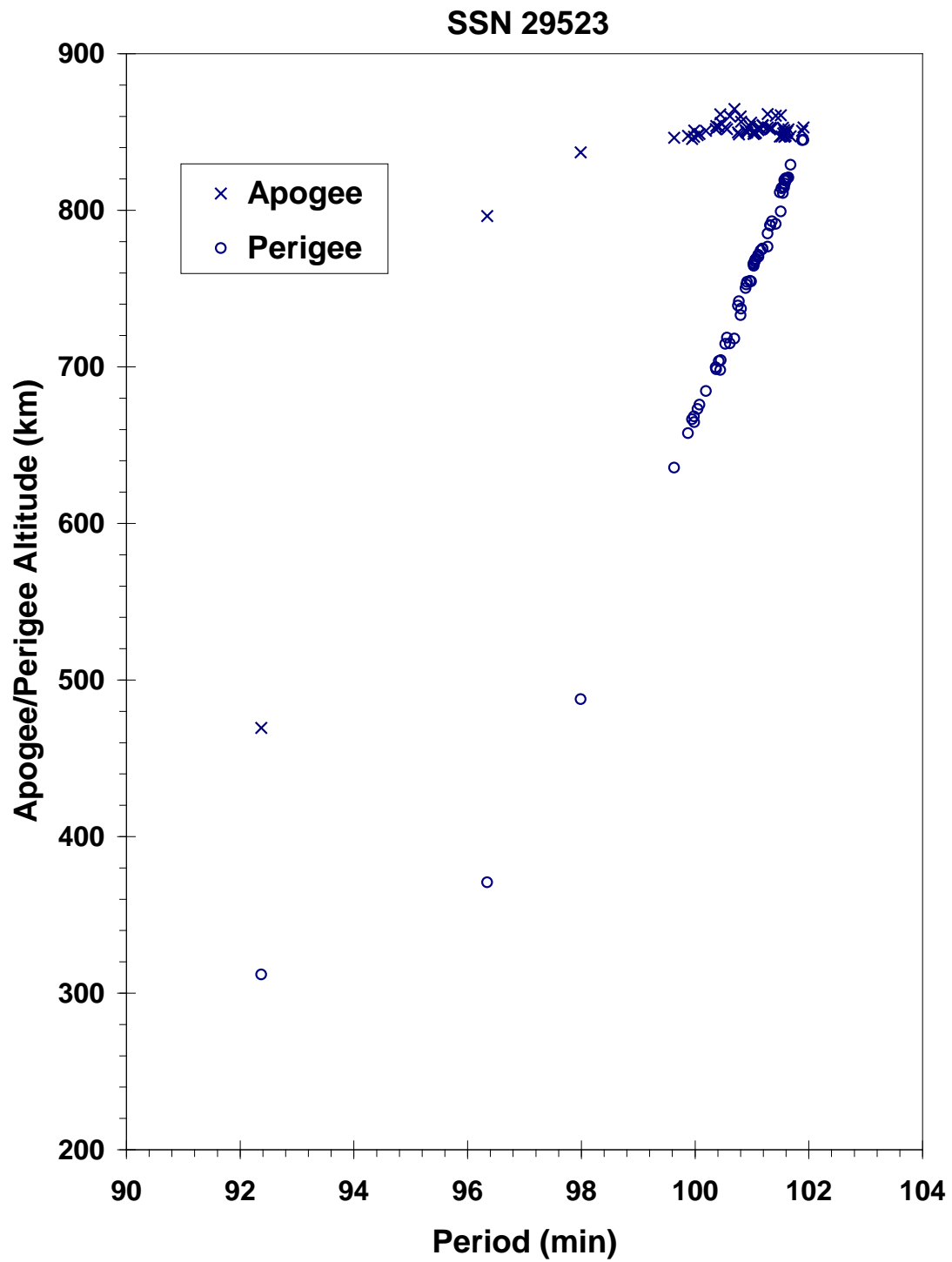
MAXIMUM ΔP : 9.5 min
MAXIMUM ΔI : 0.3 deg

COMMENTS

This was the first major event associated with a Delta IV second stage. Debris appeared to separate from the parent body in a retrograde direction soon after orbit insertion. Over 60 pieces were eventually cataloged from this event. The rocket body did not see any performance degradation and was reentered directly after payload delivery. The cause of the debris release was investigated to ensure any countermeasures can be implemented for future Delta IV missions.

REFERENCE DOCUMENTS

“Significant Increase in Satellite Breakups During 2006”, The Orbital Debris Quarterly News, NASA JSC, January 2007.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNV11i1.pdf>.



Delta 4 debris cloud of 62 fragments less than two weeks after the event.

USA 193

2006-057A

29651

SATELLITE DATA

TYPE: Payload
OWNER: USA
LAUNCH DATE: 14 Dec 2006
DRY MASS (KG): unknown
MAIN BODY: unknown
MAJOR APPENDAGES: unknown
ATTITUDE CONTROL: unknown
ENERGY SOURCES: unknown

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------------------|
| DATE: | 21 Feb 2008 | LOCATION: | 8.813N, 189.023E (asc) |
| TIME: | 0326 GMT | ASSESSED CAUSE: | Deliberate (HVI) |
| ALTITUDE: | 247.769 km | | |

PRE-EVENT ELEMENTS - unavailable

| | |
|------------------|------------------------|
| EPOCH: | MEAN ANOMALY: |
| RIGHT ASCENSION: | MEAN MOTION: |
| INCLINATION: | MEAN MOTION DOT/2: |
| ECCENTRICITY: | MEAN MOTION DOT DOT/6: |
| ARG. OF PERIGEE: | BSTAR: |

DEBRIS CLOUD DATA

MAXIMUM ΔP : unknown min
MAXIMUM ΔI : unknown deg

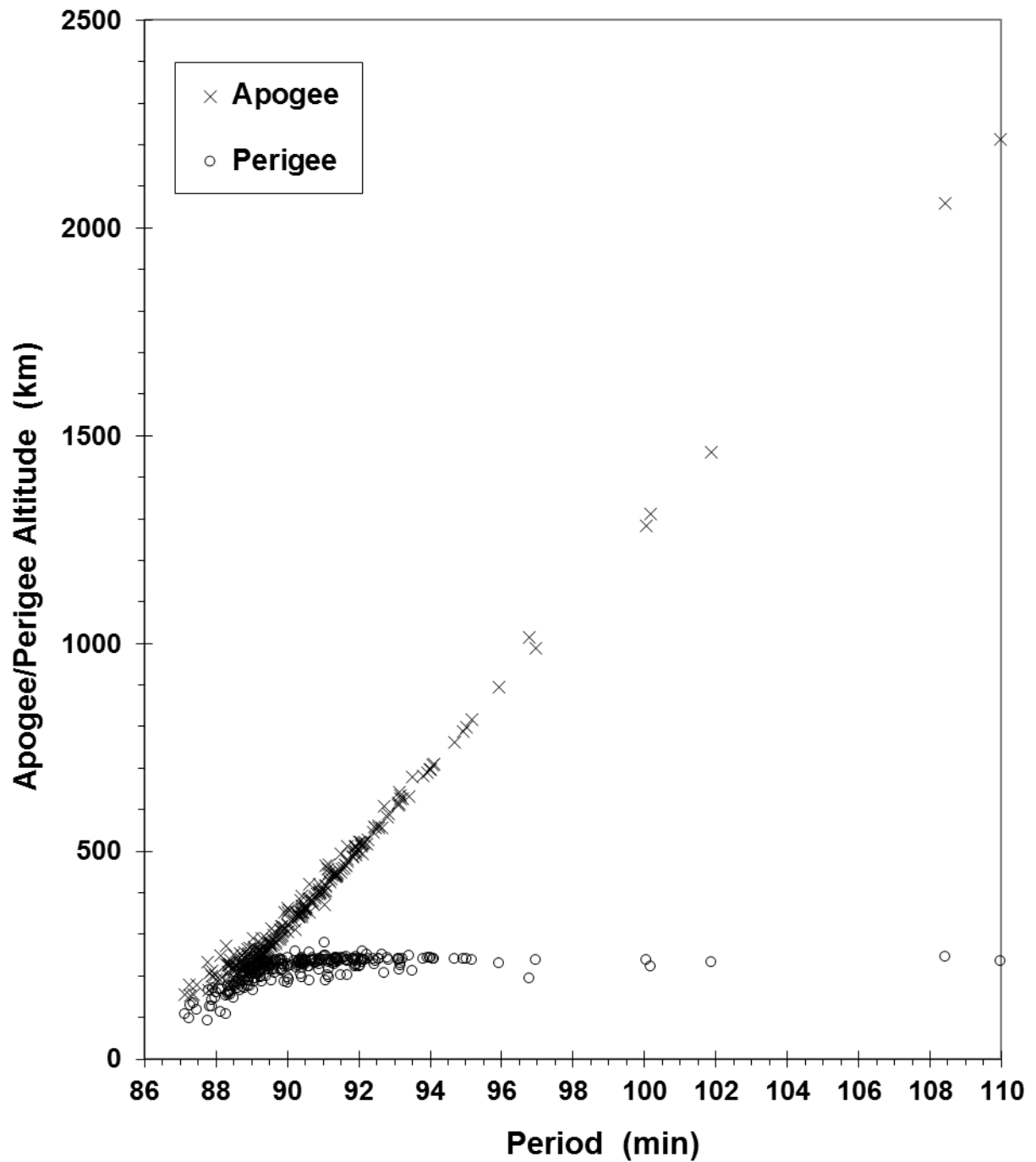
COMMENTS

This spacecraft was intentionally destroyed in order to minimize the potential ground hazard should it reenter with its fuel tank intact. NASA's standard reentry hazard analysis tool indicated a high probability that frozen/slush hydrazine monopropellant would survive intact to the ground, presenting a significant chemical hazard. The intercept was accomplished by an SM-3 missile launched by the US Navy Aegis cruiser USS *Lake Erie*.

REFERENCE DOCUMENT

"Satellite Breakups During First Quarter of 2008", The Orbital Debris Quarterly News, NASA JSC, April 2008.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/ODQNv12i2.pdf>.

SSN 29651



The USA 193 debris cloud, including 174 fragments cataloged up to 1 month after the event.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 25 Dec 2006
DRY MASS (KG): 55
MAIN BODY: ellipsoid; 0.6 m diameter x 1 m length
MAJOR APPENDAGES: none
ATTITUDE CONTROL: none
ENERGY SOURCES: on-board propellants

EVENT DATA

DATE: 27 July 2016
TIME: 0119 GMT
ALTITUDE: 18551.3 km
LOCATION: 15.38S, 124.1E (asc)
ASSESSED CAUSE: Propulsion

PRE-EVENT ELEMENTS

EPOCH: 16208.24667120
RIGHT ASCENSION: 261.9075
INCLINATION: 64.7769
ECCENTRICITY: 0.5783296
ARG. OF PERIGEE: 27.2587
MEAN ANOMALY: 356.6314
MEAN MOTION: 4.23611820
MEAN MOTION DOT/2: +0.00000237
MEAN MOTION DOT DOT/6: 0.00000
BSTAR: +0.00030726

DEBRIS CLOUD DATA

MAXIMUM ΔP : 10 min
MAXIMUM ΔI : 0.7 deg

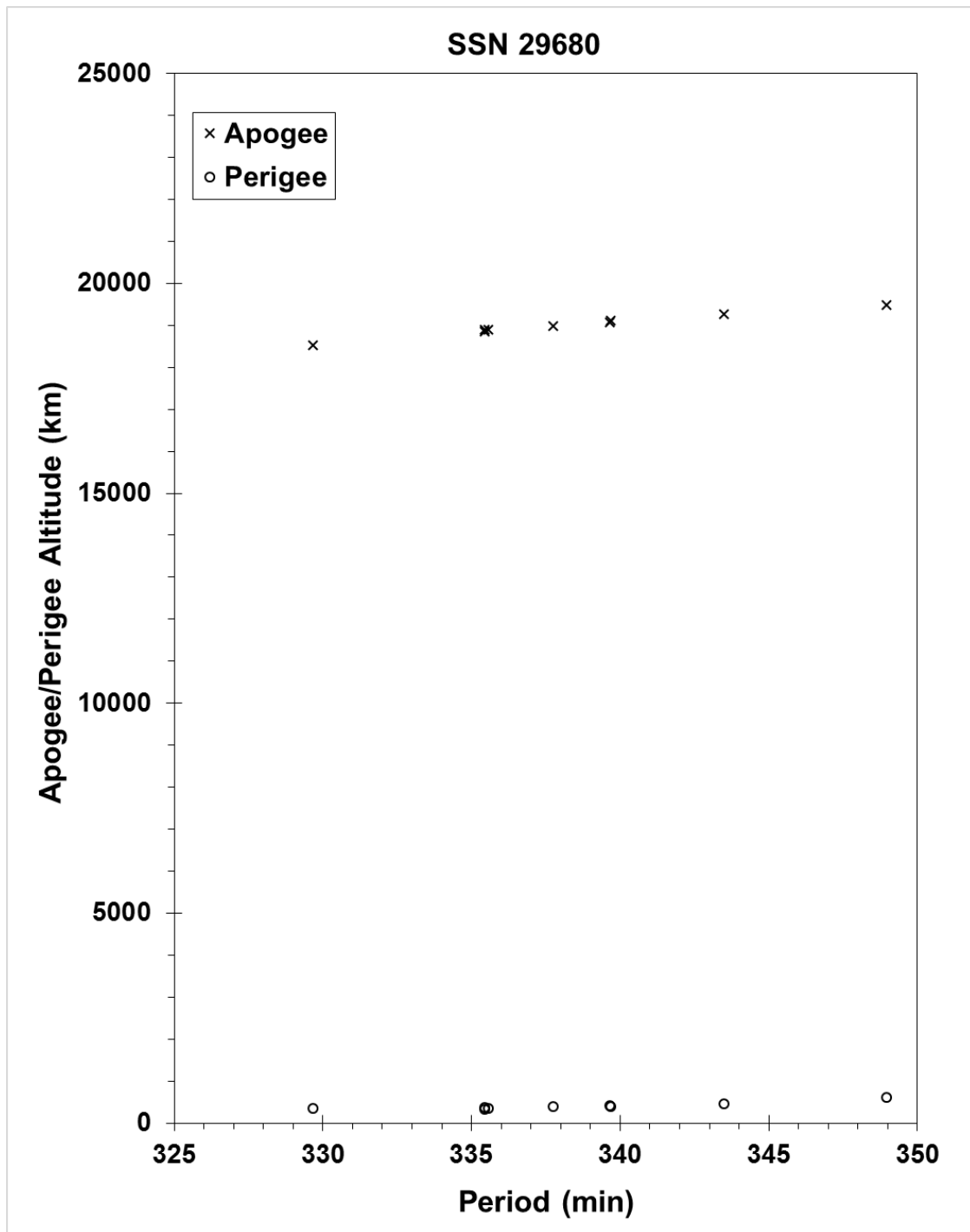
COMMENTS

This event was the 46th known fragmentation of Blok-DM ullage motors. The breakups are believed to be caused by residual propellants. Russian officials have been aware of the problem since 1992 and have made design changes, although the date of full implementation is unknown. Multiple small debris were observed by the SSN, but due to difficulties in cataloging fragments in elliptical and deep-space orbits many more fragments could be present.

REFERENCE DOCUMENTS

History of Soviet/Russian Satellite Fragmentations: A Joint U.S.-Russian Investigation. N.L. Johnson *et al.* Kaman Sciences Corporation, October 2005.

“New SOZ Breakup in July 2016”, The Orbital Debris Quarterly News, NASA JSC, October 2016. Available online at <https://www.orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/odqnv20i4.pdf>.



The COSMOS ullage motor debris cloud includes the parent body and eight debris fragments cataloged up to 75 days post-event.

SATELLITE DATA

TYPE: Rocket Body
OWNER: PRC
LAUNCH DATE: 02 Feb 2007
DRY MASS (KG): 3062
MAIN BODY: Cylinder; 3.0 m diameter x 12.4 m long
MAJOR APPENDAGES: none
ATTITUDE CONTROL: status unknown at time of event
ENERGY SOURCES: on-board propellants

EVENT DATA

| | | | |
|-----------|-------------|-----------------|---------|
| DATE: | 02 Feb 2007 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|--------------|
| EPOCH: | 07033.76400703 | MEAN ANOMALY: | 43.007 |
| RIGHT ASCENSION: | 9.4841 | MEAN MOTION: | 1.91693848 |
| INCLINATION: | 25.0136 | MEAN MOTION DOT/2: | -0.00001383 |
| ECCENTRICITY: | 0.7597957 | MEAN MOTION DOT DOT/6: | 0.0000011914 |
| ARG. OF PERIGEE: | 179.8053 | BSTAR: | 0.0 |

DEBRIS CLOUD DATA

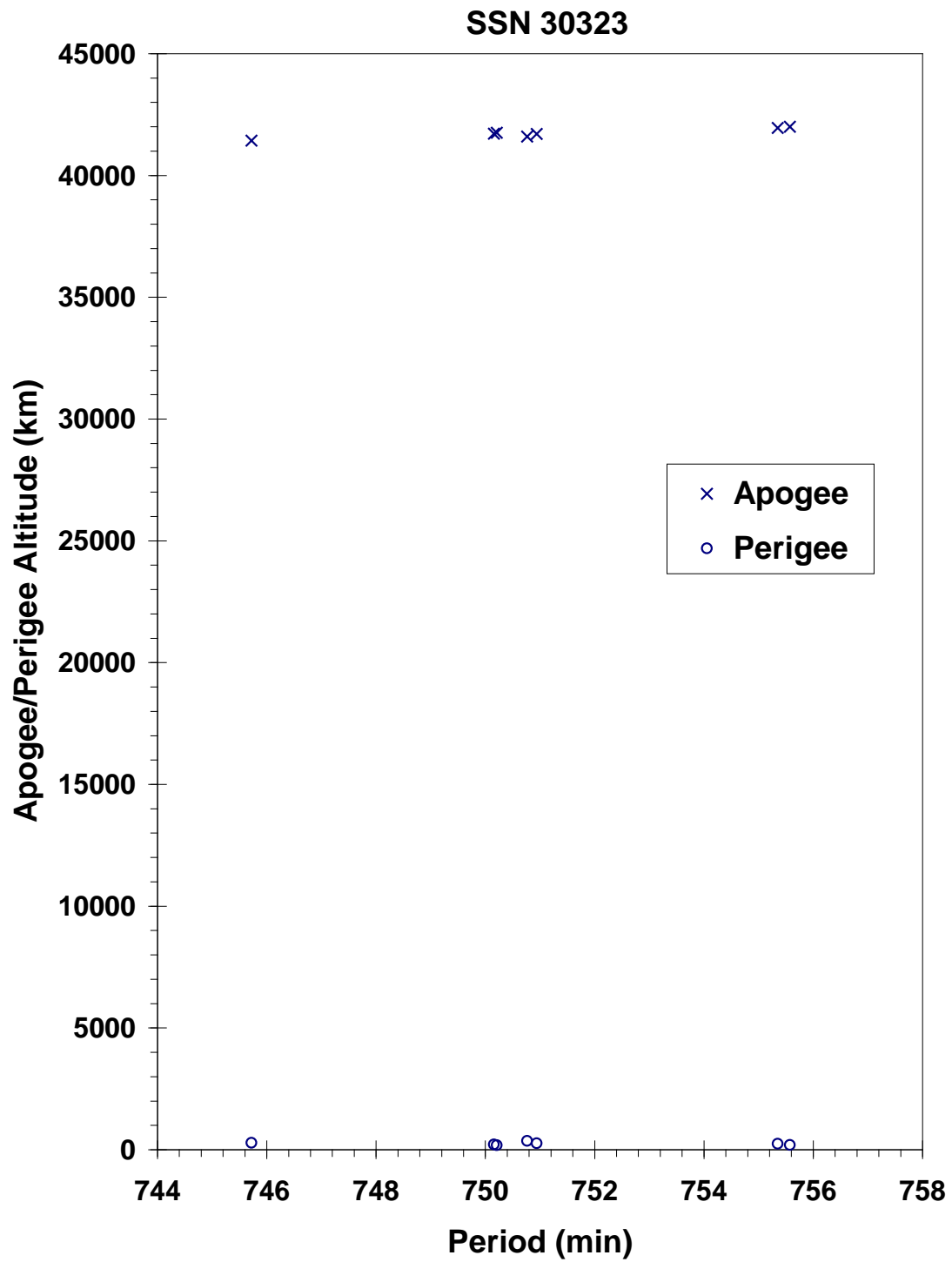
MAXIMUM ΔP : 5.5 min
MAXIMUM ΔI : 0.2 deg

COMMENTS

In the 14th edition of this work, this event was identified as being associated with the payload. To quote: “The spacecraft experienced problems soon after insertion into a geosynchronous transfer orbit. In March 2007, the spacecraft was recovered and maneuvered into GEO. As many as 100 debris were detected by the US SSN.” This event was more recently identified as being associated with the CZ-3A R/B. This is now recognized as being the first known CZ-3 third stage fragmentation event.

REFERENCE DOCUMENT

“Four Satellite Breakups in February Add to Debris Population”, The Orbital Debris Quarterly News, NASA JSC, January 2007. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv11i2.pdf>.



Identified Beidou R/B debris cloud 2 weeks after the event as reconstructed from the US SSN database.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: Japan
LAUNCH DATE: 24 Feb 2007
DRY MASS (KG): unknown
MAIN BODY: truncated cone; 4.070 m diameter tapering to < 1.98 m diameter x ~ 1.5 m length
MAJOR APPENDAGES: none
ATTITUDE CONTROL: none
ENERGY SOURCES: unknown

EVENT DATA

DATE: 23 Dec 2010
TIME: 1600 GMT
ALTITUDE: Unknown
LOCATION: Unknown
ASSESSED CAUSE: Unknown

PRE-EVENT ELEMENTS

EPOCH: 10357.22196763
RIGHT ASCENSION: 114.8768
INCLINATION: 97.3356
ECCENTRICITY: 0.0005569
ARG. OF PERIGEE: 63.1040
MEAN ANOMALY: 297.1353
MEAN MOTION: 15.43699236
MEAN MOTION DOT/2: 0.00018749
MEAN MOTION DOT DOT/6: 0.0
BSTAR: 0.0

DEBRIS CLOUD DATA

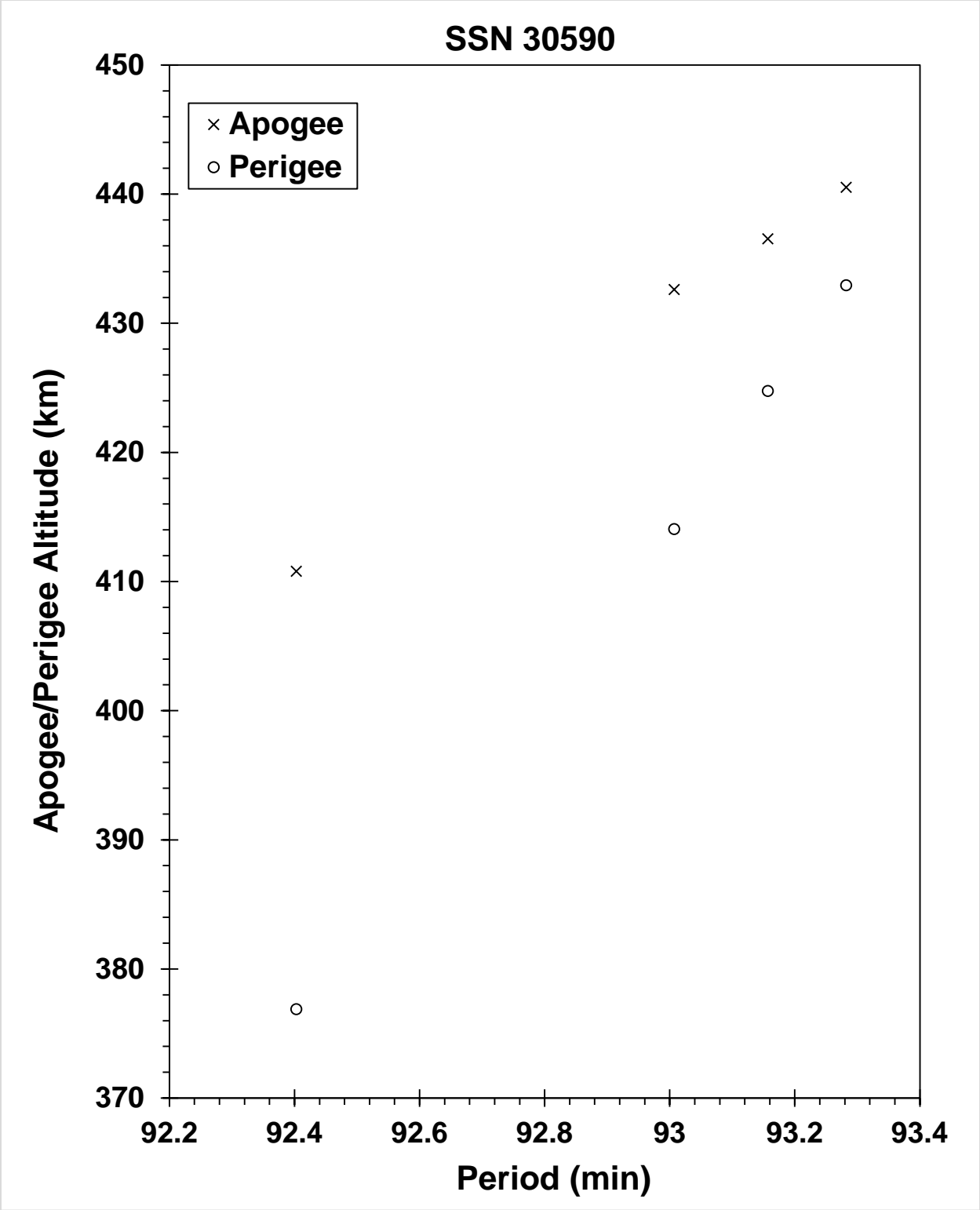
MAXIMUM ΔP : 0.9 min
MAXIMUM ΔI : 0.0 deg

COMMENTS

Parent object is the upper cap section of the three-piece Kawasaki Heavy Industries, Ltd. 4/4D-LC dual payload fairing's lower payload encapsulation system. Six objects were initially observed and three were officially cataloged. These objects displayed large area-to-mass ratios (0.38-0.85 m²/kg) and decayed rapidly from orbit. Subsequently an additional 11 pieces entered the catalog. All have decayed.

REFERENCE DOCUMENT

"New Satellite Fragmentations Add to Debris Population", The Orbital Debris Quarterly News, NASA JSC, January 2011.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv15i1.pdf>.



Four HII-A debris fragments cataloged within 5 days of the event.

SATELLITE DATA

| | |
|-------------------|----------------------------|
| TYPE: | Payload |
| OWNER: | CIS |
| LAUNCH DATE: | 26 Oct 2007 |
| DRY MASS (KG): | 3250 |
| MAIN BODY: | stepped cylinder |
| MAJOR APPENDAGES: | solar panels & antennas |
| ATTITUDE CONTROL: | gravity gradient (passive) |
| ENERGY SOURCES: | unknown at time of event |

EVENT DATA

| | | | |
|-----------|-------------|-----------------|---------|
| DATE: | 10 May 2014 | LOCATION: | TBD |
| TIME: | 2024 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | TBD km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 14129.36499359 | MEAN ANOMALY: | 236.8999 |
| RIGHT ASCENSION: | 270.8988 | MEAN MOTION: | 14.12320352 |
| INCLINATION: | 70.9495 | MEAN MOTION DOT/2: | 0.00000475 |
| ECCENTRICITY: | 0.0010837 | MEAN MOTION DOT DOT/6: | 0.00000 |
| ARG. OF PERIGEE: | 200.7382 | BSTAR: | 0.0028345 |

DEBRIS CLOUD DATA

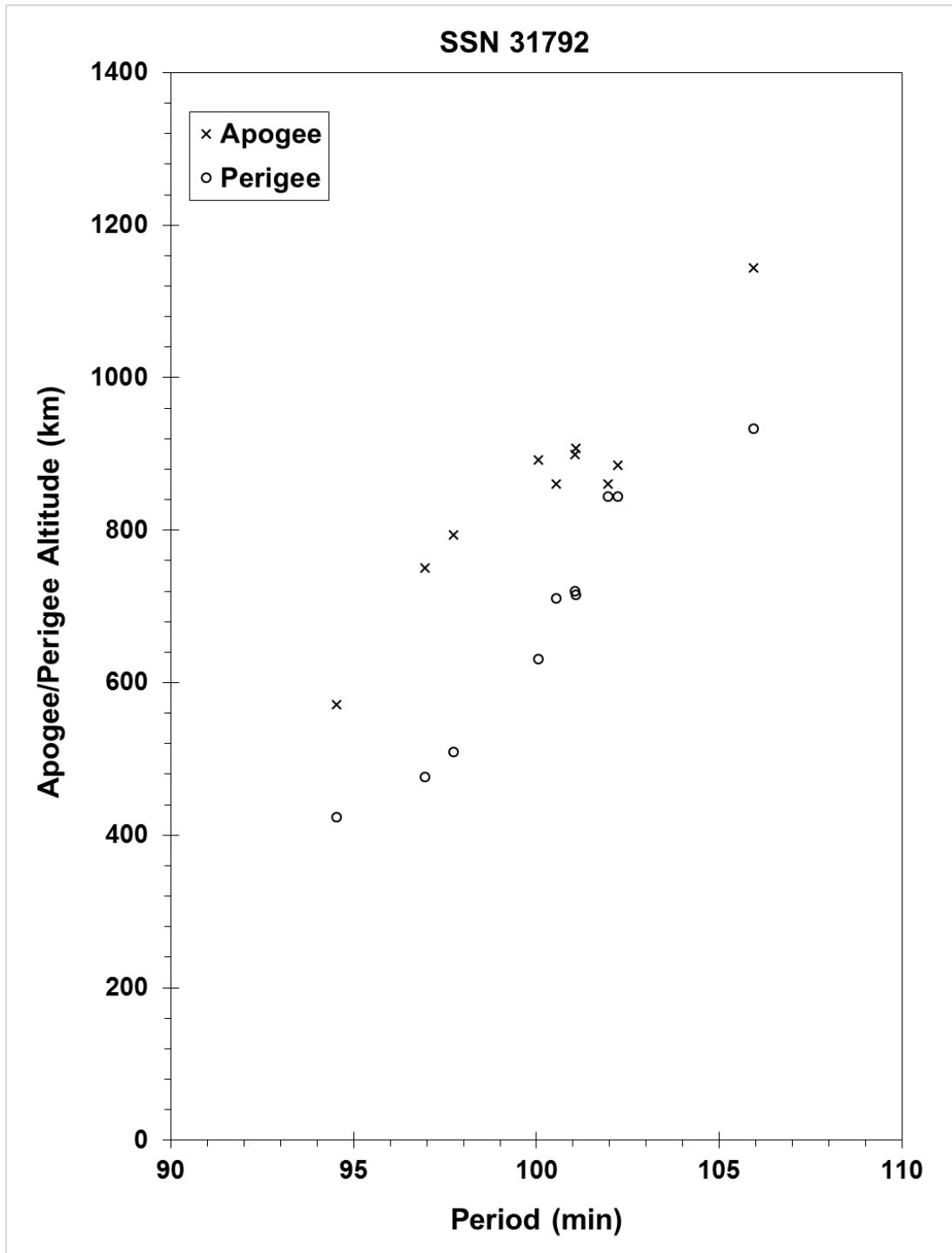
| | |
|----------------------|---------|
| MAXIMUM ΔP : | TBD min |
| MAXIMUM ΔI : | TBD deg |

COMMENTS

Cosmos 2428 is believed to be the last Tselina-2 class spacecraft flown. Approximately 17 fragments have been observed, but only nine have been officially cataloged. Of these, eight have decayed from orbit by 23 September 2014. The remaining fragment is in an orbit similar to four pieces of previously-cataloged SL-16 launch vehicle debris, and may not be related to this event.

REFERENCE DOCUMENTS

“Flurry of Small Breakups in First Half of 2014”, The Orbital Debris Quarterly News, NASA JSC, July 2014.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv18i3.pdf>.



Cosmos 2428 debris cloud cataloged approximately one week after event.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 26 Oct 2007
DRY MASS (KG): 56
MAIN BODY: ellipsoid; 0.6 m diameter x 1 m length
MAJOR APPENDAGES: none
ATTITUDE CONTROL: none
ENERGY SOURCES: on-board propellants

EVENT DATA

DATE: 13 Aug 2014 LOCATION: 30.7S, 201.4E (dsc)
TIME: 1340 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 6877.5 km

PRE-EVENT ELEMENTS

EPOCH: 14223.45723491 MEAN ANOMALY: 338.9903
RIGHT ASCENSION: 24.9798 MEAN MOTION: 4.23372123
INCLINATION: 65.3851 MEAN MOTION DOT/2: -0.00001815
ECCENTRICITY: 0.5593625 MEAN MOTION DOT DOT/6: 0.0
ARG. OF PERIGEE: 72.9539 BSTAR: -0.19915

DEBRIS CLOUD DATA

MAXIMUM ΔP : 0 min
MAXIMUM ΔI : 0 deg

COMMENTS

This is the 43rd known breakup of a Proton Blok-DM SOZ ullage unit since 1984. The breakups are believed to be caused by residual propellants. Russian officials have been aware of the problem since 1992 and have made design changes, although the date of full implementation is unknown. Approximately 70 fragments have been observed, but none have been officially cataloged. Due to the difficulty in observing and cataloging debris in deep-space, highly elliptical orbits, a much larger debris ensemble may have been created by this event.

REFERENCE DOCUMENTS

History of Soviet/Russian Satellite Fragmentations: A Joint U.S.-Russian Investigation. N.L. Johnson et al. Kaman Sciences Corporation, October 2005.

“Three Additional Breakups Mar 2014”, The Orbital Debris Quarterly News, NASA JSC, October 2014.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv18i4.pdf>.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Rocket Body
OWNER: USA
LAUNCH DATE: 11 Nov 2007
DRY MASS (KG): 3,490.0
MAIN BODY: stepped cylinder: 5 m (max) diameter x 13.7 m length
MAJOR APPENDAGES: none
ATTITUDE CONTROL: active; 3 axis RCS
ENERGY SOURCES: on-board propellants

EVENT DATA

DATE: 11 Nov 2007
TIME: Unknown
ALTITUDE: Unknown
LOCATION: Unknown
ASSESSED CAUSE: Unknown

PRE-EVENT ELEMENTS - unavailable

EPOCH:
RIGHT ASCENSION:
INCLINATION:
ECCENTRICITY:
ARG. OF PERIGEE:
MEAN ANOMALY:
MEAN MOTION:
MEAN MOTION DOT/2:
MEAN MOTION DOT DOT/6:
BSTAR:

DEBRIS CLOUD DATA

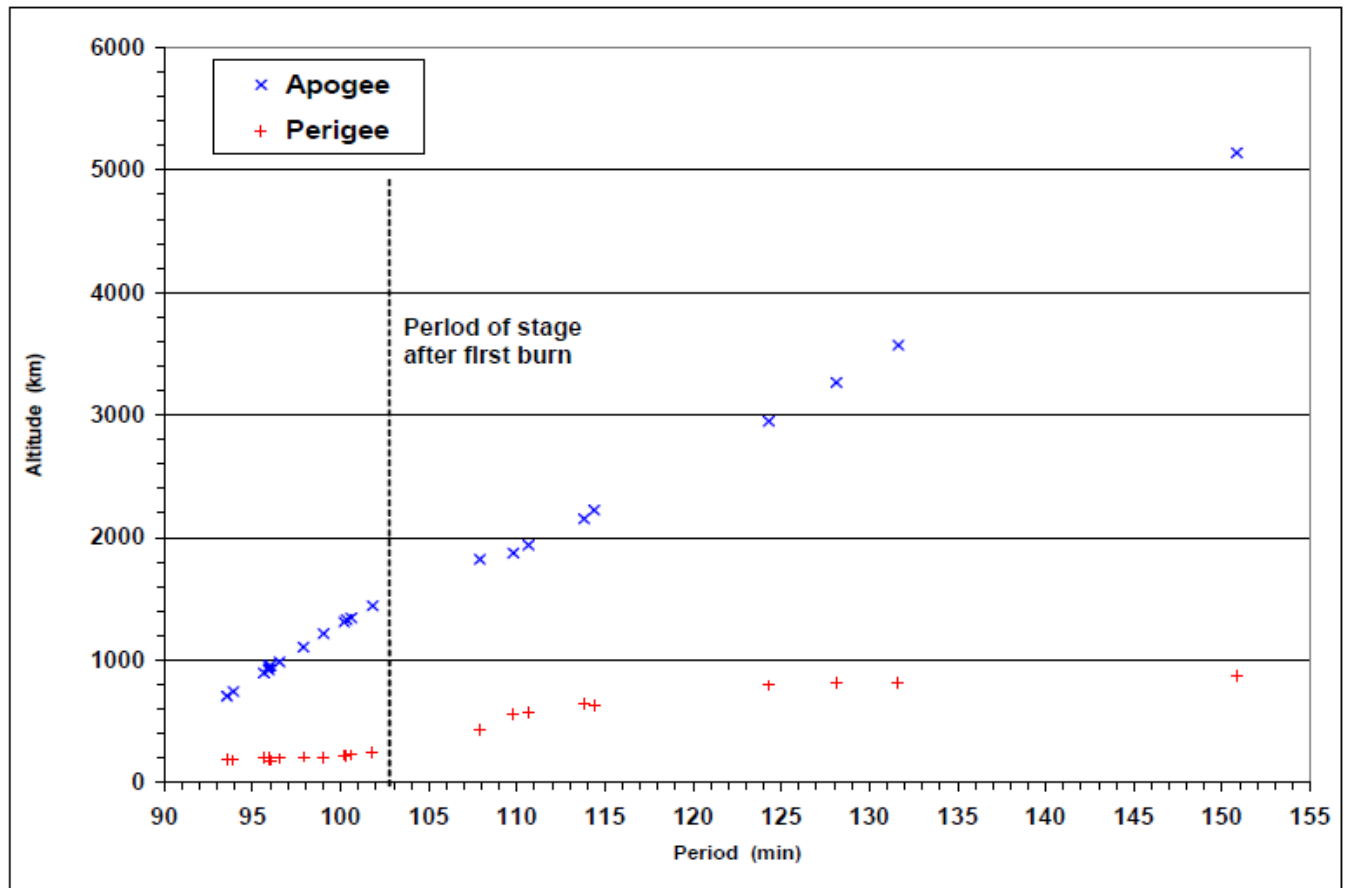
MAXIMUM ΔP : unknown min
MAXIMUM ΔI : unknown deg

COMMENTS

First operational launch of a Delta IV Heavy (Delta 9250H), though second launch of the 5 m-diameter cryogenic second stage. The flight plan called for the second stage to insert itself and the payload into three distinct orbits: a LEO parking orbit; a geosynchronous transfer orbit; and a near geosynchronous final orbit. At least two dozen debris appear to have been released in the parking orbit of approximately 1575 km x 220 km. The debris release event had no apparent effect on subsequent orbit maneuvers, and the payload was successfully delivered to the planned final orbit.

REFERENCE DOCUMENT

“Two Minor Fragmentations End Worst Debris Year Ever”, [The Orbital Debris Quarterly News](http://orbitaldebris.jsc.nasa.gov/newsletter/pdfs/ODQNv12i1.pdf), NASA JSC, January 2008.
Available online at <http://orbitaldebris.jsc.nasa.gov/newsletter/pdfs/ODQNv12i1.pdf>.



The Delta IV second stage R/B debris cloud of two dozen uncatalogued fragments were identified soon after launch.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 25 Dec 2007
DRY MASS (KG): 56.0
MAIN BODY: ellipsoid; 0.6 m diameter x 1 m length
MAJOR APPENDAGES: none
ATTITUDE CONTROL: none
ENERGY SOURCES: on-board propellants

EVENT DATA

| | | | |
|-----------|-------------|-----------------|---------|
| DATE: | 18 Aug 2011 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|------------|
| EPOCH: | 11230.12943848 | MEAN ANOMALY: | 279.9120 |
| RIGHT ASCENSION: | 286.1835 | MEAN MOTION: | 4.23764448 |
| INCLINATION: | 65.0463 | MEAN MOTION DOT/2: | 0.00000228 |
| ECCENTRICITY: | 0.5712685 | MEAN MOTION DOT DOT/6: | 0.0 |
| ARG. OF PERIGEE: | 140.2548 | BSTAR: | 0.0 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : 0.0 min
MAXIMUM ΔI : 0.0 deg

COMMENTS

This event was the 38th known fragmentation of Blok-DM ullage motors. The breakups are believed to be caused by residual propellants. Russian officials have been aware of the problem since 1992 and have made design changes, although the date of full implementation is unknown. Although some small debris were observed by the SSN, none have been officially cataloged.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations: A Joint U.S.-Russian Investigation. N.L. Johnson et al. Kaman Sciences Corporation, October 2005.

“Only a Few Minor Satellite Breakups in 2011”, The Orbital Debris Quarterly News, NASA JSC, January 2012.
Available online at <http://orbitaldebris.jsc.nasa.gov/newsletter/pdfs/ODQNV16i1.pdf>.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Rocket Body
OWNER: CIS
LAUNCH DATE: 14 Mar 2008
DRY MASS (KG): 2510
MAIN BODY: Cylinder + toroid; 4.1 m diameter x 2.65 m length
MAJOR APPENDAGES: none
ATTITUDE CONTROL: none at time of event
ENERGY SOURCES: on-board propellants

EVENT DATA

DATE: 13 Oct 2010
TIME: 0553 GMT
ALTITUDE: 10,799.1 km
LOCATION: 33.768N, 120.670E (asc)
ASSESSED CAUSE: Propulsion

PRE-EVENT ELEMENTS

EPOCH: 10285.89612768
RIGHT ASCENSION: 195.2356
INCLINATION: 48.9364
ECCENTRICITY: 0.6485876
ARG. OF PERIGEE: 287.1455
MEAN ANOMALY: 15.0680
MEAN MOTION: 3.07372467
MEAN MOTION DOT/2: 0.00000252
MEAN MOTION DOT DOT/6: 0.0
BSTAR: 0.0025757

DEBRIS CLOUD DATA

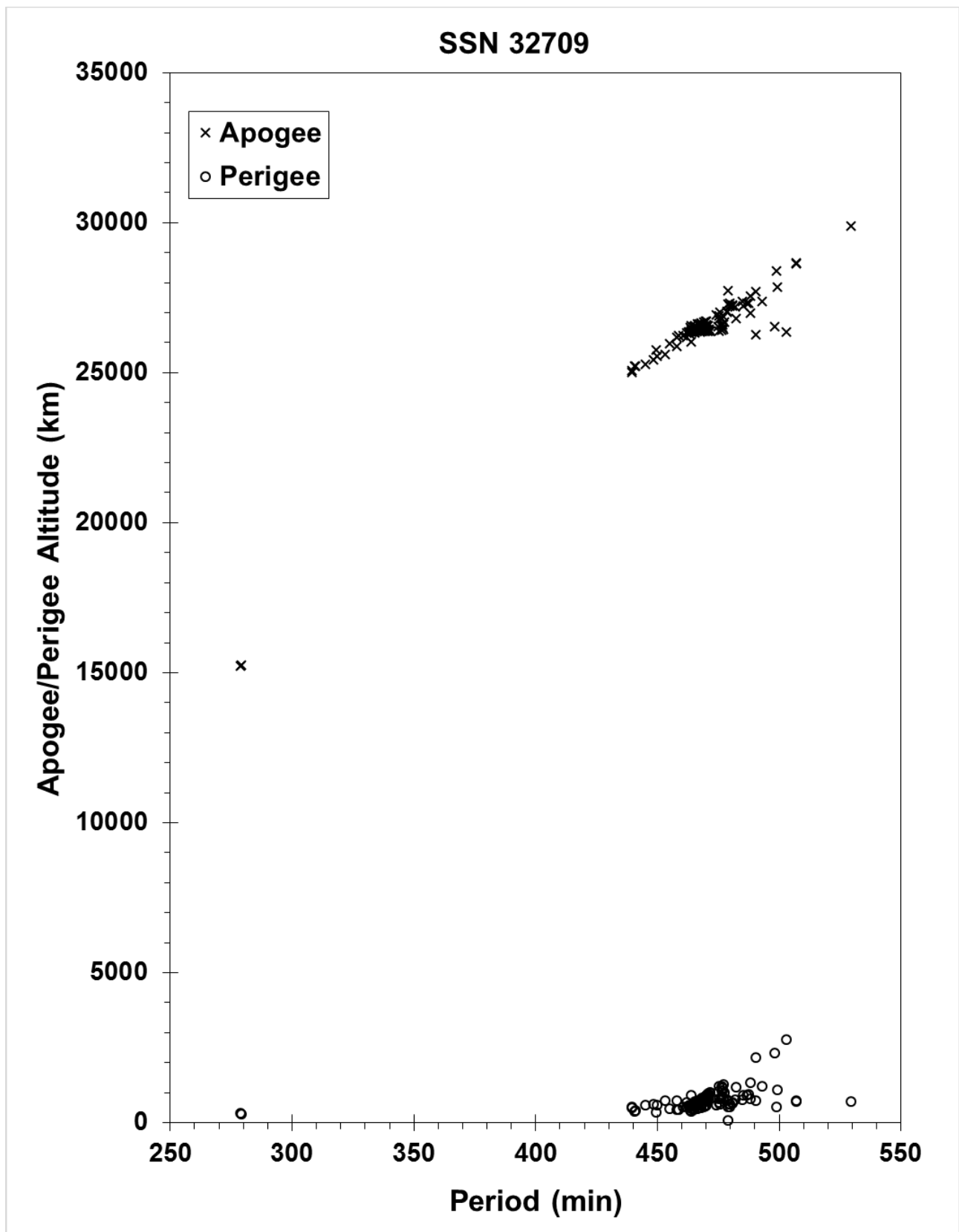
MAXIMUM ΔP : 189.6 min
MAXIMUM ΔI : 2.2 deg

COMMENTS

Event occurred 31 months after launch. The dry mass quoted above includes that of the *Briz-M* core stage and the toroidal Auxiliary Propulsion Tank (APT)—sometimes referred to as the Additional Fuel Tank (AFT); the malfunction stranding the stage in its elliptical orbit occurred prior to APT jettison. An estimated five metric tons of propellant was aboard.

REFERENCE DOCUMENT

“New Satellite Fragmentations Add to Debris Population”, The Orbital Debris Quarterly News, NASA JSC, January 2011.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv15i1.pdf>.



The *Briz-M* R/B debris cloud of 116 fragments cataloged up to 3.6 years after the event.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 25 Sep 2008
DRY MASS (KG): 56
MAIN BODY: ellipsoid; 0.6 m diameter x 1 m length
MAJOR APPENDAGES: none
ATTITUDE CONTROL: none
ENERGY SOURCES: on-board propellants

EVENT DATA

DATE: 20 May 2014 LOCATION: 62.4S, 148.5E (asc)
TIME: 0032 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 16,288.3 km

PRE-EVENT ELEMENTS

EPOCH: 14135.66838656 MEAN ANOMALY: 322.5973
RIGHT ASCENSION: 154.3825 MEAN MOTION: 4.23142676
INCLINATION: 65.3219 MEAN MOTION DOT/2: -0.00000189
ECCENTRICITY: 0.5517601 MEAN MOTION DOT DOT/6: 0.00000
ARG. OF PERIGEE: 101.3049 BSTAR: 0.00000

DEBRIS CLOUD DATA

MAXIMUM ΔP : 12.7 min
MAXIMUM ΔI : 0.3 deg

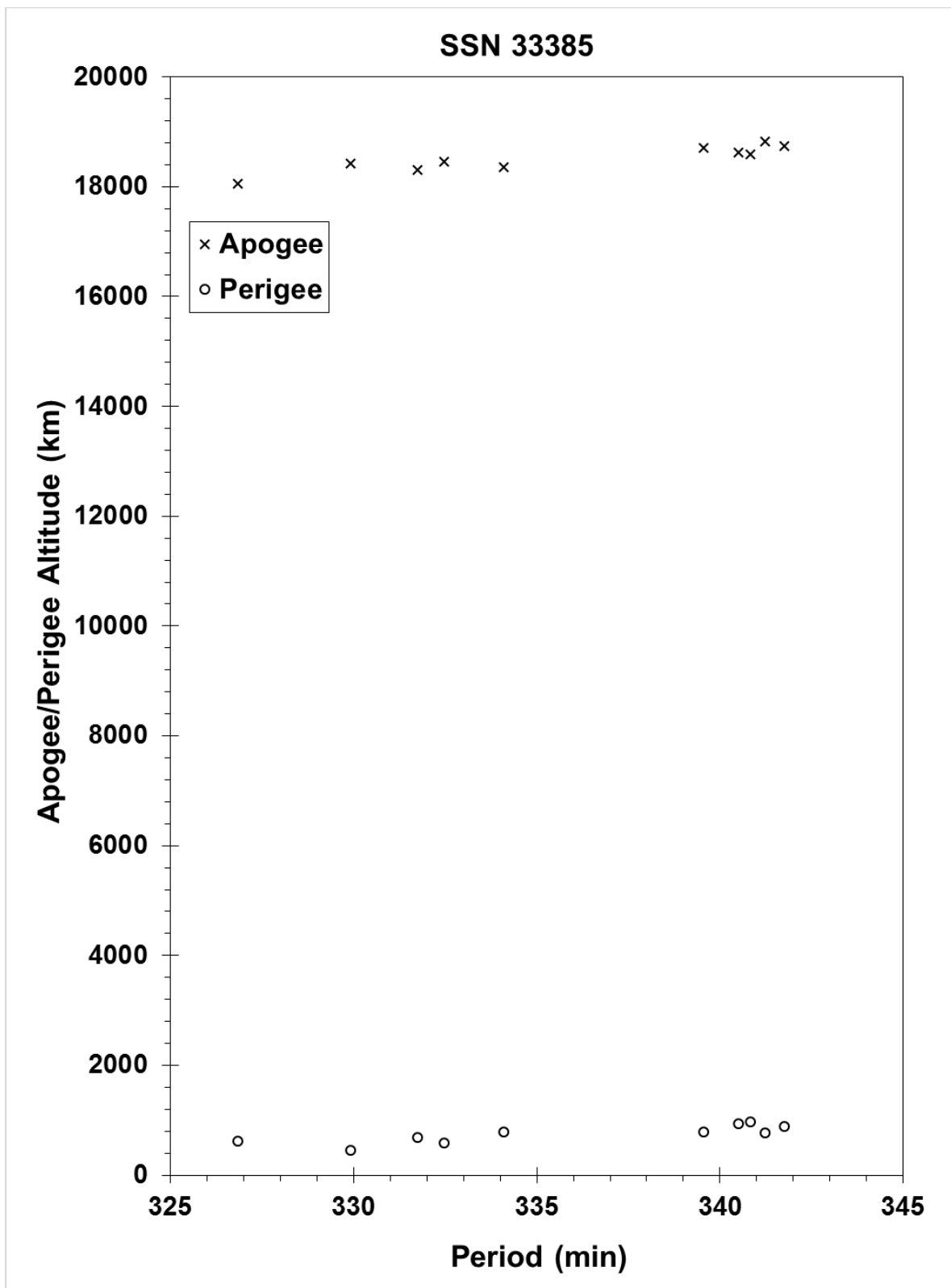
COMMENTS

This is the 40th known breakup of a Proton Blok-DM SOZ ullage unit since 1984. The breakups are believed to be caused by residual propellants. Russian officials have been aware of the problem since 1992 and have made design changes, although the date of full implementation is unknown. Due to the difficulty in observing and cataloging debris in deep-space, highly elliptical orbits, a debris ensemble much larger than that cataloged may have been created by this event.

REFERENCE DOCUMENT

History of Soviet/Russian Satellite Fragmentations: A Joint U.S.-Russian Investigation. N.L. Johnson et al. Kaman Sciences Corporation, October 2005.

“Flurry of Small Breakups in First Half of 2014”, The Orbital Debris Quarterly News, NASA JSC, July 2014.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv18i3.pdf>.



The COSMOS ullage motor debris cloud includes the parent body and 10 debris fragments (this plot omits 2008-046S, SSN#40121).

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 25 Dec 2008
DRY MASS (KG): 56.0
MAIN BODY: ellipsoid; 0.6 m diameter x 1 m length
MAJOR APPENDAGES: none
ATTITUDE CONTROL: none
ENERGY SOURCES: on-board propellants

EVENT DATA

DATE: 26 Mar 2016 LOCATION: 46.91S, 285.50E (asc)
TIME: 1212 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 18075.34 km

PRE-EVENT ELEMENTS

EPOCH: 16084.51547544 MEAN ANOMALY: 341.8531
RIGHT ASCENSION: 57.2039 MEAN MOTION: 4.23930172
INCLINATION: 65.3219 MEAN MOTION DOT/2: +0.00000377
ECCENTRICITY: 0.5630529 MEAN MOTION DOT DOT/6: 0.00000
ARG. OF PERIGEE: 66.1764 BSTAR: +0.0029410

DEBRIS CLOUD DATA

MAXIMUM ΔP : 11.4 min
MAXIMUM ΔI : 0.2 deg

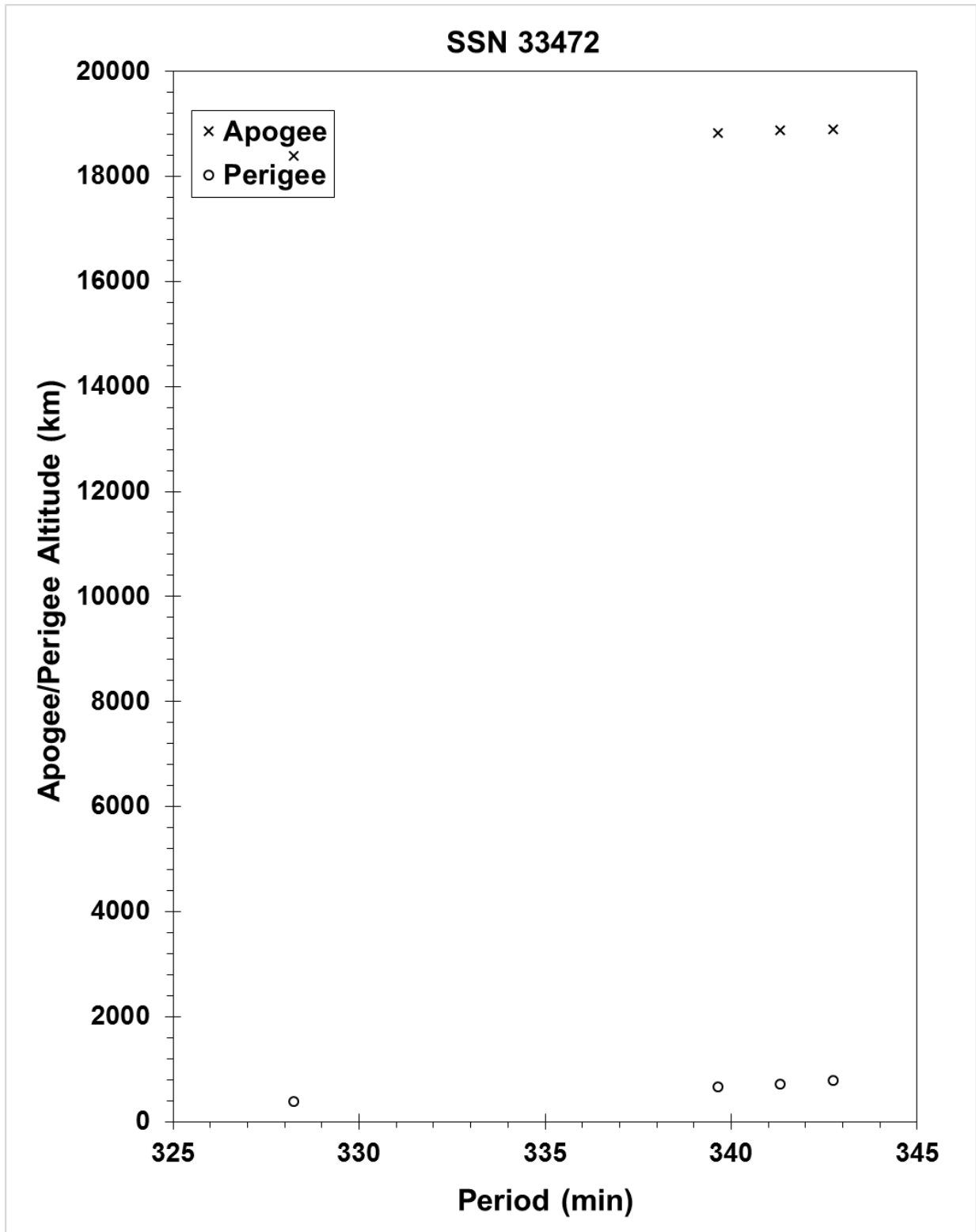
COMMENTS

This event was the 44th known fragmentation of Blok-DM ullage motors. The breakups are believed to be caused by residual propellants. Russian officials have been aware of the problem since 1992 and have made design changes, although the date of full implementation is unknown. Twenty-one pieces were initially observed. Due to the difficulty of cataloging debris in elliptical and deep space orbits there may be many more debris fragments than this number or those cataloged.

REFERENCE DOCUMENTS

History of Soviet/Russian Satellite Fragmentations: A Joint U.S.-Russian Investigation. N.L. Johnson *et al.* Kaman Sciences Corporation, October 2005.

“Russian SOZ Unit Breakup Up in March”, The Orbital Debris Quarterly News, NASA JSC, April 2016.
Available online at <https://www.orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/odqnv20i1-2.pdf>.



The COSMOS ullage motor debris cloud includes the parent body and three debris fragments.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 25 Dec 2008
DRY MASS (KG): 56.0
MAIN BODY: ellipsoid; 0.6 m diameter x 1 m length
MAJOR APPENDAGES: none
ATTITUDE CONTROL: none
ENERGY SOURCES: on-board propellants

EVENT DATA

DATE: 1 June 2016
TIME: 0925 GMT
ALTITUDE: 7148.03 km
LOCATION: 32.86S, 339.90E (dsc)
ASSESSED CAUSE: Propulsion

PRE-EVENT ELEMENTS

EPOCH: 16152.95219263
RIGHT ASCENSION: 28.8315
INCLINATION: 65.2630
ECCENTRICITY: 0.5609396
ARG. OF PERIGEE: 72.5352
MEAN ANOMALY: 5.7190
MEAN MOTION: 4.23951640
MEAN MOTION DOT/2: +0.00005748
MEAN MOTION DOT DOT/6: 0.00000
BSTAR: +0.053896

DEBRIS CLOUD DATA

MAXIMUM ΔP : 8.9 min
MAXIMUM ΔI : 0.4 deg

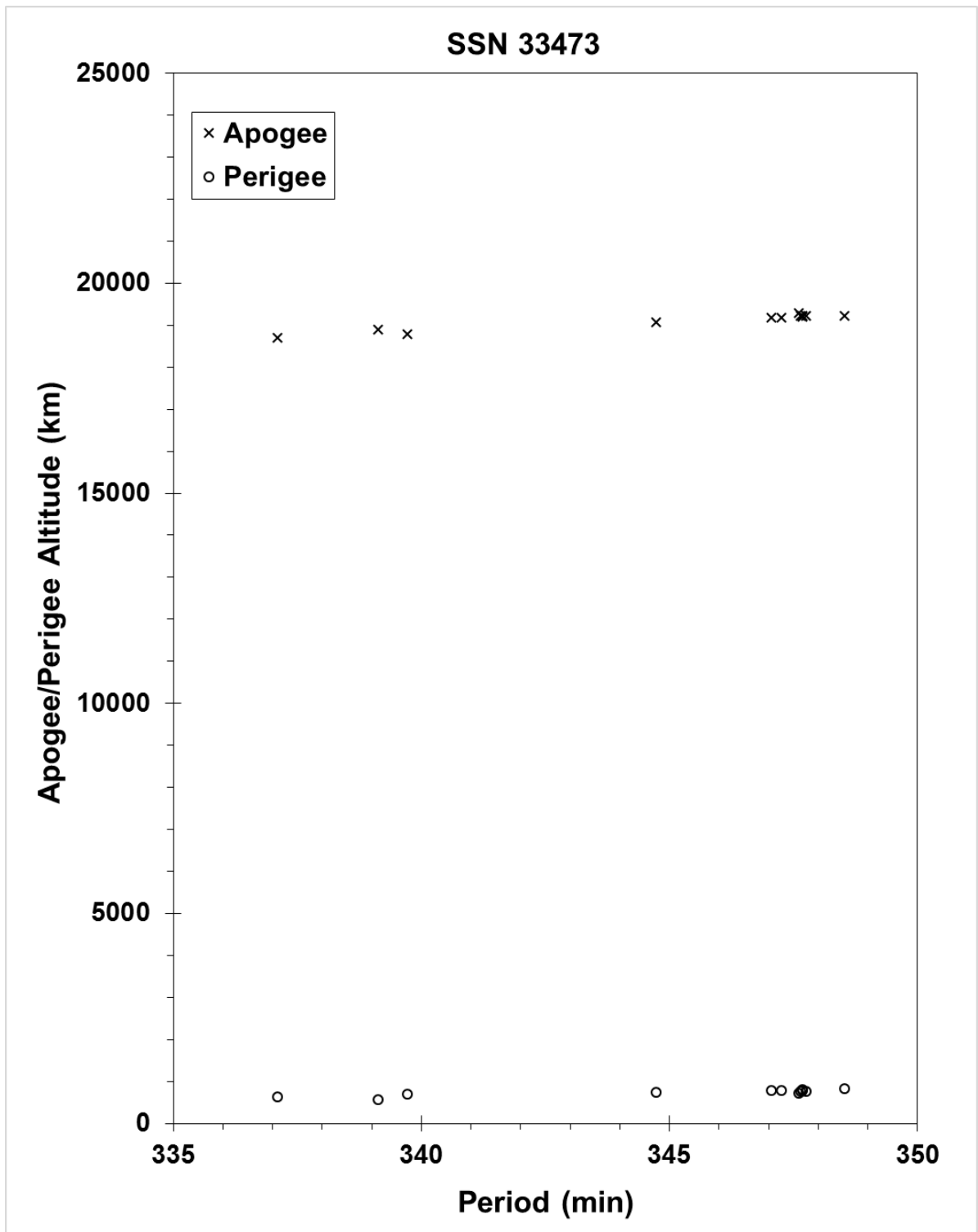
COMMENTS

This event was the 45th known fragmentation of Blok-DM ullage motors. The breakups are believed to be caused by residual propellants. Russian officials have been aware of the problem since 1992 and have made design changes, although the date of full implementation is unknown. Due to the difficulty of cataloging debris in elliptical and deep space orbits there may be many more debris fragments than those cataloged.

REFERENCE DOCUMENTS

History of Soviet/Russian Satellite Fragmentations: A Joint U.S.-Russian Investigation. N.L. Johnson *et al.* Kaman Sciences Corporation, October 2005.

“Two Additional Russian Breakups in 2016”, The Orbital Debris Quarterly News, NASA JSC, July 2016.
Available online at <https://www.orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/odqnv20i3.pdf>.



The COSMOS ullage motor debris cloud includes the parent body and 11 debris fragments cataloged up to 37 days post-event.

SATELLITE DATA

TYPE: payload
OWNER: PRC
LAUNCH DATE: 14 Apr. 2009
DRY MASS (KG): 1100
MAIN BODY: DFH-3 bus derivative; box 2.2 m long x 1.72 m wide x 2.0 m high
MAJOR APPENDAGES: solar arrays and antennas
ATTITUDE CONTROL: three-axis stabilized
ENERGY SOURCES: propellant, batteries, reaction wheels

EVENT DATA

| | | | |
|-----------|--------------|-----------------|---------|
| DATE: | 29 June 2016 | LOCATION: | unknown |
| TIME: | unknown | ASSESSED CAUSE: | unknown |
| ALTITUDE: | unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 16180.12570702 | MEAN ANOMALY: | 164.3864 |
| RIGHT ASCENSION: | 61.3828 | MEAN MOTION: | 1.00365151 |
| INCLINATION: | 4.7106 | MEAN MOTION DOT/2: | -0.00000325 |
| ECCENTRICITY: | 0.0089308 | MEAN MOTION DOT DOT/6: | 0.00000 |
| ARG. OF PERIGEE: | 195.1803 | BSTAR: | 0.00000 |

DEBRIS CLOUD DATA

MAXIMUM ΔP : unknown
MAXIMUM ΔI : unknown

COMMENTS

This spacecraft was the first launch of the People's Republic of China (PRC) BeiDou 2nd generation regional navigation satellites in the Compass Navigation Satellite System, and is sometimes labeled as "Beidou-2 G2" to indicate 2nd generation, 2nd Geosynchronous (GEO) spacecraft (the -G1 was launched in 2010). This designation ("G") identifies it as being apart from the PRC's middle Earth orbit (MEO, "M") or inclined GEO ("IG") spacecraft constellations. The spacecraft uses the Chinese Academy of Spacecraft Technology (CAST) Dong Fang Hong 3 (DFH-3) communication satellite-heritage bus with the specialized navigational payload.

The spacecraft fragmented into at least five pieces, though no fragments have entered the catalog to accompany the parent body. Due to the difficulty in cataloging fragments in deep-space orbits there may be many more fragments resident at or near GEO.

REFERENCE DOCUMENTS

"BeiDou G2 Spacecraft Fragments in Geosynchronous Orbit", The Orbital Debris Quarterly News, NASA JSC, October 2016. Available online at <https://www.orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/odqnv20i4.pdf>.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 01 Mar 2010
DRY MASS (KG): 56.0
MAIN BODY: ellipsoid; 0.6 m diameter x 1 m length
MAJOR APPENDAGES: none
ATTITUDE CONTROL: none
ENERGY SOURCES: on-board propellants

EVENT DATA

DATE: 9 July 2014
TIME: 0828 GMT
ALTITUDE: 755.2 km
LOCATION: 33.96N, 41.3E (asc)
ASSESSED CAUSE: Propulsion

PRE-EVENT ELEMENTS

EPOCH: 14188.41970654
RIGHT ASCENSION: 294.1777
INCLINATION: 65.1872
ECCENTRICITY: 0.5623059
ARG. OF PERIGEE: 128.6500
MEAN ANOMALY: 296.6908
MEAN MOTION: 4.23407390
MEAN MOTION DOT/2: -0.00000214
MEAN MOTION DOT DOT/6: 0.00000
BSTAR: -0.013117

DEBRIS CLOUD DATA

MAXIMUM ΔP : 16.4 min
MAXIMUM ΔI : 0.6 deg

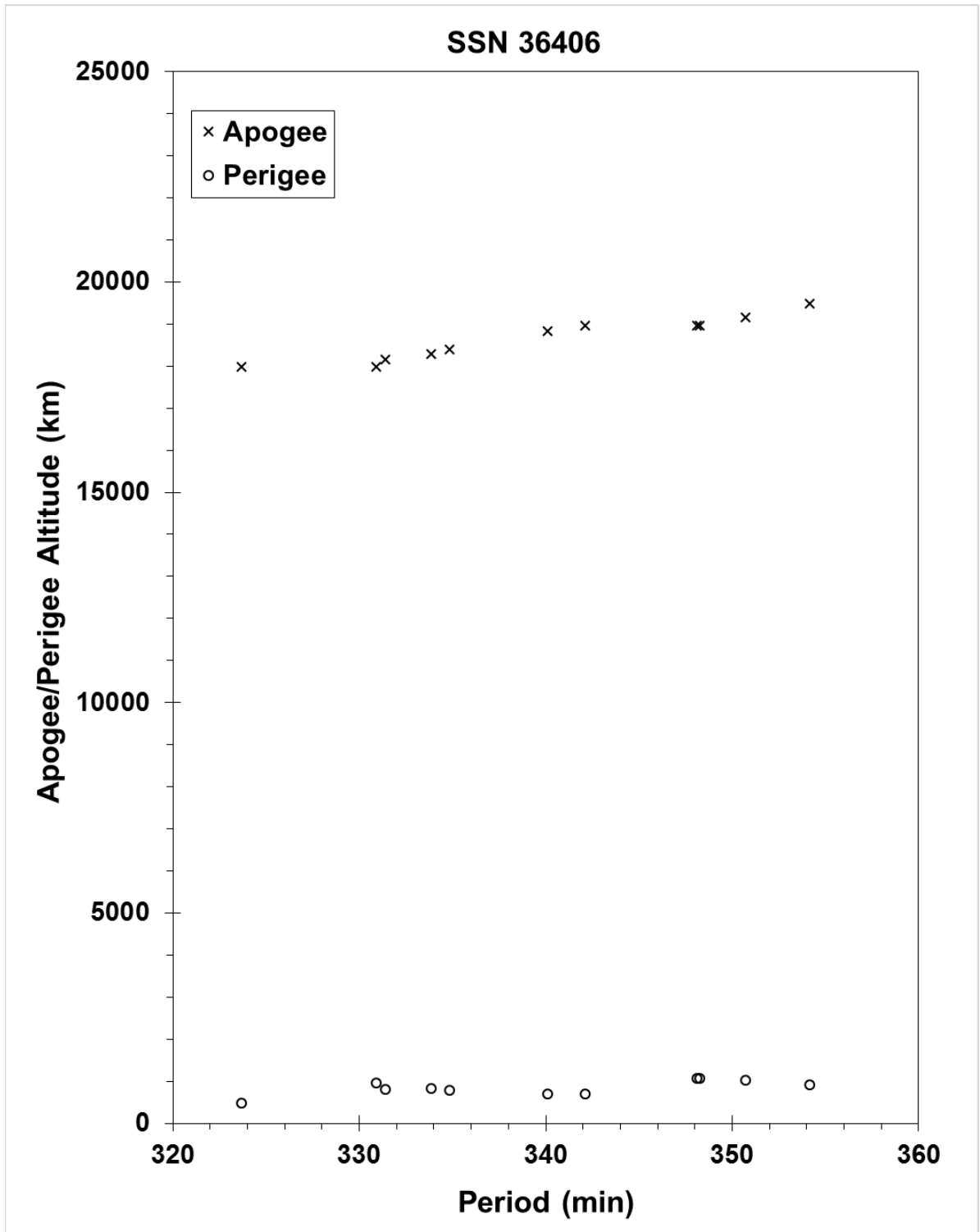
COMMENTS

This event was the 42nd known fragmentation of Blok-DM ullage motors. The breakups are believed to be caused by residual propellants. Russian officials have been aware of the problem since 1992 and have made design changes, although the date of full implementation is unknown. Although at least 16 small debris were observed by the SSN, none have been officially cataloged.

REFERENCE DOCUMENTS

History of Soviet/Russian Satellite Fragmentations: A Joint U.S.-Russian Investigation. N.L. Johnson *et al.* Kaman Sciences Corporation, October 2005.

“Three Additional Breakups Mar 2014”, The Orbital Debris Quarterly News, NASA JSC, October 2014.
Available online at <https://www.orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/odqnv18i4.pdf>.



The COSMOS ullage motor debris cloud includes the parent body and 10 debris fragments cataloged up to two years post-event.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 01 Mar 2010
DRY MASS (KG): 56.0
MAIN BODY: ellipsoid; 0.6 m diameter x 1 m length
MAJOR APPENDAGES: none
ATTITUDE CONTROL: none
ENERGY SOURCES: on-board propellants

EVENT DATA

DATE: 22 May 2018 LOCATION: 8.59S, 90.0E (asc)
TIME: 0206 GMT ASSESSED CAUSE: Propulsion
ALTITUDE: 13745.8 km

PRE-EVENT ELEMENTS

EPOCH: 18141.55091406 MEAN ANOMALY: 348.7596
RIGHT ASCENSION: 177.3536 MEAN MOTION: 4.23280497
INCLINATION: 65.0720 MEAN MOTION DOT/2: -0.00001771
ECCENTRICITY: 0.5676424 MEAN MOTION DOT DOT/6: 0.00000
ARG. OF PERIGEE: 45.9342 BSTAR: -0.0089882

DEBRIS CLOUD DATA

MAXIMUM ΔP : 23.7 min
MAXIMUM ΔI : 1 deg

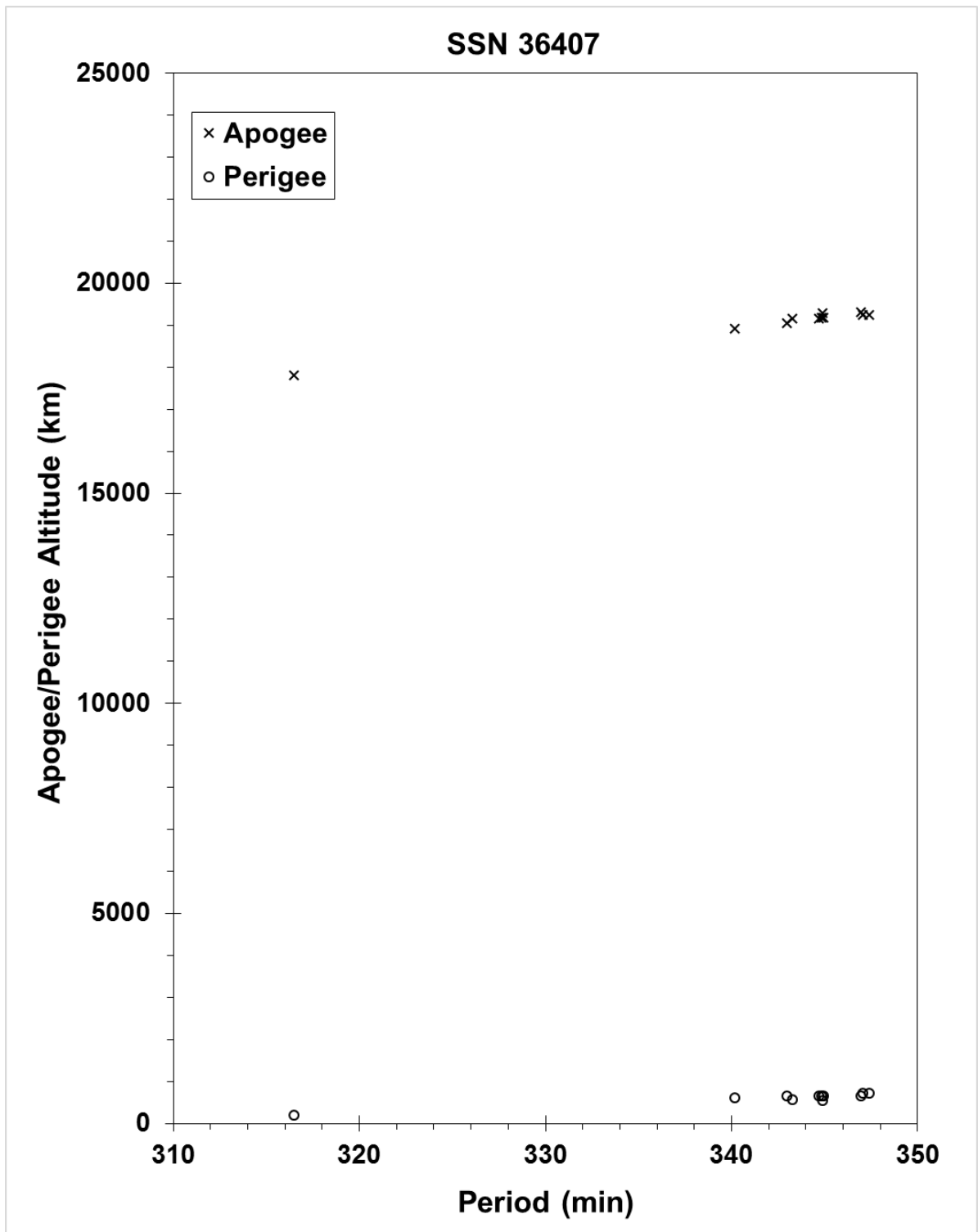
COMMENTS

This event was the 48th known fragmentation of Blok-DM ullage motors. The breakups are believed to be caused by residual propellants. Russian officials have been aware of the problem since 1992 and have made design changes, although the date of full implementation is unknown. Although at least 60 small debris were observed by the SSN, many more fragments could be present due to difficulties in cataloging fragments in elliptical and deep-space orbits.

REFERENCE DOCUMENTS

History of Soviet/Russian Satellite Fragmentations: A Joint U.S.-Russian Investigation. N.L. Johnson *et al.* Kaman Sciences Corporation, October 2005.

“A SOZ Unit Breakup Predicted and Observed in May 2018”, The Orbital Debris Quarterly News, NASA JSC, September 2018. Available online at <https://www.orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/odqnv22i3.pdf>.



The COSMOS ullage motor debris cloud includes the parent body and 10 debris fragments cataloged up to 64 days post-event.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 2 Sep 2010
DRY MASS (KG): 55
MAIN BODY: ellipsoid; 0.6 m diameter x 1 m length
MAJOR APPENDAGES: none
ATTITUDE CONTROL: none
ENERGY SOURCES: on-board propellants

EVENT DATA

DATE: 03 Sept. 2017
TIME: 0237 GMT
ALTITUDE: 14887.2 km
LOCATION: 54.96S, 115.3E (dsc)
ASSESSED CAUSE: Propulsion

PRE-EVENT ELEMENTS

EPOCH: 17245.45043553
RIGHT ASCENSION: 178.2175
INCLINATION: 65.2305
ECCENTRICITY: 0.5568611
ARG. OF PERIGEE: 84.7390
MEAN ANOMALY: 333.2513
MEAN MOTION: 4.25073663
MEAN MOTION DOT/2: +0.00009868
MEAN MOTION DOT DOT/6: 0.00000
BSTAR: +0.12084

DEBRIS CLOUD DATA

MAXIMUM ΔP : 19.2 min
MAXIMUM ΔI : 0.6 deg

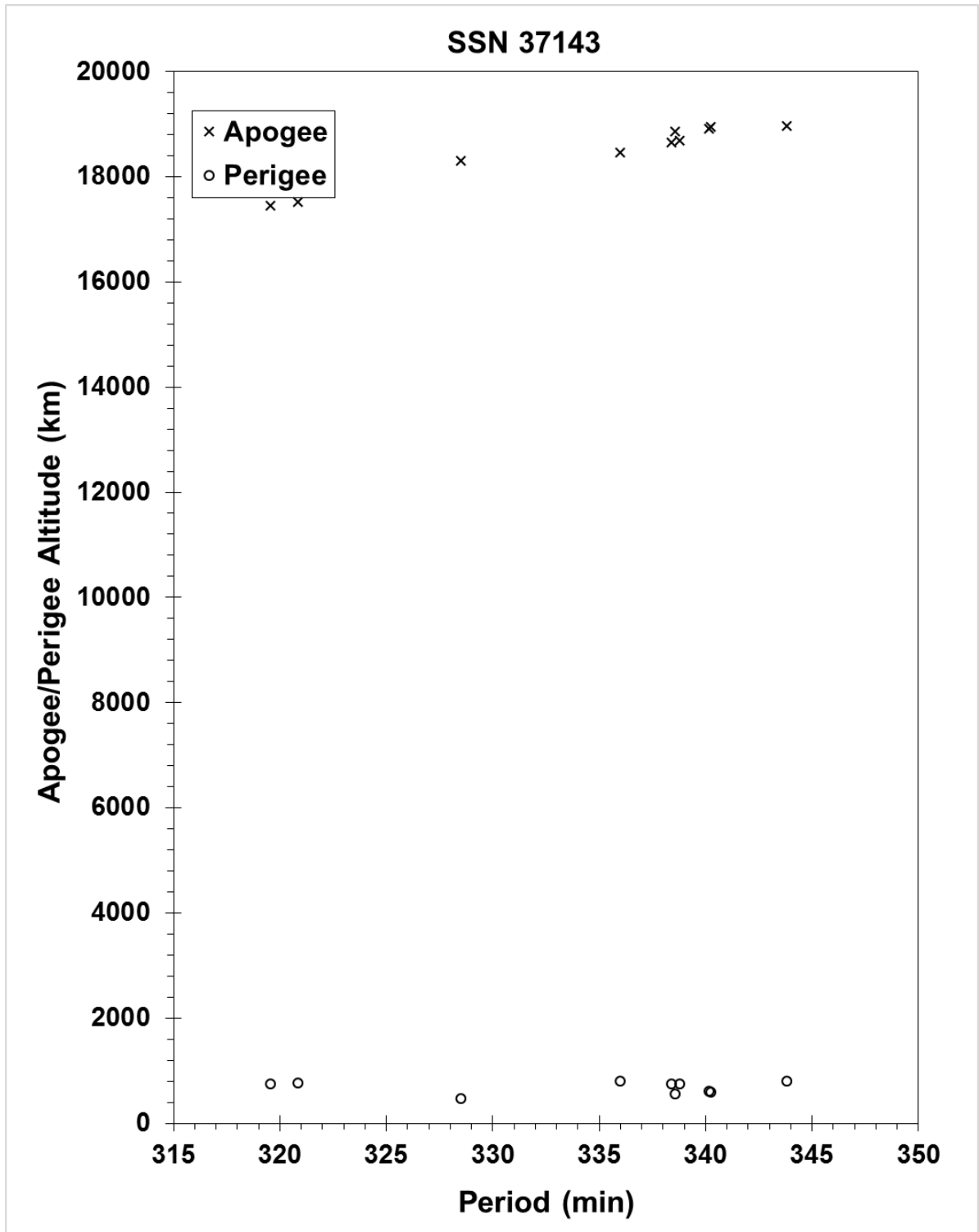
COMMENTS

This event was the 47th known fragmentation of Blok-DM ullage motors. The breakups are believed to be caused by residual propellants. Russian officials have been aware of the problem since 1992 and have made design changes, although the date of full implementation is unknown. Multiple small debris were observed by the SSN, but due to difficulties in cataloging fragments in elliptical and deep-space orbits many more fragments could be present.

REFERENCE DOCUMENTS

History of Soviet/Russian Satellite Fragmentations: A Joint U.S.-Russian Investigation. N.L. Johnson *et al.* Kaman Sciences Corporation, October 2005.

“Latest SOZ Breakup Occurs in September 2017”, The Orbital Debris Quarterly News, NASA JSC, November 2017. Available online at <https://www.orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/odqnv21i4.pdf>.



The COSMOS ullage motor debris cloud includes the parent body and nine debris fragments cataloged up to 45 days post-event.

SATELLITE DATA

TYPE: Rocket Body
OWNER: PRC
LAUNCH DATE: 4 Sep 2010
DRY MASS (KG): 3062
MAIN BODY: Cylinder; 3.0 m diameter x 12.4 m long
MAJOR APPENDAGES: none
ATTITUDE CONTROL: status unknown at time of event
ENERGY SOURCES: on-board propellants

EVENT DATA

| | | | |
|-----------|-------------|-----------------|---------|
| DATE: | ~4 Sep 2010 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 10247.43584147 | MEAN ANOMALY: | 183.2812 |
| RIGHT ASCENSION: | 224.5993 | MEAN MOTION: | 1.91160163 |
| INCLINATION: | 25.2307 | MEAN MOTION DOT/2: | 0.00001931 |
| ECCENTRICITY: | 0.7603361 | MEAN MOTION DOT DOT/6: | -0.11660e-5 |
| ARG. OF PERIGEE: | 179.3215 | BSTAR: | 0.0001 |

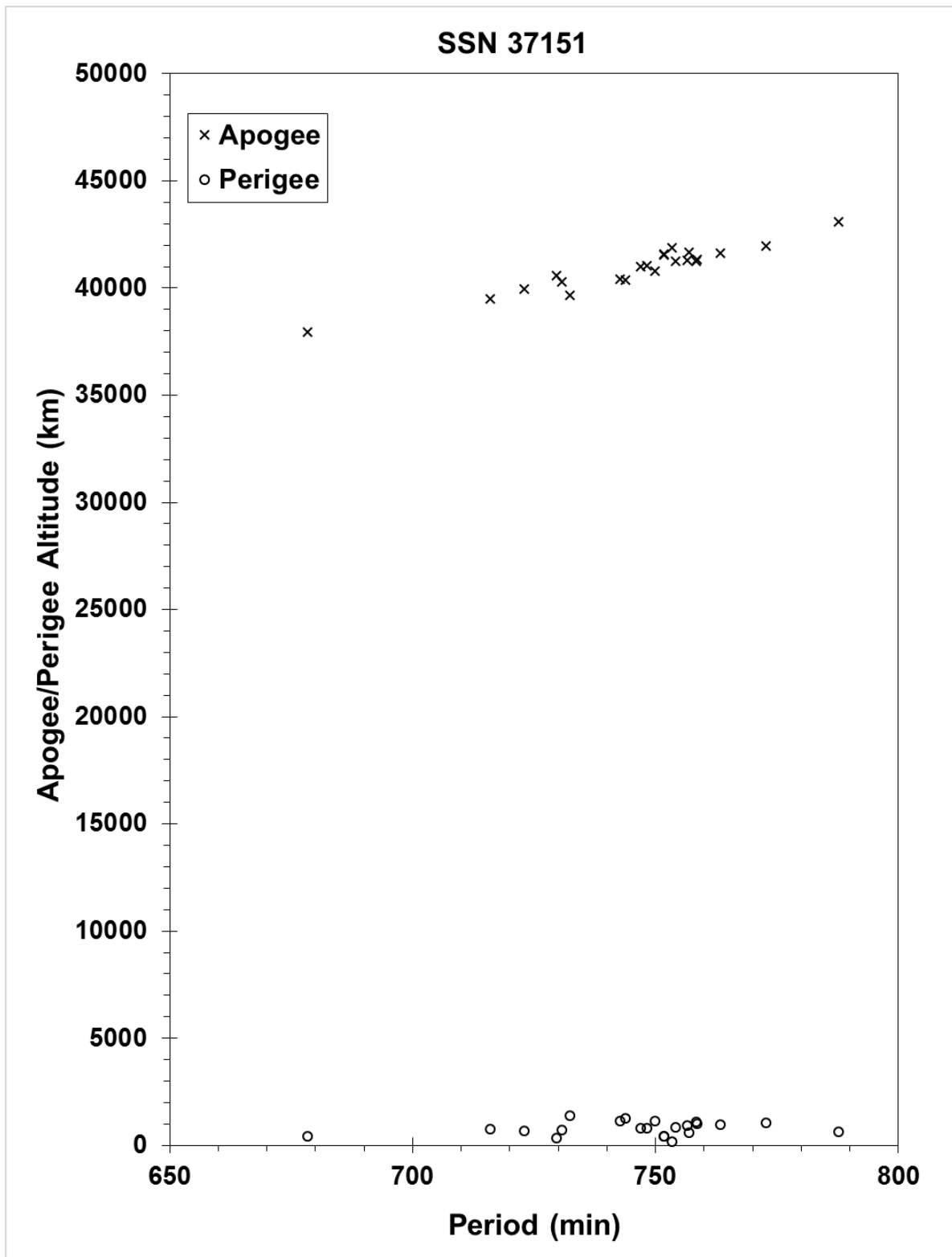
DEBRIS CLOUD DATA

MAXIMUM ΔP : 74.8 min
MAXIMUM ΔI : 3.2 deg

COMMENTS

Detected by software.

This was the second fragmentation of a CZ-3 third stage. Like the prior event, this event occurred within hours of launch and separation of the payload. Cataloging was delayed by high eccentricity of orbit.



The Chinasat 6A CZ-3B R/B debris cloud of 22 fragments cataloged up to 5 years after the event.

SATELLITE DATA

TYPE: Rocket Body
OWNER: PRC
LAUNCH DATE: 1 Nov 2010
DRY MASS (KG): 3062
MAIN BODY: Cylinder; 3.0 m diameter x 12.4 m long
MAJOR APPENDAGES: none
ATTITUDE CONTROL: status unknown at time of event
ENERGY SOURCES: on-board propellants

EVENT DATA

| | | | |
|-----------|-------------|-----------------|------------------------|
| DATE: | 1 Nov 2010 | LOCATION: | 4.407S, 328.867E (asc) |
| TIME: | 1731 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 33,072.2 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 10305.30000000 | MEAN ANOMALY: | 132.3340 |
| RIGHT ASCENSION: | 284.6090 | MEAN MOTION: | 2.28504000 |
| INCLINATION: | 20.4700 | MEAN MOTION DOT/2: | 0.00000669 |
| ECCENTRICITY: | 0.7313927 | MEAN MOTION DOT DOT/6: | -0.74391e-6 |
| ARG. OF PERIGEE: | 179.8990 | BSTAR: | 0.0 |

DEBRIS CLOUD DATA

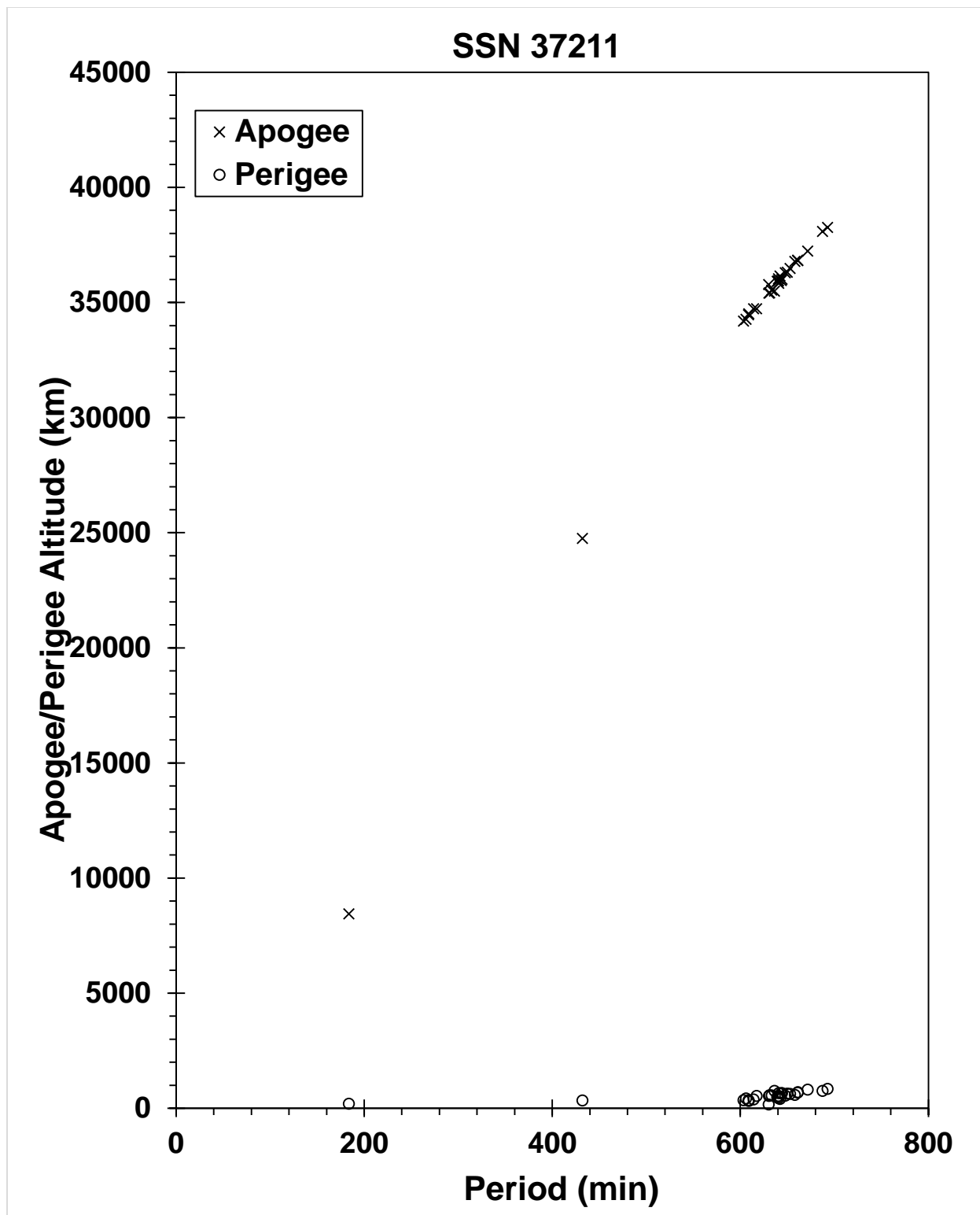
MAXIMUM ΔP : 446.6 min
MAXIMUM ΔI : 3.1 deg

COMMENTS

This was the third known fragmentation of a CZ-3 third stage. Like the prior event, this event occurred within hours of launch and separation of the payload.

REFERENCE DOCUMENT

"New Satellite Fragmentations Add to Debris Population", The Orbital Debris Quarterly News, NASA JSC, January 2011.
Available online at <http://orbitaldebris.jsc.nasa.gov/newsletter/pdfs/ODQNv15i1.pdf>.



The CZ-3C R/B debris cloud of 30 fragments cataloged up to 4 years after the event.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 21 Sep 2000
DRY MASS (KG): 375
MAIN BODY: toroidal fuel tank; 0.821 m tank diameter x 3.44 m overall diameter
MAJOR APPENDAGES: none
ATTITUDE CONTROL: none
ENERGY SOURCES: unknown

EVENT DATA

DATE: 3-4 Aug 2015
TIME: unknown
ALTITUDE: unknown
LOCATION: unknown
ASSESSED CAUSE: unknown

PRE-EVENT ELEMENTS

EPOCH: 15214.44003403
RIGHT ASCENSION: 218.9517
INCLINATION: 51.4462
ECCENTRICITY: 0.1913343
ARG. OF PERIGEE: 234.4912
MEAN ANOMALY: 298.4148
MEAN MOTION: 11.24281384
MEAN MOTION DOT/2: 0.00000437
MEAN MOTION DOT DOT/6: 0.00000
BSTAR: 0.0011788

DEBRIS CLOUD DATA

MAXIMUM ΔP : TBD min
MAXIMUM ΔI : TBD deg

COMMENTS

The parent body is assumed to be the jettisonable auxiliary fuel tank discarded by the Fregat-SB upper stage. Twenty-four debris were observed but none have entered the SSN catalog.

REFERENCE DOCUMENTS

“Fragmentation of *Fregat* Upper Stage Debris”, The Orbital Debris Quarterly News, NASA JSC, April 2016.
Available online at <http://orbitaldebris.jsc.nasa.gov/newsletter/pdfs/ODQNV20i1-2.pdf>.

Insufficient data to construct a Gabbard diagram.

SATELLITE DATA

TYPE: Rocket Body
OWNER: PRC
LAUNCH DATE: 19 Dec 2011
DRY MASS (KG): 3062
MAIN BODY: Cylinder; 3.0 m diameter x 12.4 m length
MAJOR APPENDAGES: none
ATTITUDE CONTROL: status unknown at time of event
ENERGY SOURCES: on-board propellants

EVENT DATA

| | | | |
|-----------|-------------|-----------------|---------|
| DATE: | 21 Dec 2011 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 11355.16500251 | MEAN ANOMALY: | 285.0544 |
| RIGHT ASCENSION: | 335.3965 | MEAN MOTION: | 1.91939558 |
| INCLINATION: | 24.3335 | MEAN MOTION DOT/2: | -0.00001209 |
| ECCENTRICITY: | 0.7584148 | MEAN MOTION DOT DOT/6: | 0.0 |
| ARG. OF PERIGEE: | 179.8362 | BSTAR: | -0.00031775 |

DEBRIS CLOUD DATA

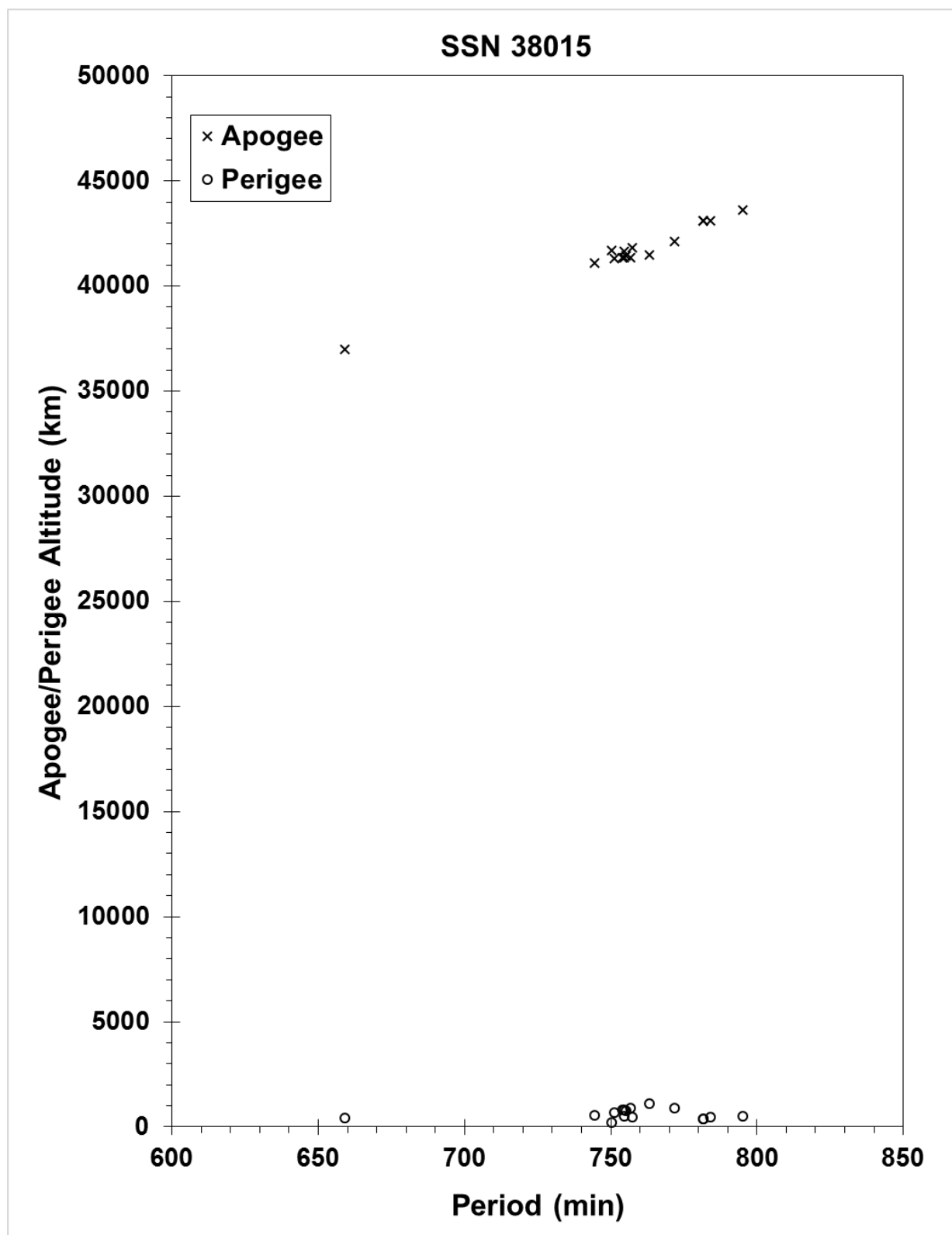
MAXIMUM ΔP : 91.3 min
MAXIMUM ΔI : 2.4 deg

COMMENTS

This was the fourth fragmentation of a CZ-3 third stage. Like prior events, this event occurred within two days of launch and separation of the payload.

REFERENCE DOCUMENT

“Only a few minor satellite breakups in 2011”, The Orbital Debris Quarterly News, NASA JSC, January 2012.
Available online at <http://orbitaldebris.jsc.nasa.gov/newsletter/pdfs/ODQNv16i1.pdf>.



The CZ-3C R/B debris cloud of 16 fragments cataloged up to 2.7 years after the event.

SATELLITE DATA

| | |
|-------------------|--|
| TYPE: | Rocket Body |
| OWNER: | PRC |
| LAUNCH DATE: | 24 Feb 2012 |
| DRY MASS (KG): | 3062 |
| MAIN BODY: | Cylinder; 3.0 m diameter x 12.4 m length |
| MAJOR APPENDAGES: | none |
| ATTITUDE CONTROL: | status unknown at time of event |
| ENERGY SOURCES: | on-board propellants |

EVENT DATA

| | | | |
|-----------|-------------|-----------------|---------|
| DATE: | 26 Feb 2012 | LOCATION: | Unknown |
| TIME: | Unknown | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | Unknown | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|--------------|
| EPOCH: | 12056.56279906 | MEAN ANOMALY: | 353.3665 |
| RIGHT ASCENSION: | 35.2298 | MEAN MOTION: | 2.27377203 |
| INCLINATION: | 20.6965 | MEAN MOTION DOT/2: | 0.00007641 |
| ECCENTRICITY: | 0.7327583 | MEAN MOTION DOT DOT/6: | 0.0000012139 |
| ARG. OF PERIGEE: | 180.1005 | BSTAR: | 0.00010000 |

DEBRIS CLOUD DATA

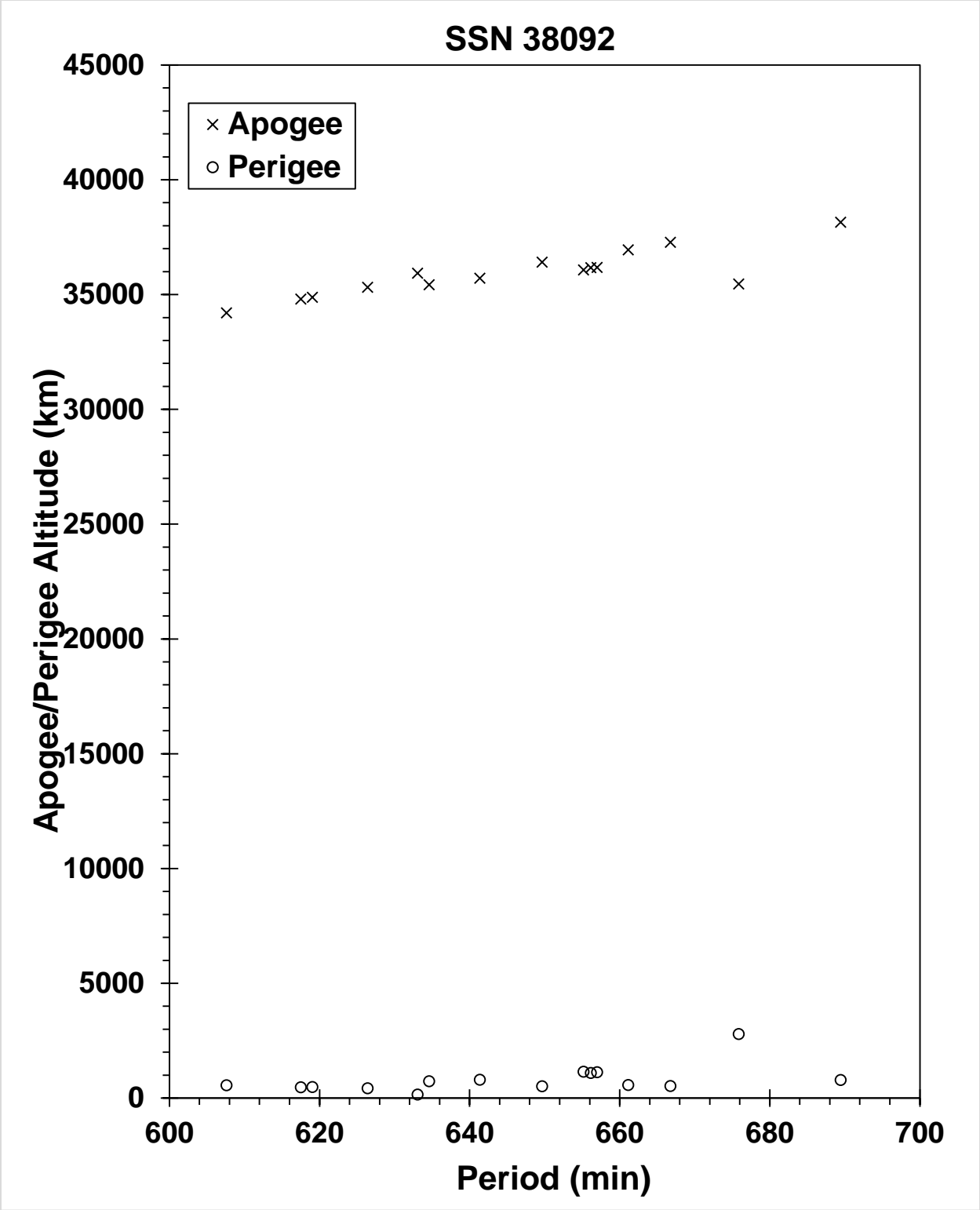
| | |
|----------------------|----------|
| MAXIMUM ΔP : | 56.1 min |
| MAXIMUM ΔI : | 2.6 deg |

COMMENTS

This was the fifth fragmentation of a CZ-3 third stage. Like its preceding event, this event occurred within two days of launch and separation of the Beidou G5 spacecraft.

REFERENCE DOCUMENT

“Chinese Rocket Body Explosions Continue”, The Orbital Debris Quarterly News, NASA JSC, April 2012.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv16i2.pdf>.



The CZ-3C R/B debris cloud of 15 fragments cataloged up to 2.6 years after the event.

SATELLITE DATA

TYPE: *Briz-M* Upper Stage Core
OWNER: CIS
LAUNCH DATE: 17 May 2012
DRY MASS (KG): 1220
MAIN BODY: cylinder 2.49 m diameter x 2.65 m length
MAJOR APPENDAGES: none
ATTITUDE CONTROL: none at time of event
ENERGY SOURCES: unknown

EVENT DATA

| | | | |
|-----------|-------------|-----------------|-------------------|
| DATE: | 23 Dec 2015 | LOCATION: | 11.9N, 178E (dsc) |
| TIME: | 1600 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 24,310 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 15356.29062075 | MEAN ANOMALY: | 128.0880177 |
| RIGHT ASCENSION: | 68.3779 | MEAN MOTION: | 1.76901584 |
| INCLINATION: | 12.0183 | MEAN MOTION DOT/2: | -0.00000143 |
| ECCENTRICITY: | 0.4187335 | MEAN MOTION DOT DOT/6: | 0.00000 |
| ARG. OF PERIGEE: | 203.9714 | BSTAR: | 0.00000 |

DEBRIS CLOUD DATA

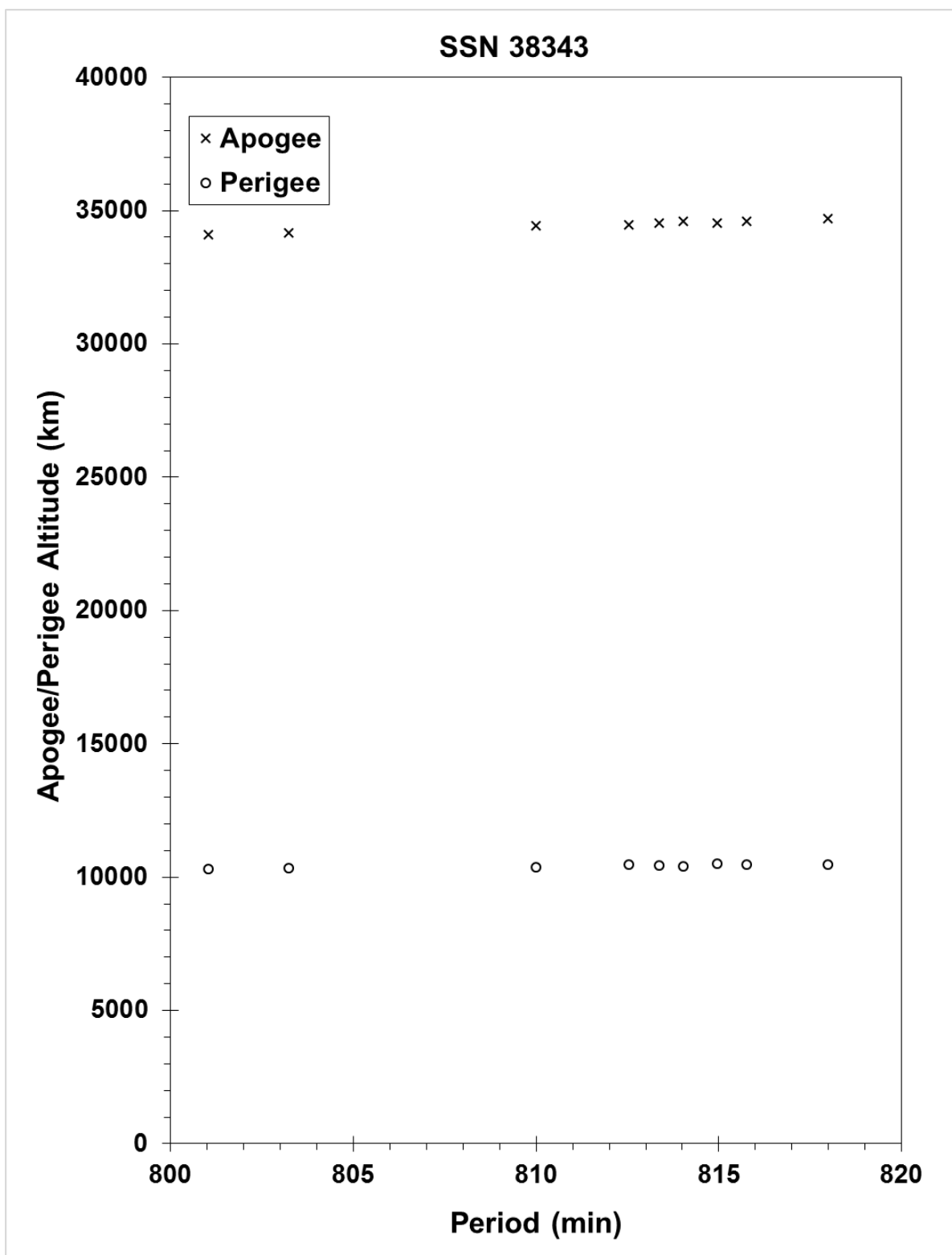
MAXIMUM ΔP : 13.9 min
MAXIMUM ΔI : 0.09 deg

COMMENTS

Eight debris (in addition to the parent object) have been officially cataloged. Debris in deep-space orbits are difficult for the SSN to track and catalog. Hundreds of additional fragments could be on-orbit.

REFERENCE DOCUMENTS

“Briz-M Core Stage Fragments in Elliptical Orbit”, [The Orbital Debris Quarterly News](http://orbitaldebris.jsc.nasa.gov/newsletter/pdfs/ODQNV20i1-2.pdf), NASA JSC, April 2016.
Available online at <http://orbitaldebris.jsc.nasa.gov/newsletter/pdfs/ODQNV20i1-2.pdf>.



The *Briz-M* debris cloud; debris cataloged approximately 39 days after the event.

SATELLITE DATA

TYPE: Rocket Body
OWNER: CIS
LAUNCH DATE: 6 August 2012
DRY MASS (KG): 2510
MAIN BODY: Cylinder + toroid; 4.1 m diameter x 2.65 m length
MAJOR APPENDAGES: none
ATTITUDE CONTROL: none at the time of the event
ENERGY SOURCES: on-board propellants

EVENT DATA

DATE: 16 Oct 2012
TIME: 1631 GMT
ALTITUDE: 273.061 km
LOCATION: 32.447N, 335.711E (asc)
ASSESSED CAUSE: Propulsion

PRE-EVENT ELEMENTS

EPOCH: 12289.67349823
RIGHT ASCENSION: 103.4658
INCLINATION: 49.8913
ECCENTRICITY: 0.2633226
ARG. OF PERIGEE: 127.8077
MEAN ANOMALY: 259.1600
MEAN MOTION: 10.14046751
MEAN MOTION DOT/2: 0.00003146
MEAN MOTION DOT DOT/6: 0.0000019595
BSTAR: 0.00011788

DEBRIS CLOUD DATA

MAXIMUM ΔP : 38.9 min
MAXIMUM ΔI : 0.8 deg

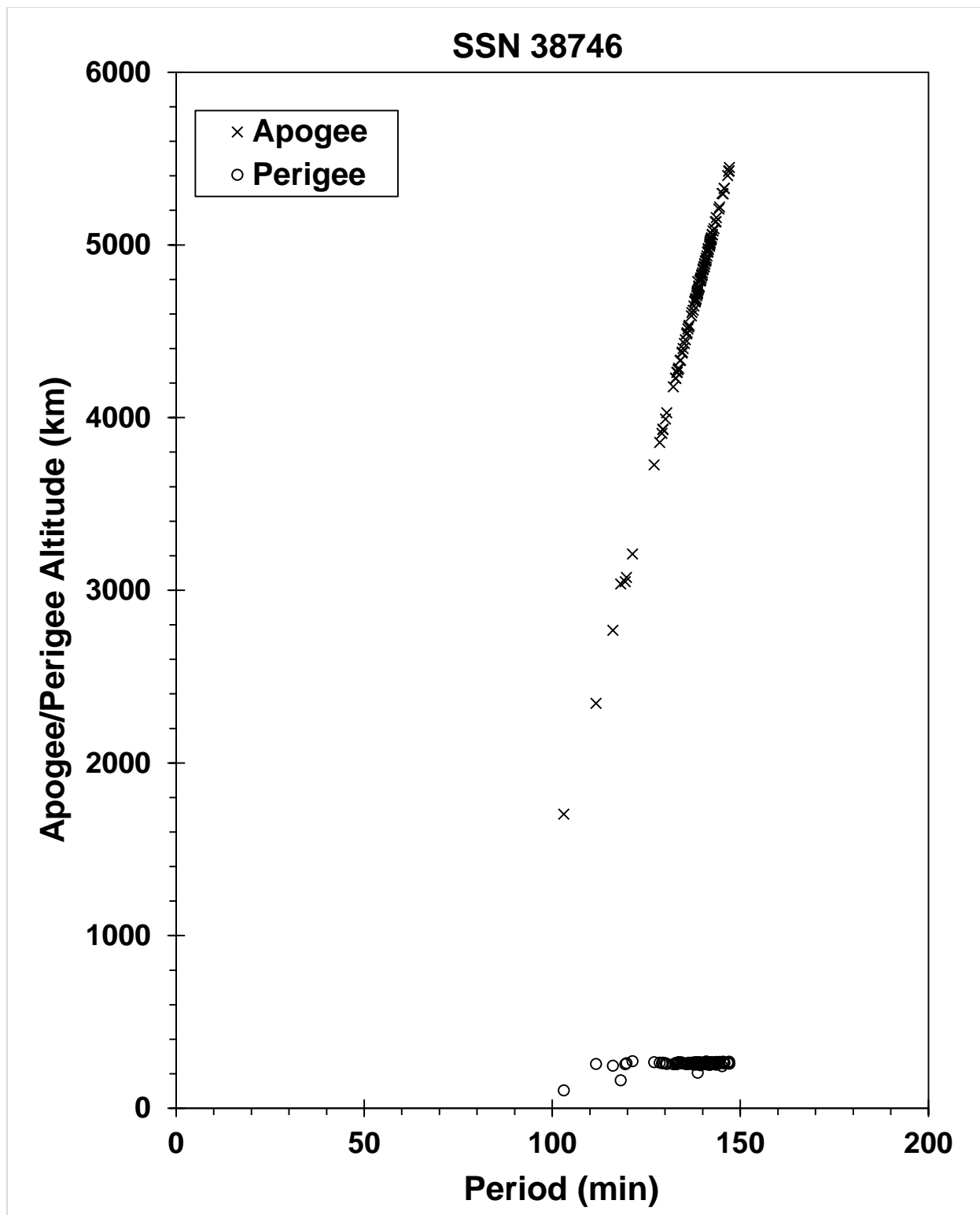
COMMENTS

The dry mass quoted above includes that of the *Briz-M* core stage and the toroidal Auxiliary Propulsion Tank (APT)—sometimes referred to as the Additional Fuel Tank (AFT); the malfunction stranding the stage in its elliptical orbit occurred prior to APT jettison. The event occurred almost 70 days after launch. An estimated 10 metric tons of propellant was aboard.

REFERENCE DOCUMENT

“Upper Stage Explosion Places LEO Satellites at Risk”, The Orbital Debris Quarterly News, NASA JSC, January 2013. Available online at <http://orbitaldebris.jsc.nasa.gov/newsletter/pdfs/ODQNv17i1.pdf>.

“New Russian Launch Failure Raises Breakup Concern”, The Orbital Debris Quarterly News, NASA JSC, October 2012. Available online at <http://orbitaldebris.jsc.nasa.gov/newsletter/pdfs/ODQNv16i4.pdf>.



The *Briz-M* R/B debris cloud of 113 fragments cataloged up to 3 weeks after the event.

SATELLITE DATA

TYPE: Rocket Body
OWNER: USA
LAUNCH DATE: 29 Sep 2013
DRY MASS (KG): 3900 kg
MAIN BODY: cylinder; 3.66 m diameter x ~ 14 m length
MAJOR APPENDAGES: none
ATTITUDE CONTROL: active, 3 axis RCS
ENERGY SOURCES: on-board propellants

EVENT DATA

| | | | |
|-----------|-------------|-----------------|-----------------------|
| DATE: | 29 Sep 2013 | LOCATION: | 78.026S, 8.302E (asc) |
| TIME: | 1641 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 1204.12 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------------|
| EPOCH: | 13272.73704787 | MEAN ANOMALY: | 327.9211 |
| RIGHT ASCENSION: | 315.1956 | MEAN MOTION: | 13.93734602 |
| INCLINATION: | 80.9876 | MEAN MOTION DOT/2: | 0.00000033 |
| ECCENTRICITY: | 0.0693087 | MEAN MOTION DOT DOT/6: | 0.0 |
| ARG. OF PERIGEE: | 153.5198 | BSTAR: | 0.0 (unavailable) |

DEBRIS CLOUD DATA

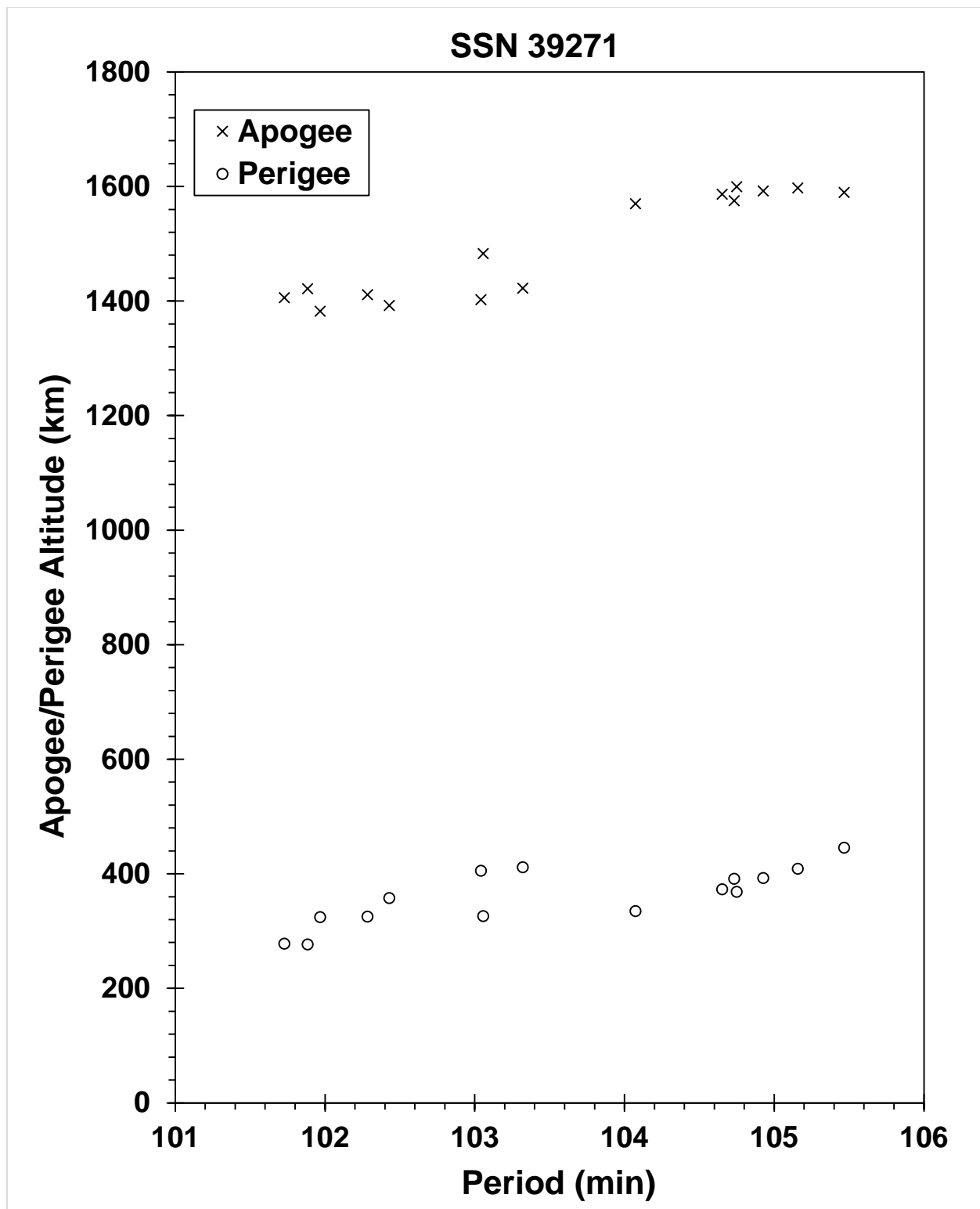
MAXIMUM ΔP : 2.1 min
MAXIMUM ΔI : 0.16 deg

COMMENTS

The elements listed above are from the first published TLE; the epoch time is approximately one hour and 21 seconds after the debris-producing event. This was the first flight of the Falcon 9 v1.1 second stage. The second stage engine restart, intended to cast the vehicle into its disposal orbit, failed.

REFERENCE DOCUMENT

“Upper Stage of New Falcon 9 Rocket Did Not Explode After Launch, SpaceX Says”,
<http://www.space.com/23038-spacex-falcon-9-rocket-explosion-rumors.html>, retrieved 23 April 2015.



The Falcon 9 R/B debris cloud of 15 fragments cataloged within 2 weeks of the event.

SATELLITE DATA

TYPE: payload
OWNER: ESA
LAUNCH DATE: 3 Apr 2014
DRY MASS (KG): 2170
MAIN BODY: cubical 3.9 m x 2.6 m x 2.5 m spacecraft envelope
MAJOR APPENDAGES: dual solar arrays and synthetic aperture radar antenna
ATTITUDE CONTROL: three-axis stabilized
ENERGY SOURCES: monopropellant, batteries, gyroscope, reaction wheels

EVENT DATA

| | | | |
|-----------|-------------|-----------------|-----------------------|
| DATE: | 23 Aug 2016 | LOCATION: | 72.03S, 39.59E (dsc) |
| TIME: | 1707 GMT | ASSESSED CAUSE: | Collision, accidental |
| ALTITUDE: | 723.20 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|--------------|
| EPOCH: | 16235.56155967 | MEAN ANOMALY: | 283.0743 |
| RIGHT ASCENSION: | 241.4752 | MEAN MOTION: | 14.59198146 |
| INCLINATION: | 98.1816 | MEAN MOTION DOT/2: | +0.00000017 |
| ECCENTRICITY: | 0.0001365 | MEAN MOTION DOT DOT/6: | 0.00000 |
| ARG. OF PERIGEE: | 77.0610 | BSTAR: | +0.000013361 |

DEBRIS CLOUD DATA

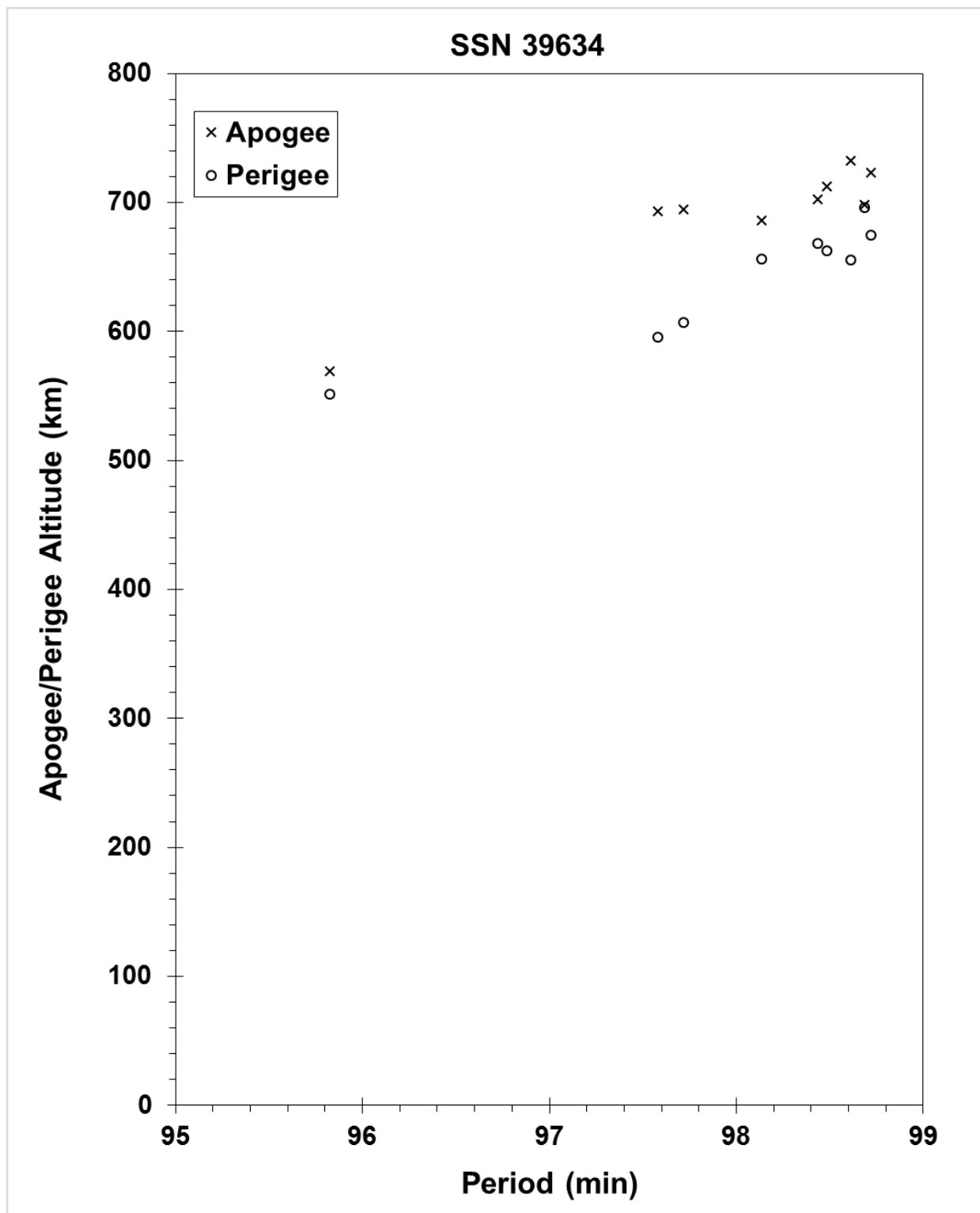
MAXIMUM ΔP : 2.9 min
MAXIMUM ΔI : 0.1 deg

COMMENTS

This event is categorized as an accidental collision with a small particle. Normally, an event of this nature would have been categorized as an anomalous event; however, spacecraft attitude control was upset by the solar array impact and on-board cameras were able to document the before/after state of the penetrated solar array. While either small, un-cataloged debris or a micrometeoroid could have been this event's projectile, the source of the resulting tracked and cataloged debris was unambiguous.

REFERENCE DOCUMENT

Collecte Localisation Satellites, "Sentinel-1A Debris Collision August 2016," MPC-S1 (18 October 2016). Retrieved 28 September 2018 from https://sentinel.esa.int/documents/247904/2142675/Sentinel-1A_Debris_Collision_August_2016_MPC.pdf.



The Sentinel 1A debris cloud; debris cataloged within approximately 18 days of the event.

SATELLITE DATA

TYPE: Soyuz-U (SL-4) Third Stage Rocket Body
OWNER: CIS
LAUNCH DATE: 28 April 2015
DRY MASS (KG): 2400
MAIN BODY: 2.66 m diameter x 8.1 m long cylinder
MAJOR APPENDAGES: none
ATTITUDE CONTROL: three-axis
ENERGY SOURCES: on-board propellants

EVENT DATA

| | | | |
|-----------|---------------|-----------------|---------|
| DATE: | 28 April 2015 | LOCATION: | TBD |
| TIME: | 07:18:35 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | TBD km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|-----|------------------------|-----|
| EPOCH: | TBD | MEAN ANOMALY: | TBD |
| RIGHT ASCENSION: | TBD | MEAN MOTION: | TBD |
| INCLINATION: | TBD | MEAN MOTION DOT/2: | TBD |
| ECCENTRICITY: | TBD | MEAN MOTION DOT DOT/6: | TBD |
| ARG. OF PERIGEE: | TBD | BSTAR: | TBD |

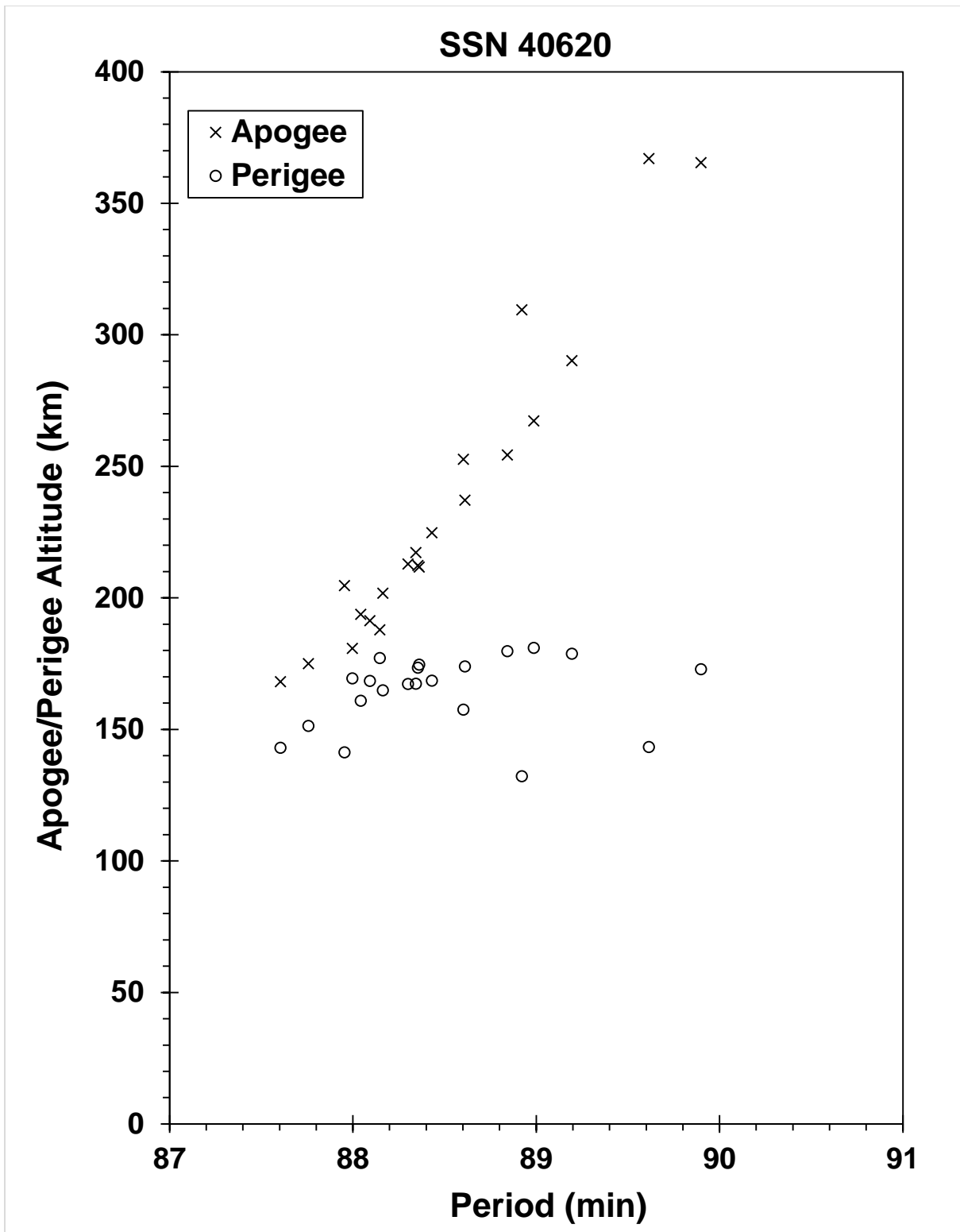
DEBRIS CLOUD DATA

MAXIMUM ΔP : unknown
MAXIMUM ΔI : unknown

COMMENTS

In addition to the parent body, 20 debris have been cataloged. Due to the low altitude of the event, all had decayed from orbit by 7 May 2015.

REFERENCE DOCUMENTS



The SL-4 R/B debris cloud cataloged between 28 April and 1 May 2015.

COSMOS 2513 R/B

2015-075B

41122

SATELLITE DATA

TYPE: *Briz-M* Upper Stage Core
OWNER: CIS
LAUNCH DATE: 13 Dec 2015
DRY MASS (KG): 1220
MAIN BODY: cylinder 2.49 m diameter x 2.65 m long
MAJOR APPENDAGES: none
ATTITUDE CONTROL: none at time of event
ENERGY SOURCES: unknown

EVENT DATA

| | | | |
|-----------|--------------|-----------------|---------------------|
| DATE: | 16 Jan 2016 | LOCATION: | 0.18S, 223.0E (dsc) |
| TIME: | 0350 GMT | ASSESSED CAUSE: | Unknown |
| ALTITUDE: | 34,880.09 km | | |

PRE-EVENT ELEMENTS

| | | | |
|------------------|----------------|------------------------|-------------|
| EPOCH: | 16015.10700804 | MEAN ANOMALY: | 221.1059 |
| RIGHT ASCENSION: | 135.1430 | MEAN MOTION: | 1.04652118 |
| INCLINATION: | 0.1737 | MEAN MOTION DOT/2: | -0.00000109 |
| ECCENTRICITY: | 0.0286832 | MEAN MOTION DOT DOT/6: | 0.00000 |
| ARG. OF PERIGEE: | 5.8561 | BSTAR: | 0.00000 |

DEBRIS CLOUD DATA

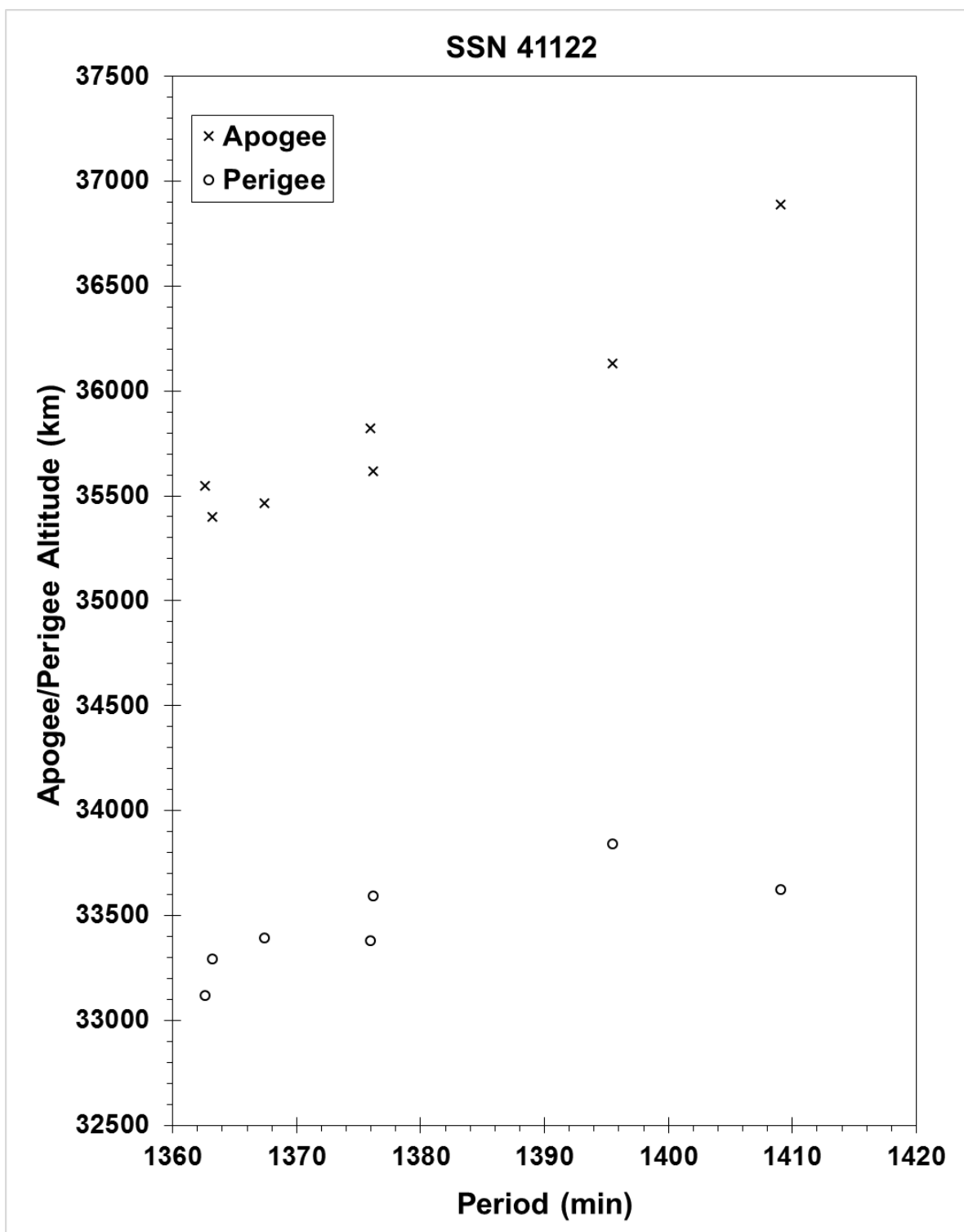
MAXIMUM ΔP : 33.1 min
MAXIMUM ΔI : 0.7 deg

COMMENTS

Seven debris (including the parent object) have been officially cataloged. Debris in deep-space orbits are difficult for the SSN to track and catalog. Hundreds of additional fragments could be on-orbit.

REFERENCE DOCUMENTS

“Briz-M Core Stage Fragments Near Geosynchronous Orbit”, [The Orbital Debris Quarterly News](https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/odqnv20i1-2.pdf), NASA JSC, April 2016. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/odqnv20i1-2.pdf>.



The *Briz-M* debris cloud; debris cataloged up to 127 days after the event.

SATELLITE DATA

TYPE: payload
OWNER: Japan
LAUNCH DATE: 17 Feb 2016
DRY MASS (KG): < 2700
MAIN BODY: cylinder 1.8 m diameter x 7.9 m long (13.6 m length overall after EOB erection)
MAJOR APPENDAGES: Solar Array Paddles (SAP) and Extensible Optical Bench (EOB)
ATTITUDE CONTROL: three-axis stabilized
ENERGY SOURCES: propellant, batteries, reaction wheels

EVENT DATA

DATE: 26 Mar 2016
TIME: 0142 GMT
ALTITUDE: 573.92 km
LOCATION: 20.82N, 267.36E (asc)
ASSESSED CAUSE: Propulsion

PRE-EVENT ELEMENTS

EPOCH: 16084.46380907
RIGHT ASCENSION: 87.9716
INCLINATION: 31.0065
ECCENTRICITY: 0.0011434
ARG. OF PERIGEE: 273.0633
MEAN ANOMALY: 86.8693
MEAN MOTION: 14.98194942
MEAN MOTION DOT/2: +0.00001057
MEAN MOTION DOT DOT/6: 0.00000
BSTAR: +0.00008

DEBRIS CLOUD DATA

MAXIMUM ΔP : 0.47 min
MAXIMUM ΔI : 0.02 deg

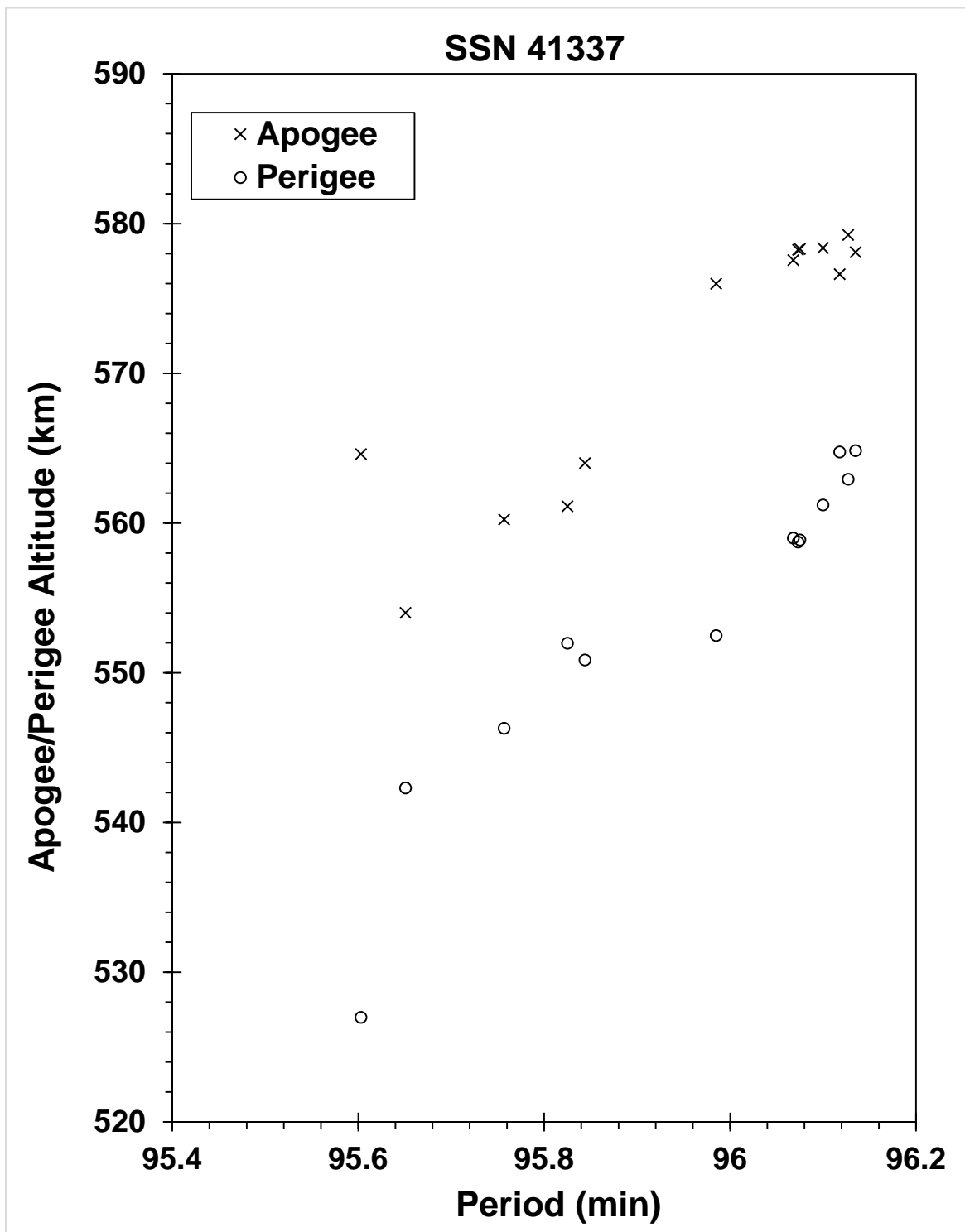
COMMENTS

This event is categorized as a propulsion-induced structural failure.

REFERENCE DOCUMENTS

“ASTRO-H Spacecraft Fragments During Payload Check-out Operations”, The Orbital Debris Quarterly News, NASA JSC, October 2016. Available online at <https://www.orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/odqnv20i4.pdf>.

JAXA, “Hitomi Experience Report: Investigation of Anomalies Affecting the X-ray Astronomy Satellite ‘Hitomi’ (ASTRO-H),” 8 June 2016 edition. Retrieved 1 October 2016 from http://global.jaxa.jp/projects/sat/astro_h/topics.html?utm_source=dlvr.it&utm_medium=twitter#topics7815



The Hitomi debris cloud; debris cataloged up to approximately one year after the event.

SATELLITE DATA

TYPE: Mission Related Debris
OWNER: CIS
LAUNCH DATE: 26 Dec 2017
DRY MASS (KG): 375
MAIN BODY: toroidal fuel tank; 0.821 m tank diameter x 3.44 m overall diameter
MAJOR APPENDAGES: none
ATTITUDE CONTROL: none
ENERGY SOURCES: unknown

EVENT DATA

DATE: 12 Feb 2018
TIME: 0957 GMT
ALTITUDE: 396.48 km
LOCATION: 38.06N, 117.94E (asc)
ASSESSED CAUSE: unknown

PRE-EVENT ELEMENTS

EPOCH: 18041.75579119
RIGHT ASCENSION: 273.8260
INCLINATION: 50.4229
ECCENTRICITY: 0.2212829
ARG. OF PERIGEE: 98.0587
MEAN ANOMALY: 287.5127
MEAN MOTION: 10.99960807
MEAN MOTION DOT/2: 0.00005800
MEAN MOTION DOT DOT/6: 0.0000034339
BSTAR: 0.00020634

DEBRIS CLOUD DATA

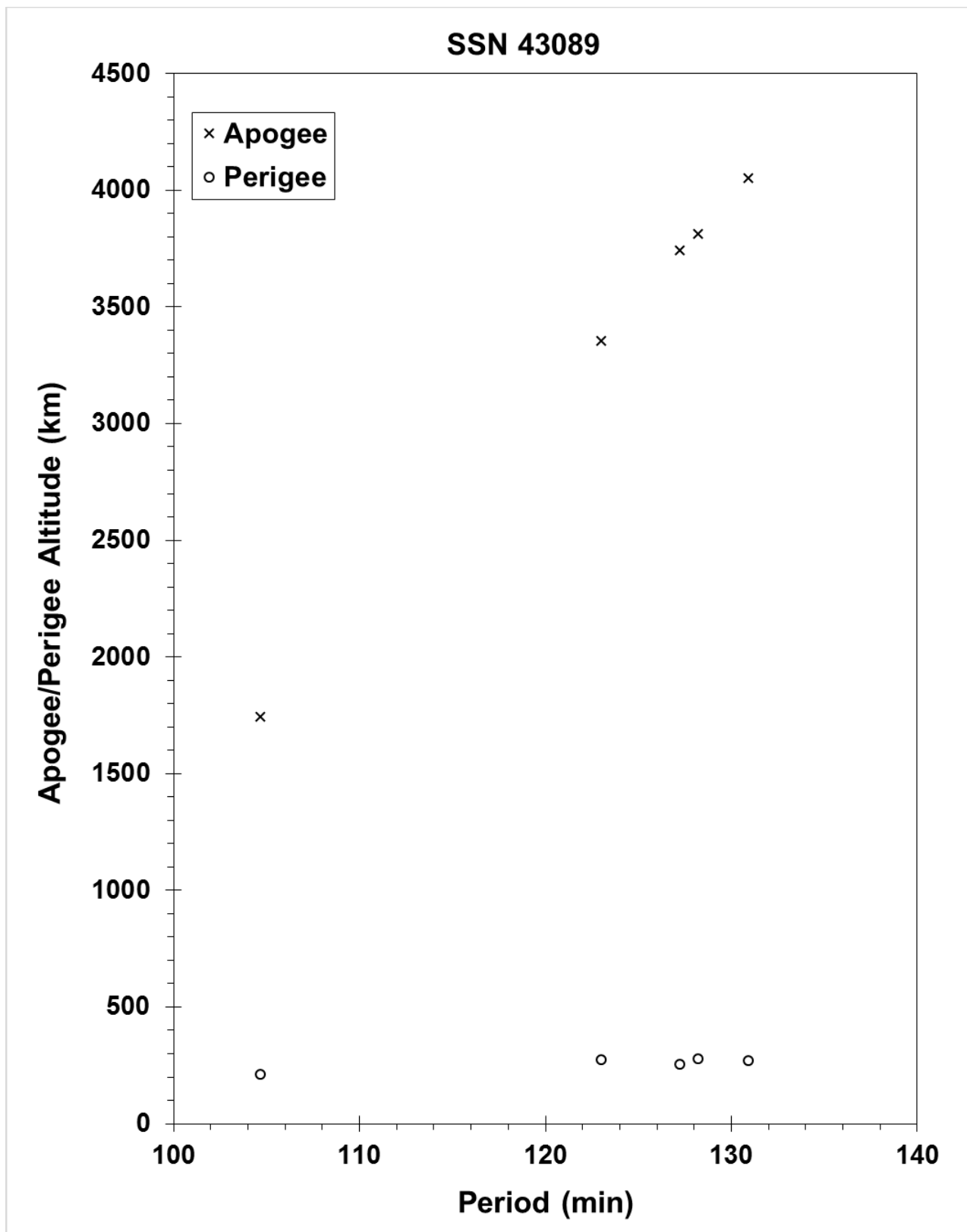
MAXIMUM ΔP : 26.3 min
MAXIMUM ΔI : 0.3 deg

COMMENTS

The parent body is assumed to be the jettisonable auxiliary fuel tank discarded by the Fregat-SB upper stage.

REFERENCE DOCUMENT

“Fragmentation of *Fregat-SB* Upper Stage Debris”, The Orbital Debris Quarterly News, NASA JSC, May 2018.
Available online at <https://www.orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/odqnv22i2.pdf>.



The Fregat-SB Auxiliary Propellant Tank debris cloud, composed of parent body and four fragments; debris cataloged up to 11 days after the event.

3.0 SATELLITE ANOMALOUS EVENTS

This section describes the identified anomalous events compiled throughout the years of the Satellite Catalog and orbital debris analysis associated with this volume. No exhaustive search for anomalous events has yet been conducted, although the following compilation should represent the most significant events noted thus far.

3.1 *Background and Status*

As defined in the introduction of this volume, an anomalous event is the unplanned separation, usually at low velocity, of one or more detectable objects from a satellite that remains essentially intact. The assessment that the configuration of the parent satellite has not changed significantly is to a degree subjective and is often based on indirect parameters and not on detailed imagery.

Anomalous events can be caused by material deterioration of items such as thermal blankets, protective shields, or solar panels and by impacts of small debris, either natural or man-made. Other satellite deteriorations, *e.g.*, paint debonding, are known to take place, but are undetectable with the sensors of the US SSN. Interestingly, 39 of the 78 satellites in this section, half, are U.S. with another (JASON 1) being a joint U.S.-French spacecraft, 25 are CIS, 3 are French, 2 each are ESA, Indian, and Canadian, and 1 each are from Japan, Indonesia, and the People's Republic of China; additionally, 1 is from a commercial organization headquartered in Luxembourg, SES. Of the 78 satellites, 57 are payloads and 21 are rocket bodies. These events are summarized in Tables 3.1 and 3.2.

Because of the usually low velocity of debris ejection and the potential delay in detecting debris liberated in small numbers, the accuracy of the calculated time of separation is often degraded. Hence, only the month and year of each event are provided, although in some cases the time of the event has been narrowed to a shorter interval. As in the previous section, orbital altitudes are cited to the nearest 5 km based on a mean Earth radius and on the last element set prior to the assessed event date.

Anomalous event debris often exhibit unusually high decay rates, which are indicative of high area-to-mass ratios. This feature, coupled with the normal small size of the debris, hinders official tracking and cataloging. Consequently, some debris are observed but are lost or decay before being assigned a permanent catalog number. The numbers of cataloged debris listed in this section are only from the anomalous events and do not include normal mission related debris identified with the particular launch nor the parent itself.

Historically, anomalous events have often been confused with satellite breakups and have not been the subject of separate, extensive analyses. The list of events in this section is known to be incomplete. Several other satellites have been tentatively tagged as sources of anomalous events. Moreover, preliminary satellite catalog surveys suggest that additional anomalous events have occurred but remain unrecognized as such. Table 3.2 suggests a potential correlation of anomalous events with high solar activity. This section will be updated as future studies warrant.

For additional information on anomalous events, see “Environmentally induced Debris Sources,” N. L. Johnson, Second World Space Congress, October 2002.

3.2 *Identified Satellite Anomalous Events*

Much like section 2.2 above, this section identifies particulars for the limited number of anomalous events thus far cataloged. There is no Gabbard Diagram included with these events, and each page often refers to multiple events. The first known date of the first anomalous event is categorized for each satellite. Where possible the best estimate of the cause and potential failure are noted in the comments section.

TABLE 3.1 HISTORY OF SATELLITE ANOMALOUS EVENTS BY LAUNCH DATE

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | FIRST EVENT DATE | KNOWN EVENTS | CATALOGED DEBRIS | ON-ORBIT DEBRIS | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) |
|-------------------------|-----------------------------|---------------------------|----------------|------------------------|-----------------|---------------------|--------------------|----------------|-----------------|----------------------|
| VANGUARD 3 | 1959-007A | 20 | 18-Sep-59 | 14-Feb-06 | 1 | 2 | 0 | 3310 | 510 | 33.4 |
| TRANSIT 5B-2 | 1963-049B | 704 | 5-Dec-63 | 9/10-Jan-98 | 1 | 2 | 2 | 1110 | 1060 | 90.1 |
| ELEKTRON 1 | 1963-006A | 746 | 30-Jan-64 | prior to 1-Jan-80 | 1 | 24 | 1 | 6900 | 416 | 60.9 |
| OPS 4412 (TRANSIT 9) | 1964-026A | 801 | 4-Jun-64 | Dec-80 | 4 | 4 | 0 | 930 | 845 | 90.5 |
| COSMOS 44 R/B | 1964-053B | 877 | 28-Aug-64 | Nov-90 | 1 | 1 | 1 | 775 | 655 | 65.1 |
| OPS 4988 (GREB 6) | 1965-016A | 1271 | 9-Mar-65 | Nov-80 | 1 | 1 | 1 | 935 | 900 | 70.1 |
| OPS 4682 (SNAPSHOT) | 1965-027A | 1314 | 3-Apr-65 | 1-Nov-79 | 7 | 158 | 158 | 1320 | 1270 | 90.3 |
| OPS 8480 (TRANSIT 5B-6) | 1965-048A | 1420 | 24-Jun-65 | Aug-80 | 4 | 9 | 0 | 1135 | 1025 | 89.9 |
| ALOUETTE 2 | 1965-098A | 1804 | 29-Nov-65 | Aug-90 | 2 | 2 | 0 | 2705 | 505 | 79.8 |
| FR-1 R/B | 1965-101B | 1815 | 6-Dec-65 | 21-Mar-03 | 1 | 2 | 0 | 660 | 655 | 75.8 |
| OPS 1509 (TRANSIT 10) | 1965-109A | 1864 | 22-Dec-65 | 30-Nov-96 | 2 | 2 | 1 | 1065 | 895 | 89.1 |
| OPS 1593 (TRANSIT 11) | 1966-005A | 1952 | 28-Jan-66 | Apr-80 | 4 | 7 | 1 | 1205 | 855 | 89.8 |
| OPS 1117 (TRANSIT 12) | 1966-024A | 2119 | 26-Mar-66 | Jul-81 | 1 | 3 | 2 | 1115 | 890 | 89.9 |
| NIMBUS 2 | 1966-040A | 2173 | 15-May-66 | Nov-97 | Many | 66 | 66 | 1175 | 1095 | 100.4 |
| OPS 0856 | 1966-077A | 2403 | 19-Aug-66 | Mar-91 | 5 | 25 | 25 | 3710 | 3660 | 89.7 |
| OPS 0100 (TRANSIT 15) | 1967-034A | 2754 | 14-Apr-67 | Sep-92 | 1 | 5 | 4 | 1065 | 1035 | 90.1 |
| OPS 7218 (TRANSIT 16) | 1967-048A | 2807 | 18-May-67 | Feb-95 | 1 | 4 | 2 | 1090 | 1060 | 89.6 |
| OPS 4947 (TRANSIT 17) | 1967-092A | 2965 | 25-Sep-67 | Apr-81 | 4 | 7 | 0 | 1110 | 1035 | 89.3 |
| COSMOS 206 R/B | 1968-019B | 3151 | 14-Mar-68 | Nov-90 | 1 | 0 | 0 | 515 | 450 | 81.2 |
| ISIS 1 | 1969-009A | 3669 | 30-Jan-69 | 24-May-07 | 1 | 2 | 2 | 3455 | 580 | 88.5 |
| TRANSIT 19 | 1970-067A | 4507 | 27-Aug-70 | 7-Mar-98 | 1 | 1 | 0 | 1205 | 945 | 90.0 |
| METEOR 1-7 R/B | 1971-003B | 4850 | 20-Jan-71 | Jun-87 | 1 | 1 | 0 | 665 | 535 | 81.2 |
| METEOR 1-12 R/B | 1972-049B | 6080 | 30-Jun-72 | Sep-89 | 1 | 1 | 1 | 935 | 860 | 81.2 |
| COSMOS 539 | 1972-102A | 6319 | 21-Dec-72 | 21-Apr-02 | 1 | 1 | 0 | 1380 | 1340 | 74.0 |
| GEOS 3 R/B | 1975-027B | 7735 | 9-Apr-75 | Mar-78 | 1 | 3 | 2 | 845 | 835 | 115.0 |
| KYOKKOH 1 (EXOS-A) | 1978-014A | 10664 | 4-Feb-78 | Jan-88 | 2 | 2 | 0 | 4220 | 760 | 65.0 |
| SEASAT | 1978-064A | 10967 | 27-Jun-78 | Jul-83 | >12 | 18 | 1 | 780 | 780 | 108.0 |
| COSMOS 1043 | 1978-094A | 11055 | 10-Oct-78 | Feb-93 | 1 | 1 | 0 | 435 | 435 | 81.2 |
| TIROS-N | 1978-096A | 11060 | 13-Oct-78 | Sep-87 | 2 | 5 | 0 | 855 | 835 | 99.0 |
| NIMBUS 7 R/B | 1978-098B | 11081 | 24-Oct-78 | May-81 | 2 | 1 | 0 | 955 | 935 | 99.3 |
| NOAA 6 | 1979-057A | 11416 | 27-Jun-79 | Sep-92 | 2 | 3 | 1 | 805 | 790 | 98.7 |
| METEOR 2-5 | 1979-095A | 11605 | 31-Oct-79 | prior to 1-Jan-05 | Multiple | 83 | 60 | 881 | 862 | 81.2 |
| METEOR 2-7 | 1981-043A | 12456 | 14-May-81 | Mar-04 | 1 | 20 | 15 | 895 | 825 | 81.3 |
| METEOR 2-7 R/B | 1981-043B | 12457 | 14-May-81 | Oct-96 | 1 | 1 | 1 | 920 | 825 | 81.3 |

TABLE 3.1 HISTORY OF SATELLITE ANOMALOUS EVENTS BY LAUNCH DATE (CONT'D)

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | FIRST EVENT DATE | KNOWN EVENTS | CATALOGED DEBRIS | ON-ORBIT DEBRIS | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) |
|-------------------------|--------------------------|---------------------|-------------|--------------------|--------------|------------------|-----------------|-------------|--------------|-------------------|
| MOLNIYA 3-16 R/B | 1981-054E | 12519 | 9-Jun-81 | Jul-98 | 1 | 0 | 0 | 33415 | 85 | 62.0 |
| NOAA 7 | 1981-059A | 12553 | 23-Jun-81 | 26-Jul-93 | 2 | 6 | 1 | 835 | 830 | 98.9 |
| COSMOS 1417 R/B | 1982-102B | 13618 | 19-Oct-82 | Early-09 | 1 | 1 | 1 | 1000 | 955 | 83.0 |
| NOAA 8 | 1983-022A | 13923 | 28-Mar-83 | Early-05 | 2 | 4 | 3 | 810 | 790 | 98.7 |
| OSCAR 30 | 1985-066A | 15935 | 3-Aug-85 | Nov-86 | 2 | 2 | 2 | 1255 | 1000 | 89.9 |
| COSMOS 1689 R/B | 1985-090B | 16111 | 3-Oct-85 | 5-May-02 | 1 | 1 | 1 | 565 | 510 | 97.7 |
| NOAA 10 | 1986-073A | 16969 | 17-Sep-86 | Jun-07 | 2 | 4 | 1 | 810 | 790 | 98.7 |
| COSMOS 1818 | 1987-011A | 17369 | 1-Feb-87 | 4-Jul-08 | 1 | 0 | 0 | 800 | 775 | 65.0 |
| COSMOS 1823 | 1987-020A | 17535 | 20-Feb-87 | Apr-May 97 | 3 | 3 | 3 | 1525 | 1480 | 73.6 |
| COSMOS 1867 | 1987-060A | 18187 | 10-Jul-87 | 21-Mar to 4-Apr-14 | 1 | 0 | 0 | 800 | 775 | 65.0 |
| METEOR 2-17 | 1988-005A | 18820 | 30-Jan-88 | 2000-2001 | 2 | 0 | 0 | 960 | 936 | 82.5 |
| COSMOS 1939 R/B | 1988-032B | 19046 | 20-Apr-88 | 30-Jul-96 | 2 | 2 | 2 | 655 | 585 | 97.6 |
| NOAA 11 | 1988-089A | 19531 | 24-Sep-88 | 24-Nov-10 | 1 | 2 | 2 | 850 | 835 | 98.8 |
| COBE | 1989-089A | 20322 | 18-Nov-89 | Mar-93 | 12 | 78 | 2 | 885 | 870 | 99.0 |
| NADEZHDA 2 R/B | 1990-017B | 20509 | 27-Feb-90 | 22-Jun-05 | 1 | 1 | 1 | 1015 | 950 | 83.0 |
| HST | 1990-037B | 20580 | 24-Apr-90 | 5-Aug-03 | 1 | 1 | 0 | 575 | 570 | 28.5 |
| NOAA 12 | 1991-032A | 21263 | 14-May-91 | 2-Oct-11 | 1 | 4 | 4 | 815 | 800 | 98.7 |
| OKEAN 3 | 1991-039A | 21397 | 4-Jun-91 | 12-Oct-98 | 1 | 1 | 0 | 655 | 620 | 82.5 |
| ERS-1 | 1991-050A | 21574 | 17-Jul-91 | 7-Jul-02 | 7 | 7 | 7 | 800 | 750 | 98.5 |
| SARA | 1991-050E | 21578 | 17-Jul-91 | 22-Aug-03 | 3 | 4 | 0 | 730 | 730 | 98.1 |
| ERS-1 R/B | 1991-050F | 21610 | 17-Jul-91 | Apr-01 | 1 | 1 | 0 | 770 | 770 | 98.2 |
| UARS | 1991-063B | 21701 | 12-Sep-91 | 10-Nov-07 | 3 | 5 | 0 | 485 | 355 | 57.0 |
| EKA 1 (START 1) | 1993-014A | 22561 | 25-Mar-93 | 4-Mar-98 | 1 | 2 | 2 | 970 | 685 | 75.8 |
| START 1 R/B | 1993-014B | 22562 | 25-Mar-93 | Late-02 | Multiple | 56 | 22 | 920 | 680 | 75.8 |
| USA 106 (DMSP 5D-2 F12) | 1994-057A | 23233 | 29-Aug-94 | 23-Oct-16 | 1 | 4 | 4 | 848 | 832 | 99.1 |
| COSMOS 2297 R/B | 1994-077B | 23405 | 24-Nov-94 | Jun-98 | 2? | 1 | 0 | 845 | 845 | 71.0 |
| NOAA 14 | 1994-089A | 23455 | 30-Dec-94 | Jul-12 | 1 | 1 | 1 | 860 | 850 | 98.8 |
| ERS-2 | 1995-021A | 23560 | 21-Apr-95 | 24-Jul-03 | Multiple | 6 | 5 | 787 | 785 | 98.6 |
| ERS-2 R/B | 1995-021B | 23561 | 21-Apr-95 | prior to 1-Jan-18 | 1 | 1 | 1 | 772 | 762 | 98.8 |
| KOREASAT 1 R/B | 1995-041B | 23640 | 5-Aug-95 | 6-Dec-95 | 1 | 1 | 0 | 1375 | 935 | 26.7 |
| RADARSAT R/B | 1995-059B | 23711 | 4-Nov-95 | 30-Jan-96 | 1 | 2 | 0 | 1495 | 935 | 100.6 |
| IRS B3 | 1996-017A | 23827 | 21-Mar-96 | Oct-00 | 1 | 1 | 0 | 822 | 820 | 98.6 |
| IRIDIUM 47 | 1997-082C | 25106 | 20-Dec-97 | 7-Jun-14 | 1 | 10 | 7 | 781 | 778 | 86.4 |
| FUSE | 1999-035A | 25791 | 24-Jun-99 | 6-Jun-04 | 1 | 9 | 0 | 760 | 745 | 25.0 |

TABLE 3.1 HISTORY OF SATELLITE ANOMALOUS EVENTS BY LAUNCH DATE (CONT'D)

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | FIRST EVENT DATE | KNOWN EVENTS | CATALOGED DEBRIS | ON-ORBIT DEBRIS | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) |
|-------------------|-----------------------------|---------------------------|----------------|------------------------|-----------------|---------------------|--------------------|----------------|-----------------|----------------------|
| TELKOM 1 | 1999-042A | 25880 | 12-Aug-99 | 25-Aug-17 | 1 | 0 | 0 | 35793 | 35781 | 0.0112 |
| IKONOS 2 | 1999-051A | 25919 | 24-Sep-99 | 19-Mar-01 | 1 | 1 | 0 | 680 | 680 | 98.2 |
| JASON 1 | 2001-055A | 26997 | 7-Dec-01 | 16-Mar-02 | 1 | 2 | 2 | 1345 | 1330 | 66.0 |
| IRIDIUM 91 | 2002-005A | 27372 | 11-Feb-02 | 30-Nov-14 | 1 | 4 | 2 | 781 | 777 | 86.4 |
| AMC-9 (GE-12) | 2003-024A | 27820 | 6-Jun-03 | 17-Jun-17 | 1 | 0 | 0 | 35798 | 35774 | 0.0174 |
| BLITS | 2009-049G | 35871 | 17-Sep-09 | 22-Jan-13 | 1 | 1 | 1 | 825 | 815 | 98.6 |
| WORLDVIEW 2 | 2009-055A | 35946 | 8-Oct-09 | 18-Jul-16 | 1 | 16 | 16 | 768 | 767 | 98.5 |
| HAIYANG 2A | 2011-043A | 37781 | 15-Aug-11 | 6/7-Jul-14 | 2 | 3 | 0 | 965 | 965 | 99.0 |
| RISAT 1 | 2012-017A | 38248 | 26-Apr-12 | 30-Sep-16 | 1 | 1 | 0 | 544 | 538 | 97.6 |
| BRIZ-KM R/B | 2015-020E | 40556 | 31-Mar-15 | 29-Apr-15 | 1 | 6 | 6 | 1342 | 1339 | 82.5 |
| | | | | | TOTALS | 722 | 446 | | | |

TABLE 3.2 HISTORY OF SATELLITE ANOMALOUS EVENTS BY EVENT DATE

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | FIRST EVENT DATE | KNOWN EVENTS | CATALOGED DEBRIS | ON-ORBIT DEBRIS | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) |
|-------------------------|-----------------------------|---------------------------|----------------|------------------------|-----------------|---------------------|--------------------|----------------|-----------------|----------------------|
| GEOS 3 R/B | 1975-027B | 7735 | 9-Apr-75 | Mar-78 | 1 | 3 | 2 | 845 | 835 | 115.0 |
| OPS 4682 (SNAPSHOT) | 1965-027A | 1314 | 3-Apr-65 | 1-Nov-79 | 7 | 158 | 158 | 1320 | 1270 | 90.3 |
| ELEKTRON 1 | 1963-006A | 746 | 30-Jan-64 | prior to 1-Jan-80 | 1 | 24 | 1 | 6900 | 416 | 60.9 |
| OPS 1593 (TRANSIT 11) | 1966-005A | 1952 | 28-Jan-66 | Apr-80 | 4 | 7 | 1 | 1205 | 855 | 89.8 |
| OPS 8480 (TRANSIT 5B-6) | 1965-048A | 1420 | 24-Jun-65 | Aug-80 | 4 | 9 | 0 | 1135 | 1025 | 89.9 |
| OPS 4988 (GREB 6) | 1965-016A | 1271 | 9-Mar-65 | Nov-80 | 1 | 1 | 1 | 935 | 900 | 70.1 |
| OPS 4412 (TRANSIT 9) | 1964-026A | 801 | 4-Jun-64 | Dec-80 | 4 | 4 | 0 | 930 | 845 | 90.5 |
| OPS 4947 (TRANSIT 17) | 1967-092A | 2965 | 25-Sep-67 | Apr-81 | 4 | 7 | 0 | 1110 | 1035 | 89.3 |
| NIMBUS 7 R/B | 1978-098B | 11081 | 24-Oct-78 | May-81 | 2 | 1 | 0 | 955 | 935 | 99.3 |
| OPS 1117 (TRANSIT 12) | 1966-024A | 2119 | 26-Mar-66 | Jul-81 | 1 | 3 | 2 | 1115 | 890 | 89.9 |
| SEASAT | 1978-064A | 10967 | 27-Jun-78 | Jul-83 | >12 | 18 | 1 | 780 | 780 | 108.0 |
| OSCAR 30 | 1985-066A | 15935 | 3-Aug-85 | Nov-86 | 2 | 2 | 2 | 1255 | 1000 | 89.9 |
| METEOR 1-7 R/B | 1971-003B | 4850 | 20-Jan-71 | Jun-87 | 1 | 1 | 0 | 665 | 535 | 81.2 |
| TIROS-N | 1978-096A | 11060 | 13-Oct-78 | Sep-87 | 2 | 5 | 0 | 855 | 835 | 99.0 |
| KYOKKOH 1 (EXOS-A) | 1978-014A | 10664 | 4-Feb-78 | Jan-88 | 2 | 2 | 0 | 4220 | 760 | 65.0 |
| METEOR 1-12 R/B | 1972-049B | 6080 | 30-Jun-72 | Sep-89 | 1 | 1 | 1 | 935 | 860 | 81.2 |
| ALOUETTE 2 | 1965-098A | 1804 | 29-Nov-65 | Aug-90 | 2 | 2 | 0 | 2705 | 505 | 79.8 |
| COSMOS 44 R/B | 1964-053B | 877 | 28-Aug-64 | Nov-90 | 1 | 1 | 1 | 775 | 655 | 65.1 |
| COSMOS 206 R/B | 1968-019B | 3151 | 14-Mar-68 | Nov-90 | 1 | 0 | 0 | 515 | 450 | 81.2 |
| OPS 0856 | 1966-077A | 2403 | 19-Aug-66 | Mar-91 | 5 | 25 | 25 | 3710 | 3660 | 89.7 |
| OPS 0100 (TRANSIT 15) | 1967-034A | 2754 | 14-Apr-67 | Sep-92 | 1 | 5 | 4 | 1065 | 1035 | 90.1 |
| NOAA 6 | 1979-057A | 11416 | 27-Jun-79 | Sep-92 | 2 | 3 | 1 | 805 | 790 | 98.7 |
| COSMOS 1043 | 1978-094A | 11055 | 10-Oct-78 | Feb-93 | 1 | 1 | 0 | 435 | 435 | 81.2 |
| COBE | 1989-089A | 20322 | 18-Nov-89 | Mar-93 | 12 | 78 | 2 | 885 | 870 | 99.0 |
| NOAA 7 | 1981-059A | 12553 | 23-Jun-81 | 26-Jul-93 | 2 | 6 | 1 | 835 | 830 | 98.9 |
| OPS 7218 (TRANSIT 16) | 1967-048A | 2807 | 18-May-67 | Feb-95 | 1 | 4 | 2 | 1090 | 1060 | 89.6 |
| KOREASAT 1 R/B | 1995-041B | 23640 | 5-Aug-95 | 6-Dec-95 | 1 | 1 | 0 | 1375 | 935 | 26.7 |
| RADARSAT R/B | 1995-059B | 23711 | 4-Nov-95 | 30-Jan-96 | 1 | 2 | 0 | 1495 | 935 | 100.6 |
| COSMOS 1939 R/B | 1988-032B | 19046 | 20-Apr-88 | 30-Jul-96 | 2 | 2 | 2 | 655 | 585 | 97.6 |
| METEOR 2-7 R/B | 1981-043B | 12457 | 14-May-81 | Oct-96 | 1 | 1 | 1 | 920 | 825 | 81.3 |
| OPS 1509 (TRANSIT 10) | 1965-109A | 1864 | 22-Dec-65 | 30-Nov-96 | 2 | 2 | 1 | 1065 | 895 | 89.1 |
| COSMOS 1823 | 1987-020A | 17535 | 20-Feb-87 | Apr-May 97 | 3 | 3 | 3 | 1525 | 1480 | 73.6 |
| NIMBUS 2 | 1966-040A | 2173 | 15-May-66 | Nov-97 | Many | 66 | 66 | 1175 | 1095 | 100.4 |
| TRANSIT 5B-2 | 1963-049B | 704 | 5-Dec-63 | 9/10-Jan-98 | 1 | 2 | 2 | 1110 | 1060 | 90.1 |

TABLE 3.2 HISTORY OF SATELLITE ANOMALOUS EVENTS BY EVENT DATE (CONT'D)

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | FIRST EVENT DATE | KNOWN EVENTS | CATALOGED DEBRIS | ON-ORBIT DEBRIS | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) |
|------------------|--------------------------|---------------------|-------------|--------------------|--------------|------------------|-----------------|-------------|--------------|-------------------|
| EKA 1 (START 1) | 1993-014A | 22561 | 25-Mar-93 | 4-Mar-98 | 1 | 2 | 2 | 970 | 685 | 75.8 |
| TRANSIT 19 | 1970-067A | 4507 | 27-Aug-70 | 7-Mar-98 | 1 | 1 | 0 | 1205 | 945 | 90.0 |
| COSMOS 2297 R/B | 1994-077B | 23405 | 24-Nov-94 | Jun-98 | 2? | 1 | 0 | 845 | 845 | 71.0 |
| MOLNIYA 3-16 R/B | 1981-054E | 12519 | 9-Jun-81 | Jul-98 | 1 | 0 | 0 | 33415 | 85 | 62.0 |
| OKEAN 3 | 1991-039A | 21397 | 4-Jun-91 | 12-Oct-98 | 1 | 1 | 0 | 655 | 620 | 82.5 |
| METEOR 2-17 | 1988-005A | 18820 | 30-Jan-88 | 2000-2001 | 2 | 0 | 0 | 960 | 936 | 82.5 |
| IRS B3 | 1996-017A | 23827 | 21-Mar-96 | Oct-00 | 1 | 1 | 0 | 822 | 820 | 98.6 |
| IKONOS 2 | 1999-051A | 25919 | 24-Sep-99 | 19-Mar-01 | 1 | 1 | 0 | 680 | 680 | 98.2 |
| ERS-1 R/B | 1991-050F | 21610 | 17-Jul-91 | Apr-01 | 1 | 1 | 0 | 770 | 770 | 98.2 |
| JASON 1 | 2001-055A | 26997 | 7-Dec-01 | 16-Mar-02 | 1 | 2 | 2 | 1345 | 1330 | 66.0 |
| COSMOS 539 | 1972-102A | 6319 | 21-Dec-72 | 21-Apr-02 | 1 | 1 | 0 | 1380 | 1340 | 74.0 |
| COSMOS 1689 R/B | 1985-090B | 16111 | 3-Oct-85 | 5-May-02 | 1 | 1 | 1 | 565 | 510 | 97.7 |
| ERS-1 | 1991-050A | 21574 | 17-Jul-91 | 7-Jul-02 | 7 | 7 | 7 | 800 | 750 | 98.5 |
| START 1 R/B | 1993-014B | 22562 | 25-Mar-93 | Late-02 | Multiple | 56 | 22 | 920 | 680 | 75.8 |
| FR-1 R/B | 1965-101B | 1815 | 6-Dec-65 | 21-Mar-03 | 1 | 2 | 0 | 660 | 655 | 75.8 |
| ERS-2 | 1995-021A | 23560 | 21-Apr-95 | 24-Jul-03 | Multiple | 6 | 5 | 787 | 785 | 98.6 |
| HST | 1990-037B | 20580 | 24-Apr-90 | 5-Aug-03 | 1 | 1 | 0 | 575 | 570 | 28.5 |
| SARA | 1991-050E | 21578 | 17-Jul-91 | 22-Aug-03 | 3 | 4 | 0 | 730 | 730 | 98.1 |
| METEOR 2-7 | 1981-043A | 12456 | 14-May-81 | Mar-04 | 1 | 20 | 15 | 895 | 825 | 81.3 |
| FUSE | 1999-035A | 25791 | 24-Jun-99 | 6-Jun-04 | 1 | 9 | 0 | 760 | 745 | 25.0 |
| METEOR 2-5 | 1979-095A | 11605 | 31-Oct-79 | prior to 1-Jan-05 | Multiple | 83 | 60 | 881 | 862 | 81.2 |
| NOAA 8 | 1983-022A | 13923 | 28-Mar-83 | Early-05 | 2 | 4 | 3 | 810 | 790 | 98.7 |
| NADEZHDA 2 R/B | 1990-017B | 20509 | 27-Feb-90 | 22-Jun-05 | 1 | 1 | 1 | 1015 | 950 | 83.0 |
| VANGUARD 3 | 1959-007A | 20 | 18-Sep-59 | 14-Feb-06 | 1 | 2 | 0 | 3310 | 510 | 33.4 |
| ISIS 1 | 1969-009A | 3669 | 30-Jan-69 | 24-May-07 | 1 | 2 | 2 | 3455 | 580 | 88.5 |
| NOAA 10 | 1986-073A | 16969 | 17-Sep-86 | Jun-07 | 2 | 4 | 1 | 810 | 790 | 98.7 |
| UARS | 1991-063B | 21701 | 12-Sep-91 | 10-Nov-07 | 3 | 5 | 0 | 485 | 355 | 57.0 |
| COSMOS 1818 | 1987-011A | 17369 | 1-Feb-87 | 4-Jul-08 | 1 | 0 | 0 | 800 | 775 | 65.0 |
| COSMOS 1417 R/B | 1982-102B | 13618 | 19-Oct-82 | Early-09 | 1 | 1 | 1 | 1000 | 955 | 83.0 |
| NOAA 11 | 1988-089A | 19531 | 24-Sep-88 | 24-Nov-10 | 1 | 2 | 2 | 850 | 835 | 98.8 |
| NOAA 12 | 1991-032A | 21263 | 14-May-91 | 2-Oct-11 | 1 | 4 | 4 | 815 | 800 | 98.7 |
| NOAA 14 | 1994-089A | 23455 | 30-Dec-94 | Jul-12 | 1 | 1 | 1 | 860 | 850 | 98.8 |
| BLITS | 2009-049G | 35871 | 17-Sep-09 | 22-Jan-13 | 1 | 1 | 1 | 825 | 815 | 98.6 |
| COSMOS 1867 | 1987-060A | 18187 | 10-Jul-87 | 21-Mar to 4-Apr-14 | 1 | 0 | 0 | 800 | 775 | 65.0 |

TABLE 3.2 HISTORY OF SATELLITE ANOMALOUS EVENTS BY EVENT DATE (CONT'D)

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | FIRST EVENT DATE | KNOWN EVENTS | CATALOGED DEBRIS | ON-ORBIT DEBRIS | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) |
|-------------------------|-----------------------------|---------------------------|----------------|------------------------|-----------------|---------------------|--------------------|----------------|-----------------|----------------------|
| IRIDIUM 47 | 1997-082C | 25106 | 20-Dec-97 | 7-Jun-14 | 1 | 10 | 7 | 781 | 778 | 86.4 |
| HAIYANG 2A | 2011-043A | 37781 | 15-Aug-11 | 6/7-Jul-14 | 2 | 3 | 0 | 965 | 965 | 99.0 |
| IRIDIUM 91 | 2002-005A | 27372 | 11-Feb-02 | 30-Nov-14 | 1 | 4 | 2 | 781 | 777 | 86.4 |
| BRIZ-KM R/B | 2015-020E | 40556 | 31-Mar-15 | 29-Apr-15 | 1 | 6 | 6 | 1342 | 1339 | 82.5 |
| WORLDVIEW 2 | 2009-055A | 35946 | 8-Oct-09 | 18-Jul-16 | 1 | 16 | 16 | 768 | 767 | 98.5 |
| RISAT 1 | 2012-017A | 38248 | 26-Apr-12 | 30-Sep-16 | 1 | 1 | 0 | 544 | 538 | 97.6 |
| USA 106 (DMSP 5D-2 F12) | 1994-057A | 23233 | 29-Aug-94 | 23-Oct-16 | 1 | 4 | 4 | 848 | 832 | 99.1 |
| AMC-9 (GE-12) | 2003-024A | 27820 | 6-Jun-03 | 17-Jun-17 | 1 | 0 | 0 | 35798 | 35774 | 0.0174 |
| TELKOM 1 | 1999-042A | 25880 | 12-Aug-99 | 25-Aug-17 | 1 | 0 | 0 | 35793 | 35781 | 0.0112 |
| ERS-2 R/B | 1995-021B | 23561 | 21-Apr-95 | prior to 1-Jan-18 | 1 | 1 | 1 | 772 | 762 | 98.8 |
| | | | | | TOTALS | 722 | 446 | | | |

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 18 Sep 1959
DRY MASS (KG): 45
MAIN BODY: 50.8 cm sphere with third stage attached
MAJOR APPENDAGES: 66 cm boom(s)
ATTITUDE CONTROL: spin stabilized

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: 14 Feb 2006

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|------------|-------------|
| 3310 | 510 | 125.14 min | 33.4 deg |

COMMENTS

At the time of the event, Vanguard 3 was the 5th oldest object in orbit. Two objects released, although the second object was not cataloged until May 2007.

REFERENCE DOCUMENT

“First Satellite Breakups of 2006”, The Orbital Debris Quarterly News, NASA JSC, July 2006.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNV10i3.pdf>.

“Detection of Debris from Chinese ASAT Test Increases; One Minor Fragmentation Event in Second Quarter of 2007”, The Orbital Debris Quarterly News, NASA JSC, July 2007.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNV11i3.pdf>.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 5 December 1963
DRY MASS (KG): 75
MAIN BODY: Octagon; 0.46 m diameter by 0.5 m length
MAJOR APPENDAGES: Boom
ATTITUDE CONTROL: None at the time of event

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: 9-10 January 1998

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------------|----------------|---------------|--------------------|
| 1110 km | 1060 km | 106.98 min | 90.1 deg |

COMMENTS

One of several Transit-class spacecraft involved in anomalous events. Spacecraft may have experienced earlier anomalous events in 1960's and 1970's. Only one object associated with January 1998 event.

SATELLITE DATA

TYPE: payload
OWNER: CIS
LAUNCH DATE: 30 Jan 1964
DRY MASS (KG): 329
MAIN BODY: Cylinder with hemispherical ends; 0.75 m diameter by 1.3 m length
MAJOR APPENDAGES: Six solar arrays, antennas
ATTITUDE CONTROL: None at the time of event

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: prior to 1 January 1980

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------------|----------------|---------------|--------------------|
| 6900 km | 416 km | 166.8 min | 60.9 deg |

COMMENTS

“Event Data” is for epoch of 3 January 1980, given the uncertainty in actual time of separation event(s). It is likely that four of the 24 anomalous debris are associated with other space missions, including the sole piece on orbit as of this edition.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 4 June 1964
DRY MASS (KG): 60
MAIN BODY: Octagonal cylinder; 0.5 m by 0.4 m
MAJOR APPENDAGES: 4 solar panels; gravity-gradient boom
ATTITUDE CONTROL: Gravity-gradient boom

EVENT DATA

KNOWN EVENTS: 4
FIRST DATE: December 1980

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|-----------|-------------|
| 930 km | 845 km | 102.7 min | 90.5 deg |

COMMENTS

Second event observed July 1982. The third event occurred in May 1994. Fourth event date not determined but also close to May 1994. First fragment decayed rapidly; the second decayed more slowly. Two latest pieces not cataloged as of publication date. One of several known Transits involved in anomalous events.

COSMOS 44 R/B

1964-053B

877

SATELLITE DATA

TYPE: Vostok Final Stage
OWNER: CIS
LAUNCH DATE: 28 August 1964
DRY MASS (KG): 1440
MAIN BODY: Cylinder; 2.6 m diameter by 3.8 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of event

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: Late-1990

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------------|----------------|---------------|--------------------|
| 775 km | 655 km | 99.1 min | 65.1 deg |

COMMENTS

Cosmos 44 was the first prototype spacecraft of the Meteor 1 program. This is one of several Vostok final stages associated with this old program to shed a piece of debris since 1987.

OPS 4988 (GREB 6)

1965-016A

1271

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 9 May 1965
DRY MASS (KG): 40
MAIN BODY: Sphere
MAJOR APPENDAGES: Unknown
ATTITUDE CONTROL: Unknown

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: November 1980

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------------|----------------|---------------|--------------------|
| 935 km | 900 km | 103.4 min | 70.1 deg |

COMMENTS

No other events observed.

SATELLITE DATA

TYPE: Payload (attached to Agena D upper stage)
OWNER: US
LAUNCH DATE: 3 April 1965
DRY MASS (KG): 2500 (approx.)
MAIN BODY: Cylinder-cone; 1.5 m by 11.6 m
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of event

EVENT DATA

KNOWN EVENTS: 7
FIRST DATE: November 1979

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------|---------|-----------|-------------|
| 1320 km | 1270 km | 111.5 min | 90.3 deg |

COMMENTS

Six additional events observed: Dec 1980, Aug 1981, Mar 1983, Aug 1983, Nov 1983, and Jan 1985. Decay rates of all debris are nominal for this altitude. One debris was administratively decayed in February 1989.

REFERENCE DOCUMENTS

Investigation of Certain Anomalies Associated with Object 1314, A US Nuclear Powered Satellite, G. T. DeVere, Technical Memorandum 85-S-001, Headquarters NORAD/ADCOM, DCS/Plans, March 1985 (Appendix TM-85-001A, Secret).

Environmentally-Induced Debris Sources, N. L. Johnson, NASA Lyndon B. Johnson Space Center, Second World Space Congress, 2002.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 24 June 1965
DRY MASS (KG): 60
MAIN BODY: Octagonal cylinder; 0.5 m by 0.4 m
MAJOR APPENDAGES: 4 solar panels; gravity-gradient boom
ATTITUDE CONTROL: Gravity-gradient

EVENT DATA

KNOWN EVENTS: Multiple
FIRST DATE: August 1980

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------|---------|-----------|-------------|
| 1135 km | 1025 km | 106.8 min | 89.9 deg |

COMMENTS

Three additional events observed: one 2 days after the initial event, one in June 1981, and the most recent in late 1999. All debris appear very small. One of several known Transits involved in anomalous events.

REFERENCE DOCUMENT

Environmentally-Induced Debris Sources, N. L. Johnson, NASA Lyndon B. Johnson Space Center, Second World Space Congress, 2002.

SATELLITE DATA

TYPE: Payload
OWNER: Canada
LAUNCH DATE: 29 November 1965
DRY MASS (KG): 146
MAIN BODY: Oblate spheroid; 0.86 m high by 1.1 m diameter
MAJOR APPENDAGES: 73 m- and 22.8 m-long dipole antennas
ATTITUDE CONTROL: spin-stabilized

EVENT DATA

KNOWN EVENTS: 2
FIRST DATE: August 1990

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------|---------|-----------|-------------|
| 2742 km | 506 km | 118.7 min | 79.8 deg |

COMMENTS

First event in August 1990 (SSN 20833); second separation event 25 July 1996 (SSN 25058). Both objects displayed a high area-to-mass ratio resulting in relatively rapid decays on 11 February 1992 and 15 December 1999 respectively.

REFERENCE DOCUMENT

Corliss, W.R., Scientific Satellites. NASA SP-133 (1967).

FR-1 R/B

1965-101B

1815

SATELLITE DATA

TYPE: Rocket Body
OWNER: US
LAUNCH DATE: 6 December 1965
DRY MASS (KG): 26
MAIN BODY: Cylinder, 0.64 m diameter by 2.53 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at the time of the event

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: 21 March 2003

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------------|----------------|---------------|--------------------|
| 660 km | 655 km | 97.89 min | 75.8 deg |

COMMENTS

There was only one piece cataloged from the relatively small Scout R/B stage.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 22 December 1965
DRY MASS (KG): 60
MAIN BODY: Octagon; 0.5 m diameter by 0.4 m length
MAJOR APPENDAGES: 4 vanes
ATTITUDE CONTROL: None at time of event

EVENT DATA

KNOWN EVENTS: 2
FIRST DATE: 30 November 1996

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------|---------|------------|-------------|
| 1065 km | 895 km | 104.66 min | 89.1 deg |

COMMENTS

One of several Transit-class spacecraft involved in anomalous events. Two debris objects (one cataloged and one not cataloged) were being tracked in 1997.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 28 January 1966
DRY MASS (KG): 60
MAIN BODY: Octagonal cylinder; 0.5 m by 0.4 m
MAJOR APPENDAGES: 4 solar panels; gravity-gradient boom
ATTITUDE CONTROL: Gravity-gradient

EVENT DATA

KNOWN EVENTS: 3
FIRST DATE: April 1980

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------|---------|-----------|-------------|
| 1205 km | 855 km | 105.8 min | 89.8 deg |

COMMENTS

Two additional events observed: Sep 1980 and Jul 1983. Last event may have originated with a piece of debris from earlier event. One of several known Transits involved in anomalous events.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 26 March 1966
DRY MASS (KG): 60
MAIN BODY: Octagonal cylinder; 0.5 m by 0.4 m
MAJOR APPENDAGES: 4 solar panels; gravity-gradient boom
ATTITUDE CONTROL: Gravity-gradient

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: July 1981

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------|---------|-----------|-------------|
| 1115 km | 890 km | 105.1 min | 89.9 deg |

COMMENTS

No other events observed. One of several known Transits involved in anomalous events.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 15 May 1966
DRY MASS (KG): 414
MAIN BODY: Conical skeleton; 1.45 m diameter by 3.0 m length
MAJOR APPENDAGES: 2 Paddles
ATTITUDE CONTROL: None at time of event

EVENT DATA

KNOWN EVENTS: Multiple
FIRST DATE: November 1997

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------|---------|------------|-------------|
| 1175 km | 1095 km | 108.03 min | 100.4 deg |

COMMENTS

A single piece of debris was detected on 16 November 1997. Separation may have occurred about 1 November. Numerous debris were released from the late 1990's to 2001. See cited reference below.

REFERENCE DOCUMENT

Environmentally-Induced Debris Sources, N. L. Johnson, NASA Lyndon B. Johnson Space Center, Second World Space Congress, 2002.

SATELLITE DATA

TYPE: Agena D Stage
OWNER: US
LAUNCH DATE: 19 August 1966
DRY MASS (KG): 600
MAIN BODY: Cylinder; 1.5 m diameter by 8 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of events

EVENT DATA

KNOWN EVENTS: 4
FIRST DATE: March 1991

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------|---------|-----------|-------------|
| 3710 km | 3660 km | 167.5 min | 89.7 deg |

COMMENTS

Second, third, and fourth events observed on 16 June 1992, 23 June 1992, and 1 November 1995 respectively. Additional events may have occurred.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 14 April 1967
DRY MASS (KG): 60
MAIN BODY: Octagonal cylinder; 0.5 m by 0.4 m
MAJOR APPENDAGES: 4 solar panels; gravity-gradient boom
ATTITUDE CONTROL: Gravity-gradient

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: September 1992

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------|---------|--------|-------------|
| 1065 km | 1035 km | 106.2 | 90.1 |

COMMENTS

Event most likely happened around 28 September 1992 based on element data near the event time and analysis using the COMBO algorithm in the SATRAK Astrodynamics Toolkit. One of several Transit-class satellites involved in anomalous events.

REFERENCE DOCUMENT

Environmentally-Induced Debris Sources, N. L. Johnson, NASA Lyndon B. Johnson Space Center, Second World Space Congress, 2002.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 18 May 1967
DRY MASS (KG): 60
MAIN BODY: Octagonal cylinder; 0.5 m by 0.4 m
MAJOR APPENDAGES: 4 solar panels; gravity-gradient boom
ATTITUDE CONTROL: Gravity-gradient boom

EVENT DATA

KNOWN EVENTS: 1
DATE: 11/12 February 1995

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------|---------|------------|-------------|
| 1090 km | 1060 km | 106.12 min | 89.6 deg |

COMMENTS

One piece of debris liberated. One of several Transit-class satellites involved in anomalous events.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 25 September 1967
DRY MASS (KG): 60
MAIN BODY: Octagonal cylinder; 0.5 m by 0.4 m
MAJOR APPENDAGES: 4 solar panels; gravity-gradient boom
ATTITUDE CONTROL: Gravity-gradient

EVENT DATA

KNOWN EVENTS: 4
FIRST DATE: April 1981

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------|---------|-----------|-------------|
| 1110 km | 1035 km | 106.7 min | 89.3 deg |

COMMENTS

Second event observed in August 1986. One of several known Transits involved in anomalous events.

SATELLITE DATA

TYPE: Vostok Final Stage
OWNER: CIS
LAUNCH DATE: 14 March 1968
DRY MASS (KG): 1440
MAIN BODY: Cylinder; 2.6 m diameter by 3.8 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at the time of event

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: Late-1990

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|----------|-------------|
| 515 km | 450 km | 94.3 min | 81.2 deg |

COMMENTS

Cosmos 206 was a prototype spacecraft of the Meteor 1 program. This is one of several Vostok final stages to shed a piece of debris since 1987. One piece of debris was released, although never officially cataloged.

SATELLITE DATA

TYPE: Payload
OWNER: Canada
LAUNCH DATE: 30 January 1969
DRY MASS (KG): 240
MAIN BODY: Cylinder; 1.27 m diameter x 1.07 m length
MAJOR APPENDAGES: Several antennae
ATTITUDE CONTROL: None at the time of event

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: 24.82 May 2007

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------|---------|------------|-------------|
| 3455 km | 580 km | 127.57 min | 88.5 deg |

COMMENTS

One piece of debris cataloged. Altitude at the time of breakup was approximately 2940 km.

REFERENCE DOCUMENT

“Two Minor Satellite Fragmentations Identified in the Third Quarter”, The Orbital Debris Quarterly News, NASA JSC, October 2007. Available online at: <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNV11i4.pdf>

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 27 August 1970
DRY MASS (KG): 60
MAIN BODY: Octagon; 0.5 m diameter by 0.4 m length
MAJOR APPENDAGES: 4 solar panels; gravity-gradient boom
ATTITUDE CONTROL: None at time of event

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: 7 March 1998

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------|---------|------------|-------------|
| 1205 km | 945 km | 106.75 min | 90.0 deg |

COMMENTS

One of several Transit-class satellites involved in anomalous events.

SATELLITE DATA

TYPE: Vostok Final Stage
OWNER: CIS
LAUNCH DATE: 20 January 1971
DRY MASS (KG): 1440
MAIN BODY: Cylinder; 2.6 m diameter by 3.8 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: June 1987

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------------|----------------|---------------|--------------------|
| 665 km | 535 km | 96.7 min | 81.2 deg |

COMMENTS

No other events observed. One of several Vostok final stages to be involved in anomalous events.

METEOR 1-12 R/B

1972-049B

6080

SATELLITE DATA

TYPE: Vostok Final Stage
OWNER: CIS
LAUNCH DATE: 30 June 1972
DRY MASS (KG): 1440
MAIN BODY: Cylinder; 2.6 m diameter by 3.8 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: September 1989

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|-----------|-------------|
| 935 km | 860 km | 102.9 min | 81.2 deg |

COMMENTS

No other events observed. One of several Vostok final stages involved in anomalous events.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 21 December 1972
DRY MASS (KG): 600
MAIN BODY: Unknown
MAJOR APPENDAGES: Unknown
ATTITUDE CONTROL: None at the time of the event

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: April 2002

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------|---------|-----------|-------------|
| 1380 km | 1340 km | 112.9 min | 74.0 deg |

COMMENTS

One piece of debris cataloged. It was concluded that because of the deduced debris ejecta velocity, Cosmos 539 was apparently struck by a small meteoroid or man-made object.

REFERENCE DOCUMENTS

Environmentally-Induced Debris Sources, N. L. Johnson, NASA Lyndon B. Johnson Space Center, Second World Space Congress, 2002.

“A New Collision in Space?”, The Orbital Debris Quarterly News, NASA JSC, July 2002. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNV7i3.pdf>

SATELLITE DATA

TYPE: Delta Second Stage (2410)
OWNER: US
LAUNCH DATE: 9 April 1975
DRY MASS (KG): 900
MAIN BODY: Cylinder-nozzle; 2.4 m diameter by 8 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: March 1978

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|-----------|-------------|
| 845 km | 835 km | 101.7 min | 115.0 deg |

COMMENTS

Only one event noted with three fragments cataloged 12 March 1978. There was repeated mis-tagging of objects during 1978 among rocket body and debris. One fragment lost in 1978 and administratively decayed in 1983. This event may be related to series of major Delta second stage breakups.

KYOKKOH 1 (EXOS-A)

1978-014A

10664

SATELLITE DATA

TYPE: Payload
OWNER: Japan
LAUNCH DATE: 4 February 1978
DRY MASS (KG): 103
MAIN BODY: Octagonal cylinder; 0.95 m by 0.8 m
MAJOR APPENDAGES: 3 small booms
ATTITUDE CONTROL: Unknown

EVENT DATA

KNOWN EVENTS: 2
FIRST DATE: January 1988

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------------|----------------|---------------|--------------------|
| 4219 km | 760 km | 134.0 min | 65.0 deg |

COMMENTS

First event (object 18816) may have occurred much earlier than the January 1988 date. Catalog actions taken at the end of 1988. The second event occurred in June 1992. Object 22008 led a short life, decaying on 2 August 1992.

SATELLITE DATA

TYPE: Payload (attached to Agena R/B)
OWNER: US
LAUNCH DATE: 27 June 1978
DRY MASS (KG): 2300
MAIN BODY: Cylinder; 1.5 m diameter by 21 m length
MAJOR APPENDAGES: 2 solar panels; 1 antenna panel; miscellaneous booms
ATTITUDE CONTROL: None at time of event

EVENT DATA

KNOWN EVENTS: Multiple
FIRST DATE: July 1983

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|-----------|-------------|
| 780 km | 780 km | 100.5 min | 108.0 deg |

COMMENTS

On average, one piece of debris is released per year, but sometimes in groups. Most debris experience very rapid decay for this altitude. Last known release was in 2007.

REFERENCE DOCUMENTS

“Environmentally-Induced Debris Sources,” N.L. Johnson. Journal of Advances in Space Research, Vol. 34, Issue 5, 2004, pp. 993-999.

“Detection of Debris from Chinese ASAT Test Increases; One Minor Fragmentation Event in Second Quarter of 2007”, The Orbital Debris Quarterly News, NASA JSC, July 2007.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv11i3.pdf>.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 10 October 1978
DRY MASS (KG): 2200 (est.)
MAIN BODY: Cylinder; dimensions ~1.5 m diameter by 5 m length
MAJOR APPENDAGES: Solar panels; payload panels; gravity-gradient boom
ATTITUDE CONTROL: None at time of the event.

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: February 1993

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|----------|-------------|
| 437 km | 435 km | 94.9 min | 81.2 deg |

COMMENTS

No other satellite of this type has experienced an anomalous event. The piece was cataloged on 28 Feb 93 and decayed on 11 Mar 93. Given prior cataloging practices, other spacecraft could have experienced similar events that went unrecorded.

TIROS N

1978-096A

11060

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 13 October 1978
DRY MASS (KG): 725
MAIN BODY: Cylinder; 1.9 m diameter by 3.7 m length
MAJOR APPENDAGES: 1 solar panel
ATTITUDE CONTROL: None at time of the event

EVENT DATA

KNOWN EVENTS: 2
FIRST DATE: September 1987

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|-----------|-------------|
| 855 km | 835 km | 101.9 min | 99.0 deg |

COMMENTS

Both fragments from the first event decayed rapidly during winter of 1988-89. A second event associated with 1978-096A occurred on 23 Feb 96 liberating 1 piece.

SATELLITE DATA

TYPE: Delta Second Stage (2910)
OWNER: US
LAUNCH DATE: 24 October 1978
DRY MASS (KG): 900
MAIN BODY: Cylinder-nozzle; 2.4 m diameter by 8 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.

EVENT DATA

KNOWN EVENTS: 2
FIRST DATE: May 1981

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|-----------|-------------|
| 955 km | 935 km | 104.0 min | 99.3 deg |

COMMENTS

Second anomalous event apparently occurred about January 1987. A more prolific event in December 1981 is tentatively categorized as a satellite breakup (see Section 2). The cataloged debris section above refers only to the new fragment observed after the second anomalous event and does not include the Delta second stage that is accounted for in the tables of Section 2. These events may be related to the series of major Delta second stage breakups.

NOAA 6

1979-057A

11416

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 27 June 1979
DRY MASS (KG): 723
MAIN BODY: Rectangular; 3.71 m by 1.88 m
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: 3-axis reaction control

EVENT DATA

KNOWN EVENTS: 2
FIRST DATE: September 1992

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|-----------|-------------|
| 810 km | 795 km | 100.8 min | 98.68 deg |

COMMENTS

One piece of debris cataloged from the first event. A second event took place in June 1995 with one piece of debris liberated, but none cataloged.

REFERENCE DOCUMENT

Environmentally-Induced Debris Sources, N. L. Johnson, NASA Lyndon B. Johnson Space Center, Second World Space Congress, 2002.

METEOR 2-5

1979-095A

11605

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 31 Oct 1979
DRY MASS (KG): 2750
MAIN BODY: Cylinder
MAJOR APPENDAGES: Large Solar Arrays
ATTITUDE CONTROL: None at the time of event

EVENT DATA

KNOWN EVENTS: Multiple
FIRST DATE: prior to 1 Jan. 2005; orbital data for 1 Jan. 2014.

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|----------|---------|------------|-------------|
| 880.7 km | 862 km | 102.35 min | 81.2 deg |

COMMENTS

Multiple anomalous debris from multiple individual events; the origination date of the pieces is not conclusive. The last event occurred as recently as late 2013 to early 2014.

METEOR 2-7

1981-043A

12456

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 14 May 1981
DRY MASS (KG): 2750
MAIN BODY: Cylinder
MAJOR APPENDAGES: Large Solar Arrays
ATTITUDE CONTROL: None at the time of event

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: March 2004

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------------|----------------|---------------|--------------------|
| 895 km | 825 km | 102.15 min | 81.3 deg |

COMMENTS

Eight pieces of debris cataloged, may have been from two individual events, the origination date of the pieces is not conclusive. The rocket body associated with the launch of this spacecraft experienced an anomalous event over seven years earlier. The events are unrelated.

METEOR 2-7 R/B

1981-043B

12457

SATELLITE DATA

TYPE: Vostok Final Stage
OWNER: CIS
LAUNCH DATE: 14 May 1981
DRY MASS (KG): 1440
MAIN BODY: Cylinder; 2.6 m diameter by 3.8 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: October 1996

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------------|----------------|---------------|--------------------|
| 920 km | 825 km | 102.41 min | 81.3 deg |

COMMENTS

One of several Vostok upper stages involved in anomalous events.

MOLNIYA 3-16 R/B

1981-054E

12519

SATELLITE DATA

TYPE: Molniya Final Stage
OWNER: CIS
LAUNCH DATE: 9 Jun 1981
DRY MASS (KG): 1100
MAIN BODY: Cylinder; 2.7 m diameter by 3 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: July 1998

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------------|----------------|---------------|--------------------|
| 33415 km | 85 km | 583.42 min | 62.0 deg |

COMMENTS

No debris was cataloged from this event.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 23 June 1981
DRY MASS (KG): 723
MAIN BODY: Rectangular; 3.71 m by 1.88 m
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: None at the time of event

EVENT DATA

KNOWN EVENTS: 2
FIRST DATE: 26.5 July 1993

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|-----------|-------------|
| 835 km | 830 km | 101.6 min | 98.9 deg |

COMMENTS

Two objects were detected by the NAVSPOC and subsequently entered in the US SSN catalog. A piece separation analysis by the NAVSPOC identified the precise time these objects separated from the parent. It is unclear whether this event is a small breakup or whether other explanations such as spacecraft degradation could explain this separation. Unless other evidence is uncovered, this event will be classified as an anomalous event. The NOAA 7 payload was inactive for 3 years prior to this event. In 1997 a second, more curious event occurred. Three new debris appeared simultaneously with a discrete decrease in the orbital period of NOAA 7 of approximately 1 second.

REFERENCE DOCUMENT

Environmentally-Induced Debris Sources, N. L. Johnson, NASA Lyndon B. Johnson Space Center, Second World Space Congress, 2002.

COSMOS 1417 R/B

1982-102B

13618

SATELLITE DATA

TYPE: Cosmos 3M/SL-8 Second Stage
OWNER: CIS
LAUNCH DATE: 19 October 1982
DRY MASS (KG): 1435
MAIN BODY: Cylinder; 2.4 m diameter by 6 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: early 2009

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------------|----------------|---------------|--------------------|
| 1000 km | 955 km | 104.6 min | 83.0 deg |

COMMENTS

One piece of debris was cataloged. The object resides in an orbit similar to its parent.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 28.66 March 1983
DRY MASS (KG): 740
MAIN BODY: Rectangular; 3.71 m by 1.88 m
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: None at the time of event

EVENT DATA

KNOWN EVENTS: 2
FIRST DATE: 2004

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|-----------|-------------|
| 813 km | 790 km | 100.9 min | 98.7 deg |

COMMENTS

Four objects, liberated in two unique events, have been cataloged. The first event occurred in in 2004, and the last in 2011 (three debris). All objects displayed very high area-to-mass ratios and have decayed from orbit. Unless other evidence is uncovered, these events will be classified as anomalous events. The NOAA 8 payload employed the Advanced TIROS-N bus and was declared lost on 29 December 1985 following a battery thermal runaway. The battery was destroyed and that event is described in Section 2.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 3 August 1985
DRY MASS (KG): 60
MAIN BODY: Octagonal cylinder; 0.5 m by 0.4 m
MAJOR APPENDAGES: 4 solar panels; gravity-gradient boom
ATTITUDE CONTROL: Gravity-gradient boom

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: 27 December 1991

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------|---------|-----------|-------------|
| 1255 km | 1000 km | 107.8 min | 89.9 deg |

COMMENTS

Other debris pieces are associated with this dual payload launch. The most recent event identified (SCC 21878) apparently originated from Oscar 30 on 27 December 1991, when a portion of the gravity-gradient boom broke off. One of several Transit-class satellites involved in anomalous events.

REFERENCE DOCUMENT

“The Transit System,” L. Lee Pryor, AIAA Paper 92-1708, Applied Physics Laboratory, 1992.

COSMOS 1689 R/B

1985-090B

16111

SATELLITE DATA

TYPE: Vostok Final Stage
OWNER: CIS
LAUNCH DATE: 3 October 1985
DRY MASS (KG): 1440
MAIN BODY: Cylinder; 2.6 m diameter by 3.8 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at the time of the event.

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: May 2002

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|----------|-------------|
| 565 km | 510 km | 95.4 min | 97.7 deg |

COMMENTS

One of several Vostok stages involved in anomalous events.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 17.66 September 1986
DRY MASS (KG): 740
MAIN BODY: Rectangular; 3.71 m by 1.88 m
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: None at the time of event

EVENT DATA

KNOWN EVENTS: 3
FIRST DATE: April 2000

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|-----------|-------------|
| 816 km | 799 km | 101.0 min | 98.6 deg |

COMMENTS

Four objects, liberated in three unique events, have been cataloged. The first event occurred in April 2000, the second in 2007, and the last in 2010. All objects displayed very high area-to-mass ratios and have decayed from orbit. Unless other evidence is uncovered, this event will be classified as an anomalous event. The NOAA 10 payload employed the Advanced TIROS-N bus, was placed into a standby status on 17 September 1991 and had been decommissioned on 30 August 2001.

SATELLITE DATA

TYPE: Payload

OWNER: CIS

LAUNCH DATE: 23.86 April 1987

DRY MASS (KG): 2500

MAIN BODY: cylinder-cone 1.4 m diameter x 9.3 m long

MAJOR APPENDAGES: ~ 3 m beam extending from base of cylindrical section

ATTITUDE CONTROL: inactive at time of event

ENERGY SOURCES: inactive nuclear power source; other sources speculative

EVENT DATA

KNOWN EVENTS: 1

FIRST DATE: 04 Jul 2008

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|------------|-------------|
| 803 | 774 | 100.62 min | 65.0 deg |

COMMENTS

Cosmos 1818 was the first of two Plazma-A spacecraft, the second being Cosmos 1867. These spacecraft used a thermionic nuclear power reactor in lieu of the thermoelectric reactors aboard the predecessor Radar Ocean Reconnaissance (RORSAT) spacecraft series. However, like RORSATs, the Plazma-A spacecraft used sodium-potassium (NaK) as a coolant. Cosmos 1818's operational lifetime is estimated to be approximately five months. The event occurred after approximately 21.4 years on orbit. Radar observations indicate signatures consistent with small, metallic spheres, most likely NaK coolant droplets.

REFERENCE DOCUMENT

"New Debris Seen from Decommissioned Satellite with Nuclear Power Source", The Orbital Debris Quarterly News, NASA JSC, January 2009. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv13i1.pdf>.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 20.20 Feb 1987
DRY MASS (KG): 1500
MAIN BODY: Cylinder; 2.4 m diameter by 4 m length
MAJOR APPENDAGES: Gravity-gradient boom; 10 small solar panels
ATTITUDE CONTROL: Gravity gradient

EVENT DATA

KNOWN EVENTS: 3
FIRST DATE: Apr-May 1997

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------|---------|-----------|-------------|
| 1525 km | 1480 km | 116.0 min | 73.6 deg |

COMMENTS

Cosmos 1823 appears to have experienced three separate anomalous events, two in 1997 and one in 1999. Because Cosmos 1823 suffered a serious fragmentation in December 1987, the anomalous debris pieces may have been loosely attached to the spacecraft, then separated after continued exposure to the space environment or change in attitude of the spacecraft remnant.

REFERENCE DOCUMENTS

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

Environmentally-Induced Debris Sources, N. L. Johnson, NASA Lyndon B. Johnson Space Center, Second World Space Congress, 2002.

METEOR 2-17

1988-005A

18820

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 30 January 1988
DRY MASS (KG): 2750
MAIN BODY: Cylinder
MAJOR APPENDAGES: Large Solar Arrays
ATTITUDE CONTROL: None at the time of event

EVENT DATA

KNOWN EVENTS: 2
FIRST DATE: 2000-2001

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|---------|-------------|
| 960 | 936 | 104 min | 82.5 deg |

COMMENTS

One debris object separated from the parent spacecraft in '00-'01 time frame, the second in the '02-'03 time frame. The debris were tracked under various analyst satellite numbers until being cataloged in 2012 as SSN 38540 and 38543.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 10.65 July 1987
DRY MASS (KG): 2500
MAIN BODY: cylinder-cone 1.4 m diameter x 9.3 m long
MAJOR APPENDAGES: ~ 3 m beam extending from base of cylindrical section
ATTITUDE CONTROL: inactive at time of event
ENERGY SOURCES: inactive nuclear power source; other sources speculative

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: 21 March to 4 April 2014

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|------------|-------------|
| 802 | 776 | 100.63 min | 65.0 deg |

COMMENTS

Cosmos 1867 was the second of two Plazma-A spacecraft, the first being Cosmos 1818. These spacecraft used a thermionic nuclear power reactor in lieu of the thermoelectric reactors aboard the predecessor Radar Ocean Reconnaissance (RORSAT) spacecraft series. However, like RORSATs, the Plazma-A spacecraft used sodium-potassium (NaK) as a coolant. Cosmos 1867's operational lifetime is estimated to be approximately five months. The event occurred after approximately 26.9 years on orbit and was an extended event. Radar observations indicate signatures consistent with small, metallic spheres, most likely NaK coolant droplets.

COSMOS 1939 R/B

1988-032B

19046

SATELLITE DATA

TYPE: Vostok Final Stage
OWNER: CIS
LAUNCH DATE: 20 April 1988
DRY MASS (KG): 1440
MAIN BODY: Cylinder; 2.6 m diameter by 3.8 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at the time of the event.

EVENT DATA

KNOWN EVENTS: 2
FIRST DATE: 30 July 1996

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------------|----------------|---------------|--------------------|
| 655 km | 585 km | 97.14 min | 97.6 deg |

COMMENTS

One of several Vostok final stages involved in anomalous events.

NOAA 11

1988-089A

19531

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 24.42 September 1988
DRY MASS (KG): 740
MAIN BODY: Rectangular; 3.71 m by 1.88 m
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: None at the time of event

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: 24 November 2010

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------------|----------------|---------------|--------------------|
| 816 km | 799 km | 101.0 min | 98.6 deg |

COMMENTS

Two low area-to-mass ratio objects have been cataloged. Unless other evidence is uncovered, this event will be classified as an anomalous event. The NOAA 11 payload employed the Advanced TIROS-N bus and was decommissioned on 16 June 2004. This event occurred approximately 6.4 years after decommissioning.

COBE

1989-089A

20322

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 18.61 November 1989
DRY MASS (KG): 2265
MAIN BODY: Cylinder; with protective shield, 4.0 m diameter by 5.8 m length
MAJOR APPENDAGES: 3 - 8.6 m solar arrays
ATTITUDE CONTROL: Spin stabilized; gyroscopes

EVENT DATA

KNOWN EVENTS: Multiple
FIRST DATE: January 1993

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------------|----------------|---------------|--------------------|
| 885 km | 870 km | 102.5 min | 99.0 deg |

COMMENTS

At least 12 separate event dates have been calculated by the NAVSPOC, and other events are certain to have occurred. Through December 1993 the satellite remained active, and the cause of the separations could be determined. No degradation of satellite performance was reported by the satellite operators.

REFERENCE DOCUMENT

Environmentally-Induced Debris Sources, N. L. Johnson, NASA Lyndon B. Johnson Space Center, Second World Space Congress, 2002.

SATELLITE DATA

TYPE: Rocket Body
OWNER: CIS
LAUNCH DATE: 27 February 1990
DRY MASS (KG): 1434
MAIN BODY: Cylinder; 2.4 m diameter x 6.0 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at the time of the event

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: 22 June 2005

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|------------|-------------|
| 1015 | 950 | 104.65 min | 83.0 deg |

COMMENTS

A piece was detected in a more eccentric and higher inclination orbit, indicating that this event may have been a collision with a small, uncataloged object or meteoroid.

REFERENCE DOCUMENT

“Recent Satellite Breakups”, The Orbital Debris Quarterly News, NASA JSC, July 2005.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNV9i3.pdf>.

HST

1990-037B

20580

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 24 April 1990
DRY MASS (KG): 10863
MAIN BODY: Cylinder
MAJOR APPENDAGES: Two Solar Array Panels
ATTITUDE CONTROL: CMG controlled

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: August 2003

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|----------|-------------|
| 575 km | 570 km | 96.1 min | 28.5 deg |

COMMENTS

The Hubble Space Telescope (HST) debris decayed rapidly after the event.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 14.66 May 1991
DRY MASS (KG): 740
MAIN BODY: Rectangular; 3.71 m by 1.88 m
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: None at the time of event

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: 2 October 2011

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|---------|-------------|
| 815 km | 800 km | 100 min | 98.7 deg |

COMMENTS

Four low area-to-mass ratio objects have been cataloged. Unless other evidence is uncovered, this event will be classified as an anomalous event. The NOAA 12 (NOAA D pre-launch) payload employed the last TIROS-N bus, No. 5, and was decommissioned on 10 August 2007. This event occurred approximately 4.1 years after decommissioning.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 4 June 1991
DRY MASS (KG): 1922
MAIN BODY: Cylinder; 1.4-0.8 m diameter by 3.5 m length
MAJOR APPENDAGES: Solar arrays, payload trays, radar antenna
ATTITUDE CONTROL: Gravity-gradient

EVENT DATA

KNOWN EVENTS: 1
DATE: 12 October 1998

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|----------|-------------|
| 665 km | 620 km | 97.5 min | 82.5 deg |

COMMENTS

First event for this type object. No other events observed.

REFERENCE DOCUMENT

Environmentally-Induced Debris Sources, N. L. Johnson, NASA Lyndon B. Johnson Space Center, Second World Space Congress, 2002.

SATELLITE DATA

TYPE: Payload

OWNER: ESA

LAUNCH DATE: 17 July 1991

DRY MASS (KG): 2084

MAIN BODY: SPOT Mk. 1 bus. Cubical box. Approximately 2 m x 2 m x 3 m long payload support module; 11.8 m length overall when antennas and solar panel deployed.

MAJOR APPENDAGES: Solar Array Panel, multiple instruments and antennas

ATTITUDE CONTROL: CMG/monopropellant thrusters; inactive at time of event

EVENT DATA

KNOWN EVENTS: 7

FIRST DATE: 7 July 2002

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------------|----------------|---------------|--------------------|
| 801.6 km | 750.8 km | 100.3 min | 98.5 deg |

COMMENTS

Seven debris objects were cataloged from seven unique production events, the most recent being in 2017. The ERS-1 mission ended on 10 March 2000 due to the failure of the attitude control system, approximately 2.3 years before the initial release event.

SARA

1991-050E

21578

SATELLITE DATA

TYPE: Payload
OWNER: France
LAUNCH DATE: 17 July 1991
DRY MASS (KG): 26
MAIN BODY: Cube; 360 mm per side
MAJOR APPENDAGES: Several deployable 5 m long antennae
ATTITUDE CONTROL: None at the time of events

EVENT DATA

KNOWN EVENTS: 3
FIRST DATE: August 2003

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------------|----------------|---------------|--------------------|
| 730 km | 730 km | 99.4 min | 98.1 deg |

COMMENTS

This French “microsat” was no longer active at the time of the events. Some objects may be a piece broken off from any of the long antennae. Follow on events occurred on 17 April 2005 and 15 October 2006. Four debris total have been cataloged from the parent object.

SATELLITE DATA

TYPE: Ariane 40 Rocket Body
OWNER: France
LAUNCH DATE: 17 July 1991
DRY MASS (KG): 1720
MAIN BODY: Cylinder; 2.6 m diameter by 10 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: April 2001

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|-----------|-------------|
| 770 km | 770 km | 100.2 min | 98.2 deg |

COMMENTS

One piece of debris cataloged. Parent object was in a sun-synchronous orbit at the time of the event. First occurrence of an anomalous event with an Ariane R/B.

REFERENCE DOCUMENT

Environmentally-Induced Debris Sources, N. L. Johnson, NASA Lyndon B. Johnson Space Center, Second World Space Congress, 2002.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 12 September 1991
DRY MASS (KG): 10863
MAIN BODY: Box; approximately 4.6 m x 4.6 m x 10.7 m long
MAJOR APPENDAGES: Solar Array Panel, multiple instruments and antennas
ATTITUDE CONTROL: CMG/torque rod controlled; inactive at time of event

EVENT DATA

KNOWN EVENTS: 3
FIRST DATE: 10 November 2007

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|----------|-------------|
| 483 km | 353 km | 92.9 min | 57 deg |

COMMENTS

Five debris objects were cataloged and all debris decayed rapidly after their production events. UARS was decommissioned on 14 December 2005, approximately 1.9 years before this series of events.

EKA 1 (START 1)

1993-014A

22561

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 25 March 1993
DRY MASS (KG): 260
MAIN BODY: Two cylinders; < 1.5 m diameter
MAJOR APPENDAGES: Solar panels; gravity-gradient boom
ATTITUDE CONTROL: Gravity-gradient

EVENT DATA

KNOWN EVENTS: Multiple
FIRST DATE: 4 March 1998

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------------|----------------|---------------|--------------------|
| 970 km | 685 km | 101.43 min | 75.8 deg |

COMMENTS

EKA 1 is a test payload prior to launches of small communications satellites. First orbital launch of Start-1 booster derived from SS-20/SS-25 missiles.

START 1 R/B

1993-014B

22562

SATELLITE DATA

TYPE: Rocket Body
OWNER: CIS
LAUNCH DATE: 25 March 1993
DRY MASS (KG): 200
MAIN BODY: Cylinder; 1.4 m diameter x 2.5 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at the time of the event

EVENT DATA

KNOWN EVENTS: Multiple
FIRST DATE: Late 2002

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------------|----------------|---------------|--------------------|
| 920 km | 680 km | 100.85 min | 75.8 deg |

COMMENTS

Dozens of pieces have been cataloged from this rocket body, starting in late 2002 through 2005. Exact time and date of the event(s) is unknown. The payload associated with this rocket body, START-1, also experienced an anomalous event in March of 1998, the events are unrelated.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 29 Aug. 1994
DRY MASS (KG): 767
MAIN BODY: TIROS N bus. Cylindrical; approximately 1.88 m diameter x 3.71 m long
MAJOR APPENDAGES: Solar Array Panel, multiple instruments and antennas
ATTITUDE CONTROL: reaction wheels/monopropellant thrusters; inactive at time of event

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: 23 Oct. 2016

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|-----------|-------------|
| 848 km | 832 km | 101.7 min | 99.1 deg |

COMMENTS

Four objects were cataloged from a single production event, prior cataloged debris being mission-related. Two line elements are not available for the payload after 2013 but are for the four anomalous debris; payload elements are taken from the 11 October 2016 public satellite catalog. This event may be similar in nature to the breakup events suffered by USA 29 (DMSP 5D-2 F9, SSN 18822), USA 73 (DMSP 5D-2 F11, SSN 21798), and USA 109 (DMSP 5D-2 F13, SSN 23533).

SATELLITE DATA

TYPE: Zenit Second Stage
OWNER: CIS
LAUNCH DATE: 24 November 1994
DRY MASS (KG): 8300
MAIN BODY: Cylinder; 3.9 m diameter by 12 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None

EVENT DATA

KNOWN EVENTS: 2?
FIRST DATE: June 1998

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|------------|-------------|
| 845 km | 845 km | 101.82 min | 71.0 deg |

COMMENTS

One piece of debris was cataloged.

REFERENCE DOCUMENTS

History of Soviet/Russian Satellite Fragmentations-A Joint U.S.-Russian Investigation, N. L. Johnson et al, Kaman Sciences Corporation, October 1995.

Environmentally-Induced Debris Sources, N. L. Johnson, NASA Lyndon B. Johnson Space Center, Second World Space Congress, 2002.

NOAA 14

1994-089A

23455

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 30 December 1994
DRY MASS (KG): 740
MAIN BODY: Rectangular; 3.71 m by 1.88 m
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: None at the time of event

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: July 2012

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------------|----------------|---------------|--------------------|
| 860 km | 850 km | 101 min | 98.8 deg |

COMMENTS

A single low area-to-mass ratio object has been cataloged. Unless other evidence is uncovered, this event will be classified as an anomalous event. The NOAA 14 payload employed the Advanced TIROS-N bus and was decommissioned on 23 May 2007.

ERS-2

1995-021A

23560

SATELLITE DATA

TYPE: Payload

OWNER: ESA

LAUNCH DATE: 21 Apr. 1995

DRY MASS (KG): 2216

MAIN BODY: SPOT Mk. 1 bus. Cubical box. Approximately 2 m x 2 m x 3 m long payload support module; 11.8 m length overall when antennas and solar panel deployed.

MAJOR APPENDAGES: Solar Array Panel, multiple instruments and antennas

ATTITUDE CONTROL: CMG/monopropellant thrusters; inactive at time of event

EVENT DATA

KNOWN EVENTS: 4

FIRST DATE: 24 July 2003

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------------|----------------|---------------|--------------------|
| 786.6 km | 784.8 km | 100.5 min | 98.6 deg |

COMMENTS

Six debris objects were cataloged from at least four production events, the most recent being in 2011. The ERS-2 spacecraft was decommissioned on 5 Sept. 2011 when it was passivated in accordance with debris mitigation guidelines.

SATELLITE DATA

TYPE: Ariane 40+ Rocket Body
OWNER: France
LAUNCH DATE: 21 Apr. 1995
DRY MASS (KG): 1720
MAIN BODY: Cylinder; 2.6 m diameter by 10 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: prior to 1 Jan. 2018

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|----------|----------|-----------|-------------|
| 772.4 km | 761.6 km | 100.1 min | 98.8 deg |

COMMENTS

One piece of debris (SSN 43351) cataloged. Parent object was in a sun-synchronous orbit at the time of the event. "Event data" epoch is 1 Jan. 2018 but actual separation time was earlier.

KOREASAT 1 R/B

1995-041B

23640

SATELLITE DATA

TYPE: Delta Second Stage
OWNER: US
LAUNCH DATE: 5 August 1995
DRY MASS (KG): 900
MAIN BODY: Cylinder; 2.4 m diameter by 8 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of the event.

EVENT DATA

KNOWN EVENTS: 1
DATE: 6 December 1995

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------|---------|-----------|-------------|
| 1375 km | 935 km | 108.5 min | 26.7 deg |

COMMENTS

One piece was liberated.

RADARSAT R/B

1995-059B

23711

SATELLITE DATA

TYPE: Delta Second Stage
OWNER: US
LAUNCH DATE: 5 November 1995
DRY MASS (KG): 900
MAIN BODY: Cylinder; 2.4 m diameter by 8 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None

EVENT DATA

KNOWN EVENTS: 1
DATE: 30 January 1996

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------|---------|-----------|-------------|
| 1495 km | 935 km | 109.7 min | 100.6 deg |

COMMENTS

One piece was liberated.

SATELLITE DATA

TYPE: Payload
OWNER: India
LAUNCH DATE: 21 March 1996
DRY MASS (KG): 838
MAIN BODY: Cubical box; 1.6 m x 1.6 m by 1.2 m high
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: three-axis stabilized; reaction wheels, torque rods, and monopropellant reaction control system

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: October 2000

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|----------|----------|-----------|-------------|
| 821.9 km | 820.3 km | 101.3 min | 98.6 deg |

COMMENTS

A single relatively high area-to-mass ratio object has been cataloged. "Event Data" epoch is 30 October 2000. Unless other evidence is uncovered, this event will be classified as an anomalous event. The payload was operational at the time of separation and was decommissioned in January 2006.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 20 December 1997
DRY MASS (KG): 556
MAIN BODY: Triangular prism, 3.6 m long
MAJOR APPENDAGES: Two solar arrays, three antennas
ATTITUDE CONTROL: Three-axis stabilized

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: 7 June 2014

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|-----------|-------------|
| 781 km | 778 km | 100.4 min | 86.4 deg |

COMMENTS

Ten pieces of debris were cataloged from this event. Three of the cataloged debris had reentered within 10 months indicating relatively large area-to-mass ratios. The event did not affect vehicle operations or performance.

REFERENCE DOCUMENT

“Iridium Anomalous Debris Events”, The Orbital Debris Quarterly News, NASA JSC, January 2015.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/odqnv19i1.pdf>.

FUSE

1999-035A

25791

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 24 June 1999
DRY MASS (KG): 1360
MAIN BODY: Box; 1.3 m by 0.9 m by 0.9 m
MAJOR APPENDAGES: Two 3.5 m² solar Arrays
ATTITUDE CONTROL: Three-axis stabilized

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: 6 June 2004

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------------|----------------|---------------|--------------------|
| 760 km | 745 km | 99.90 min | 25.0 deg |

COMMENTS

Eight pieces of debris were cataloged from this event. An additional piece was detected but never cataloged. The event might have been coincidental with a “safe mode” entry around 5 June 2004, which cause the closing and reopening of several sensor doors. Five of the cataloged debris had reentered within 8 months indicating higher than normal area-to-mass ratios. The event did not affect vehicle operations or performance.

REFERENCE DOCUMENT

“FUSE Satellite Releases Unexpected Debris”, The Orbital Debris Quarterly News, NASA JSC, July 2004. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/ODQNv8i3.pdf>.

SATELLITE DATA

TYPE: Payload
OWNER: Indonesia
LAUNCH DATE: 12 Aug. 1999
DRY MASS (KG): 1640
MAIN BODY: Lockheed-Martin A2100A bus. Cubical box; 1.8 m x 1.8 m by 3.7 m high
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: three-axis stabilized; reaction wheels and monopropellant reaction control system

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: 25 Aug. 2017

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|----------|----------|------------|-------------|
| 35793 km | 35781 km | 1436.1 min | 0.01 deg |

COMMENTS

This event occurred after approximately 18.1 years on-orbit, three years past the nominal operational lifetime. The payload was operational at the time of the event. No debris have been cataloged as of this edition.

REFERENCE DOCUMENT

“Two Anomalous Events in GEO”, The Orbital Debris Quarterly News, NASA JSC, February 2018.
Available online at <https://www.orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/odqnv22i1.pdf>.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 24.76 September 1999
DRY MASS (KG): 735
MAIN BODY: Box; 1.8 m by 1.8 m by 1.6 m
MAJOR APPENDAGES: 3 solar panels
ATTITUDE CONTROL: 3 axis stabilization

EVENT DATA

KNOWN EVENTS: 1
DATE: 19 March 2001

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|----------|-------------|
| 680 km | 678 km | 98.3 min | 98.2 deg |

COMMENTS

One piece was liberated. A very high ballistic coefficient resulted in the anomalous debris object reentering on 11 April 2001.

SATELLITE DATA

TYPE: Payload

OWNER: US-France joint mission

LAUNCH DATE: 7 Dec. 2001

DRY MASS (KG): 472

MAIN BODY: box; 0.95 m x 0.95 m x 2.2 m high

MAJOR APPENDAGES: Solar panels

ATTITUDE CONTROL: three-axis stabilized; reaction wheels and monopropellant reaction control system

EVENT DATA

KNOWN EVENTS: 1

FIRST DATE: 16 Mar. 2002

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|-----------|-----------|-----------|-------------|
| 1344.7 km | 1332.2 km | 112.4 min | 66.0 deg |

COMMENTS

On 16 March 2002 spacecraft controllers noted an attitude upset of the payload. A detailed study of the spacecraft's perturbations allowed analysts to infer that the left solar array had been impacted by a small particle (micrometeoroid or orbital debris unknown). Two anomalous debris were observed and associated with this payload, although the objects did not enter the public catalog until 2009 (SSN 35414) and 2011 (SSN 37379). A second anomaly was experienced in 2005 but there is no evidence that this event liberated debris.

REFERENCE DOCUMENTS

"New Evidence of Particle Impact on Jason-1 Spacecraft", The Orbital Debris Quarterly News, NASA JSC, July 2011. Available online at <https://www.orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/odqnv15i3.pdf>.

SATELLITE DATA

TYPE: Payload
OWNER: US
LAUNCH DATE: 11 February 2002
DRY MASS (KG): 556
MAIN BODY: Triangular prism, 3.6 m long
MAJOR APPENDAGES: Two solar arrays, three antennas
ATTITUDE CONTROL: Three-axis stabilized

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: 30 November 2014

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|-----------|-------------|
| 781 km | 777 km | 100.4 min | 86.4 deg |

COMMENTS

Four pieces of debris were cataloged from this event. Two of the cataloged debris had reentered within a year indicating relatively large area-to-mass ratios. The event did not affect vehicle operations or performance.

REFERENCE DOCUMENT

“Iridium Anomalous Debris Events”, The Orbital Debris Quarterly News, NASA JSC, January 2015.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/odqnv19i1.pdf>.

SATELLITE DATA

TYPE: Payload

OWNER: SES (formerly Société Européenne des Satellites; Luxembourg)

LAUNCH DATE: 6 June 2003

DRY MASS (KG): 1663

MAIN BODY: Thales Alenia Space Spacebus-3000B3 bus. Cubical box; 3.2 m x 2.4 m by 4 m high

MAJOR APPENDAGES: Solar panels

ATTITUDE CONTROL: three-axis stabilized; reaction wheels and bipropellant reaction control system

EVENT DATA

KNOWN EVENTS: 1

FIRST DATE: 17 June 2017

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|----------|----------|------------|-------------|
| 35798 km | 35774 km | 1436.1 min | 0.02 deg |

COMMENTS

This event occurred after approximately 14 years on-orbit. The payload was operational at the time of the event and was subsequently boosted to the so-called GEO “graveyard orbit.” No debris have been cataloged as of this edition.

REFERENCE DOCUMENT

“Two Anomalous Events in GEO”, The Orbital Debris Quarterly News, NASA JSC, February 2018.
Available online at <https://www.orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/odqnv22i1.pdf>.

SATELLITE DATA

TYPE: Payload
OWNER: CIS
LAUNCH DATE: 17 September 2009
DRY MASS (KG): 7.53
MAIN BODY: Nested spheres; 0.17 m diameter
MAJOR APPENDAGES: none
ATTITUDE CONTROL: none

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: 22 January 2013

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|-----------|-------------|
| 825 km | 818 km | 101.3 min | 98.6 deg |

COMMENTS

A single piece of debris was produced by this event. There remains uncertainty about the exact nature of this event so it is characterized as an anomalous event at this time.

REFERENCE DOCUMENT

“Small Satellite Possibly Hit by Even Smaller Object”, The Orbital Debris Quarterly News, NASA JSC, April 2013. Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/odqnv17i2.pdf>.

SATELLITE DATA

TYPE: Payload
OWNER: DigitalGlobe (US)
LAUNCH DATE: 8 Oct. 2009
DRY MASS (KG): 2385
MAIN BODY: Cylinder; 2.5 m diameter x 4.3 m high
MAJOR APPENDAGES: Solar panels
ATTITUDE CONTROL: three-axis stabilized; CMGs and monopropellant reaction control system

EVENT DATA

KNOWN EVENTS: 1
FIRST DATE: 18 July 2016

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|----------|----------|-----------|-------------|
| 767.9 km | 767.0 km | 100.2 min | 98.5 deg |

COMMENTS

Nine anomalous debris were initially observed, but sixteen have been cataloged as of this edition. An initial analysis indicated a separation velocity of approximately 3 m/s for the longest-period debris object, and debris were observed with a maximum change, with respect to the parent body, in period of 0.8 minutes and inclination 0.02 deg. Debris were found at semimajor axes both larger and smaller than the parent body, indicating this event was not a simple shedding event. The spacecraft remained operational after the event.

REFERENCE DOCUMENTS

“WorldView 2 Spacecraft Fragments in July 2016”, The Orbital Debris Quarterly News, NASA JSC, October 2016. Available online at <https://www.orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/odqnv20i4.pdf>.

SATELLITE DATA

TYPE: Payload
OWNER: PRC
LAUNCH DATE: 15 August 2011
DRY MASS (KG): 1500?
MAIN BODY: Box, 3 m x 2 m x 2 m?
MAJOR APPENDAGES: one solar array
ATTITUDE CONTROL: Unknown at time of event

EVENT DATA

KNOWN EVENTS: 2
FIRST DATE: 6-7 July 2014

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|--------|---------|-----------|-------------|
| 969 km | 967 km | 100.4 min | 99.4 deg |

COMMENTS

Four pieces of debris were produced from these events, of which three entered the catalog. All three of the cataloged debris had reentered by 15 December 2014 indicating relatively large area-to-mass ratios. This spacecraft uses the CAST 968 bus; however, actual physical parameters are highly uncertain.

REFERENCE DOCUMENT

“Three Additional Breakups Mar 2014”, The Orbital Debris Quarterly News, NASA JSC, October 2014.
Available online at <https://orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/odqnv18i4.pdf>.

SATELLITE DATA

TYPE: Payload

OWNER: India

LAUNCH DATE: 26 Apr. 2012

DRY MASS (KG): 1858 at launch (wet mass)

MAIN BODY: Cubical box; 2 m x 2 m by 5 m high

MAJOR APPENDAGES: Solar panels

ATTITUDE CONTROL: three-axis stabilized; reaction wheels, gyros, and monopropellant reaction control system

EVENT DATA

KNOWN EVENTS: 1

FIRST DATE: 30 Sept. 2016

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|----------|---------|----------|-------------|
| 543.7 km | 538 km | 95.4 min | 97.6 deg |

COMMENTS

Over 12 anomalous debris were initially observed, but only one (SSN 41797) has been cataloged as of this edition. All known debris have decayed.

REFERENCE DOCUMENTS

“Indian RISAT-1 Spacecraft Experiences Possible Fragmentation”, The Orbital Debris Quarterly News, NASA JSC, October 2016. Available online at <https://www.orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/odqnv20i4.pdf>.

“Indian RISAT-1 Spacecraft Fragments in Late September - Update”, The Orbital Debris Quarterly News, NASA JSC, February 2017. Available online at <https://www.orbitaldebris.jsc.nasa.gov/quarterly-news/pdfs/odqnv21i1.pdf>.

BRIZ-KM R/B

2015-020E

40556

SATELLITE DATA

TYPE: rocket body
OWNER: CIS
LAUNCH DATE: 31 Mar. 2015
DRY MASS (KG): 1220
MAIN BODY: Cylinder; 2.5 m diameter by 2.6 m length
MAJOR APPENDAGES: None
ATTITUDE CONTROL: None at time of event

EVENT DATA

KNOWN EVENTS: 1
DATE: 29 Apr. 2015

| APOGEE | PERIGEE | PERIOD | INCLINATION |
|---------------|----------------|---------------|--------------------|
| 1342.5 km | 1339.2 km | 112.5 min | 82.5 deg |

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4.0 OTHER SATELLITES ASSOCIATED WITH FRAGMENTATIONS

4.1 *Aerodynamic Associations with Fragmentation Events*

Aerodynamic breakups are associated with the breakup of a resident space object caused by interacting with Earth's atmosphere. Forty such events have occurred between 1994 and July 2018. Because of the orbit elements of the parent object at the time of fragmentation, only seven of these events showed any cataloged debris and all parent objects reentered within 1 year of the event (most reentered within a few days). It is understood that only a fraction of these fragmentations can be detected, because of the short remaining life of the parent and debris created. These events have no impact to the mid- or long-term debris environment and therefore, it was deemed more appropriate to separate these from the fragmentations in Chapter 2. The parent object for these aerodynamic events shall not be considered "fragmentation debris" when discussing object categorization. As mentioned, seven of these events produced cataloged debris other than the parent, and these debris objects represent the difference between the decayed fragmentation debris count in Table 1.3.2 and the decayed fragmentation debris count in Tables 2.1 and 2.2.

The following missions, listed by international designator in Table 4.1-1 and by event date in 4.1-2, have been determined to be solely related to aerodynamic effects at the time of reentry and therefore, did not contribute to the orbital environment. Note that the tables strictly interpret cataloging; in the case of there being no cataloged debris there were one to multiple objects observed and decayed prior to entry into the public satellite catalog.

TABLE 4.1-1 HISTORY OF SATELLITE AERODYNAMIC EVENTS BY LAUNCH DATE

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | BREAKUP DATE | REENTRY DATE | DEBRIS CATALOGED | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) | ADDITIONAL INFORMATION |
|------------------|--------------------------|---------------------|-------------|--------------|--------------|------------------|-------------|--------------|-------------------|------------------------|
| ELEKTRON 1/2 R/B | 1964-006D | 751 | 30-Jan-64 | 13-Feb-98 | 15-Feb-98 | 1 | 56315 | 90 | 56.2 | VOSTOK FINAL STAGE |
| COSMOS 41 | 1964-049E | 898 | 22-Aug-64 | Apr-04 | 7-May-04 | 0 | ~35750 | ~115 | 64.5 | |
| COSMOS 862 DEB | 1976-105F | 9889 | 22-Oct-76 | 29-Mar-14 | 29-May-14 | 2 | 14990 | 110 | 62.0 | |
| COSMOS 1030 | 1978-083A | 11015 | 6-Sep-78 | 14-Aug-04 | 17-Aug-04 | 1 | ~4560 | ~95 | 61.9 | |
| COSMOS 1172 | 1980-028A | 11758 | 12-Apr-80 | 23-Dec-97 | 26-Dec-97 | 1 | 5125 | 75 | 61.8 | |
| COSMOS 1188 | 1980-050A | 11844 | 14-Jun-80 | 23-May-13 | 24-May-13 | 0 | 1745 | 100 | 62.2 | |
| MOLNIYA 3-16 | 1981-054A | 12512 | 9-Jun-81 | 5-Feb-98 | 10-Feb-98 | 0 | 7670 | 85 | 62.1 | |
| MOLNIYA 3-16 R/B | 1981-054E | 12519 | 9-Jun-81 | 28-Jul-98 | 30-Apr-99 | 0 | 33415 | 85 | 62.0 | MOLNIYA FINAL STAGE |
| COSMOS 41 DEB | 1964-049F | 13091 | 22-Aug-64 | 30-Dec-02 | 31-Dec-02 | 0 | 1200 | 85 | 64.4 | |
| MOLNIYA 3-19 | 1982-083A | 13432 | 27-Aug-82 | 13-Jan-02 | 13-Jan-02 | 0 | 2075 | 95 | 62.3 | |
| COSMOS 1658 | 1985-045A | 15808 | 11-Jun-85 | 12-Nov-05 | 12-Nov-05 | 0 | 1730 | 80 | 62.1 | |
| MOLNIYA 3-26 | 1985-091A | 16112 | 3-Oct-85 | 21-Feb-01 | 22-Feb-01 | 0 | 5690 | 80 | 62.6 | |
| MOLNIYA 1-66 R/B | 1985-103D | 16223 | 28-Oct-85 | 13-Jan-03 | 13-Jan-03 | 0 | ~1600 | ~120 | 62.4 | MOLNIYA FINAL STAGE |
| COSMOS 1701 | 1985-105A | 16235 | 9-Nov-85 | 29-Apr-01 | 11-May-01 | 0 | 25570 | 85 | 62.9 | |
| COSMOS 1849 | 1987-048A | 18083 | 4-Jun-87 | 27-Jan-03 | 4-Feb-03 | 0 | 7450 | 95 | 62.1 | |
| COSMOS 1966 | 1988-076A | 19445 | 30-Aug-88 | ~02-Nov-05 | 10-Nov-05 | 0 | 11535 | 90 | 62.9 | |
| MOLNIYA 3-35 | 1989-043A | 20052 | 8-Jun-89 | 14-Dec-01 | 14-Dec-01 | 0 | 593 | 65 | 61.9 | |
| MOLNIYA 3-36 | 1989-094A | 20338 | 28-Nov-89 | 19-May-00 | 20-May-00 | 0 | 1795 | 80 | 63.4 | |
| MOLNIYA 3-36 R/B | 1989-094B | 20339 | 28-Nov-89 | 28-Jun-00 | 4-Jul-00 | 0 | 7145 | 75 | 63.6 | MOLNIYA FINAL STAGE |
| MOLNIYA 1-77 | 1990-039A | 20583 | 26-Apr-90 | 24-Feb-05 | 25-Feb-05 | 0 | 1710 | 75 | 62.0 | |
| MOLNIYA 3-38 R/B | 1990-052D | 20649 | 13-Jun-90 | ~Sep-06 | 13-Sep-06 | 0 | 37710 | 130 | 62.4 | MOLNIYA FINAL STAGE |
| COSMOS 2105 | 1990-099A | 20941 | 20-Nov-90 | 16-Jan-08 | 21-Jan-08 | 0 | 2470 | 65 | 62.6 | |
| MOLNIYA 1-82 | 1991-053A | 21630 | 1-Aug-91 | 8-Oct-04 | 9-Oct-04 | 0 | 1510 | 75 | 61.7 | |
| COSMOS 2176 | 1992-003A | 21847 | 24-Jan-92 | 16-Jan-12 | 17-Jan-12 | 0 | 2555 | 75 | 62.1 | |
| MOLNIYA 1-83 R/B | 1992-011D | 21900 | 4-Mar-92 | 26-Sep-06 | 26-Sep-06 | 0 | 1090 | 70 | 62.0 | MOLNIYA FINAL STAGE |
| MOLNIYA 1-84 | 1992-050A | 22068 | 6-Aug-92 | 3-Apr-08 | 4-Apr-08 | 0 | 2600 | 80 | 61.5 | |
| MOLNIYA 3-44 | 1993-025A | 22633 | 21-Apr-93 | 25-Jan-04 | 25-Jan-04 | 0 | ~1000 | ~90 | 63.4 | |
| MOLNIYA 3-46 R/B | 1994-051D | 23214 | 23-Aug-94 | 17-Feb-08 | 19-Feb-08 | 0 | 5530 | 115 | 62.3 | MOLNIYA FINAL STAGE |
| ETS-VI R/B | 1994-056B | 23231 | 28-Aug-94 | 31-Mar-95 | 2-Apr-95 | 0 | 4840 | 100 | 28.6 | H-II SECOND STAGE |
| MOLNIYA 3-48 | 1996-060A | 24640 | 24-Oct-96 | 13-Oct-07 | 18-Oct-07 | 0 | 7825 | 100 | 63.4 | |
| MOLNIYA 3-52 | 2001-050A | 26970 | 25-Oct-01 | 5-Dec-11 | 6-Dec-11 | 0 | 2745 | 85 | 63.9 | |
| HELLAS SAT-2 R/B | 2003-020B | 27812 | 13-May-03 | 11-Dec-04 | 12-Dec-04 | 0 | 10300 | 90 | 17.5 | ATLAS V |
| MOLNIYA 1-93 | 2004-005A | 28163 | 18-Feb-04 | 15-Apr-16 | 16-Apr-16 | 0 | 2415 | 77 | 62.9 | |

TABLE 4.1-1 HISTORY OF SATELLITE AERODYNAMIC EVENTS BY LAUNCH DATE

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | BREAKUP DATE | REENTRY DATE | DEBRIS CATALOGED | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) | ADDITIONAL INFORMATION |
|---------------------|-----------------------------|------------------------|----------------|-----------------|-----------------|---------------------|----------------|-----------------|----------------------|---------------------------|
| USA 195 R/B | 2007-046B | 32259 | 11-Oct-07 | 21-Mar-08 | 21-Mar-08 | 0 | 59015 | 95 | 19.2 | ATLAS V |
| CHANG'E R/B | 2007-051B | 32274 | 24-Oct-07 | 27-Jan-08 | 28-Jan-08 | 0 | 6035 | 80 | 30.7 | CZ-3A FINAL STAGE |
| CTDRS R/B | 2008-019B | 32780 | 25-Apr-08 | 5-Mar-11 | 5-Mar-11 | 1 | 1065 | 110 | 17.9 | CZ-3C FINAL STAGE |
| BRIZ-M TANK | 2009-042C | 35698 | 11-Aug-09 | 21-Jun-10 | 22-Jun-10 | 88 | 1490 | 90 | 48.4 | Briz-M APT |
| USA 230 R/B | 2011-019B | 37482 | 7-May-11 | 17-Aug-11 | 17-Aug-11 | 1 | 2285 | 95 | 21.0 | ATLAS V |
| BEIDOU IGSO4 R/B | 2011-038B | 37764 | 26-Jul-11 | 2-Mar-12 | 3-Mar-12 | 0 | 1580 | 95 | 54.4 | CZ-3A FINAL STAGE |
| BREEZE-M DEB (TANK) | 2014-064C | 40279 | 21-Oct-14 | 17-Jun-15 | 18-Jun-15 | 0 | 4690 | 100 | 48.6 | Briz-M APT |
| TOTAL | | | | | | 95 | | | | |

TABLE 4.1-2 HISTORY OF SATELLITE AERODYNAMIC EVENTS BY EVENT DATE

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | BREAKUP DATE | REENTRY DATE | DEBRIS CATALOGED | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) | ADDITIONAL INFORMATION |
|------------------|--------------------------|---------------------|-------------|--------------|--------------|------------------|-------------|--------------|-------------------|------------------------|
| ETS-VI R/B | 1994-056B | 23231 | 28-Aug-94 | 31-Mar-95 | 2-Apr-95 | 0 | 4840 | 100 | 28.6 | H-II SECOND STAGE |
| COSMOS 1172 | 1980-028A | 11758 | 12-Apr-80 | 23-Dec-97 | 26-Dec-97 | 1 | 5125 | 75 | 61.8 | |
| MOLNIYA 3-16 | 1981-054A | 12512 | 9-Jun-81 | 5-Feb-98 | 10-Feb-98 | 0 | 7670 | 85 | 62.1 | |
| ELEKTRON 1/2 R/B | 1964-006D | 751 | 30-Jan-64 | 13-Feb-98 | 15-Feb-98 | 1 | 56315 | 90 | 56.2 | VOSTOK FINAL STAGE |
| MOLNIYA 3-16 R/B | 1981-054E | 12519 | 9-Jun-81 | 28-Jul-98 | 30-Apr-99 | 0 | 33415 | 85 | 62.0 | MOLNIYA FINAL STAGE |
| MOLNIYA 3-36 | 1989-094A | 20338 | 28-Nov-89 | 19-May-00 | 20-May-00 | 0 | 1795 | 80 | 63.4 | |
| MOLNIYA 3-36 R/B | 1989-094B | 20339 | 28-Nov-89 | 28-Jun-00 | 4-Jul-00 | 0 | 7145 | 75 | 63.6 | MOLNIYA FINAL STAGE |
| MOLNIYA 3-26 | 1985-091A | 16112 | 3-Oct-85 | 21-Feb-01 | 22-Feb-01 | 0 | 5690 | 80 | 62.6 | |
| COSMOS 1701 | 1985-105A | 16235 | 9-Nov-85 | 29-Apr-01 | 11-May-01 | 0 | 25570 | 85 | 62.9 | |
| MOLNIYA 3-35 | 1989-043A | 20052 | 8-Jun-89 | 14-Dec-01 | 14-Dec-01 | 0 | 593 | 65 | 61.9 | |
| MOLNIYA 3-19 | 1982-083A | 13432 | 27-Aug-82 | 13-Jan-02 | 13-Jan-02 | 0 | 2075 | 95 | 62.3 | |
| COSMOS 41 DEB | 1964-049F | 13091 | 22-Aug-64 | 30-Dec-02 | 31-Dec-02 | 0 | 1200 | 85 | 64.4 | |
| MOLNIYA 1-66 R/B | 1985-103D | 16223 | 28-Oct-85 | 13-Jan-03 | 13-Jan-03 | 0 | ~1600 | ~120 | 62.4 | MOLNIYA FINAL STAGE |
| COSMOS 1849 | 1987-048A | 18083 | 4-Jun-87 | 27-Jan-03 | 4-Feb-03 | 0 | 7450 | 95 | 62.1 | |
| MOLNIYA 3-44 | 1993-025A | 22633 | 21-Apr-93 | 25-Jan-04 | 25-Jan-04 | 0 | ~1000 | ~90 | 63.4 | |
| COSMOS 41 | 1964-049E | 898 | 22-Aug-64 | Apr-04 | 7-May-04 | 0 | ~35750 | ~115 | 64.5 | |
| COSMOS 1030 | 1978-083A | 11015 | 6-Sep-78 | 14-Aug-04 | 17-Aug-04 | 1 | ~4560 | ~95 | 61.9 | |
| MOLNIYA 1-82 | 1991-053A | 21630 | 1-Aug-91 | 8-Oct-04 | 9-Oct-04 | 0 | 1510 | 75 | 61.7 | |
| HELLAS SAT-2 R/B | 2003-020B | 27812 | 13-May-03 | 11-Dec-04 | 12-Dec-04 | 0 | 10300 | 90 | 17.5 | ATLAS V |
| MOLNIYA 1-77 | 1990-039A | 20583 | 26-Apr-90 | 24-Feb-05 | 25-Feb-05 | 0 | 1710 | 75 | 62.0 | |
| COSMOS 1966 | 1988-076A | 19445 | 30-Aug-88 | ~02-Nov-05 | 10-Nov-05 | 0 | 11535 | 90 | 62.9 | |
| COSMOS 1658 | 1985-045A | 15808 | 11-Jun-85 | 12-Nov-05 | 12-Nov-05 | 0 | 1730 | 80 | 62.1 | |
| MOLNIYA 3-38 R/B | 1990-052D | 20649 | 13-Jun-90 | ~Sep-06 | 13-Sep-06 | 0 | 37710 | 130 | 62.4 | MOLNIYA FINAL STAGE |
| MOLNIYA 1-83 R/B | 1992-011D | 21900 | 4-Mar-92 | 26-Sep-06 | 26-Sep-06 | 0 | 1090 | 70 | 62.0 | MOLNIYA FINAL STAGE |
| MOLNIYA 3-48 | 1996-060A | 24640 | 24-Oct-96 | 13-Oct-07 | 18-Oct-07 | 0 | 7825 | 100 | 63.4 | |
| COSMOS 2105 | 1990-099A | 20941 | 20-Nov-90 | 16-Jan-08 | 21-Jan-08 | 0 | 2470 | 65 | 62.6 | |
| CHANG'E R/B | 2007-051B | 32274 | 24-Oct-07 | 27-Jan-08 | 28-Jan-08 | 0 | 6035 | 80 | 30.7 | CZ-3A FINAL STAGE |
| MOLNIYA 3-46 R/B | 1994-051D | 23214 | 23-Aug-94 | 17-Feb-08 | 19-Feb-08 | 0 | 5530 | 115 | 62.3 | MOLNIYA FINAL STAGE |
| USA 195 R/B | 2007-046B | 32259 | 11-Oct-07 | 21-Mar-08 | 21-Mar-08 | 0 | 59015 | 95 | 19.2 | ATLAS V |
| MOLNIYA 1-84 | 1992-050A | 22068 | 6-Aug-92 | 3-Apr-08 | 4-Apr-08 | 0 | 2600 | 80 | 61.5 | |
| BRIZ-M TANK | 2009-042C | 35698 | 11-Aug-09 | 21-Jun-10 | 22-Jun-10 | 88 | 1490 | 90 | 48.4 | Briz-M APT |
| CTDRS R/B | 2008-019B | 32780 | 25-Apr-08 | 5-Mar-11 | 5-Mar-11 | 1 | 1065 | 110 | 17.9 | CZ-3C FINAL STAGE |
| USA 230 R/B | 2011-019B | 37482 | 7-May-11 | 17-Aug-11 | 17-Aug-11 | 1 | 2285 | 95 | 21.0 | ATLAS V |

TABLE 4.1-2 HISTORY OF SATELLITE AERODYNAMIC EVENTS BY EVENT DATE

| SATELLITE NAME | INTERNATIONAL DESIGNATOR | US SATELLITE NUMBER | LAUNCH DATE | BREAKUP DATE | REENTRY DATE | DEBRIS CATALOGED | APOGEE (KM) | PERIGEE (KM) | INCLINATION (DEG) | ADDITIONAL INFORMATION |
|---------------------|-----------------------------|------------------------|----------------|-----------------|-----------------|---------------------|----------------|-----------------|----------------------|---------------------------|
| MOLNIYA 3-52 | 2001-050A | 26970 | 25-Oct-01 | 5-Dec-11 | 6-Dec-11 | 0 | 2745 | 85 | 63.9 | |
| COSMOS 2176 | 1992-003A | 21847 | 24-Jan-92 | 16-Jan-12 | 17-Jan-12 | 0 | 2555 | 75 | 62.1 | |
| BEIDOU IGSO4 R/B | 2011-038B | 37764 | 26-Jul-11 | 2-Mar-12 | 3-Mar-12 | 0 | 1580 | 95 | 54.4 | CZ-3A FINAL STAGE |
| COSMOS 1188 | 1980-050A | 11844 | 14-Jun-80 | 23-May-13 | 24-May-13 | 0 | 1745 | 100 | 62.2 | |
| COSMOS 862 DEB | 1976-105F | 9889 | 22-Oct-76 | 29-Mar-14 | 29-May-14 | 2 | 14990 | 110 | 62.0 | |
| BREEZE-M DEB (TANK) | 2014-064C | 40279 | 21-Oct-14 | 17-Jun-15 | 18-Jun-15 | 0 | 4690 | 100 | 48.6 | Briz-M APT |
| MOLNIYA 1-93 | 2004-005A | 28163 | 18-Feb-04 | 15-Apr-16 | 16-Apr-16 | 0 | 2415 | 77 | 62.9 | |

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4.2 Reactor Coolant Debris

The Soviet Union conducted the Radar Ocean Reconnaissance satellite (RORSAT) program with developmental and operational flights spanning 1965 (Cosmos 102) to 1988 (Cosmos 1932). Starting with Cosmos 198 (1967) large operational debris objects were left in their operational orbit while the nuclear reactor power source was boosted to a higher storage (or “graveyard”) orbit. Beginning with Cosmos 1176 (1980) RORSATs routinely ejected their reactor core. This practice resulted in the venting of all or part of the reactor’s Sodium-Potassium (NaK) liquid metal coolant. This population is largely undetectable by the sensors of the Space Surveillance Network (SSN). However, some coolant of sufficient size to be detected by at least some sensors has been cataloged by the SSN. Table 4.2 lists the coolant objects cataloged as of 4 July 2018.

TABLE 4.2: CATALOGED RORSAT PROGRAM NAK COOLANT DEBRIS

| <i>international designator</i> | <i>common name</i> | <i>Total coolant cataloged</i> | <i>coolant on- orbit</i> |
|-------------------------------------|--------------------|--|----------------------------------|
| 1976-103 | COSMOS 860 | 3 | 3 |
| 1980-034 | COSMOS 1176 | 2 | 2 |
| 1981-021 | COSMOS 1249 | 17 | 17 |
| 1981-037 | COSMOS 1266 | 1 | 1 |
| 1982-099 | COSMOS 1412 | 1 | 1 |
| 1984-069 | COSMOS 1579 | 31 | 31 |
| 1987-011 | COSMOS 1818* | 29 | 29 |
| 1987-060 | COSMOS 1867* | 40 | 40 |

*not operational RORSAT program element; discussed in Section 3.

4.3 Spurious Associations with Fragmentation Events

Satellite fragmentation lists compiled by other organizations, in particular by the National Security Council and NAVSPASUR, were carefully reviewed during the preparation of the fourth edition of the History of On-Orbit Satellite Fragmentations. However, due to the frequent exchange of information within the small orbital debris and space operations community and the long period during which satellite fragmentation lists have been maintained, no current list is completely independent from all others.

These reviews also revealed the need to define better the terms "satellite breakup" and "anomalous event" as discussed in Section 1.0. Many "breakup" lists have historically included entries related to normal launch and mission activities which resulted in numbers of debris in excess of the handful usually observed on these occasions. Some researchers have been misled by tracking difficulties and cataloging procedures that may cause late cataloging or misidentification of debris, superficially giving the appearance of fragmentations. A higher than average number of debris alone is not sufficient to assume a satellite fragmentation. Such pitfalls can generally be avoided

by conducting analyses with complete satellite element set data rather than the limited orbital data available in the U.S. Satellite Catalog.

The following space missions, listed by international designator, have been examined in detail and have failed to qualify as either satellite breakup or anomalous event as set forth in Section 1.0. The source of debris associated with nearly all of these flights is of a mission-related nature. Bolded items indicate the alleged source of the debris when unique or appropriate. The Common Name prefix C indicates a Cosmos-series satellite.

TABLE 4.3: SPURIOUS ASSOCIATION WITH FRAGMENTATIONS BY LAUNCH DATE

| INT'L DES. | COMMON NAME | S/C | R/B | TOTAL DEBRIS | DEBRIS ON-ORBIT |
|------------|------------------|------------------|------------------------------|--------------|-----------------|
| 1963-014 | FTV 1169 | Payload | Agena B spacecraft | 147 | 48 |
| 1965-073 | C 86-90 | Payload | Cosmos 3 | 5 | 5 |
| 1965-112 | C 103 | Payload | Cosmos 3 | 13 | 0 |
| 1967-001 | INTELSAT 2-F2 | INTELSAT II | Delta 1 R/B (2): FW-4 | 17 | 1 |
| 1967-011 | Diademe 1 | Payload | Diamant | 13 | 0 |
| 1967-014 | Diademe 2 | Payload | Diamant | 12 | 3 |
| 1967-024 | C 149 | DS-MO | Cosmos 2 | 16 | 0 |
| 1967-086 | C 176 | DS-P1-Yu | Cosmos 2 | 9 | 0 |
| 1968-117 | C 261 | DS-U2-GK | Cosmos 2 | 22 | 0 |
| 1969-021 | C 269 | <i>Tselina-O</i> | Cosmos 3 | 21 | 0 |
| 1970-005 | C 320 | DS-MO | Cosmos 2 | 5 | 0 |
| 1970-033 | C 334 | DS-P1-Yu | Cosmos 2 | 3 | 0 |
| 1970-065 | C 359 | Venera | Molniya | 2 | 0 |
| 1972-078 | C 523 | DS-P1-Yu | Cosmos 2 | 10 | 0 |
| 1973-027 | Skylab 1 | Skylab workshop | Saturn V | 22 | 0 |
| 1973-075 | C 601 | DS-P1-Yu | Cosmos 2 | 12 | 0 |
| 1974-074 | C 686 | DS-P1-Yu | Cosmos 2 | 18 | 0 |
| 1974-104 | Salyut 4 | Manned station | Proton | 17 | 0 |
| 1976-012 | C 801 | DS-P1-I | Cosmos 2 | 15 | 0 |
| 1976-037 | C 816 | Romb | Cosmos 3 | 23 | 0 |
| 1976-057 | Salyut 5 | Manned station | Proton | 8 | 0 |
| 1976-124 | C 885 | <i>Romb</i> | Cosmos 3 | 17 | 0 |
| 1977-042 | C 913 | <i>Romb</i> | Cosmos 3 | 20 | 0 |
| 1977-097 | Salyut 6 | Manned station | Proton | 104 | 0 |
| 1977-111 | C 965 | <i>Romb</i> | Cosmos 3 | 25 | 0 |
| 1978-043 | C 1004 | <i>Zenit-2M</i> | Soyuz | 5 | 0 |
| 1978-120 | C 1065 | Romb | Cosmos 3 | 6 | 0 |

| INT'L DES. | COMMON NAME | S/C | R/B | TOTAL DEBRIS | DEBRIS ON- ORBIT |
|---------------|-----------------|-----------------------------|--------------------------------|-----------------|---------------------|
| 1979-008 | C 1074 | <i>Soyuz T test vehicle</i> | Soyuz | 5 | 0 |
| 1979-063 | C 1112 | Romb | Cosmos 3 | 24 | 0 |
| 1980-047 | C 1186 | <i>Romb</i> | Cosmos 3 | 25 | 0 |
| 1980-067 | C 1204 | <i>Romb</i> | Cosmos 3 | 22 | 0 |
| 1980-083 | C 1215 | Payload | Cosmos 3 | 2 | 0 |
| 1981-093 | SJ-2/-2A/-2B | Payloads | CZ-2B | 6 | 0 |
| 1981-097 | C 1311 | Romb | Cosmos 3 | 24 | 0 |
| 1982-006 | OPS 2849 | Payload | Titan 3B Agena | 4 | 3 (?) |
| 1982-007 | C 1335 | Romb | Cosmos 3 | 22 | 0 |
| 1982-033 | Salyut 7 | Manned station | Proton | 197 | 0 |
| 1982-034 | C 1351 | <i>Romb</i> | Cosmos 3 | 24 | 0 |
| 1982-076 | C 1397 | <i>Romb</i> | Cosmos 3 | 22 | 0 |
| 1983-034 | C 1453 | <i>Romb</i> | Cosmos 3 | 22 | 0 |
| 1983-049 | C 1465 | <i>Romb</i> | Cosmos 3 | 8 | 0 |
| 1983-091 | C 1494 | <i>Romb</i> | Cosmos 3 | 25 | 0 |
| 1983-101 | C 1501 | <i>Romb</i> | Cosmos 3 | 24 | 0 |
| 1984-008 | STTW-T1 | Payload | CZ-3 | 2 | 0 |
| 1984-104 | C 1601 | Romb | Cosmos 3 | 28 | 0 |
| 1985-021 | GEOSAT | Payload | Atlas 41E (OIS R/B) | 5 | 3 |
| 1985-050 | C 1662 | Romb | Cosmos 3 | 27 | 0 |
| 1985-075 | C 1677 | RORSAT | Tsyklon | 2 | 0 |
| 1985-097 | C 1697 | <i>Tselina-2</i> | Zenit | 4 | 4 |
| 1986-017 | Mir | Manned station | Proton | 323 | 0 |
| 1986-024 | C 1736 | RORSAT | Tsyklon | 28 | 1 |
| 1986-030 | C 1741 | Payload | Cosmos 3 | 2 | 2 |
| 1986-052 | C 1763 | Payload | Cosmos 3 | 4 | 3 |
| 1986-067 | C 1776 | Romb | Cosmos 3 | 28 | 0 |
| 1986-101 | C 1809 | Payload | Tsyklon | 9 | 9 |
| 1988-019 | C 1932 | RORSAT | Tsyklon | 3 | 2 |
| 1988-065 | C 1960 | Romb | Cosmos 3 | 28 | 0 |
| 1988-067 | FSW-1 2 | Payload | CZ-2C | 5 | 0 |
| 1988-113 | C 1985 | Duga-K | Tsyklon | 36 | 0 |
| 1989-012 | C 2002 | <i>Romb</i> | Cosmos 3 | 10 | 0 |
| 1989-100 | C 2053 | <i>Duga-K</i> | Tsyklon | 37 | 0 |
| 1990-012 | C 2059 | <i>Romb</i> | Cosmos 3 | 10 | 0 |
| 1990-038 | C 2075 | <i>Romb</i> | Cosmos 3 | 14 | 0 |
| 1990-104 | C 2106 | <i>Romb</i> | Tsyklon | 28 | 0 |

| INT'L DES. | COMMON NAME | S/C | R/B | TOTAL DEBRIS | DEBRIS ON- ORBIT |
|---------------|----------------|---|----------|-----------------|---------------------|
| 1995-008 | C 2306 | <i>Romb</i> | Cosmos 3 | 23 | 0 |
| 1998-067 | Zarya | International Space Station (ISS) | Proton | 107 | 4 |

Note: the debris count associated with 1998-067, the International Space Station (ISS), does not include payloads deployed from the ISS by the U.S., the Commonwealth of Independent States (CIS), partners, or participant countries and non-governmental organizations.

For more information on these events, see History of On-orbit Satellite Fragmentations, 4th Ed., Jan. 1990; the Interagency Group (Space) Report on Orbital Debris, 1989; and Soviet Space Programs, 1976-80, Part 3, May 1985.

5.0 SATELLITES NOT ASSOCIATED WITH BREAKUPS

The table below identifies specific SSN numbers of objects, which possess the same International Designator year and number but are not associated with the indicated event. For example, 1961-015C was an Ablestar rocket body, which broke up. The mission deployed two objects (Transit 4A and Solrad 3/Injun 1) that were not associated with the rocket body explosion. Those two objects are not counted in the 1961-015 totals, although they definitely are associated with the 1961-015 international designator.

Occasionally it is not obvious whether an object should be included in a fragmentation event. In those cases historical research and historical Satellite Catalogs usually reveal whether an object should be included in the count. The list below represents the best summary of excluded objects. The parent object is always considered a fragment. Aerodynamic breakups are included in this list if they produced cataloged fragmentation other than the parent object.

The list below is formatted as follows: The international designator and number of excluded debris in parenthesis are followed by the SSN numbers, which are not debris. A blank line separates years.

International Designator

1961-015 (2) - 116 117

1962-057 (0) -

1963-047 (0) -

1964-006 (28) - 746 748 750 751 14427 14428 15786 16544 16545 16546 16547 16548 18589 18686 19010
19173 19990 19991 19992 19993 19994 19995 19996 19997 19998 20101 20224 21621

1964-070 (1) - 920

1965-012 (1) - 1095

1965-020 (3) - 1267 1268 1269

1965-082 (1) - 1641

1965-088 (23) 1707 1708 1740 1741 1784 1785 1786 1787 1788 1789 1790 1791 1792 1793 1794 1795 1796
1797 1798 1799 1800 1801 1802

1965-108 (4) - 1870 1902 1941 13912

1966-012 (2) - 2012 2014

1966-046 (3) - 2186 2189 2190

1966-056 (3) - 2255 2256 2511

1966-059 (1) - 2291

1966-088 (1) - 2438

1966-101 (0) -

1967-116 (1) - 3048

1968-003 (1) - 3096

1968-025 (1) - 3170

1968-081 (5) - 3428 3429 3430 3431 5999

1968-090 (0) -

1968-091 (1) - 3505

1968-097 (0) -

1968-114 (1) - 3615

 1969-013 (1) - 3691
 1969-029 (1) - 3835
 1969-064 (1) - 4051
 1969-082 (10) - 4111 4132 4166 4168 4237 4247 4256 4257 4259 4295

 1970-025 (2) - 4362 4363
 1970-089 (1) - 4597
 1970-091 (0) -

 1971-015 (1) - 4965
 1971-106 (4) - 5650 5664 5665 5672

 1972-058 (1) - 6126

 1973-017 (1) - 6398
 1973-021 (4) - 6434 6436 6442 6443
 1973-086 (1) - 6920

 1974-015 (1) - 7218
 1974-089 (3) - 7529 7530 7531
 1974-103 (1) - 7588

 1975-004 (1) - 7615
 1975-052 (2) - 7924 7965
 1975-080 (1) - 8192
 1975-102 (1) - 8417

 1976-063 (1) - 8933
 1976-067 (2) - 9013 9016
 1976-072 (1) - 9048
 1976-077 (1) - 9057
 1976-105 (3) - 9496 9497 9506
 1976-120 (2) - 9604 9605
 1976-123 (4) - 9623 9624 9639 9640
 1976-126 (3) - 9643 9644 9645

 1977-027 (3) - 9912 9913 9921
 1977-047 (3) - 10060 10066 10089
 1977-065 (3) - 10143 10145 10156
 1977-068 (3) - 10151 10152 10167
 1977-092 (6) - 10366 10367 10368 10408 10484 11571
 1977-121 (1) - 10532

 1978-026 (2) - 10702 10703
 1978-083 (3) - 11016 11017 11076
 1978-098 (2) - 11080 18605
 1978-100 (4) - 11084 11085 11086 11177

 1979-017 (3) - 11279 11291 11322
 1979-033 (2) - 11334 11367
 1979-058 (3) - 11418 11423 11555
 1979-077 (3) - 11512 11513 11550
 1979-101 (2) - 11636 11637
 1979-104 (3) - 11645 24754 25098

1980-021 (1) - 11730
 1980-028 (5) - 11758 11759 11760 11761 11762
 1980-030 (1) - 11766
 1980-050 (3) - 11845 11846 11847
 1980-057 (3) - 11872 11873 11888
 1980-085 (3) - 12033 12034 12035
 1980-089 (1) - 12055

 1981-016 (4) - 12304 12305 12306 12311
 1981-028 (1) - 12365
 1981-031 (3) - 12377 12378 12384
 1981-053 (1) - 12508
 1981-058 (3) - 12548 12549 12561
 1981-071 (3) - 12629 12630 12680
 1981-072 (1) - 12632
 1981-088 (5) - 12818 12819 12820 12821 12822
 1981-089 (1) - 12829
 1981-108 (3) - 12934 12935 12940

 1982-025 (1) - 13114
 1982-029 (3) - 13125 13126 13169
 1982-038 (1) - 13151
 1982-055 (2) - 13260 13261
 1982-088 (1) - 13509
 1982-115 (4) - 13685 13686 13692 13693

 1983-020 (3) - 13901 13903 20413
 1983-022 (8) - 13924 14477 16502 16503 28604 38839 38840 38841
 1983-038 (6) - 14036 14037 14038 14041 14042 14043
 1983-044 (1) - 14065
 1983-070 (3) - 14183 14184 14191
 1983-075 (5) - 14208 14209 14229 14631 14928
 1983-127 (7) - 14590 14591 14592 14593 14594 14595 14607

 1984-011 (6) - 14681 14688 14689 14692 14695 14696
 1984-083 (1) - 15168
 1984-106 (6) - 15333 15334 15335 15336 15337 17358
 1984-114 (2) - 15385 15386

 1985-030 (1) - 15654
 1985-037 (7) - 15697 15698 15699 15700 15701 15702 15715
 1985-039 (1) - 15735
 1985-042 (5) - 15755 15770 15771 15772 15774
 1985-082 (1) - 16055
 1985-094 (6) - 16138 16140 16141 16142 16143 16144
 1985-108 (1) - 16262
 1985-118 (10) - 16396 16397 16398 16399 16403 16404 16405 16406 16407 16445
 1985-121 (5) - 16434 16435 16436 16437 16438

 1986-019 (3) - 16613 16614 16616
 1986-059 (1) - 16896
 1986-069 (4) - 16946 16947 16948 16949

 1987-004 (1) - 17298
 1987-020 (4) - 17536 26111 26601 26982

1987-059 (2) - 18185 18186
 1987-062 (1) - 18215
 1987-068 (1) - 18312
 1987-078 (3) - 18350 18351 18353
 1987-079 (6) - 18355 18356 18357 18358 18359 18360
 1987-108 (1) - 18714
 1987-109 (5) - 18715 18716 18717 18718 18722

 1988-005 (3) - 18821 38540 38543
 1988-006 (4) - 18845 18846 18855 18984
 1988-007 (1) - 18824
 1988-023 (1) - 18986
 1988-040 (1) - 19121
 1988-085 (6) - 19501 19502 19503 19504 19505 21751
 1988-109 (3) - 19687 19688 19690

 1989-001 (6) - 19749 19750 19751 19752 19753 19754
 1989-004 (5) - 19765 19766 19767 19768 19776
 1989-006 (1) - 19772
 1989-039 (7) - 20024 20025 20026 20027 20028 20044 20082
 1989-052 (5) - 20107 20108 20109 20110 20115
 1989-054 (1) - 20125
 1989-056 (2) - 20137 20138
 1989-089 (79) - 20322 20324 20328 22625 22683 22695 22747 22748 22749 22750 22751 22752 22753 22754
 22755 22756 22757 22758 22759 22760 22761 22762 22763 22764 22765 22766 22767 22768
 22769 22770 22771 22772 22773 22774 22775 22776 22820 22852 22853 22854 22855 22856
 22857 22858 22972 23053 23054 23055 23056 23057 23058 23059 23060 23061 23062 23063
 23064 23065 23066 23067 23068 23069 23070 23071 23072 23073 23074 23075 23076 23077
 23078 23079 23080 23081 23082 23083 23084 23085 23086
 1989-100 (38) - 20389 20397 20398 20408 20467 20468 20515 20522 20531 20532 20637 20640 20802 20803
 20821 20822 20823 20911 21020 21021 21022 21023 21042 21043 21064 21205 21206 21207
 21537 21540 21767 21768 21769 21770 21771 21772 21773 21774
 1989-101 (6) - 20391 20392 20393 20394 20400 21648

 1990-045 (5) - 20619 20620 20621 20622 20623
 1990-081 (7) - 20788 20789 20790 20792 20793 20797 20798
 1990-087 (1) - 20829
 1990-102 (5) - 20953 20954 20955 20958 21046
 1990-105 (1) - 20978
 1990-110 (6) - 21006 21007 21008 21009 21010 21011

 1991-003 (3) - 21055 21056 21058
 1991-009 (8) - 21100 21101 21102 21103 21104 21105 21106 21107
 1991-010 (5) - 21111 21112 21113 21122 21129
 1991-015 (4) - 21139 21140 21142 21904
 1991-025 (6) - 21216 21217 21218 21219 21220 21221
 1991-068 (6) - 21728 21729 21730 21731 21732 21733
 1991-071 (1) - 21742
 1991-075 (1) - 21765
 1991-082 (4) - 21800 21801 21825 21836

 1992-021 (3) - 21939 21940 21942
 1992-041 (8) - 22027 22028 22033 27484 27485 27486 27487 27675
 1992-047 (6) - 22056 22057 22058 22059 22060 22061
 1992-082 (5) - 22245 22246 22247 22248 22249
 1992-088 (5) - 22269 22270 22271 22272 22273

1992-091 (1) - 22281
 1992-093 (5) - 22284 22290 22291 22292 22293

 1993-016 (3) - 22565 22575 22576
 1993-018 (1) - 22586
 1993-028 (1) - 22642
 1993-036 (3) - 22676 23007 25028
 1993-045 (1) - 22717
 1993-057 (2) - 22790 22953
 1993-072 (5) - 22907 22908 22909 22910 22926

 1994-004 (2) - 22973 22987
 1994-029 (1) - 23105
 1994-038 (5) - 23168 23169 23170 23171 23172
 1994-069 (5) - 23327 23328 23329 23330 23339
 1994-076 (7) - 23396 23397 23398 23399 23400 23401 23403
 1994-085 (1) - 23439

 1995-015 (4) - 23534 23535 23594 23595
 1995-028 (1) - 23597
 1995-033 (3) - 23605 23607 23608
 1995-037 (9) - 23620 23621 23622 23623 23624 23625 23626 23627 23630

 1996-010 (5) - 23794 23795 23796 23824 24736
 1996-034 (5) - 23880 23881 23882 23883 23886

 1997-024 (1) - 24806
 1997-051 (7) - 24944 24945 24947 24948 24949 24950 24951
 1997-070 (5) - 25045 25046 25047 25048 25053
 1997-079 (1) - 25089
 1997-086 (3) - 25126 25127 25128

 1998-011 (1) - 25175

 1999-008 (3) - 25634 25635 25636
 1999-025 (5) - 25731 25732 25733 37580 37581
 1999-057 (1) - 25941
 1999-072 (1) - 26041

 2000-036 (5) - 26394 26395 26396 26397 26399
 2000-055 (1) - 27477

 2001-049 (3) - 26957 26958 26959
 2001-057 (1) - 27054

 2002-037 (6) - 27470 27471 27472 27473 27476 27494

 2003-035 (6) - 27857 28084 28085 28086 28087 28088

 2006-002 (1) - 28931
 2006-006 (1) - 28943
 2006-015 (5) - 29093 29536 29537 29538 29539
 2006-026 (1) - 29248
 2006-037 (4) - 29393 29395 29396 29493
 2006-039 (2) - 29397 29403
 2006-050 (5) - 29522 29524 29525 29600 29637

2006-057 (1) - 29652
 2006-062 (7) - 29670 29671 29672 29673 29674 29675 29682

 2007-003 (2) - 30323 30479
 2007-005 (14) - 30586 30587 30588 30589 30591 30651 31105 31106 31107 31108 31109 31110
 31111 31112
 2007-029 (7) - 31793 31794 31795 31796 31799 43374 43375
 2007-052 (6) - 32275 32276 32277 32278 32279 32281
 2007-054 (1) - 32287
 2007-065 (10) - 32393 32394 32395 32396 32397 32398 32400 32401 32402 32403

 2008-011 (1) - 32708
 2008-019 (2) - 32779 32780
 2008-046 (7) - 33378 33379 33380 33381 33382 33383 33384
 2008-067 (6) - 33466 33467 33468 33469 33470 33471

 2009-018 (1) - 34780
 2009-042 (3) - 35696 35697 35698

 2010-007 (9) - 36400 36401 36402 36403 36404 36405 36408 36409 36410
 2010-041 (11) - 37137 37138 37139 37140 37141 37142 37144 37145 37146 37147 37149
 2010-042 (1) - 37150
 2010-057 (1) - 37210

 2011-019 (2) - 37481 37482
 2011-037 (6) - 37755 37757 37758 37759 37760 37761
 2011-077 (1) - 38014

 2012-008 (2) - 38091 38095
 2012-026 (2) - 38342 38344
 2012-044 (2) - 38744 38745

 2013-055 (11) - 39265 39266 39267 39268 39269 39270 39280 39290 39292 39293 39369

 2014-016 (1) - 39634

 2015-024 (1) - 40619
 2015-075 (2) - 41121 41123

 2016-012 (5) - 41338 41339 41340 41341 41342

 2017-086 (7) - 43087 43088 43090 43091 43092 43093 43094

6.0 SATELLITES NOT ASSOCIATED WITH ANOMALOUS EVENTS

The table below identifies specific SSN numbers of objects, which possess the same International Designator year and number but are not associated with the indicated anomalous event. The list below represents the best summary of excluded objects. Parent object is not considered a fragment.

The list below is formatted as follows: The international designator and number of excluded debris in parenthesis are followed by the SSN numbers that are not debris.

International Designator

1959-007 (1) - 20

1963-049 (12) - 703 704 705 706 715 753 2432 2620 2930 4586 6182 6283

1964-006 (5) - 746 748 750 751 25278
1964-026 (5) - 801 805 806 809 2986
1964-053 (2) - 876 877

1965-016 (9) - 1208 1244 1245 1271 1272 1291 1292 1293 1310
1965-027 (3) - 1314 1315 1316
1965-048 (6) - 1420 1425 1428 1435 2701 3592
1965-098 (10) - 1804 1806 1807 1808 1944 1948 1951 2092 2153 20833
1965-101 (4) - 1814 1815 1934 1935
1965-109 (5) - 1864 1865 2086 2226 2353

1966-005 (6) - 1952 1953 2140 2141 2889 2989
1966-024 (3) - 2119 2120 3590
1966-040 (2) - 2173 2174
1966-077 (3) - 2403 2411 2412

1967-034 (6) - 2754 2755 2777 2778 6718 7670
1967-048 (4) - 2807 2811 17723 19222
1967-092 (4) - 2965 2967 2994 3122

1968-019 (2) - 3150 3151

1969-009 (2) - 3669 3670

1970-067 (5) - 4507 4515 5036 5447 6372

1971-003 (2) - 4849 4850

1972-049 (2) - 6079 6080
1972-102 (2) - 6319 6320

1975-027 (2) - 7734 7735

1978-014 (6) - 10664 10665 12329 12330 12331 12406
1978-064 (1) - 10967
1978-094 (2) - 11055 11056
1978-096 (3) - 11060 11061 11062
1978-098 (2) - 11080 11081

| | |
|-----------------|--|
| 1979-057 (3) - | 11416 11419 11634 |
| 1979-095 (2) - | 11605 11608 |
| 1981-043 (4) - | 12456 12457 15769 25255 |
| 1981-054 (5) - | 12512 12513 12514 12515 12519 |
| 1981-059 (3) - | 12553 12559 12560 |
| 1982-102 (3) - | 13617 13618 37865 |
| 1983-022 (9) - | 13923 13924 14477 16442 16443 16444 16502 16503 16504 |
| 1985-066 (6) - | 15935 15936 15938 15950 15951 16020 |
| 1985-090 (2) - | 16110 16111 |
| 1986-073 (4) - | 16969 16982 16983 26303 |
| 1987-011 (4) - | 17369 17370 17399 17400 |
| 1987-020 (2) - | 17535 17536 (there are over 100 pieces of fragmentation as well) |
| 1987-060 (4) - | 18187 18188 18191 18524 |
| 1988-005 (46) - | 18821 28988 28999 28990 28991 28992 28993 28994 28995 29066 29067 29299 29300 29301 29302 31397 31398 31399 31400 31401 31402 35364 35365 35366 35367 35368 35369 35370 35371 35372 35373 38281 38282 38283 38284 38285 38286 38287 38330 38539 38541 38542 38544 38655 38656 18820 |
| 1988-032 (2) - | 19045 19046 |
| 1988-089 (3) - | 19531 19532 19534 |
| 1989-089 (2) - | 20322 20323 |
| 1990-017 (2) - | 20508 20509 |
| 1990-037 (3) - | 20579 20580 22920 |
| 1991-032 (3) - | 21263 21267 21298 |
| 1991-039 (3) - | 21397 21398 21842 |
| 1991-050 (6) - | 21574 21575 21576 21577 21578 21610 |
| 1991-063 (2) - | 21700 21701 |
| 1993-014 (5) - | 22561 22562 22567 22568 22599 |
| 1994-057 (5) - | 23233 23234 23235 12150 23277 |
| 1994-077 (11) - | 23404 23405 23406 23407 23408 23409 23410 23417 23418 23419 27760 |
| 1994-089 (3) - | 23455 23457 23458 |
| 1995-021 (3) - | 23560 23561 28066 |
| 1995-041 (3) - | 23639 23640 23641 |
| 1995-059 (2) - | 23710 23711 |
| 1996-017 (3) - | 23827 23828 31403 |
| 1997-082 (7) - | 25104 25105 25106 25107 25108 25109 25141 |
| 1999-035 (2) - | 25791 25792 |
| 1999-042 (2) - | 25880 25881 |
| 1999-051 (3) - | 25919 25920 25921 |

2001-055 (5) - 26997 26998 26999 27000 27497
2002-005 (6) - 27372 27373 27374 27375 27376 27377
2003-024 (3) - 27820 27821 28998
2009-049 (8) - 35865 35866 35867 35868 35869 35870 35871 35872
2009-055 (2) - 35946 35947
2011-043 (6) - 37781 37782 38313 38314 38315 38316
2012-017 (2) - 38248 38249
2015-020 (5) - 40552 40553 40554 40555 40556

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| 14. ABSTRACT <p>The History of On-Orbit Satellite Fragmentations chronicles all known satellite fragmentation events, this 15th edition complete through a suspense date of 4 July 2018. Since the 14th edition breakups, in addition to launch activity, have resulted in an approximately 36% increase in the number of cataloged space objects. More significantly, breakup and anomalous debris accounted for 65% of the catalog growth observed. The reason for these large increases was the first accidental collision of two intact spacecraft, Iridium 33 and Cosmos 2251, and the continued cataloging of debris created by the intentional destruction of the Fengyun 1C spacecraft-the most environmentally harmful fragmentation to date.</p> | | | | | | |
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